TYPE CERTIFICATE
DATA SHEET

N° EASA.R.123

for
SE 316 / SA 315

Type Certificate Holder
Airbus Helicopters

Aéroport International Marseille – Provence
13725 Marignane CEDEX
France

For Models: SE 3160, SA 316 B, SA 316 C, SA 319 B, SA 315 B
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SECTION 1: SE 3160 & SA 316 B

I. General

1. Type/ Model/ Variant
   1.1 Type
   SE 316
   1.2 Model
   SE 3160, SA 316 B
   1.3 Variant
   - - -

2. Airworthiness Category
   Small Rotorcraft

3. Manufacturer
   Airbus Helicopters
   Aéroport International Marseille – Provence
   13725 Marignane CEDEX, France

4. Type Certification Application Date to DGAC
   not recorded

5. State of Design Authority
   EASA
   (pre EASA: DGAC, France)

6. Type Certificate Date by DGAC FR
   for SE 3160 15 December 1961
   for SA 316B 17 March 1970

7. Type Certificate n°
   DGAC FR: n° 14
   EASA: EASA.R.123

8. Type Certificate Data Sheet n°
   n° 61 (until issue 7, dated March 1993)
   EASA.R.123 (since 27 January 2010)

9. EASA Type Certification Date
   28 September 2003,
   in accordance with CR (EU) 1702/2003, Article 2, 3., (a),
   (i), 2nd indented bullet, 1st indented bullet.

II. Certification Basis

1. Reference Date for determining the applicable requirements
   not recorded

2. Airworthiness Requirements
   CAR-6, edition dated 20 December 1956 (including
   Amdts. 6-1 to 6-3) with additional Special Conditions for
   turbine helicopter notified at the DGAC FR by the
   Government of the United States (FAA letter, dated
   3 May 1960)

3. Special Conditions
   Refer to §1 certification basis (see II.2)

4. Exemptions
   none

5. Deviations
   none

6. Equivalent Safety Findings
   none

7. Requirements elected to comply
   none

8. Environmental Protection Requirements
   8.1 Noise Requirements
   Complies with the essential requirements by virtue of early TC date, see also TCDSN N° EASA.R.123

   8.2 Emission Requirements
   n/a

9. Operational Suitability Data (OSD)
   Not required for rotorcraft that are no longer in production. CR (EU) 748/2012, as amended by CR (EU)
   69/2014 does not require OSD elements for this model
   (see Article 7a, 1.).
III. Technical Characteristics and Operational Limitations

1. Type Design Definition

SE 3160: basic SE 3160 definition
SA 316 B: definition of SA 316 B is obtained by applying the SE 3160 modifications (structural reinforcements and mechanical assy. improvement) which allows extending the maximal weight. All SE 3160 aircraft could be upgraded to SA 316 B version by applying Aérospatiale Service Bulletin Alouette N° 01-20.

2. Description

Main rotor: three-bladed main rotor
Tail rotor: three-bladed tail rotor
Fuselage: airframe of conventional structure
Landing gear: three-wheeled fixed landing gear, skids or float gear
Powerplant: single turbine

3. Equipment

As per compliance with applicable airworthiness requirements defined here above and referenced within approved Rotorcraft Flight Manual.

4. Dimensions

4.1 Fuselage

Length: 10.18 m (33.38 ft)
Width: 2.59 m (8.50 ft) with wheel gear
2.65 m (8.69 ft) with skid gear
3.92 m (12.86 ft) with float gear
Height: 2.97 m (9.74 ft)
3.46 m (11.35 ft) with float gear

4.2 Main Rotor

Diameter: 11.00 m (36.09 ft)

4.3 Tail Rotor

Diameter: 1.91 m (6.27 ft)

5. Engine

5.1 Model

SAFRAN Helicopter Engines (Turbomeca)
1 x Model Artouste III B, or,
1 x Model Artouste III B1

5.2 Type Certificate

EASA TC/TCDS n°: EASA.E.091
(DGAC-FR TC/TCDS n°: M12)

5.3 Limitations

5.3.1 Installed Engine Limitations

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
<th>Gas generator [min$^{-1}$]</th>
<th>Temperature $T_4$ [°C]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Max rpm</td>
<td>---</td>
<td>33 500$^{1)}$</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (thermic) on GND</td>
<td>640</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (reducer) with Artouste III B</td>
<td>420</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (reducer) with Artouste III B1</td>
<td>440</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>MCP used</td>
<td>405 (reducer)</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max T4 at Start-up</td>
<td>---</td>
<td>---</td>
<td>630</td>
</tr>
<tr>
<td>Max T4 (5 min)</td>
<td>---</td>
<td>---</td>
<td>550</td>
</tr>
<tr>
<td>Max T4 continuous</td>
<td>---</td>
<td>---</td>
<td>500</td>
</tr>
</tbody>
</table>

Note: $^{1)}$ rpm ± 200 (rpm ±1 000 during transitory variation allowed)
5.3.2 Transmission Torque Limits

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
</tr>
</thead>
<tbody>
<tr>
<td>TOP (SE 3160)</td>
<td>405</td>
</tr>
<tr>
<td>TOP (SA 316 B)</td>
<td>440</td>
</tr>
<tr>
<td>MCP (SE 3160, SA 316 B)</td>
<td>420</td>
</tr>
</tbody>
</table>

6. Fluids (Fuel/ Oil/ Additives)

6.1 Fuel Refer to approved RFM

6.2 Oil Refer to approved RFM for engine and gearboxes

6.3 Additives Refer to approved RFM

6.4 Hydraulic Refer to approved RFM

7. Fluid capacities

7.1 Fuel Cubic tank:
Fuel tank capacity: 565 (±3.1) litres (149.2 ±0.82 US gal)
Usable fuel: 555 litres (146.6 US gal)
(Total capacity: 595 litres (157.1 US gal))

Quadrilobic tank:
Fuel tank capacity: 575 (±3.1) litres (151.9 ±0.82 US gal)
Usable fuel: 573 litres (151.3 US gal)
(Total capacity: 590 litres (155.8 US gal))

7.2 Oil 10 litres ±3.6 litres (2.6 ±0.95 US gal)

7.3 Coolant System Capacity n/a

8. Air Speed Limitations

For centre of gravity between 2.78 m and 3.08 m:

