European Aviation Safety Agency

EASA

TYPE-CERTIFICATE DATA SHEET

/EASA TC No.: R.008/

AS 350 B
AS 350 D
AS 350 B1
AS 350 B2
AS 350 BA
AS 350 BB
AS 350 B3
EC 130 B4
EC 130 T2

Type Certificate Holder: Eurocopter
Aéroport International Marseille – Provence
13725 Marignane cedex
France

Issue 6: 25.05.2012

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AS 350 B

I. General

1. Data Sheet No: EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005

2. Type / Model / Variant
   AS 350 B (commercial designation: ECUREUIL)

3. Airworthiness Category: Small Rotorcraft

4. Manufacturer:
   EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France

5. DGAC-F Certification Date: October 27, 1977

6. DGAC-F Application Date: June 19, 1974

II. Certification Basis

1. Airworthiness Requirements: FAR Part 27, Amendments 1 to 10 included

2. Special Conditions:
   Complementary and special conditions defined in DGAC-F letters 6518 dated
   August 17, 1976 and 6437 dated July 28, 1977

III. Technical Characteristics and Operational Limitations

1. Description: Small single-engine helicopter

2. Equipment:
   The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320.
   
   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must
   be installed on the aircraft at certification time and at every time after certification.

3. Dimensions:
   Fuselage:
   - Length 10.93 m (35.86 ft)
   - Width 1.87 m (6.14 ft)
   - Height 3.14 m (10.30 ft)
   
   Main Rotor:
   - 3 blades – Diameter 10.69 m (35.07 ft)
   
   Tail Rotor:
   - 2 blades – Diameter 1.86 m (6.10 ft)

4. Engine:
   TURBOMECA ARRIEL 1B
   EASA Engine TCDS E.073 (Superseding DGAC-F Engine TCDS No M5)

4.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed (NG)**</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td></td>
<td></td>
<td>105 %</td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min.)</td>
<td>829 m.N</td>
<td>478 kW</td>
<td>100 %</td>
<td>810°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>829 m.N</td>
<td>440 kW</td>
<td>98 %</td>
<td>775°C</td>
</tr>
</tbody>
</table>
Maximum T4 on starting: 840°C
* ISA, ground level
** 100% = 51 800 rpm

4.2. Transmission Torque Limits:

Maximum torque: 83% (100% torque corresponds to 396 kW power output at 386 rpm main rotor speed)

5. Fluids (Fuel/Oil/Additives):

5.1. Fuel: Refer to Flight Manual

5.2. Oil: Refer to Flight Manual

6. Fluid capacities:

6.1. Fuel: Total: 540 l (142.7 US gal)
Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289
Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

6.2. Oil: Engine reservoir: 5.2 l (1.37 US gal)
BTP: 6.5 l (1.72 US gal) - circuit included
BTA: 0.33 l (0.09 US gal)

7. Airspeed limits:

Never exceed speed: Vne - 272 km/h (147 kt) from 0 to 330 m (1083 ft), then decreasing by 20 km/h (11 kt) per 1000 m (3281 ft) from 330 m (1083 ft).

When OAT is between -30°C and -40°C, 18.5 km/h have to be subtracted from the above decreasing law.

8. Rotor Speed Limits:

Power-on flight: 385 +1 -5 rpm
In autorotation:
- Maximum: 424 rpm
- Minimum: 320 rpm (audio warning at 335 rpm)

The audio warning sounds when rotor speed drops below:
- 335 rpm pre-modification 07.1891
- 360 rpm post-modification 07.1891

9. Maximum Operating Altitude and Temperature:

En route: 4875 m (16000ft)

9.2. Temperature: Refer to Flight Manual

10. Operating Limitations:

- Day VFR flight
- Night VFR flight, when the additional equipment required by operational regulations is installed and serviceable.

(For more information refer to Flight Manual)

11. Maximum Certified Weight: 1950 kg (4300 lb)

12. Centre of Gravity Range:

Longitudinal: Forward limit: 3.17 m (124.8 in)
At limit: 3.55 m (139.8 in) up to 1300 kg (2866 lb)
3.43 m (135 in) for 1900 kg (4189 lb) and up to 1950 kg (4300 lb)
Linear variation between the points

Lateral: L.H. limit: 0.15 m (5.90 in)
R.H. limit: 0.08 m (3.14 in)

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

13. Datum:

Longitudinal: 3.4 m (134 in) forward of the MRH centreline
Lateral: Aircraft symmetry plane

14. Levelling Means: Transmission deck
15. Minimum Flight Crew: 1 pilot in R.H. seat


When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

17. Passenger Emergency Exit: Refer to Flight Manual

18. Maximum Baggage/Cargo Loads:

<table>
<thead>
<tr>
<th>Location</th>
<th>Max Load</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum load in R.H. side hold</td>
<td>100 kg (220 lb)</td>
</tr>
<tr>
<td>Maximum load in L.H. side hold</td>
<td>120 kg (264 lb)</td>
</tr>
<tr>
<td>Maximum load in rear hold</td>
<td>80 kg (176 lb)</td>
</tr>
<tr>
<td>Maximum load in forward cabin floor</td>
<td>150 kg (331 lb)</td>
</tr>
<tr>
<td>Maximum load in rear cabin floor</td>
<td>310 kg (683 lb)</td>
</tr>
</tbody>
</table>

19. Rotor blade and control movement: Refer to aircraft Maintenance Manual

20. Life-limited parts:

The AS 350 Maintenance Manual Chapter 5 "Master Servicing Recommendations" have been deemed acceptable to carry out satisfactory maintenance operations on the AS 350 helicopter; part 5.99 "Airworthiness Limitations", originally approved by DGAC-F and subsequently by EASA, contains limitations which are mandatory.

IV. Operating and Service Instructions


2. Maintenance Manual, Document No:
   - AS 350 Service Manual
   - AS 350 Overhaul Manual
   - AS 350 Repair Manual
   - AS 350 Illustrated Parts Catalogue

Compatibility between optional items of equipment is described:
   - From an installation aspect: in the "Master Servicing Recommendations".

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by DGAC-F and EASA.

V. Notes

1. Eligible serial numbers:
   - AS 350 B aircraft S/N 1003 and up.
   - AS 350 D aircraft converted into AS 350 B by application of Service Bulletin 01.00.12.

2. Placards:
   1. The following placard must be fitted in a way that the pilot can see it clearly:

   - LES REPERES ET PLAQUETTES INDICATRICES INSTALLES SUR CET HELICOPTERE CONTIENNENT LES LIMITATIONS D'UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L'UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D'UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L'UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION "LIMITATIONS DE NAVIGABILITE" DU MANUEL D'ENTRETIEN DU GIRAVION DOIT ETRE RESPECTEE.

   - THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE "AIRWORTHINESS LIMITATIONS" SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH.

   2. Refer to the Flight Manual as regards the other placards.
AS 350 C

For memory:

Certification issued on September 2, 1977 and cancelled on June 1, 1997 following the cancellation of the one of LYCOMING LTS 101-600 A turboshaft engine on April 20, 1987.
AS 350 D

I. General

1. Data Sheet No: EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005

2. Type / Model / Variant
   AS 350 D (commercial designation: ASTAR)

3. Airworthiness Category: Small Rotorcraft

4. Manufacturer:
   EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France

5. DGAC-F Certification Date: July 4, 1978

6. DGAC-F Application Date: March 28, 1978

II. Certification Basis

1. Airworthiness Requirements: FAR Part 27, Amendments 1 to 10 included

2. Special Conditions: Complementary and special conditions defined in DGAC-F letters 6518 dated August 17, 1976 and 6437 dated July 28, 1977

III. Technical Characteristics and Operational Limitations

1. Description: Small single-engine helicopter

2. Equipment: The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320
   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must be installed on the aircraft at certification time and at every time after certification.

3. Dimensions:
   Fuselage: Length 10.93 m (35.86 ft)
              Width 1.87 m (6.14 ft)
              Height 3.14 m (10.30 ft)
   Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)
   Tail Rotor: 2 blades – Diameter 1.83 m (6.10 ft)

4. Engine:
   LYCOMING LTS 101-600A-2
   DGAC-F Engine TCDS No M.IM 5

4.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power</th>
<th>Generator speed (NG)**</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient</td>
<td>105.6 %</td>
<td>843°C***</td>
<td></td>
<td></td>
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<tr>
<td>(5 sec.)</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff</td>
<td>103.7 %</td>
<td>782°C</td>
<td></td>
<td></td>
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<tr>
<td>(5 min.)</td>
<td>733 m.N 459 kW</td>
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</tr>
<tr>
<td>Maximum continuous</td>
<td>102.2 %</td>
<td>763°C</td>
<td></td>
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<tr>
<td>704 m.N 440 kW</td>
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</tbody>
</table>
The installed engine limitations at maximum continuous power are:

- **NG = 48 930 rpm**
- **T4 = 755°C**

Maximum T4 on take-off: 899°C***

* ISA, ground level
** 100% = 47 866 rpm
*** Operating time with temperature above 818°C: 1.2 s

4.2. Transmission Torque Limits:

- Maximum torque: 101% (100% torque corresponds to 396 kW power output at 386 rpm main rotor speed)

5. Fluids (Fuel/Oil/Additives):

5.1. Fuel: Refer to Flight Manual

5.2. Oil: Refer to Flight Manual

6. Fluid capacities:

6.1. Fuel:
- Total: 540 l (142.7 US gal)
- Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289
- Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

6.2. Oil:
- Engine reservoir: 4 l (0.88 US gal)
- BTP: 6.5 l (1.72 US gal) - circuit included
- BTA: 0.33 l (0.09 US gal)

7. Airspeed limits:

Never exceed speed: Vne - 272 km/h (147 kt) from 0 to 330 m (1083 ft), then decreasing by 20 km/h (11 kt) per 1000 m (3281 ft) from 330 m (1083 ft).