<table>
<thead>
<tr>
<th>Altitude [m]</th>
<th>Mass [kg]</th>
<th>0 to 1 000</th>
<th>2 000</th>
<th>3 000</th>
<th>4 000</th>
<th>5 000</th>
<th>6 000</th>
</tr>
</thead>
<tbody>
<tr>
<td>2 200*</td>
<td>210</td>
<td>195</td>
<td>172</td>
<td>---</td>
<td>---</td>
<td>---</td>
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<tr>
<td>2 100</td>
<td>210</td>
<td>200</td>
<td>180</td>
<td>140</td>
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<tr>
<td>1 900</td>
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<td>115</td>
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</tr>
<tr>
<td>1 700</td>
<td>210</td>
<td>210</td>
<td>195</td>
<td>173</td>
<td>143</td>
<td>105</td>
<td></td>
</tr>
<tr>
<td>≤1 500</td>
<td>210</td>
<td>210</td>
<td>203</td>
<td>185</td>
<td>160</td>
<td>130</td>
<td></td>
</tr>
</tbody>
</table>

* SA 316 B only

SE 3160 (for instruments with imperial units):

<table>
<thead>
<tr>
<th>Altitude [ft]</th>
<th>Mass [lb]</th>
<th>0 to 3 000</th>
<th>6 000</th>
<th>9 000</th>
<th>12 000</th>
<th>15 000</th>
<th>18 000</th>
<th>21 000</th>
</tr>
</thead>
<tbody>
<tr>
<td>4 630</td>
<td>113</td>
<td>109</td>
<td>101</td>
<td>85</td>
<td>---</td>
<td>---</td>
<td>---</td>
<td></td>
</tr>
<tr>
<td>4 400</td>
<td>113</td>
<td>110</td>
<td>103</td>
<td>89</td>
<td>---</td>
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<td>4 000</td>
<td>113</td>
<td>112</td>
<td>106</td>
<td>94</td>
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<td>113</td>
<td>108</td>
<td>99</td>
<td>86</td>
<td>72</td>
<td>51</td>
<td></td>
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<tr>
<td>≤3 300</td>
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<td>113</td>
<td>110</td>
<td>102</td>
<td>92</td>
<td>78</td>
<td>62</td>
<td></td>
</tr>
</tbody>
</table>
SA 316 B (for instruments with imperial units):

<table>
<thead>
<tr>
<th>Altitude [ft]</th>
<th>0 to 3 000</th>
<th>6 000</th>
<th>9 000</th>
<th>12 000</th>
<th>15 000</th>
<th>18 000</th>
<th>21 000</th>
</tr>
</thead>
<tbody>
<tr>
<td>Mass [lb]</td>
<td></td>
<td></td>
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<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>4 850</td>
<td>113</td>
<td>107</td>
<td>97</td>
<td>---</td>
<td>---</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>4 630</td>
<td>113</td>
<td>109</td>
<td>101</td>
<td>85</td>
<td>75</td>
<td>50</td>
<td>---</td>
</tr>
<tr>
<td>4 200</td>
<td>113</td>
<td>111</td>
<td>104</td>
<td>92</td>
<td>97</td>
<td>84</td>
<td>68</td>
</tr>
<tr>
<td>3 750</td>
<td>113</td>
<td>113</td>
<td>107</td>
<td>97</td>
<td>84</td>
<td>68</td>
<td>45</td>
</tr>
<tr>
<td>≤3 300</td>
<td>113</td>
<td>113</td>
<td>110</td>
<td>102</td>
<td>92</td>
<td>78</td>
<td>62</td>
</tr>
<tr>
<td>VNE [kt]</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
</tbody>
</table>

For centre of gravity between 3.08 m and 3.15 m (single-seat flight):

- metric units: same limitation as above but V_{NE} limited to 190 km/h.
- imperial units: same limitation as above but V_{NE} limited to 103 kt.

9. Rotor Speed Limitations
   - Maximum: 420 rpm
   - Minimum: 270 rpm
   - Max continuous: 353 rpm

10. Maximum Operating Altitude and Temperature
   10.1 Altitude
   - Enroute: 21 300 ft (6 500 m) PA
   - TKOF: 19 000 ft (5 800 m) PA
   10.2 Temperature
   - -40 °C to +55 °C

11. Operating Limitations
   VFR day
   VFR night, when the additional equipment required by operational regulations is installed and serviceable.
   For more information refer to approved RFM.

12. Maximum Mass
   TKOF/LDG
   - SE 3160: 2 100 kg (4 630 lb)
   - SE316B: 2 200 kg (4 850 lb)

13. Centre of Gravity Range
   Longitudinal C.G. limits
   - Forward limit: 2 780 mm
   - Aft limit: 3 080 mm
   Or:
   - Forward limit: 3 080 mm
   - Aft limit: 3 150 mm
   with the following maximum translation speed limitation:
   - C.G. between 2 800 mm to 3 080 mm: 210 km/h (113 kt)
   - C.G. between 3 080 mm to 3 150 mm: 190 km/h (103 kt)
   Lateral C.G Limits
   - LH limit: 140 mm
   - RH limit: 120 mm

14. Datum
   Longitudinal:
   - 3 000 mm (9.84 ft) forward of main rotor centre line
   Lateral: aircraft symmetry plane

15. Levelling Means
   - 4 levelling legs on the central structure:
   - 2 on the front of the aircraft
   - 2 on the rear of the aircraft

16. Minimum Flight Crew
   1 pilot
17. Maximum Passenger Seating Capacity
   Six
   Front seats: 2 passengers
   Rear seats: 4 passengers

18. Passenger Emergency Exit
   Refer to approved RFM

19. Maximum Baggage/ Cargo Loads
   Refer to approved RFM

20. Rotor Blade Control Movement
   For rigging information refer to the Maintenance Manual

21. Auxiliary Power Unit (APU)
   n/a

22. Life-limited Parts
   The periods specified in the latest approved revision of the Airworthiness Limitations section of the Maintenance Manual must not be exceeded.

IV. Operating and Service Instructions

1. Flight Manual
   SE 3160 and SA 316 B Flight Manual,
   original edition approved by DGAC-FR for type certification, or later DGAC-FR / EASA approved revision

   SE 3160 and SA 316 B Maintenance Manual

   not recorded

   not recorded

5. Illustrated Parts Catalogue
   not recorded

6. Miscellaneous Manuals
   not recorded

7. Service Letters and Service Bulletins
   As published by Aérospatiale, Eurocopter or Airbus Helicopters

8. Required Equipment
   As per compliance with applicable requirements and in accordance with the original Type Design standard; refer to approved Flight Manual.

V. Notes

1. Manufacturer’s eligible serial numbers:
s/n 1048 and subsequent.

2. The certified “optional” installations are each approved independently of the basic helicopter and an approved Flight Manual Supplement is associated to each optional installation, if necessary.

3. Commercial designation:
   ALOUETTE III

* * *
SECTION 2: SA 316 C

I. General

1. Type/ Model/ Variant
   1.1 Type
   SE 316
   1.2 Model
   SA 316 C
   1.3 Variant
   - - -

2. Airworthiness Category
   Small Rotorcraft

3. Manufacturer
   Airbus Helicopters
   Aéroport International Marseille – Provence
   13725 Marignane CEDEX, France

4. Type Certification Application Date to DGAC
   not recorded

5. State of Design Authority
   EASA
   (pre EASA: DGAC, France)

6. Type Certificate Date by DGAC FR
   14 May 1971

7. Type Certificate n°
   DGAC FR: n° 14
   EASA: EASA.R.123

8. Type Certificate Data Sheet n°
   n° 61 (until issue 7, dated March 1993)
   EASA.R.123 (since 27 January 2010)

9. EASA Type Certification Date
   28 September 2003,
   in accordance with CR (EU) 1702/2003, Article 2, 3., (a),
   (i), 2nd bullet, 1st indented bullet.

II. Certification Basis

1. Reference Date for determining the
   applicable requirements
   not recorded

2. Airworthiness Requirements
   CAR-6, edition dated 20 December 1956 (including
   Amdts. 6-1 to 6-3) with additional Special Conditions for
   turbine helicopter notified at the DGAC FR by the
   Government of the United States (FAA letter dated 3 May
   1960)

3. Special Conditions
   Refer to §1 certification basis (see II.2)

4. Exemptions
   none

5. Deviations
   none

6. Equivalent Safety Findings
   none

7. Requirements elected to comply
   none

8. Environmental Protection Requirements
   8.1 Noise Requirements
   Complies with the essential requirements by virtue of
   early TC date, see also TCDSN N° EASA.R.123

   8.2 Emission Requirements
   n/a

9. Operational Suitability Data (OSD)
   Not required for rotorcraft that are no longer in
   production. CR (EU) 748/2012, as amended by CR (EU)
   69/2014 does not require OSD elements for this model
   (see Article 7a, 1.).
III. Technical Characteristics and Operational Limitations

1. Type Design Definition
   Definition of SA 316 C is obtained by applying the SE 3160 modifications according to document Aérospatiale SA 319 A 04.00.025

2. Description
   Main rotor: three-bladed main rotor
   Tail rotor: three-bladed tail rotor
   Fuselage: airframe of conventional structure
   Landing gear: three-wheeled fixed landing gear
   Powerplant: single turbine

3. Equipment
   As per compliance with applicable airworthiness requirements defined here above and referenced within approved Rotorcraft Flight Manual.

4. Dimensions
   4.1 Fuselage
      Length: 10.18 m (33.38 ft)
      Width: 2.60 m (8.54 ft)
      Height: 3.00 m (9.84 ft)
   4.2 Main Rotor
      Diameter: 11.02 m (36.15 ft)
   4.3 Tail Rotor
      Diameter: 1.92 m (6.30 ft)

5. Engine
   5.1 Model
      SAFRAN Helicopter Engines (Turbomeca)
      1 x Model Artouste III D
   5.2 Type Certificate
      EASA TC/TCDS n°: EASA.E.091
      (DGAC-FR TC/TCDS n°: M12)
   5.3 Limitations
      5.3.1 Installed Engine Limitations

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
<th>Gas generator [min⁻¹]</th>
<th>Temperature T4 [°C]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Max rpm</td>
<td>---</td>
<td>33 500¹)</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (thermic) on GND</td>
<td>640</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (reducer)</td>
<td>440</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>MCP used (reducer)</td>
<td>405</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max T4 at Start-up</td>
<td>---</td>
<td>---</td>
<td>630</td>
</tr>
<tr>
<td>Max T4 (5 min)</td>
<td>---</td>
<td>---</td>
<td>550</td>
</tr>
<tr>
<td>Max T4 continuous</td>
<td>---</td>
<td>---</td>
<td>500</td>
</tr>
</tbody>
</table>

Note: ¹) rpm ± 200 (rpm ±1 000 during transitory variation allowed)

5.3.2 Transmission Torque Limits

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
</tr>
</thead>
<tbody>
<tr>
<td>TOP</td>
<td>440</td>
</tr>
<tr>
<td>MCP</td>
<td>368</td>
</tr>
</tbody>
</table>

6. Fluids (Fuel/ Oil/ Additives)
   6.1 Fuel
      Refer to approved RFM
   6.2 Oil
      Refer to approved RFM for engine and gearboxes
   6.3 Additives
      Refer to approved RFM
   6.4 Hydraulic
      Refer to approved RFM
7. Fluid capacities

7.1 Fuel

Cubic tank:
- Fuel tank capacity: 565 (±3.1) litres (149.2 ±0.82 US gal)
- Usable fuel: 555 litres (146.6 US gal)
- (Total capacity: 595 litres (157.1 US gal))

Quadrilobic tank:
- Fuel tank capacity: 575 (±3.1) litres (151.9 ±0.82 US gal)
- Usable fuel: 573 litres (151.3 US gal)
- (Total capacity: 590 litres (155.8 US gal))

7.2 Oil

12 litres (3.2 US gal)

7.3 Coolant System Capacity

n/a

8. Air Speed Limitations

For centre of gravity between 2.78 m and 3.08 m:

(for instruments with metric units):

<table>
<thead>
<tr>
<th>Altitude [m]</th>
<th>0 to 1 000</th>
<th>2 000</th>
<th>3 000</th>
<th>4 000</th>
<th>5 000</th>
<th>6 000</th>
<th>6 500</th>
</tr>
</thead>
<tbody>
<tr>
<td>Mass [kg]</td>
<td>V\textsubscript{NE} [km/h]</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
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<tr>
<td>2 250</td>
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<td>130</td>
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<td>2 100</td>
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<td>184</td>
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<td>192</td>
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</tr>
<tr>
<td>1 700</td>
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<td>198</td>
<td>175</td>
<td>145</td>
<td>110</td>
<td>---</td>
</tr>
<tr>
<td>≤1 500</td>
<td>220</td>
<td>218</td>
<td>202</td>
<td>184</td>
<td>160</td>
<td>132</td>
<td>120</td>
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</tbody>
</table>

(for instruments with imperial units):

<table>
<thead>
<tr>
<th>Altitude [ft]</th>
<th>0 to 3 000</th>
<th>6 000</th>
<th>9 000</th>
<th>12 000</th>
<th>15 000</th>
<th>18 000</th>
<th>21 000</th>
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<tbody>
<tr>
<td>Mass [lb]</td>
<td>V\textsubscript{NE} [kt]</td>
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<td></td>
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<td>4 980</td>
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<tr>
<td>4 650</td>
<td>118</td>
<td>113</td>
<td>103</td>
<td>95</td>
<td>64</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>4 200</td>
<td>118</td>
<td>116</td>
<td>107</td>
<td>95</td>
<td>78</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>3 750</td>
<td>118</td>
<td>118</td>
<td>109</td>
<td>99</td>
<td>86</td>
<td>70</td>
<td>---</td>
</tr>
<tr>
<td>≤3 300</td>
<td>118</td>
<td>118</td>
<td>112</td>
<td>102</td>
<td>92</td>
<td>80</td>
<td>67</td>
</tr>
</tbody>
</table>

For centre of gravity between 3.08 m and 3.15 m (single-seat flight):

Instruments with:
- metric units: same limitation as above but V\textsubscript{NE} limited to 180 km/h up to 3 000 m, and to 160 km/h over 3 000 m.
- imperial units: same limitation as above but V\textsubscript{NE} limited to 97 kt up to 10 000 ft, and to 85 kt over 10 000 ft.