When OAT is between -30°C and -40°C, 18.5 km/h have to be subtracted from the above decreasing law.

8. Rotor Speed Limits:

Power-on flight: 385 rpm

\[ +1 \text{ rpm} \]

In autorotation:
- Maximum: 424 rpm
- Minimum: 320 rpm (audio warning below 335 rpm)

9. Maximum Operating Altitude and Temperature:

9.1. Altitude:
- Take-off and landing: Refer to Flight Manual
- En route: 4575 m (15 000 ft)

9.2. Temperature: Refer to Flight Manual

10. Operating Limitations:

- Day VFR flight
- Night VFR flight, when the additional equipment required by operational regulations is installed and serviceable.

(For more information refer to Flight Manual)

11. Maximum Certified Weight: 1950 kg (4300 lb)

12. Centre of Gravity Range:

| Longitudinal: | Forward limit: | 3.17 m (124.8 in) |
|              | Aft limit:     | 3.55 m (139.8 in) up to 1300 kg (2866 lb) |
|              | 3.43 m (135 in) for 1900 kg (4189 lb) and up to 1950 kg (4300 lb) |
| Linear variation between the points |

| Lateral: | L.H. limit: | 0.15 m (5.90 in) |
|          | R.H. limit: | 0.08 m (3.14 in) |

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

13. Datum:

| Longitudinal: | 3.4 m (134 in) forward of the MRH centreline |
| Lateral:      | Aircraft symmetry plane |
14. Levelling Means:  
Transmission deck

15. Minimum Flight Crew:  
1 pilot in R.H. seat

16. Maximum Passenger Seating Capacity:  
5

When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

17. Passenger Emergency Exit:  
Refer to Flight Manual

18. Maximum Baggage/Cargo Loads:

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19. Rotor blade and control movement:  
Refer to aircraft Maintenance Manual

20. Life-limited parts:

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IV. Operating and Service Instructions

1. Rotorcraft Flight Manual, Document No:  

2. Maintenance Manual, Document No:
- AS 350 Service Manual
- AS 350 Overhaul Manual
- AS 350 Repair Manual
- AS 350 Illustrated Parts Catalogue

Compatibility between optional items of equipment is described:
- From an installation aspect: in the "Master Servicing Recommendations".

3. Service Letters and Service Bulletins:  
As published by Eurocopter and approved by DGAC-F and EASA.

V. Notes

1. Eligible serial numbers:
- AS 350 D aircraft S/N 1028 and up.
- AS 350 C aircraft converted into AS 350 D by application of Service Bulletin 01.01

2. Placards:

1 - The following placard must be fitted in a way that the pilot can see it clearly:

- LES REPERES ET PLAQUETTES INDICATRICES INSTALLES SUR CET HELICOPTERE CONTIENNENT LES LIMITATIONS D’UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L’UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D’UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L’UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION “LIMITATIONS DE NAVIGABILITE” DU MANUEL D’ENTRETIEN DU GIRAVION DOIT ETRE RESPECTEE.
- THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THE ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE “AIRWORTHINESS LIMITATIONS” SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH

2 - Refer to the Flight Manual as regards the other placards.

Page 9 of 42
AS 350 D1

For memory:

AS 350 B1

I. General

1. Data Sheet No: EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005

2. Type / Model / Variant AS 350 B1 (commercial designation: ECUREUIL)

3. Airworthiness Category: Small Rotorcraft

4. Manufacturer: EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France

5. DGAC-F Certification Date: January 9, 1986

6. DGAC-F Application Date: December 13, 1984

II. Certification Basis

1. Airworthiness Requirements: FAR Part 27, Amendments 1 to 10 included

2. Special Conditions: Complementary and special conditions defined in DGAC-F letters 6518 dated August 17, 1976, 6437 dated July 28, 1977 and 53639 dated June 25, 1985

III. Technical Characteristics and Operational Limitations

1. Type design definition: Documents 350A.04.4455

2. Description: Small single-engine helicopter designed as a derivative product of the former type certified model AS 350 B.

3. Equipment:

   The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320.

   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must be installed on the aircraft at certification time and at every time after certification.

4. Dimensions:

   Fuselage: Length 10.93 m (35.86 ft)
              Width 1.87 m (6.14 ft)
              Height 3.14 m (10.30 ft)

   Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)

   Tail Rotor: 2 blades – Diameter 1.86 m (6.10 ft)

5. Engine: TURBOMECA ARRIEL 1D
   EASA Engine TCDS E.073 (Superseding DGAC-F Engine TCDS No M5)
5.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed (NG)**</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td>83 m.daN</td>
<td>510 kW</td>
<td>100.8 +0.4 +0 %</td>
<td>845°C</td>
</tr>
<tr>
<td>Maximum at takeoff (5 min.)</td>
<td>83 m.daN</td>
<td>450 kW</td>
<td>98 %</td>
<td>795°C</td>
</tr>
</tbody>
</table>

* ISA, ground level  
** 100 % = 51 800 rpm

5.2. Transmission Torque Limits:

Maximum torque:
- IAS 40kt – 74 km/h – 46 MPH or higher: 94%
- IAS below 40kt – 74 km/h – 46 MPH: 100%
(100% torque corresponds to 488 kW power output at 394 rpm main rotor speed/or 478 kW at 386 rpm/)

6. Fluids (Fuel/Oil/Additives):

6.2. Oil: Refer to Flight Manual

7. Fluid capacities:

7.1. Fuel:  
- Total: 540 l (142.7 US gal)  
- Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289  
- Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

7.2. Oil:  
- Engine reservoir: 5.2 l (1.37 US gal)  
- BTP: 6.5 l (1.72 US gal) - circuit included  
- BTA: 0.33 l (0.09 US gal)

8. Airspeed limits:

Power-on VNE:  
- Absolute VNE: 287 km/h (155 kt) for HP = 0  
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).  
- In cold weather, when OAT is below - 30°C, subtract 19 km/h (10 kt) from the above VNE.

Power-off VNE:  
- Absolute VNE: 231 km/h (125 kt) for HP = 0.  
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).  
- In cold weather, subtract the following values from the above VNE:  
  - 19 km/h (10 kt), when OAT is between - 20°C and - 30°C  
  - 37 km/h (20 kt), when OAT is below - 30°, with out VNE being less than 120 km/h (65 kt).

9. Rotor Speed Limits:

Power-on flight: 390 +4 -5 rpm  
In autorotation:  
- Maximum: 430 rpm  
- Minimum: 320 rpm (audio warning below 365 rpm)

10. Maximum Operating Altitude and Temperature:

10.1. Altitude HP:  
- Take-off and landing: 4267 m (14 000 ft)  
- En route: 6096 m (20 000 ft)

10.2. Temperature: Refer to Flight Manual

11. Operating Limitations:  
- Day VFR flight  
- Night VFR flight, when the additional equipment required by operational regulations is installed and serviceable.  
(For more information refer to Flight Manual)

12. Maximum Certified Weight: 2200 kg (4850 lb)
13. Centre of Gravity Range:

**Longitudinal:**
- **Forward limit:** 3.17 m (124.8 in) up to 2000 kg (4409 lb)
- Linear variation from 3.17 m (124.8 in) to 3.20 m (126 in) between 2000 kg (4409 lb) and 2200 kg (4850 lb)
- 3.20 m (126 in) at 2200 kg (4850 lb)
- **Aft limit:** 3.50 m (137.8 in) up to 1200 kg (2645 lb)
- Linear variation from 3.50 m (137.8 in) to 3.43 m (135 in) between 1200 kg (2645 lb) and 2200 kg (4850 lb)
- 3.43 m (135 in) at 2200 kg (4850 lb)

**Lateral:**
- **L.H. limit:** 0.18 m (7.08 in)
- **R.H. limit:** 0.14 m (5.51 in)

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

14. Datum:
- **Longitudinal:** 3.4 m (134 in) forward of the MRH centreline
- **Lateral:** Aircraft symmetry plane

15. Levelling Means:
- Transmission deck

16. Minimum Flight Crew:
- 1 pilot in R.H. seat

17. Maximum Passenger Seating Capacity:
- 5

When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

18. Passenger Emergency Exit:
- Refer to Flight Manual

19. Maximum Baggage/Cargo Loads:

<table>
<thead>
<tr>
<th>Location</th>
<th>Max Load</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum load in R.H. side hold:</td>
<td>100 kg (220 lb)</td>
</tr>
<tr>
<td>Maximum load in L.H. side hold:</td>
<td>120 kg (264 lb)</td>
</tr>
<tr>
<td>Maximum load in rear hold:</td>
<td>80 kg (176 lb)</td>
</tr>
<tr>
<td>Maximum load in forward cabin floor:</td>
<td>150 kg (331 lb)</td>
</tr>
<tr>
<td>Maximum load in rear cabin floor:</td>
<td>310 kg (683 lb)</td>
</tr>
</tbody>
</table>

20. Rotor blade and control movement:
- Refer to aircraft Maintenance Manual

21. Life-limited parts:

The AS 350 Maintenance Manual Chapter 5 "Master Servicing Recommendations" have been deemed acceptable to carry out satisfactory maintenance operations on the AS 350 helicopter; part 5.99 "Airworthiness Limitations", originally approved by DGAC-F and subsequently by EASA, contains limitations which are mandatory.