9. Rotor Speed Limitations

- Maximum: 420 rpm
- Minimum: 270 rpm
- Max continuous: 353 rpm

10. Maximum Operating Altitude and Temperature

10.1 Altitude

- Enroute: 21 300 ft (6 500 m) PA
- Take-off: 13 000 ft (4 000 m) PA
- Restart in flight: 17 000 ft (5 200 m) PA for engine equipped with barostatic controller (modification Turbomeca ref. TU 81)
10.2 Temperature

-40 °C to +55 °C

11. Operating Limitations

- VFR day
- VFR night, when the additional equipment required by operational regulations is installed and serviceable

For more information refer to approved RFM.

12. Maximum Mass

TKOF/LDG: 2 250 kg (4 960 lb)

13. Centre of Gravity Range

- Longitudinal C.G. limits
  - Forward limit: 2 780 mm
  - Aft limit: 3 080 mm

  Or:
  - Forward limit: 3 080 mm
  - Aft limit: 3 150 mm

  with the following maximum translation speed limitation:
  - For C.G. between 2 800 mm to 3 080 mm:
    - 220 km/h (119 kt)

  Note: For altitude over 4 920 ft (1 500 m) and for temperature ≤ISA -20°C VNE must be decreased by 15 km/h (8 kt).

  - For C.G. between 3 080 mm to 3 150 mm:
    - 180 km/h (97 kt) up to 9 840 ft (3 000 m)
    - 160 km/h (85 kt) above 9 840 ft (3 000 m)

- Lateral C.G. Limits
  - LH limit: 140 mm
  - RH limit: 120 mm

14. Datum

- Longitudinal: 3 000 mm (9.84 ft) forward of main rotor centre line
- Lateral: aircraft symmetry plane

15. Levelling Means

- 4 levelling legs on the central structure:
  - 2 on the front of the aircraft
  - 2 on the rear of the aircraft

16. Minimum Flight Crew

- 1 pilot at STA +1 385 mm

17. Maximum Passenger Seating Capacity

- Six
  - Front seats: 2 passengers at STA +1 385 mm
  - Rear seats: 4 passengers at STA +2 195 mm

18. Passenger Emergency Exit

- Refer to approved RFM

19. Maximum Baggage/ Cargo Loads

- Refer to approved RFM

20. Rotor Blade Control Movement

- For rigging information refer to the Maintenance Manual

21. Auxiliary Power Unit (APU)

- n/a

22. Life-limited Parts

- The periods specified in the latest approved revision of the Airworthiness Limitations section of the Maintenance Manual must not be exceeded.
IV. Operating and Service Instructions

1. Flight Manual
   SA 316 C Flight Manual, original edition approved by DGAC-FR for type certification, or later DGAC-FR / EASA approved revision

   SA 316 C Maintenance Manual

   not recorded

   not recorded

5. Illustrated Parts Catalogue
   not recorded

6. Miscellaneous Manuals
   not recorded

7. Service Letters and Service Bulletins
   As published by Aérospatiale, Eurocopter or Airbus Helicopters

8. Required Equipment
   As per compliance with applicable requirements and in accordance with the original Type Design standard; refer to approved Flight Manual.

V. Notes

1. Manufacturer’s eligible serial numbers:
   All s/n complying with SA 316 C type (refer to III.1. Type Design Definition).

2. The certified “optional” installations are each approved independently of the basic helicopter and an approved Flight Manual Supplement is associated to each optional installation, if necessary.

3. Commercial designation:
   ALOUETTE III

   * * *
SECTION 3: SA 319 B

I. General

1. Type/ Model/ Variant
   1.1 Type
   SE 316
   1.2 Model
   SA 319 B
   1.3 Variant
   - - -

2. Airworthiness Category
   Small Rotorcraft

3. Manufacturer
   Airbus Helicopters
   Aéroport International Marseille – Provence
   13725 Marignane Cedex, France

4. Type Certification Application Date to DGAC
   not recorded

5. State of Design Authority
   EASA
   (pre EASA: DGAC, France)

6. Type Certificate Date by DGAC FR
   14 May 1971

7. Type Certificate n°
   DGAC FR: n° 14
   EASA: EASA.R.123

8. Type Certificate Data Sheet n°
   n° 61 (until issue 7, dated March 1993)
   EASA.R.123 (since 27 January 2010)

9. EASA Type Certification Date
   28 September 2003,
   in accordance with CR (EU) 1702/2003, Article 2, 3., (a), (i), 2nd bullet, 1st indented bullet.

II. Certification Basis

1. Reference Date for determining the applicable requirements
   not recorded

2. Airworthiness Requirements
   CAR-6, edition dated 20 December 1956 (including
   Amdts. 6-1 to 6-3) with additional Special Conditions for
   turbine helicopter notified at the DGAC FR by the
   Government of the United States (FAA letter dated 3 May
   1960)

3. Special Conditions
   Refer to §1 certification basis (see II.2)

4. Exemptions
   none

5. Deviations
   none

6. Equivalent Safety Findings
   none

7. Requirements elected to comply
   none

8. Environmental Protection Requirements
   8.1 Noise Requirements
   Complies with the essential requirements by virtue of
   early TC date, see also TCDSN n° EASA.R.123

   8.2 Emission Requirements
   n/a

9. Operational Suitability Data (OSD)
   Not required for rotorcraft that are no longer in
   production. CR (EU) 748/2012, as amended by CR (EU)
   69/2014 does not require OSD elements for this model
   (see Article 7a, 1.).
III. Technical Characteristics and Operational Limitations

1. Type Design Definition
Definition of SA 319 B is obtained by applying the SE 3160 modifications according to document Aérospatiale SA 319 A 04.00.025

2. Description
Main rotor: three-bladed main rotor
Tail rotor: three-bladed tail rotor
Fuselage: airframe of conventional structure
Landing gear: three-wheeled fixed landing gear
Powerplant: single turbine

3. Equipment
As per compliance with applicable airworthiness requirements defined here above and referenced within approved Rotorcraft Flight Manual.