IV. Operating and Service Instructions

1. Rotorcraft Flight Manual, Document No:

2. Maintenance Manual, Document No:
- AS 350 Service Manual
- AS 350 Overhaul Manual
- AS 350 Repair Manual
- AS 350 Illustrated Parts Catalogue

Compatibility between optional items of equipment is described:
- From an installation aspect: in the "Master Servicing Recommendations".

3. Service Letters and Service Bulletins:
- As published by Eurocopter and approved by DGAC-F and EASA.
V. Notes


2. Placards:

   1 - The following placard must be fitted in a way that the pilot can see it clearly:

   - LES REPÆRES ET PLAQUETTES INDICATRICES INSTALLEES SUR CET HELICOPTÈRE CONTIENNENT LES LIMITATIONS D'UTILISATION QUI DOIVENT ÊTRE RESPECTÉES LORS DE L'UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D'UTILISATION QUI DOIVENT ÊTRE RESPECTÉES LORS DE L'UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION "LIMITATIONS DE NAVIGABILITE" DU MANUEL D'ENTRETIEN DU GIRAVION DOIT ÊTRE RESPECTÉE.

   - THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THE ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE "AIRWORTHINESS LIMITATIONS" SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH

   2 - Refer to the Flight Manual as regards the other placards.
AS 350 B2

I. General

1. Data Sheet No: EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005
2. Type / Model / Variant AS 350 B2 (commercial designation: ECUREUIL)
3. Airworthiness Category: Small Rotorcraft
4. Manufacturer: For helicopters manufactured by Type Certificate holder:
   EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France
   For helicopters manufactured under license - see sub-paragraph V.1 – Eligible
   serial numbers
5. DGAC-F Certification Date: April 26, 1989
6. DGAC-F Application Date: October 6, 1988

II. Certification Basis

1. Airworthiness Requirements: FAR Part 27, Amendments 1 to 10 included
   For a/c equipped with VEMD major modification, as above plus:
   • special conditions on protection against the effects of High Intensity
     Radiated Fields (HIRF) and Lightning
   • equivalent safety findings for powerplant instrument markings
   See AS350B2 (VEMD) CRI n. A-1 issue 3 dated 17/11/06 for details

III. Technical Characteristics and Operational Limitations

1. Type design definition: Documents 350A.04.4541
2. Description: Small single-engine helicopter designed as a derivative product of the former type certified model AS 350 B1.
3. Equipment:
   The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320.
   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must be installed on the aircraft at certification time and at every time after certification.
4. Dimensions:
   - Fuselage: Length 10.93 m (35.86 ft)
     Width 1.87 m (6.14 ft)
     Height 3.14 m (10.30 ft)
   - Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)
   - Tail Rotor: 2 blades – Diameter 1.86 m (6.10 ft)
5. Engine: TURBOMECA ARRIEL 1 D1  
EASA Engine TCDS E.073 (Superseding DGAC-F Engine TCDS No M5)

5.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>B2 without VEMD Generator speed (NG)** (Ng)</th>
<th>B2 with VEMD Generator speed (NG)** (Ng)</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td>83 m.daN</td>
<td>478 kW***</td>
<td>107.5 % (+6)</td>
<td>103.1 % (+1)</td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min.)</td>
<td>83 m.daN</td>
<td>449 kW</td>
<td>98 % (-3.5)</td>
<td>98 % (-4)</td>
<td>795°C</td>
</tr>
<tr>
<td>Maximum Continuous</td>
<td>83 m.daN</td>
<td></td>
<td></td>
<td></td>
<td>845°C</td>
</tr>
</tbody>
</table>

* ISA, ground level  
** 100 % = 51 800 rpm  
*** The mechanical power has been limited to this value taking the fuel flow limit into account.

5.2. Transmission Torque Limits:

Max continuous torque: 94 %  
Take-off torque range from 0 to 40kt 94 to 100%  
Max take-off torque: 100 %  
Max transient torque (5s): 107 %

(100% torque corresponds to 478kW (641 shp) at 386 rpm main rotor speed)

6. Fluids (Fuel/Oil/Additives):

6.2. Oil: Refer to Flight Manual

7. Fluid capacities:

7.1. Fuel:  
Total: 540 l (142.7 US gal)  
Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289  
Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

7.2. Oil:  
Engine reservoir: 5.2 l (1.37 US gal)  
BTP: 6.5 l (1.72 US gal) - circuit included  
BTA: 0.33 l (0.09 US gal)

8. Airspeed limits:

Power-on VNE:  
- Absolute VNE: 287 km/h (155 kt) for HP = 0  
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft)  
- In cold weather, when OAT is below - 30°C, sub tract 19 km/h (10 kt) from the above VNE

Power-off VNE:  
- Absolute VNE: 231 km/h (125 kt) for HP = 0.  
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).  
- In cold weather, subtract the following values from the above VNE:  
  - 19 km/h (10 kt), when OAT is between -20°C a nd -30°C  
  - 37 km/h (20 kt), when OAT is below -30°, with out VNE being less than 120 km/h (65 kt)

9. Rotor Speed Limits:

Power-on flight: 390 +4 -5 rpm  
In autorotation:  
- Maximum: 430 rpm (audio warning above 410 rpm)  
- Minimum: 320 rpm (audio warning below 360 rpm)

10. Maximum Operating Altitude and Temperature:

10.1. Altitude HP:  
Take-off and landing: Refer to Flight Manual  
En route: 6096 m (20 000 ft)

10.2. Temperature: Refer to Flight Manual

11. Operating Limitations:

- Day VFR flight  
- Night VFR flight, when additional equipment required by operational regulations is installed and serviceable  
- Flight in falling snow: refer to Flight Manual

(For more information refer to Flight Manual)
12. Maximum Certified Weight: 2250 kg (4960 lb)

13. Centre of Gravity Range:

Longitudinal:

\[ \text{C.G. (m)} \]

\[ \begin{align*}
\text{Weight (kg)} & \quad 1.4 \quad 1.7 \quad 2.1 \quad 2.5 \quad 3.0 \quad 3.5 \quad 4.0 \quad 4.5 \\
\text{Weight (lbs)} & \quad 3.07 \quad 4.63 \quad 6.88 \quad 10.98 \quad 16.55 \quad 22.05 \quad 31.17 \quad 66.10
\end{align*} \]

Lateral: L.H. limit: 0.18 m (7.08 in)
R.H. limit: 0.14 m (5.51 in)

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

14. Datum:

Longitudinal: 3.4 m (134 in) forward of the MRH centreline
Lateral: aircraft symmetry plane

15. Levelling Means:

Transmission deck

16. Minimum Flight Crew:

1 pilot in R.H. seat

17. Maximum Passenger Seating Capacity: 5

When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

18. Passenger Emergency Exit:

Refer to Flight Manual

19. Maximum Baggage/Cargo Loads:

<table>
<thead>
<tr>
<th>Location</th>
<th>Max Load</th>
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<tr>
<td>Maximum load in R.H. side hold:</td>
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<td>Maximum load in L.H. side hold:</td>
<td>120 kg (264 lb)</td>
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<tr>
<td>Maximum load in rear hold:</td>
<td>80 kg (176 lb)</td>
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<tr>
<td>Maximum load in forward cabin floor:</td>
<td>150 kg (331 lb)</td>
</tr>
<tr>
<td>Maximum load in rear cabin floor:</td>
<td>310 kg (683 lb)</td>
</tr>
</tbody>
</table>

20. Rotor blade and control movement:

Refer to aircraft Maintenance Manual

21. Life-limited parts:

The AS 350 Maintenance Manual Chapter 5 "Master Servicing Recommendations" have been deemed acceptable to carry out satisfactory maintenance operations on the AS 350 helicopter ; part 5.99 "Airworthiness Limitations", originally approved by DGAC-F and subsequently by EASA, contains limitations which are mandatory.
IV. Operating and Service Instructions

1. Rotorcraft Flight Manual, Document No:
   
   For VEMD major modification:
   - AS 350 B2 (VEMD) Flight Manual, approved under ref. EASA.R.C 01396 on 22nd November 2006 or later approved revision (reference in English language)
   - AS 350 B2 (VEMD) Flight Manual, approved under ref. 10029919 on 03rd May 2010 or later approved revision (reference in French language)

2. Maintenance Manual, Document No:
   - AS 350 Overhaul Manual
   - AS 350 Repair Manual
   - AS 350 B2 Illustrated Parts Catalogue

 Compatibility between optional items of equipment is described:

   - From an installation aspect: in the "Master Servicing Recommendations"
   - From an operational aspect: in "Supplements" chapter of the Flight Manual

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by DGAC-F and EASA.

V. Notes

1. Eligible serial numbers:
   - AS 350 B2 aircraft S/N 2100 and up
   - AS 350 B1 aircraft converted into AS 350 B2 by application of Service Bulletin n°01.26 or 01.00.26
   - AS 350 B aircraft converted into AS 350 B2 by application of Service Bulletin n°01.00.51
   - AS 350 BA aircraft converted into AS 350 B2 by application of Service Bulletin n°01.00.50 or Service Bulletin n°01.90.61
   - AS 350 B2 with VEMD major modification: aircraft S/N 4129 and up

The aircrafts whose s/n is listed in Eurocopter document L102-001 are manufactured under Helibras license and those in L 102-002 under AE-MS license.