4. Dimensions

4.1 Fuselage
Length: 10.18 m (33.38 ft)
Width: 2.60 m (8.54 ft)
Height: 3.00 m (9.84 ft)

4.2 Main Rotor
Diameter: 11.02 m (36.15 ft)

4.3 Tail Rotor
Diameter: 1.92 m (6.30 ft)

5. Engine

5.1 Model
SAFRAN Helicopter Engines (Turbomeca)
1 x Model Astazou XIV B

5.2 Type Certificate
EASA TC/TCDS n°: EASA.E.075
(DGAC-FR TC/TCDS n°: M3)

5.3 Limitations

5.3.1 Installed Engine Limitations

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
<th>Gas generator [min⁻¹]</th>
<th>Temperature T4 [°C]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Max rpm</td>
<td>---</td>
<td>43 500¹)</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (thermic) on GND</td>
<td>640</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (reducer)</td>
<td>440</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>MCP used (reducer)</td>
<td>405</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max T4 at Start-up</td>
<td>---</td>
<td>---</td>
<td>700</td>
</tr>
<tr>
<td>Max T4 at Start-up (5 sec)</td>
<td>---</td>
<td>---</td>
<td>750</td>
</tr>
<tr>
<td>Max T4 (5 min)</td>
<td>---</td>
<td>---</td>
<td>550</td>
</tr>
<tr>
<td>Max T4 continuous without modification AB 60/N°40</td>
<td>---</td>
<td>---</td>
<td>470</td>
</tr>
<tr>
<td>Max T4 continuous with modification AB 60/N°40</td>
<td>---</td>
<td>---</td>
<td>500</td>
</tr>
</tbody>
</table>

Note: ¹) rpm ± 200 (rpm ±1 500 during transitory variation allowed)

5.3.2 Transmission Torque Limits

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
</tr>
</thead>
<tbody>
<tr>
<td>TOP</td>
<td>440</td>
</tr>
<tr>
<td>MCP</td>
<td>368</td>
</tr>
</tbody>
</table>
6. Fluids (Fuel/ Oil/ Additives)

6.1 Fuel Refer to approved RFM
6.2 Oil Refer to approved RFM for engine and gearboxes
6.3 Additives Refer to approved RFM
6.4 Hydraulic Refer to approved RFM

7. Fluid capacities

7.1 Fuel Cubic tank:
Fuel tank capacity: 565.0 (±3.1) litres (149.2 ±0.82 US gal)
Usable fuel: 555.0 litres (146.6 US gal)
(Total capacity: 595.0 litres (157.1 US gal))

Quadrilobic tank:
Fuel tank capacity: 575.0 (±3.1) litres (151.9 ±0.82 US gal)
Usable fuel: 573.0 litres (151.3 US gal)
(Total capacity: 590.0 litres (155.8 US gal))

7.2 Oil 12 litres (3.2 US gal)

7.3 Coolant System Capacity n/a

8. Air Speed Limitations

For centre of gravity between 2.78 m and 3.08 m:

(for instruments with metric units):

<table>
<thead>
<tr>
<th>Altitude [m]</th>
<th>0 to 1 000</th>
<th>2 000</th>
<th>3 000</th>
<th>4 000</th>
<th>5 000</th>
<th>6 000</th>
<th>6 500</th>
</tr>
</thead>
<tbody>
<tr>
<td>Mass [kg]</td>
<td>V_{NE} [km/h]</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>2 250</td>
<td>220</td>
<td>198</td>
<td>175</td>
<td>130</td>
<td>---</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>2 100</td>
<td>220</td>
<td>206</td>
<td>184</td>
<td>154</td>
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<td>---</td>
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<tr>
<td>1 900</td>
<td>220</td>
<td>212</td>
<td>192</td>
<td>165</td>
<td>122</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>1 700</td>
<td>220</td>
<td>216</td>
<td>198</td>
<td>175</td>
<td>145</td>
<td>110</td>
<td>---</td>
</tr>
<tr>
<td>≤1 500</td>
<td>220</td>
<td>218</td>
<td>202</td>
<td>184</td>
<td>160</td>
<td>132</td>
<td>120</td>
</tr>
</tbody>
</table>

(for instruments with imperial units):

<table>
<thead>
<tr>
<th>Altitude [ft]</th>
<th>0 to 3 000</th>
<th>6 000</th>
<th>9 000</th>
<th>12 000</th>
<th>15 000</th>
<th>18 000</th>
<th>21 000</th>
</tr>
</thead>
<tbody>
<tr>
<td>Mass [lb]</td>
<td>V_{NE} [kt]</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>4 980</td>
<td>118</td>
<td>108</td>
<td>98</td>
<td>82</td>
<td>---</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>4 650</td>
<td>118</td>
<td>113</td>
<td>103</td>
<td>90</td>
<td>64</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>4 200</td>
<td>118</td>
<td>116</td>
<td>107</td>
<td>95</td>
<td>78</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>3 750</td>
<td>118</td>
<td>118</td>
<td>109</td>
<td>99</td>
<td>86</td>
<td>70</td>
<td>---</td>
</tr>
<tr>
<td>≤3 300</td>
<td>118</td>
<td>118</td>
<td>112</td>
<td>102</td>
<td>92</td>
<td>80</td>
<td>67</td>
</tr>
</tbody>
</table>

For centre of gravity between 3.08 m and 3.15 m (single-seat flight):

Instruments with:
- metric units: same limitation as above but V_{NE} limited to 180 km/h up to 3 000 m, and to 160 km/h over 3 000 m.
- imperial units: same limitation as above but V_{NE} limited to 97 kt up to 10 000 ft, and to 85 kt over 10 000 ft.

9. Rotor Speed Limitations

Maximum 420 rpm
Minimum 270 rpm
Max continuous 353 rpm
10. Maximum Operating Altitude and Temperature

10.1 Altitude

Enroute: 21 300 ft (6 500 m) PA
Take-off: 13 000 ft (4 000 m) PA
Restart in flight: 16 500 ft (5 000 m) PA

10.2 Temperature

-40 °C to +55 °C

11. Operating Limitations

VFR day

VFR night, when the additional equipment required by operational regulations is installed and serviceable
For more information refer to approved RFM.

12. Maximum Mass

TKOF/LDG: 2 250 kg (4 960 lb)

13. Centre of Gravity Range

Longitudinal C.G. limits
Forward limit: 2 780 mm
Aft limit: 3 080 mm

Or:
Forward limit: 3 080 mm
Aft limit: 3 150 mm

with the following maximum translation speed limitation:

For C.G. between 2 800 mm to 3 080 mm:
- 220 km/h (119 kt)

Note: For altitude over 4 920 ft (1 500 m) and for temperature ≤ISA -20°C V_{NE} must be decreased by 15 km/h (8 kt).

For C.G. between 3 080 mm to 3 150 mm:
- 180 km/h (97 kt) up to 9 840 ft (3 000 m)
- 160 km/h (85 kt) above 9 840 ft (3 000 m)

Lateral C.G Limits
LH limit: 140 mm
RH limit: 120 mm

14. Datum

Longitudinal:
3 000 mm (9.84 ft) forward of main rotor centre line

Lateral: aircraft symmetry plane

15. Levelling Means

4 levelling legs on the central structure:
- 2 on the front of the aircraft
- 2 on the rear of the aircraft

16. Minimum Flight Crew

1 pilot at STA +1 385 mm

17. Maximum Passenger Seating Capacity

Six
Front seats: 2 passengers at STA +1 385 mm
Rear seats: 4 passengers at STA +2 195 mm

18. Passenger Emergency Exit

Refer to approved RFM

19. Maximum Baggage/ Cargo Loads

Refer to approved RFM

20. Rotor Blade Control Movement

For rigging information refer to the Maintenance Manual

21. Auxiliary Power Unit (APU)

n/a

22. Life-limited Parts

The periods specified in the latest approved revision of the Airworthiness Limitations section of the Maintenance Manual must not be exceeded.
IV. Operating and Service Instructions

1. Flight Manual
   SA 319 B Flight Manual, original edition approved by DGAC-FR for type certification, or later DGAC-FR / EASA approved revision

   SA 319 B Maintenance Manual

   not recorded

   not recorded

5. Illustrated Parts Catalogue
   not recorded

6. Miscellaneous Manuals
   not recorded

7. Service Letters and Service Bulletins
   As published by Aérospatiale, Eurocopter or Airbus Helicopters

8. Required Equipment
   As per compliance with applicable requirements and in accordance with the original Type Design standard; refer to approved Flight Manual.

V. Notes

1. Manufacturer’s eligible serial numbers:
   All s/n complying with SA 319 B type (refer to III.1. Type Design Definition).

2. The certified “optional” installations are each approved independently of the basic helicopter and an approved Flight Manual Supplement is associated to each optional installation, if necessary.

3. Commercial designation:
   ALOUETTE III

   * * *
SECTION 4: SA 315 B

I. General

1. Type/ Model/ Variant
   1.1 Type	SA 315
   1.2 Model	SA 315 B
   1.3 Variant	-

2. Airworthiness Category
   Small Rotorcraft

3. Manufacturer
   Airbus Helicopters
   Aéroport International Marseille – Provence
   13725 Marignane CEDEX, France

4. Type Certification Application Date to DGAC
   not recorded

5. State of Design Authority
   EASA
   (pre EASA: DGAC, France)

6. Type Certificate Date by DGAC FR
   29 September 1970

7. Type Certificate n°
   DGAC FR: n° 14
   EASA: EASA.R.123

8. Type Certificate Data Sheet n°
   n° 61 (until issue 7, dated March 1993)
   EASA.R.123 (since 27 January 2010)

9. EASA Type Certification Date
   28 September 2003,
   in accordance with CR (EU) 1702/2003, Article 2, 3., (a), (i), 2\textsuperscript{nd} bullet, 1\textsuperscript{st} indented bullet.

II. Certification Basis

1. Reference Date for determining the applicable requirements
   not recorded

2. Airworthiness Requirements
   CAR-6, edition dated 20 December 1956 (including Amdts. 6-1 to 6-3) with additional Special Conditions for turbine helicopter notified at the DGAC FR by the Government of the United States (FAA letter, dated 3 May 1960)

3. Special Conditions
   Refer to §1 certification basis (see II.2.)

4. Exemptions
   none

5. Deviations
   none

6. Equivalent Safety Findings
   none

7. Requirements elected to comply
   none

8. Environmental Protection Requirements
   8.1 Noise Requirements
   Complies with the essential requirements by virtue of early TC date, see also TCDSN n° EASA.R.123

   8.2 Emission Requirements
   n/a

9. Operational Suitability Data (OSD)
   Not required for rotorcraft that are no longer in production. CR (EU) 748/2012, as amended by CR (EU) 69/2014 does not require OSD elements for this model (see Article 7a, 1.).
III. Technical Characteristics and Operational Limitations

1. Type Design Definition
   SA 315 B:
   Basic SA 315 B definition according to drawing Aérospatiale 315 A.00.10 000.
   SA 315 B:
   Could also be obtained from Alouette II/Alouette Astazou helicopter by applying drawing Aérospatiale 315 A.00.02 000.1.
   Note: SA 315 B obtained from Alouette II are deemed approved by EASA if this transformation was done before 7 March 2007 when the Alouette II was officially declared to satisfy the definition of the Annex II of Basic Regulation EC 1592/2002.

2. Description
   Main rotor: three-bladed main rotor
   Tail rotor: three-bladed tail rotor
   Fuselage: airframe of conventional structure
   Landing gear: three-wheeled fixed landing gear
   Powerplant: single turbine

3. Equipment
   As per compliance with applicable airworthiness requirements defined here above and referenced within approved Rotorcraft Flight Manual.

4. Dimensions
   4.1 Fuselage
      Length: 10.24 m (33.58 ft)
      Width: 2.38 m (7.80 ft)
      Height: 3.09 m (10.14 ft)
   4.2 Main Rotor
      Diameter: 11.02 m (36.15 ft)
   4.3 Tail Rotor
      Diameter: 1.91 m (6.27 ft)

5. Engine
   5.1 Model
      SAFRAN Helicopter Engines (Turbomeca)
      1 x Model Artouste III B, or,
      1 x Model Artouste III B1
   5.2 Type Certificate
      EASA TC/TCDS n°: EASA.E.091
      (DGAC-FR TC/TCDS n°: M12)
   5.3 Limitations