2. Placards:

   1 - The following placard must be fitted in a way that the pilot can see it clearly:

   2 - Refer to the Flight Manual as regards the other placards.

   - LES REPERES ET PLAQUETTES INDICATRICES INSTALLES SUR CET HELICOPTERE CONTIENNENT LES LIMITATIONS D’UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L’UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D’UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L’UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION "LIMITATIONS DE NAVIGABILITE" DU MANUEL D’ENTRETIEN DU GIRAVION DOIT ETRE RESPECTEE.

   - THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THE ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE "AIRWORTHINESS LIMITATIONS" SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH.

   2 - Refer to the Flight Manual as regards the other placards.
I. General

1. Data Sheet No: EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005
2. Type / Model / Variant: AS 350 BA (commercial designation: ECUREUIL)
3. Airworthiness Category: Small Rotorcraft
4. Manufacturer:
   For helicopters manufactured by Type Certificate holder:
   EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France
   For helicopters manufactured under license - see sub-paragraph V.1 – Eligible serial numbers
5. DGAC-F Certification Date: November 26, 1991
6. DGAC-F Application Date: May 17, 1991

II. Certification Basis

1. Airworthiness Requirements: FAR Part 27, Amendments 1 to 10 included

III. Technical Characteristics and Operational Limitations

1. Type design definition: Document 350A.04.4685
2. Description: Small single-engine helicopter designed as a derivative product of the former type certified models AS 350 B1 and B2.
3. Equipment:
   The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320.
   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must be installed on the aircraft at certification time and at every time after certification.
4. Dimensions:
   Fuselage: Length 10.93 m (35.86 ft)
   Width 1.87 m (6.14 ft)
   Height 3.14 m (10.30 ft)
   Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)
   Tail Rotor: 2 blades – Diameter 1.86 m (6.10 ft)
5. **Engine:**

TURBOMECA ARRIEL 1B

EASA Engine TCDS E.073 (Superseding DGAC-F Engine TCDS No M5)

### 5.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed (NG)**</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td>105 %</td>
<td></td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min.)</td>
<td>478 kW</td>
<td>100 %</td>
<td>810°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>440 kW</td>
<td>98 %</td>
<td>775°C</td>
</tr>
</tbody>
</table>

Maximum T4 starting: 840°C

* ISA, ground level

** 100 % = 51 800 rpm

### 5.2. Transmission Torque Limits:

Maximum torque:

- IAS 40kt – 74 km/h – 46 MPH or higher: 83%
- IAS below 40kt – 74 km/h – 46 MPH: 88%

(100% torque corresponds to 478 kW at 386 rpm main rotor speed)

### 6. Fluids (Fuel/Oil/Additives):

**6.1. Fuel:** Refer to Flight Manual.

**6.2. Oil:** Refer to Flight Manual.

### 7. Fluid capacities:

**7.1. Fuel:**

- Total: 540 l (142.7 US gal)
- Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289
- Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

**7.2. Oil:**

- Engine reservoir: 5.2 l (1.37 US gal)
- BTP: 6.5 l (1.72 US gal) - circuit included
- BTA: 0.33 l (0.09 US gal)

### 8. Airspeed limits:

**Power-on VNE:**

- Absolute VNE: 287 km/h (155 kt) for HP = 0
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).
- In cold weather, when OAT is below - 30°C, subtract 19 km/h (10 kt) from the above VNE.

**Power-off VNE**

- Absolute VNE: 231 km/h (125 kt) for HP = 0.
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).
- In cold weather, subtract the following values from the above VNE:
  - 19 km/h (10 kt), when OAT is between - 20°C and - 30°C
  - 37 km/h (20 kt), when OAT is below - 30°C, with out VNE being less than 120 km/h (65 kt).

### 9. Rotor Speed Limits:

**Power-on flight:**

- Maximum: 430 rpm (audio warning above 410 rpm)
- Minimum: 320 rpm (audio warning below 360 rpm)

**In autorotation:**

- Maximum: 430 rpm (audio warning above 410 rpm)
- Minimum: 320 rpm (audio warning below 360 rpm)

### 10. Maximum Operating Altitude and Temperature:

**10.1. Altitude HP:**

- Take-off and landing: Refer to Flight Manual
- En route: 4875 m (16 000 ft)

**10.2. Temperature:** Refer to Flight Manual

### 11. Operating Limitations:

- Day VFR flight
- Night VFR flight, when the additional equipment required by operational regulations is installed and serviceable.

(For more information refer to Flight Manual)

### 12. Maximum Certified Weight:

2100 kg (4630 lb)
13. Centre of Gravity Range:

Longitudinal:

Lateral:  
L.H. limit:  0.18 m (7.08 in)  
R.H. limit:  0.14 m (5.51 in)

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

14. Datum:

Longitudinal:  3.4 m (134 in) forward of the MRH centreline
Lateral:  Aircraft symmetry plane

15. Levelling Means:

Transmission deck

16. Minimum Flight Crew:

1 pilot in R.H. seat

17. Maximum Passenger Seating Capacity:

5

When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

18. Passenger Emergency Exit:

Refer to Flight Manual

19. Maximum Baggage/Cargo Loads:

<table>
<thead>
<tr>
<th>Location</th>
<th>Max Load</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum load in R.H. side hold:</td>
<td>100 kg (220 lb)</td>
</tr>
<tr>
<td>Maximum load in L.H. side hold:</td>
<td>120 kg (264 lb)</td>
</tr>
<tr>
<td>Maximum load in rear hold:</td>
<td>80 kg (176 lb)</td>
</tr>
<tr>
<td>Maximum load in forward cabin floor:</td>
<td>150 kg (331 lb)</td>
</tr>
<tr>
<td>Maximum load in rear cabin floor:</td>
<td>310 kg (683 lb)</td>
</tr>
</tbody>
</table>

20. Rotor blade and control movement:

refer to aircraft Maintenance Manual

21. Life-limited parts:

The AS 350 Maintenance Manual Chapter 5 "Master Servicing Recommendations" have been deemed acceptable to carry out satisfactory maintenance operations on the AS 350 helicopter ; part 5.99 "Airworthiness Limitations", originally approved by DGAC-F and subsequently by EASA, contains limitations which are mandatory.
IV. Operating and Service Instructions


   - AS 350 Overhaul Manual
   - AS 350 Repair Manual
   - AS 350 Illustrated Parts Catalogue

Compatibility between optional items of equipment is described:
   - From an installation aspect: in the "Master Servicing Recommendations".

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by DGAC-F and EASA.

V. Notes

1. Eligible serial numbers: - AS 350 BA aircraft S/N 2588 and up.
   - AS 350 B aircraft converted into AS 350 BA by application of Service Bulletin n°01.00.35
   - AS 350 D aircraft converted into AS 350 BA by application of Service Bulletin n°01.00.40.

The aircrafts whose s/n is listed in Eurocopter document L102-001 are manufactured under Helibras and those in L102-002 under AE-MS license.

2. Placards:
   1. The following placard must be fitted in a way that the pilot can see it clearly:

   - LES REPERES ET PLAQUETTES INDICATRICES INSTALLES SUR CET HELICOPTERE CONTIENNENT LES LIMITATIONS D'UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L'UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D'UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L'UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION "LIMITATIONS DE NAVIGABILITE" DU MANUEL D'ENTRETIEN DU GIRAVION DOIT ETRE RESPECTEE.
   - THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THE ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE "AIRWORTHINESS LIMITATIONS" SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH.

   2. Refer to the Flight Manual as regards the other placards.
AS 350 BB

I. General

1. Data Sheet No: EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005

2. Type / Model / Variant AS 350 BB (commercial designation: ECUREUIL)

3. Airworthiness Category: Small Rotorcraft

4. Manufacturer: EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIIGNANE Cedex
   France

5. DGAC-F Certification Date: November 15, 1996

6. DGAC-F Application Date: July 23, 1996

II. Certification Basis

1. Airworthiness Requirements: FAR Part 27, Amendments 1 to 10 included


III. Technical Characteristics and Operational Limitations

1. Description: Small single-engine helicopter designed as a derivative product of the former types certified model AS 350 BA.

2. Equipment:

   The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320.

   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must be installed on the aircraft at certification time and at every time after certification.

3. Dimensions:

   Fuselage:
   - Length 10.93 m (35.86 ft)
   - Width 1.87 m (6.14 ft)
   - Height 3.14 m (10.30 ft)

   Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)

   Tail Rotor: 2 blades – Diameter 1.86 m (6.10 ft)

4. Engine:
   TURBOMECA ARRIEL 1D1 with TU 221
   EASA Engine TCDS E.073 (Superseding DGAC-F Engine TCDS No M5)
4.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed (NG)**</th>
<th>T4*** temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td></td>
<td>107.5 %</td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min.)</td>
<td>478 kW</td>
<td>98.5%</td>
<td>845°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>428 kW</td>
<td>96.5 %</td>
<td>795°C</td>
</tr>
</tbody>
</table>

* ISA, ground level
** Minimum stabilized rating: 67 % - 100 % = 51 800 rpm
*** Maximum transient during starting: 865°C

4.2. Transmission Torque Limits:

- Maximum continuous: 88 % for IAS < 60 kt
- Maximum transient: 107 % for IAS < 40 kt
  (88% torque corresponds to 420 kW power output at 386 rpm main rotor speed /or 429 kW at 394 rpm/)

5. Fluids (Fuel/Oil/Additives):


5.2. Oil: refer to Flight Manual.

6. Fluid capacities:

6.1. Fuel:
- Total: 540 l (142.7 US gal)
- Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289
- Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

6.2. Oil:
- Engine reservoir: 5.2 l (1.37 US gal)
- BTP: 6.5 l (1.72 US gal) - circuit included
- BTA: 0.33 l (0.09 US gal)

7. Airspeed limits:

Power-on VNE:
- Absolute VNE: 287 km/h (155 kt) for HP = 0
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).
- In cold weather, when OAT is below - 30°C, subtract 19 km/h (10 kt) from the above VNE.

Power-off VNE:
- Absolute VNE: 231 km/h (125 kt) for HP = 0.
- In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).
- In cold weather, subtract the following values from the above VNE:
  - 19 km/h (10 kt), when OAT is between - 20°C and - 30°C
  - 37 km/h (20 kt), when OAT is below - 30°C; with out VNE being less than 120 km/h (65 kt).

8. Rotor Speed Limits:

Power-on flight:
- Maximum: 430 rpm (audio warning above 410 rpm)
- Minimum: 320 rpm (audio warning below 360 rpm)

9. Maximum Operating Altitude and Temperature:

9.1. Altitude HP:
- Take-off and landing: Refer to Flight Manual
- En route: 4875 m (16 000 ft)

9.2. Temperature:
- Refer to Flight Manual

10. Operating Limitations:
- Day VFR flight
- Night VFR flight, when the additional equipment required by operational regulations is installed and serviceable.
(For more information refer to Flight Manual)

12. Maximum Certified Weight: 2100 kg (4630 lb)
13. Centre of Gravity Range:

Longitudinal:

```
1.7
3.45
5.18
6.20
7.30
8.45
9.59
10.00
12.00
14.00
16.00
18.00
20.00
22.00
24.00

Weight (kg)
```

Lateral:

```
L.H. limit:  0.18 m (7.08 in)
R.H. limit:  0.14 m (5.51 in)
```

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

14. Datum:

Longitudinal:  3.4 m (134 in) forward of the MRH centreline
Lateral:  Aircraft symmetry plane

15. Levelling Means:

Transmission deck

16. Minimum Flight Crew:

1 pilot in R.H. seat

17. Maximum Passenger Seating Capacity:

5

When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

18. Passenger Emergency Exit:

Refer to Flight Manual

19. Maximum Baggage/Cargo Loads:

<table>
<thead>
<tr>
<th>Location</th>
<th>Max Load</th>
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<tr>
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</tr>
<tr>
<td>Maximum load in forward cabin floor:</td>
<td>150 kg (331 lb)</td>
</tr>
<tr>
<td>Maximum load in rear cabin floor:</td>
<td>310 kg (683 lb)</td>
</tr>
</tbody>
</table>

20. Rotor blade and control movement:

Refer to aircraft Maintenance Manual

21. Life-limited parts:

The AS 350 Maintenance Manual Chapter 5 “Master Servicing Recommendations” have been deemed acceptable to carry out satisfactory maintenance operations on the AS 350 helicopter; part 5.99 "Airworthiness Limitations", originally approved by DGAC-F and subsequently by EASA, contains limitations which are mandatory.
IV. Operating and Service Instructions


2. Maintenance Manual, Document No:  
   - AS 350 Service Manual  
   - AS 350 Overhaul Manual  
   - AS 350 Repair Manual  
   - AS 350 Illustrated Parts Catalogue  

Compatibility between optional items of equipment is described:  
- From an installation aspect: in the "Master Servicing Recommendations".  

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by DGAC-F and EASA.

V. Notes

1. Eligible serial numbers: AS 350 BB aircraft S/N 2945 and up.

2. Placards:  
   1 - The following placard must be fitted in a way that the pilot can see it clearly:

   - LES REPERES ET PLAQUETTES INDICATRICES INSTALLES SUR CET HELICOPTERE CONTIENNENT LES LIMITATIONS D’UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L’UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D’UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L’UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION “LIMITATIONS DE NAVIGABILITE” DU MANUEL D’ENTRETIEN DU GIRAVION DOIT ETRE RESPECTEE.  
   - THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THE ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE “AIRWORTHINESS LIMITATIONS” SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH

   2 - Refer to the Flight Manual as regards the other placards.
**AS 350 B3**

**I. General**

1. **Data Sheet No:** EASA R.008
   Superseding DGAC-F n°157, issue 16 dated February 2005

2. **Type / Model / Variant**
   AS 350 B3 (commercial designation: ECUREUIL)

3. **Airworthiness Category:** Small Rotorcraft

4. **Manufacturer:**
   For helicopters manufactured by Type Certificate holder:
   EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France
   For helicopters manufactured under license - see sub-paragraph V.1 – Eligible serial numbers

5. **DGAC-F Certification Date:** December 24, 1997

6. **DGAC-F Application Date:** October 14, 1996

**II. Certification Basis**

1. **Airworthiness Requirements:** FAR Part 27, Amendments 1 to 10 included

2. **Special Conditions:** Complementary and special conditions defined in DGAC-F letter 971726 dated April 3, 1997

3. **Airworthiness Requirements for a/c incorporating mod. OP-3369 (2370 kg weight extension)** - as above with the following requirements of CS 27 first issue of 14/11/2003 (ED Decision 2003/15/RM) to replace the same numbered paragraphs of FAR 27:
   - CS27 §1; §21; §25; §33; §45; §51; §65; §71; §73; §75; §79; §141; §143; §173; §175; §177; §241; §301; §303; §305; §307; §309; §321; §337; §339; §341; §361; §471; §473; §501; §505; §521; §547; §549; §563(b); §571; §602; §661; §663; §695; §723; §725; §727; §737; §751; §753; §801(b)(d); §927(c); §1041; §1043; §1045; §1301; §1501; §1519; §1529; §1581; §1583; §1585; §1587; §1589.

4. **Airworthiness Requirements for a/c incorporating mod. OP-4305 (Arriel2D engine installation)** - as above and modified as per CRI A-01(X1)

**III. Technical Characteristics and Operational Limitations**

1. **Type design definition:** Document 350A.04.4805
   Document 350A045426 for a/c incorporating mod. OP-3369 (2370 kg weight extension).
   Document 350A047343 for a/c incorporating mod. OP-4305 (Arriel2D engine installation)

2. **Description:** Small single-engine helicopter designed as a derivative product of the former type certified model AS 350 B2.

3. **Equipment:** The approved items of equipment are listed in EUROCOPTER document No. 350A04.4320.
   The basic required equipment specified in the applicable airworthiness regulations (see certification bases) must be installed on the aircraft at certification time and at every time after certification.

4. **Dimensions:**
   - **Fuselage:** Length 10.93 m (35.86 ft)
     - Width 1.87 m (6.14 ft)
     - Height 3.14 m (10.30 ft)
   - **Main Rotor:** 3 blades – Diameter 10.69 m (35.07 ft)
   - **Tail Rotor:** 2 blades – Diameter 1.86 m (6.10 ft)
5. Engine:

**TURBOMECA ARRIEL 2B**
EASA Engine TCDS E.001 (Superseding DGAC-F Engine TCDS No M 19)

**TURBOMECA ARRIEL 2B1**
EASA Engine TCDS E.001 (Superseding DGAC-F Engine TCDS No M 19)

**TURBOMECA ARRIEL 2D**
EASA Engine TCDS E.001

5.1. Installed Engine Limits:

- On AS350 B3 ARRIEL 2B (before modifications AMS 072803 and 072808):

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed (NG)** (ΔNg)</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td>102.3 % (+1)</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min)</td>
<td>85.3 m.daN</td>
<td>535 kW</td>
<td>101.1 % (0)</td>
<td>915°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>71.6 m.daN</td>
<td>450 kW</td>
<td>94.8 % (-4) Vi &gt; 70 kt 97.1 % (-4) Vi &lt; 70 kt</td>
<td>849°C</td>
</tr>
</tbody>
</table>

* ISA, ground level at rotor speed 386 rpm
** 100 % = 52 110 rpm - without electrical bleed, nor P2, ISA, ground level.