5.3.1 Installed Engine Limitations

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
<th>Gas generator [min⁻¹]</th>
<th>Temperature T4 [°C]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Max rpm</td>
<td>---</td>
<td>33 500¹)</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (thermic) on GND</td>
<td>640</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (reducer) with Artouste III B</td>
<td>420</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max TOP (reducer) with Artouste III B1</td>
<td>440</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>MCP used</td>
<td>405 (reducer)</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>Max T4 at Start-up</td>
<td>---</td>
<td>---</td>
<td>630</td>
</tr>
<tr>
<td>Max T4 (5 min)</td>
<td>---</td>
<td>---</td>
<td>550</td>
</tr>
<tr>
<td>Max T4 continuous</td>
<td>---</td>
<td>---</td>
<td>500</td>
</tr>
</tbody>
</table>

Note: ¹) rpm ± 200 (rpm ±1 000 during transitory variation allowed)
5.3.2 Transmission Torque Limits

<table>
<thead>
<tr>
<th></th>
<th>PWR [kW]</th>
</tr>
</thead>
<tbody>
<tr>
<td>TOP</td>
<td>440</td>
</tr>
<tr>
<td>MCP</td>
<td>368</td>
</tr>
</tbody>
</table>

6. Fluids (Fuel/ Oil/ Additives)

6.1 Fuel Refer to approved RFM
6.2 Oil Refer to approved RFM for engine and gearboxes
6.3 Additives Refer to approved RFM
6.4 Hydraulic Refer to approved RFM

7. Fluid capacities

7.1 Fuel
Cubic tank:
Fuel tank capacity: 565 (±3.1) litres (149.2 ±0.82 US gal)
Usable fuel: 555 litres (146.6 US gal)
(Total capacity: 595 litres (157.1 US gal))
Quadrilobic tank:
Fuel tank capacity: 575 (±3.1) litres (151.9 ±0.82 US gal)
Usable fuel: 573 litres (151.3 US gal)
(Total capacity: 590 litres (155.8 US gal))

7.2 Oil
Engine: 10 litres (2.64 US gal)
MGB: 6 litres (1.58 US gal)
RGB: 0.5 litres (1.05 US pint)

7.3 Coolant System Capacity n/a

8. Air Speed Limitations

For centre of gravity between 2.76 m and 3.00 m:

(for instruments with metric units):

<table>
<thead>
<tr>
<th>Altitude [m]</th>
<th>0 to 1 000</th>
<th>2 000</th>
<th>3 000</th>
<th>4 000</th>
<th>5 000</th>
<th>6 000</th>
<th>6 500</th>
<th>7 000</th>
</tr>
</thead>
<tbody>
<tr>
<td>Mass [kg]</td>
<td>V(_{NE}) [km/h]</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>1 950</td>
<td>210</td>
<td>205</td>
<td>188</td>
<td>156</td>
<td>106</td>
<td>---</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>1 750</td>
<td>210</td>
<td>208</td>
<td>192</td>
<td>167</td>
<td>135</td>
<td>93</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>1 600</td>
<td>210</td>
<td>197</td>
<td></td>
<td>152</td>
<td>120</td>
<td>---</td>
<td>88</td>
<td></td>
</tr>
<tr>
<td>1 400</td>
<td>210</td>
<td>204</td>
<td></td>
<td>188</td>
<td>167</td>
<td>140</td>
<td>---</td>
<td>110</td>
</tr>
<tr>
<td>≤1 200</td>
<td>210</td>
<td>210</td>
<td>210</td>
<td>200</td>
<td>182</td>
<td>160</td>
<td>---</td>
<td>130</td>
</tr>
</tbody>
</table>

(for instruments with imperial units):

<table>
<thead>
<tr>
<th>Altitude [ft]</th>
<th>0 to 3 000</th>
<th>6 000</th>
<th>9 000</th>
<th>12 000</th>
<th>15 000</th>
<th>18 000</th>
<th>21 000</th>
<th>23 000</th>
</tr>
</thead>
<tbody>
<tr>
<td>Mass [lb]</td>
<td>V(_{NE}) [kt]</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>4 300</td>
<td>113</td>
<td>113</td>
<td>104</td>
<td>91</td>
<td>72</td>
<td>---</td>
<td>---</td>
<td>---</td>
</tr>
<tr>
<td>3 850</td>
<td>113</td>
<td>113</td>
<td>106</td>
<td>95</td>
<td>79</td>
<td>58</td>
<td>38</td>
<td>---</td>
</tr>
<tr>
<td>3 500</td>
<td>113</td>
<td>113</td>
<td>110</td>
<td>102</td>
<td>89</td>
<td>75</td>
<td>59</td>
<td>47</td>
</tr>
<tr>
<td>3 000</td>
<td>113</td>
<td>113</td>
<td>113</td>
<td>107</td>
<td>97</td>
<td>84</td>
<td>70</td>
<td>61</td>
</tr>
<tr>
<td>≤2 500</td>
<td>113</td>
<td>113</td>
<td>113</td>
<td>112</td>
<td>104</td>
<td>95</td>
<td>84</td>
<td>76</td>
</tr>
</tbody>
</table>

For centre of gravity between 3.00 m and 3.15 m (single-seat flight):

Instruments with:
- metric units: same limitation as above but V\(_{NE}\) limited to 200 km/h.
- imperial units: same limitation as above but V\(_{NE}\) limited to 108 kt.
9. Rotor Speed Limitations
   Maximum 420 rpm
   Minimum 270 rpm
   Max continuous 353 rpm

10. Maximum Operating Altitude and Temperature
   10.1 Altitude
       Enroute: 23 000 ft (7 000 m) PA
       Take-off: 19 000 ft (5 800 m) PA
       Restart in flight: 19 000 ft (5 800 m) PA
   10.2 Temperature
       -40 °C to +55 °C

11. Operating Limitations
    VFR day
    VFR night, when the additional equipment required by operational regulations is installed and serviceable
    For more information refer to approved RFM.

12. Maximum Mass
    TKOF/LD (with non-releasable loads):
        1 950 kg (4 299 lb)
        with gear 315 A 46.10.000
    TKOF/LD (with releasable loads):
        2 300 kg (5 071 lb)
    Max sling load: 1 000 kg (2 205 lb)

13. Centre of Gravity Range
    Longitudinal C.G. limits
    Forward limit: 2 760 mm
    Aft limit: 3 000 mm
    Or:
    Forward limit: 3 000 mm
    Aft limit: 3 150 mm
    with the following Maximum translation speed limitation and mass limited to 1 750 kg (3 858 lb):
    For C.G. between 2 760 mm to 3 000 mm:
        - 210 km/h (113 kt)
    For C.G. between 3 000 mm to 3 150 mm:
        - 200 km/h (108 kt)
    Lateral C.G Limits
    LH limit: 135 mm
    RH limit: 43 mm

14. Datum
    Longitudinal:
    3 000 mm (9.84 ft) forward of main rotor centre line
    Lateral: aircraft symmetry plane