- On AS350 B3 ARRIEL 2B (after modifications AMS 072803 and 072808) and on AS 350 B3 ARRIEL 2B1:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed (NG)** (ΔNg)</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td>102.3 % (+1)</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min)</td>
<td>85.3 m.daN</td>
<td>535 kW</td>
<td>101.1 % (0)</td>
<td>915°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>79.1 m.daN</td>
<td>497 kW</td>
<td>97.1 % (-4)</td>
<td>849°C</td>
</tr>
</tbody>
</table>

- On AS350 B3 ARRIEL 2D ****:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed*** ΔNg (NG)**</th>
<th>T4 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td></td>
<td></td>
<td>+1 (101.9 %)</td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min)</td>
<td>85.3 m.daN</td>
<td>535 kW</td>
<td>0 (100.9 %)</td>
<td>949°C</td>
</tr>
<tr>
<td>Maximum at takeoff (30 min) (HiP)*****</td>
<td>85.3 m.daN</td>
<td>535 kW</td>
<td>0 100.9 %</td>
<td>949°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>79.1 m.daN</td>
<td>497 kW</td>
<td>-4 98.0 %</td>
<td>905°C</td>
</tr>
</tbody>
</table>

* ISA, ground level at rotor speed 386 rpm
** 100 % = 52 110 rpm.
*** As the actual Ng limitations depend on ambient conditions, the operational limitations are the ΔNg values. Ng values correspond to the maximum Ng reached in the whole flight domain.
**** Note: The engine is not physically derated but its performance is limited when installed in the AS350B3. Specific limitations have been implemented in the VEMD, allowing the pilot to control the installed Arriel2D at the same power limitations as when an Arriel2B1 is installed, for each aircraft rating (MCP, MTOP and MTP).
***** Use of HIP (Hover Increased Power, TOP 30 min) is only allowed when enhanced thermal protection is fitted on the AS350B3 tail boom (modification OP-4309)
5.2. Transmission Torque Limits:

On AS350 B3 ARRIEL 2B (before modifications AMS 072803 and 072808):

When airspeed is lower than 40kt (74 km/h – 46 MPH):
  Max transient torque (10s): 104 %
  Max continuous torque: 100 %

When airspeed is equal to or greater than 40kt (74 km/h – 46 MPH):
  Max continuous torque: 84 %

On AS350 B3 ARRIEL 2B (after modifications AMS 072803 and 072808):

When airspeed is lower than 40kt (74 km/h – 46 MPH):
  Max transient torque (10s): 104 %
  Max continuous torque: 100 %

When airspeed is equal to or greater than 40kt (74 km/h – 46 MPH):
  Max continuous torque: 92.7 %

On AS350 B3 ARRIEL 2B1:

  Max continuous torque: 92.7 %
  Take-off torque range from 0 to 40kt 92.7 to 100%
  Max take-off torque: 100 %
  Max transient torque (5s): 104 %

On AS350 B3 ARRIEL 2D:

  Max continuous torque: 92.7 %
  Take-off torque range from 0 to 40kt 92.7 to 100%
  Max take-off torque: 100 %
  Max transient torque (5s): 104 %

(100% torque corresponds to 535 kW at 386 rpm main rotor speed)

6. Fluids (Fuel/Oil/Additives):


6.2. Oil: Refer to Flight Manual.

7. Fluid capacities:

7.1. Fuel:
  Total: 540 l (142.7 US gal)
  Usable fuel: 538.75 l (142.34 US gal) - post AMS 07 0289
  Unusable fuel: 1.25 l (0.33 US gal) - post AMS 07 0289

7.2. Oil:
  Engine reservoir: 5.2 l (1.37 US gal)
  BTP: 6.5 l (1.72 US gal) - circuit included
  BTA: 0.33 l (0.09 US gal)
8. Airspeed limits:

8.1. For AS 350 B3 ARRIEL 2B (before modifications AMS 072803 and 072808) and for AS 350 B3 ARRIEL 2B1:

- **Power-on VNE:**  
  - Absolute VNE: 287 km/h (155 kt) for HP = 0.  
  - In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).  
  - In cold weather, when OAT is below -30°C, subtract 19 km/h (10 kt) from the above VNE.

- **Power-off VNE:**  
  - Absolute VNE: 231 km/h (125 kt) for HP = 0.  
  - In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft), without VNE being less than 120 km/h (65 kt).  
  - In cold weather, subtract 37 km/h (20 kt) from the above VNE, when OAT is below -20°C, without VNE being less than 120 km/h (65 kt).

8.2. For AS 350 B3 ARRIEL 2B (after modifications AMS 072803 and 072808):

- **Power-on VNE:**  
  - Absolute VNE: 287 km/h (155 kt) for HP = 0.  
  - In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).  
  - In cold weather, when OAT is below -30°C, subtract 19 km/h (10 kt) from the above VNE.  
  - In the cross-hatched area in the C of G graph, VNE is limited to 246 km/h (133 kt) or the VNE defined above (the lowest value).

- **Power-off VNE:**  
  - Absolute VNE: 231 km/h (125 kt) for HP = 0.  
  - In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft), without VNE being less than 120 km/h (65 kt).  
  - In cold weather, subtract 37 km/h (20 kt) from the above VNE, when OAT is below -20°C, without VNE being less than 120 km/h (65 kt).

8.3. For AS 350 B3 ARRIEL 2D:

- **Power-on VNE:**  
  - Absolute VNE: 287 km/h (155 kt) for HP = 0.  
  - In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft).  
  - In cold weather, when OAT is below -30°C, subtract 19 km/h (10 kt) from the above VNE.

- **Power-off VNE:**  
  - Absolute VNE: 231 km/h (125 kt) for HP = 0.  
  - In altitude, speed decreases by 18 km/h per 1000 m (3 kt per 1000 ft), without VNE being less than 120 km/h (65 kt).  
  - In cold weather, subtract 37 km/h (20 kt) from the above VNE, when OAT is below -20°C, without VNE being less than 120 km/h (65 kt).

9. Rotor Speed Limits:

- **Power-on flight:**  
  - For AS 350 B3 ARRIEL 2B: 390 rpm  
  - For AS 350 B3 ARRIEL 2B1: 375 to 405 rpm  
  - For AS 350 B3 ARRIEL 2D: 375 to 405 rpm

- **In autorotation:**  
  - Maximum: 430 rpm (audio warning above 410 rpm)  
  - Minimum: 320 rpm (audio warning below 360 rpm)

10. Maximum Operating Altitude and Temperature:

10.1. **Altitude HP:**

- Take-off and landing: Refer to Flight Manual  
- En route: 7010 m (23 000 ft)

10.2. **Temperature:**

Refer to Flight Manual

11. Operating Limitations:

- **Day VFR flight**
- **Night VFR flight,** when the additional equipment required by operational regulations is installed and serviceable.

(For more information refer to Flight Manual)

12. **Maximum Certified Weight:**

- 2250 kg (4960 lb)  
- 2370 kg (5220 lb) for a/c incorporating mod. OP-3369.
13. Centre of Gravity Range:

Longitudinal: AS350 B3 ARRIEL 2B1 a/c incorporating mod. OP-3369:

--- Diagram of Centre of Gravity Range for AS350 B3 ARRIEL 2B1 ---

Longitudinal: AS350 B3 ARRIEL 2B (before modifications AMS 072803 and 072808) and for AS 350 B3 ARRIEL 2B1:

--- Diagram of Centre of Gravity Range for AS350 B3 ARRIEL 2B and 2B1 ---
Longitudinal: AS350 B3 ARRIEL 2B (after modifications AMS 072803 and 072808):

Weight (kg)

Longitudinal: AS350 B3 ARRIEL 2D a/c incorporating mod. OP-3369:
**Longitudinal:** AS350 B3 ARRIEL 2D:

![Longitudinal Diagram](image)

Lateral:
- L.H. limit: 0.18 m (7.08 in) up to 2250 kg and 0.08 m (3.15 in) from 2250 up to 2370 kg for a/c incorporating mod. OP-3369
- R.H. limit: 0.14 m (5.51 in) up to 2250 kg and 0.08 m (3.15 in) from 2250 up to 2370 kg for a/c incorporating mod. OP-3369

The weight breakdown and C.G. limit document containing the list of equipment included in the certificated empty weight and the loading instructions shall accompany the helicopter at the time of the initial certification and on a permanent basis from that period on.

In order to obtain the most correct weight and C.G. data, the helicopter shall be jacked up its lifting points rather than using the skids. Should modifications affecting weight and C.G. position to be incorporated, the Flight Manual instructions shall be referred to.

**14. Datum:**
- Longitudinal: 3.4 m (134 in) forward of the MRH centrelines
- Lateral: Aircraft symmetry plane

**15. Levelling Means:**
- Transmission deck

**16. Minimum Flight Crew:**
- 1 pilot in R.H. seat

**17. Maximum Passenger Seating Capacity:**
- 5

When the aircraft is fitted with the forward two-place seat optional equipment, the maximum number of passengers is increased to six (pilot not included). This optional is to be used in accordance with the corresponding Flight Manual supplement.

**18. Passenger Emergency Exit:**
- Refer to Flight Manual

**19. Maximum Baggage/Cargo Loads:**

<table>
<thead>
<tr>
<th>Location</th>
<th>Max Load</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum load in R.H. side hold:</td>
<td>100 kg (220 lb)</td>
</tr>
<tr>
<td>Maximum load in L.H. side hold:</td>
<td>120 kg (264 lb)</td>
</tr>
<tr>
<td>Maximum load in rear hold:</td>
<td>80 kg (176 lb)</td>
</tr>
<tr>
<td>Maximum load in forward cabin floor:</td>
<td>150 kg (331 lb)</td>
</tr>
<tr>
<td>Maximum load in rear cabin floor:</td>
<td>310 kg (683 lb)</td>
</tr>
</tbody>
</table>

**20. Rotor blade and control movement:**
- Refer to aircraft Maintenance Manual

**21. Life-limited parts:**

The AS 350 Maintenance Manual Chapter 5 "Master Servicing Recommendations" have been deemed acceptable to carry out satisfactory maintenance operations on the AS 350 helicopter. MSM Chapter 04 "Airworthiness Limitations", originally approved by DGAC-F and subsequently by EASA, contains limitations which are mandatory.
IV. Operating and Service Instructions

1. Rotorcraft Flight Manual, Document No:
   - AS 350 B3 ARRIEL 2B Flight Manual, approved by DGAC-F on December 24, 1997 plus rapid revision RR 1A (after modifications AMS 072803 and 072808), or later (DGAC-F and subsequently EASA) approved revisions (reference in English language).
   - AS 350 B3 ARRIEL 2B1 Flight Manual, approved by DGAC-F on July 16, 2004 or later (DGAC-F and subsequently EASA) approved revision (reference in English language).
   - AS 350 B3e Flight Manual, in English (for a/c incorporating mod. OP-4305 - Arriel2D engine installation – and additional modifications to the tail rotor control system – see point 3 in section V. Notes), approved by EASA on June 17, 2011, or later approved revisions
   - AS 350 B3e Flight Manual, in French (for a/c incorporating mod. OP-4305 - Arriel2D engine installation – and additional modifications to the tail rotor control system – see point 3 in section V. Notes), approved by EASA on June 17, 2011, or later approved revisions

2. Maintenance Manual, Document No:
   - AS 350 B3 Service Manual
   - AS 350 Overhaul Manual
   - AS 350 Repair Manual
   - AS 350 B3 Illustrated Parts Catalogue

Compatibility between optional items of equipment is described:
   - From an installation aspect: in the "Master Servicing Recommendations".

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by DGAC-F (prior to EASA) or under the Eurocopter DOA.

V. Notes

1. Eligible serial numbers:
   - AS 350 B3 aircraft S/N 2968, S/N 3063 and up.
   - AS 350 B3 aircraft S/N 4201 and up for a/c incorporating mod. OP-3369 (2370 kg weight extension)
   - AS 350 B3 aircraft S/N 4767 and up for a/c incorporating mod. OP-4305 (without or with mod. OP-3369)

The aircraft whose s/n is listed in Eurocopter documents L102-001 are manufactured under Helibras license and those in L102-002 under AE-MS license.

2. Placards:
   1. The following placard must be fitted in a way that the pilot can see it clearly:

   - LES REPERES ET PLAQUETTES INDICATRICES INSTALLES SUR CET HELICOPTERE CONTIENNENT LES LIMITATIONS D'UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L'UTILISATION DE CE GIRAVION. LES AUTRES LIMITATIONS D'UTILISATION QUI DOIVENT ETRE RESPECTEES LORS DE L'UTILISATION DE CE GIRAVION, SONT CONTENUES DANS LE MANUEL DE VOL DU GIRAVION. LA SECTION "LIMITATIONS DE NAVIGABILITE" DU MANUEL D'ENTRETIEN DU GIRAVION DOIT ETRE RESPECTEE.

   - THE MARKINGS AND PLACARDS INSTALLED ON THIS HELICOPTER CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THE ROTORCRAFT. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS ROTORCRAFT ARE CONTAINED IN THE ROTORCRAFT FLIGHT MANUAL. THE "AIRWORTHINESS LIMITATIONS" SECTION OF THE ROTORCRAFT MAINTENANCE MANUAL MUST BE COMPLIED WITH

   2. Refer to the Flight Manual as regards the other placards.

3. Commercial designation related to particular modifications:

   The AS350B3 model a/c, which type design incorporates the following modifications:
   - OP-4305 (Arriel2D engine installation), and additionally
   - 07-5601 (Tail rotor control mechanism modification)
   - 07-5600 (Tail rotor blade reinforcement)
   - 07-8551 (Tail Gear Box control lever modification)

has a commercial designation <AS350B3e>
EC 130 B4

I. General

1. Data Sheet No: EASA R.008

2. Type / Model / Variant
   EC 130 B4

3. Airworthiness Category: Small Rotorcraft

4. Manufacturer:
   EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France

5. JAA recommendation date: December 14, 2000

6. DGAC-F Certification Date: December 14, 2000

7. JAA Application Date: March 23, 1998

II. Certification Basis


3. Special Conditions: High intensity radiated field

4. Exemptions: For:
   - The rear seat bench regarding JAR 27-562 and JAR 27-785 (a), (b), (j)
   - Fuel systems regarding JAR 27.952 (a), (c), (d), (f), (g)

5. Equivalent safety findings: On:
   - Main gearbox oil filter by pass
   - Powerplant instrument markings

III. Technical Characteristics and Operational Limitations

1. Type design definition: Document 350A.04.7053

2. Description: Small single-engine helicopter designed as a derivative product of the former type certified model AS 350 B3.

3. Equipment: As per compliance with JAR 27 requirements and included in the original Type Design Standard or indicated on the section 2 - limitations of the Flight Manual.

4. Dimensions:
   - Fuselage: Length 10.68 m (35.03 ft), Width 2.03 m (6.66 ft), Height 3.61 m (11.84 ft)
   - Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)
   - Tail Rotor: Fan-in-fan 10 blades – Diameter 1.00 m (3.28 ft)

5. Engine: TURBOMECA ARRIEL 2B1
   EASA Engine TCDS E.001 (Superseding DGAC-F Engine TCDS No M 19)
5.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Ratings</th>
<th>Gas generator Speed (NG)* (ΔNg)</th>
<th>Exhaust gas Temperature (T4)**</th>
</tr>
</thead>
<tbody>
<tr>
<td>Max transient</td>
<td>102.3 % (+1)</td>
<td>865°C (10 s)</td>
</tr>
<tr>
<td>Max take-off (5 min)</td>
<td>101.1 % (0)</td>
<td>915°C</td>
</tr>
<tr>
<td>Max Continuous</td>
<td>97.1 % (-4)</td>
<td>849°C</td>
</tr>
</tbody>
</table>

* 100%: 52110 rpm  
** Max Continuous during starting: 750 °C

5.2. Transmission Torque Limits:  
Max transient (5s): 104 %  
Max Take off: 100 %  
Max Continuous: 92.7 %

(100% torque corresponds to 536 kW at 6000 rpm engine speed = 386 rpm main rotor speed)

6. Fluids (Fuel/Oil/Additives):

6.2. Oil:  Refer to Flight Manual

7. Fluid capacities:

7.1. Fuel:  Total: 540 l (142.7 US gal)  
Usable fuel: 538.7 l (142.32 US gal)

7.2. Oil:  Refer to Flight Manual

8. Airspeed limits:

- VNE power on: 155 kt at sea level less 3 kt/1000 ft  
- VNE power off: 125 kt at sea level less 3 kt/1000 ft

9. Rotor Speed Limits:

- Normal range power on: 375 to 405 rpm  
- Maximum power off: 430 rpm (aural warning ≥ 410 rpm)  
- Minimum power off: 320 rpm (aural warning = 360 rpm)

10. Maximum Operating Altitude and Temperature:

10.1. Altitude HP:  
- Take-off and landing: Refer to Flight Manual  
- En route: 7010 m (23 000 ft)

10.2. Temperature:  
- Minimum: -20°C or -40°C after modification 076302  
- Maximum: ISA +35°C limited to +50°C

11. Operating Limitations:

- Day VFR  
- Night VFR if modification 07-3664 is installed  
- Flights under icing and freezing rain are prohibited  
- Aerobatic manoeuvres are prohibited  
- Flights under falling snow are prohibited except if sand filter is installed (see Flight Manual Supplement 9-14)

12. Maximum Certified Weight: 2427 kg (5350 lb)
13. Centre of Gravity Range:

**Longitudinal:**

![Diagram showing Centre of Gravity Range](image)

- L.H. limit: 0.10 m (3.94 in)
- R.H. limit: 0.10 m (3.94 in)

14. Datum:
- Longitudinal: 3.4 m (134 in) forward of the MRH centerline
- Lateral: Symmetrical plane of the aircraft

15. Levelling Means:
- Mechanical floor

16. Minimum Flight Crew:
- 1 pilot in left seat

17. Maximum number of occupants (including flight crew):
- 7 (3 at front and 4 at rear)
- or 8 (4 at front and 4 at rear) after modification OP-3673

18. Passenger Emergency Exit:
- One door on each side of the fuselage

19. Maximum Baggage/Cargo Loads:
- Maximum load in R.H. cargo compartment: 130 kg (287 lb) / 300 kg/m² (62.5 lb/ft²)
- Maximum load in L.H. cargo compartment: 155 kg (342 lb) / 300 kg/m² (62.5 lb/ft²)
- Maximum load in rear cargo compartment: 80 kg (176 lb) / 145 kg/m² (30 lb/ft²)
- Maximum load in forward cabin floor: 495 kg (1091 lb) / 300 kg/m² (62.5 lb/ft²)
- Maximum load in rear cabin floor: 405 kg (893 lb) / 300 kg/m² (62.5 lb/ft²)

20. Rotor blade and control movement:
- Refer to aircraft Maintenance Manual

**IV. Operating and Service Instructions**

1. Rotorcraft Flight Manual, Document No:
   - EC 130 B4 Flight Manual (in English), approved by DGAC-F on November 29, 2000 or later approved revision
   - EC 130 B4 Flight Manual (in French), approved by DGAC-F on May 27, 2002 or later approved revision.

2. Maintenance Manual, Document No:
   - EC 130 B4 Master Servicing Manual – chapter 04 (Airworthiness Limitations), approved by DGAC-F on December 6, 2000 or later approved revision/edition (in English).
   - Maintenance Manual
   - Overhaul Manual
   - Illustrated Parts Catalogue

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by DGAC-F and EASA.