15. Levelling Means
    4 levelling legs on the central structure:
    - 2 on the front of the aircraft
    - 2 on the rear of the aircraft

16. Minimum Flight Crew
    1 pilot at STA +1 340 mm

17. Maximum Passenger Seating Capacity
    Four
    Front seats: 1 passenger at STA +1 340 mm
    Rear seats: 3 passengers at STA +2 100 mm

18. Passenger Emergency Exit
    Refer to approved RFM
19. Maximum Baggage/ Cargo Loads

<table>
<thead>
<tr>
<th>Configuration</th>
<th>Baggage/Cargo location</th>
<th>Max load</th>
<th>Station</th>
</tr>
</thead>
<tbody>
<tr>
<td>5 passengers – 80 kg each</td>
<td>Under the rear bench</td>
<td>100 kg (220 lb)</td>
<td>+2 200 mm</td>
</tr>
<tr>
<td>1 pilot + 1 passenger on front seats</td>
<td>Behind the front seats with the rear bench folded up</td>
<td>230 kg (507 lb)</td>
<td>+1 900 mm</td>
</tr>
</tbody>
</table>

20. Rotor Blade Control Movement

For rigging information refer to the Maintenance Manual

21. Auxiliary Power Unit (APU)

n/a

22. Life-limited Parts

The periods specified in the latest approved revision of the Airworthiness Limitations section of the Maintenance Manual must not be exceeded.

IV. Operating and Service Instructions

1. Flight Manual

SA 315 B Flight Manual, original edition approved by DGAC-FR for type certification, or later DGAC-FR / EASA approved revision


SA 315 B Maintenance Manual


not recorded


not recorded

5. Illustrated Parts Catalogue

not recorded

6. Miscellaneous Manuals

not recorded

7. Service Letters and Service Bulletins

As published by Aérospatiale, Eurocopter or Airbus Helicopters

8. Required Equipment

As per compliance with applicable requirements and in accordance with the original Type Design standard; refer to approved Flight Manual.

V. Notes

1. Manufacturer’s eligible serial numbers:

All s/n complying with SA 315 B type (refer to III.1. Type Design Definition).

2. The certified “optional” installations are each approved independently of the basic helicopter and an approved Flight Manual Supplement is associated to each optional installation, if necessary.

3. Commercial designation:

ALOUETTE III LAMA

* * *
SECTION 5: ADMINISTRATIVE

I. Acronyms and Abbreviations

<table>
<thead>
<tr>
<th>Abbreviation</th>
<th>Definition</th>
</tr>
</thead>
<tbody>
<tr>
<td>°C</td>
<td>Degree Celsius</td>
</tr>
<tr>
<td>C.G.</td>
<td>Centre of Gravity</td>
</tr>
<tr>
<td>CR</td>
<td>(European) Commission Regulation</td>
</tr>
<tr>
<td>EU</td>
<td>European Union</td>
</tr>
<tr>
<td>LDG</td>
<td>Landing</td>
</tr>
<tr>
<td>Max</td>
<td>Maximum</td>
</tr>
<tr>
<td>MCP</td>
<td>Maximum Continuous Power</td>
</tr>
<tr>
<td>MGB</td>
<td>Main Gear Box</td>
</tr>
<tr>
<td>n/a</td>
<td>not applicable</td>
</tr>
<tr>
<td>n°</td>
<td>Number</td>
</tr>
<tr>
<td>OSD</td>
<td>Operational Suitability Data</td>
</tr>
<tr>
<td>PA</td>
<td>Pressure Altitude</td>
</tr>
<tr>
<td>PWR</td>
<td>Power</td>
</tr>
<tr>
<td>RFM</td>
<td>Rotorcraft Flight Manual</td>
</tr>
<tr>
<td>RGB</td>
<td>Rear Gear Box</td>
</tr>
<tr>
<td>rpm</td>
<td>Rounds per minute</td>
</tr>
<tr>
<td>s/n</td>
<td>Serial Number</td>
</tr>
<tr>
<td>sec</td>
<td>Seconds</td>
</tr>
<tr>
<td>STA</td>
<td>Station</td>
</tr>
<tr>
<td>TC</td>
<td>Type Certificate</td>
</tr>
<tr>
<td>TCDS</td>
<td>Type Certificate Data Sheet</td>
</tr>
<tr>
<td>TKOF</td>
<td>Take-Off</td>
</tr>
<tr>
<td>TOP</td>
<td>Take-Off Power</td>
</tr>
<tr>
<td>VFR</td>
<td>Visual Flight Rules</td>
</tr>
<tr>
<td>V_{NE}</td>
<td>Never Exceed Speed</td>
</tr>
<tr>
<td>OSD</td>
<td>Operational Suitability Data</td>
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<tr>
<td>PA</td>
<td>Pressure Altitude</td>
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<td>PWR</td>
<td>Power</td>
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II. Type Certificate Holder Record

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<th>Type Certificate Holder</th>
<th>Period</th>
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<tr>
<td>Sud Aviation</td>
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<tr>
<td>37, Boulevard de Montmorency</td>
<td>75016 Paris, France</td>
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<tr>
<td>Aérospatiale</td>
<td>From 1 January 1970 until 31 December 1991</td>
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<tr>
<td>37, Boulevard de Montmorency</td>
<td>75781 Paris CEDEX 16, France</td>
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<tr>
<td>Eurocopter France</td>
<td>From 1 January 1992 until 30 May 1997</td>
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<tr>
<td>Aéroport International Marseille – Provence</td>
<td>13725 Marignane CEDEX, France</td>
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<tr>
<td>Eurocopter</td>
<td>From 1 June 1997 until 6 January 2014</td>
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<tr>
<td>Aéroport International Marseille – Provence</td>
<td>13725 Marignane CEDEX, France</td>
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<tr>
<td>Airbus Helicopters</td>
<td>Since 7 January 2014</td>
</tr>
<tr>
<td>Aéroport International Marseille – Provence</td>
<td>13725 Marignane CEDEX, France</td>
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III. Change Record

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<th>Issue</th>
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<th>TC issue</th>
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<td>Issue 01</td>
<td>27 Jan 2010</td>
<td>Initial issue of EASA TCDS</td>
<td>Re-issued on 27 January 2010</td>
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<tr>
<td>Issue 02</td>
<td>7 Jan 2014</td>
<td>The company name has been changed to AIRBUS HELICOPTERS</td>
<td>Re-issued on 7 January 2014</td>
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<tr>
<td>Issue 03</td>
<td>14 Feb 2017</td>
<td>New TCDS template, reference to OSD, minor editorial corrections</td>
<td>- - -</td>
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- end of file -