**V. Notes**

1. Eligible serial numbers: EC 130 B4 aircraft S/N 3358 and up.
EC 130 T2

I. General

1. Data Sheet No: EASA R.008
2. Type / Model / Variant: EC 130 T2
3. Airworthiness Category: Small Rotorcraft
4. Manufacturer: EUROCOPTER
   Aéroport International Marseille Provence
   13725 MARIGNANE Cedex
   France
5. EASA Certification Date: 25.05.2012
6. EASA Application Date: 14 October, 2010

II. Certification Basis

1. EASA certification basis: As defined in CRI A-01(X2) issue 6 dated 23.05.2012
2. Airworthiness Requirements: JAR 27 first issue dated September 6, 1993, and
   orange paper amendment 27/98/1 effective February 16, 1998
3. Special Conditions: High intensity radiated field (CRI F-01(X2))
   Rotor drive system endurance test for HIP rating (CRI E-02(X2))
4. Exemptions: N/A
5. Equivalent safety findings: Main gearbox oil filter by pass (EC130B4 CRI E4, CRI A-01(X2))
   Powerplant instrument markings (CRI G-01(X2))

III. Technical Characteristics and Operational Limitations

1. Type design definition: Document 350A.04.7422
2. Description: Small single-engine helicopter designed as a derivative product of the former type certified model EC 130 B4.
3. Equipment: As per compliance with EC 130 T2 certification basis and included in the original Type Design Standard or
   indicated on the section 2 - Limitations of the Flight Manual.
4. Dimensions: Fuselage: Length 10.68 m (35.03 ft)
   Width 2.03 m (6.66 ft)
   Height 3.61 m (11.84 ft)
   Main Rotor: 3 blades – Diameter 10.69 m (35.07 ft)
   Tail Rotor: Fan-in-fan 10 blades – Diameter 1.00 m (3.28 ft)
5. Engine: TURBOMECA ARRIEL 2D
   EASA Engine TCDS E.001
5.1. Installed Engine Limits:

<table>
<thead>
<tr>
<th>Power rating</th>
<th>Limit torque on shaft</th>
<th>Minimum guaranteed power*</th>
<th>Generator speed***</th>
<th>T45 temperature</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum transient (5 sec.)</td>
<td></td>
<td></td>
<td>+1 (102.8 %)</td>
<td></td>
</tr>
<tr>
<td>Maximum at takeoff (5 min) ****</td>
<td>95.1 m.daN</td>
<td>597.5 kW</td>
<td>0</td>
<td>949°C</td>
</tr>
<tr>
<td>Maximum at takeoff (30 min) (HIP) ****</td>
<td>95.1 m.daN</td>
<td>597.5 kW</td>
<td>0</td>
<td>949°C</td>
</tr>
<tr>
<td>Maximum continuous</td>
<td>77.3 m.daN</td>
<td>485.7 kW</td>
<td>-4</td>
<td>905°C</td>
</tr>
</tbody>
</table>

* ISA, ground level at rotor speed 386 rpm
** 100 % = 52 110 rpm.
*** As the actual Ng limitations depend on ambient conditions, the operational limitations are the ΔNg values. Ng values correspond to the maximum Ng reached in the whole flight domain.
**** Use of <takeoff (30 min)> power is limited to 30 min. continuous use. Cumulated use per flight of <takeoff (5 min)> and <takeoff (30 min)> powers is limited to 60 min.

5.2. Transmission Torque Limits:
- Max transient (5s): 104 %
- Max Take off: 100 %
- Max Continuous: 81.3 %

(100% torque corresponds to 598 kW at 386 rpm main rotor speed)

6. Fluids (Fuel/Oil/Additives):

6.2. Oil: Refer to Flight Manual

7. Fluid capacities:

7.1. Fuel: Total: 540 l (142.7 US gal)
Usable fuel: 538 l (142.32 US gal)

7.2. Oil: Refer to Flight Manual

8. Airspeed limits:

VNE power on: 155 kt at sea level less 3 kt/1000 ft
136 kt at sea level less 3 kt/2000 ft below Hp 12750 ft for reduced VNE area (Refer to Flight Manual)

VNE power off: 125 kt at sea level less 3 kt/1000 ft

9. Rotor Speed Limits:

Normal range power on: 375 to 405 rpm
Maximum power off: 430 rpm (aural warning ≥ 410 rpm)
Minimum power off: 320 rpm (aural warning ≤ 360 rpm)

10. Maximum Operating Altitude and Temperature:

10.1. Altitude HP:
- Take-off and landing: Refer to Flight Manual
- En route: 7010 m (23 000 ft)

10.2. Temperature:
- Minimum: -40°C
- Maximum: ISA +35°C limited to +50°C

11. Operating Limitations:

- Day VFR
- Night VFR, when additional equipment required by operational regulations is installed and serviceable
- Aerobatic manoeuvres are prohibited
- Flights in freezing rain or under icing conditions forbidden
- Flights in falling snow are prohibited except if sand filter is installed (see Flight Manual Supplement SUP.14)

(For more information refer to Flight Manual)
12. Maximum Certified Weight: 2500 kg (5512 lb)

13. Centre of Gravity Range:

Longitudinal:

Lateral:
14. Datum:  
Longitudinal: 3.4 m (134 in) forward of the MRH centerline  
Lateral: Symmetrical plane of the aircraft

15. Levelling Means: Mechanical floor

16. Minimum Flight Crew: 1 pilot in left seat

17. Maximum number of occupants (including flight crew):  
7 (3 at front and 4 at rear) or 8 (4 at front and 4 at rear) if modification OP-3673 is installed

18. Passenger Emergency Exit: One door on each side of the fuselage

19. Maximum Baggage/Cargo Loads:  
Maximum load in R.H. cargo compartment: 130 kg (287 lb) / 300 kg/m² (62.5 lb/ft²)  
Maximum load in L.H. cargo compartment: 155 kg (342 lb) / 300 kg/m² (62.5 lb/ft²)  
Maximum load in rear cargo compartment: 80 kg (176 lb) / 145 kg/m² (30 lb/ft²)  
Maximum load in rear cabin floor: 495 kg (1091 lb) / 300 kg/m² (62.5 lb/ft²)  
Maximum load in forward cabin floor: 405 kg (893 lb) / 300 kg/m² (62.5 lb/ft²)

20. Rotor blade and control movement: Refer to aircraft Maintenance Manual

IV. Operating and Service Instructions

1. Rotorcraft Flight Manual, Document No:  
- EC 130 T2 Flight Manual (in English), approved by EASA on 25.05.2012 or later approved revision

2. Maintenance Manual, Document No:  
- EC 130 T2 Master Servicing Manual – chapter 04 (Airworthiness Limitations Section), approved by EASA on 25.05.2012 or later approved revision/edition (in English).  
- EC 130 T2 Maintenance Manual  
- EC 130 T2 Overhaul Manual  
- EC 130 T2 Illustrated Parts Catalogue

3. Service Letters and Service Bulletins: As published by Eurocopter and approved by EASA.

V. Notes

1. Eligible serial numbers: EC 130 T2 aircraft S/N 7355 and up.
ADMINISTRATIVE SECTION

I. Acronyms

HIP Hover-Increased Power

II. Type Certificate Holder Record

<table>
<thead>
<tr>
<th>Type Certificate Holder</th>
<th>Period</th>
</tr>
</thead>
<tbody>
<tr>
<td>AEROSPATIALE</td>
<td>Before January 1, 1992</td>
</tr>
<tr>
<td>37, Boulevard de Montmorency</td>
<td></td>
</tr>
<tr>
<td>75781 PARIS Cedex 16</td>
<td></td>
</tr>
<tr>
<td>France</td>
<td></td>
</tr>
<tr>
<td>EUROCOPTER France</td>
<td>Before June 1, 1997</td>
</tr>
<tr>
<td>Aéroport International Marseille Provence</td>
<td></td>
</tr>
<tr>
<td>13725 MARIGNANE Cedex</td>
<td></td>
</tr>
<tr>
<td>France</td>
<td></td>
</tr>
<tr>
<td>EUROCOPTER</td>
<td>Since June 1, 1997</td>
</tr>
<tr>
<td>Aéroport International Marseille Provence</td>
<td></td>
</tr>
<tr>
<td>13725 MARIGNANE Cedex</td>
<td></td>
</tr>
<tr>
<td>France</td>
<td></td>
</tr>
</tbody>
</table>

III. Change record

<table>
<thead>
<tr>
<th>Issue</th>
<th>Date</th>
<th>Details</th>
</tr>
</thead>
<tbody>
<tr>
<td>01</td>
<td>18 October 2005</td>
<td>initial issue to supersede DGAC-F TCDS No. 157 issue 16</td>
</tr>
<tr>
<td>02</td>
<td>23 January 2007</td>
<td>AS350B3 OP-3369 added, AS350B2 VEMD added</td>
</tr>
<tr>
<td>04</td>
<td>23 November 2009</td>
<td>engine TCDS references corrected, §865 removed from airworthiness reqs for AS350B3 OP-3369, transmission torque limits definitions corrected, SB for AS350BA conversion into AS350B2 added</td>
</tr>
<tr>
<td>06</td>
<td>25.05.2012</td>
<td>EC130T2 model added, note *** under Arriel 2D installed engine limits table in AS350B3 section corrected, superseded DGAC-F TCDS issuance date corrected, TCDS format and editorial changes</td>
</tr>
</tbody>
</table>