European Aviation Safety Agency

EASA

TYPE-CERTIFICATE DATA SHEET

EASA.A.353

ZLIN Z 26 - SERIES

Type Certificate Holder:

ZLIN AIRCRAFT A.S.
Letiště 1578
765 81 Otrokovice
CZECH REPUBLIC

For Models: Z 126, Z 126 T, Z 226 B, Z 226 T, Z 226 A, Z 226 M, Z 226 MS;

Z 326, Z 326 A, Z 326 M;

Z 526, Z 526 A, Z 526 F, Z 526 L, Z 526 AFS, Z 526 AFS-V, Z 526 M

Z 726, Z 726 K

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SECTION A: Z 126

AI. General

1. a) Type:	Z 26
b) Model:	Z 126
2. Airworthiness category:	Aerobatic (A) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n. p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: 701 – 870
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	August 15, 1956

7. EASA Type Certificate replaces CAA CZ Type Certificate No. 4-4445-1956.

AII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K5
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None

9. EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

AIII. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of aircraft Z 126, No. S 126.000.	
2. Description:	The Z 126 aircraft is two-seat, low wing, single-engine monoplane.	
3. Equipment:	Approved equipment list is stated in document Maintenance Manual Z 126, Chapter 5.	
4. Dimensions:	Wing Span:	10.282 m
	Length:	7.555 m
	Height:	2.770 m
	Wing Area:	14.900 m^2
5. Engine:		
5.1 Model:	Walter Minor 4-III	
5.2 Type Certificate:	No. 19/4-IV A/3 ai 1946, Ministry of Transport, CZ	
5.3 Limitations:	Max. Continuous power	
	Max. Power	77 kW (105 HP)
	Max. Engine speed	2 500 RPM
	Max. Consumption	35 l/h
	Max. Cruising power	
	Max. Power	59 kW (80 HP)
	Max. Engine speed	2 300 RPM
	Max. Consumption	26 l/h
6. Load factors:	For category Aerobatic (A)	+6.0 g, -3.0 g
	For category Normal (N)	+3.5 g, -1.0 g
7. Propellers:		
7.1.1 Model:	V-126	
7.1.2 Type Certificate:	---	
7.1.3 Number of blades:	2	
7.1.4 Diameter:	2 000 mm	
7.1.5 Sense of Rotation:	Anticlockwise in flight direction.	

or

7.2.1 Model:	L 26.1.8100.5

7.2.2 Type Certificate: ---

7.2.3 Number of blades: 2

7.2.4 Diameter: 2 000 mm

7.2.5 Sense of Rotation: Anticlockwise in flight direction.

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline, with min. 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.
Recommended kinds of fuel:
LBZ 72
LBZ 78
LBE 80
Shell 80
Esso 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose per-centual carbon residue does not exceed the value of 0.4.
MS 20
Aero Shell W100
Aero Shell W120 (in tropical climates)

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: 80 litres
Usable: 78 litres

2 x 35 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never exceed speed limit	V_{NE}	320 km/h IAS
Maximum speed limit near the ground	V_H	205 km/h IAS
Maximum flaps extended speed limit	V_{FE}	130 km/h IAS

11. Maximum Operating Altitude:	4 800 m
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.
13. Maximum Weights:	Max. Take-off and Landing weight: 765 kg Max. Variable Load: 194 kg
14. Centre of Gravity Range:	23 % – 26.5 % M.A.C. M.A.C. is 1 532 mm; 0 % M.A.C. is 621 mm aft reference datum.
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.
16. Control surface deflections:	Elevator deflection up 25° ± 2° down 20° ± 2° Elevator trim up 23° ± 2° down 40° ± 2° Rudder deflection left an right 30° ± 2° Ailerons deflection up 108 mm (+5, -3) mm down 98 mm (+5, -3) mm Wing flaps position: retracted 0° take-off 15° landing 43° (+ 2°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.
18. Minimum Flight Crew:	1 (Pilot)
19. Maximum Passenger Seating Capacity:	2 (including crew)
20. (Reserved)	
21. Baggage/Cargo Compartments:	Maximum 10 kg
22. Wheels and Tyres:	The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 Tl with tube 420x150; The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL;

The wheels K 34-990 or K 13-0000.00 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

AIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Směrnice pro pilota letounu Z 126
   - In English language — Instruction for pilot on the use and handling of the Z 126 Aircraft

2. Maintenance and operating manual:
   - In Czech language — Návod k obsluze a ošetřování letounu Z 126, date of issue 1954
   - In German language — Flugzeug Z 126 Bedienungsanleitung, date of issue 1956

3. Repair manual:
   - In Czech language — Opravárenská příručka Z 126, Z 226, date of issue 1994

4. Illustrated parts catalogue:
   - In Czech language — Seznam náhradních součástí draku letounu Z 126 Trenér, date of issue 1954

AV. Notes:

Following Z 26 aircraft have been converted to the model Z 126 by the aircraft manufacturer:
S/N: 518, 524, 525, 535, 580, 591 595, 611

Following Z 126 aircraft have been converted to the models:
T 126 M S/N: 841
Z 226 S/N: 817
Z 226 B S/N: 830, 831
Z 226 T S/N: 870
Z 226 MS: S/N: 741, 750, 804, 839
by the aircraft manufacturer.

SECTION B: Z 126 T

BI. General

1. a) Type:	Z 26
b) Model:	Z 126 T
2. Airworthiness category:	Utility (U) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan n.p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: 780
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	February 10, 1977

7. EASA Type Certificate replaces CAA CZ Type Certificate No. 55.470 – 1957.

BII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K4, K5
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None

9.	EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

BIII. Technical Characteristics and Operational Limitations

1.	Type Design Definition:	The specification list of aircraft Z 126 T, No. T-S 126.000.
2.	Description:	The Z-126 T aircraft is two-seat, low wing, single-engine monoplane.
3.	Equipment:	Approved equipment list is stated in document Maintenance Manual Z 126, Chapter 5.
4.	Dimensions:	Wing Span: 10.282 m Length: 7.555 m Height: 2.770 m Wing Area: 14.900 m^2
5.	Engine:	
	5.1 Model:	Walter Minor 4-III
	5.2 Type Certificate:	No. 19/4-IV A/3 a.i. 1946 Ministry of Transport, CZ
	5.3 Limitations:	Max. Continuous power Max. Power 77 kW (105 HP) Max. Engine speed 2 500 RPM Max. Comsumption 35 l/h Max. Cruising power Max. Power 59 kW (80 HP) Max. Engine speed 2 300 RPM Max. Consumption 26 l/h
6.	Load factors:	For category Utility (U) +4.4 g, -1.8 g For category Normal (N) +3.5 g, -1.0 g
7.	Propellers:	
	7.1.1 Model:	V-126
	7.1.2 Type Certificate:	---
	7.1.3 Number of blades:	2
	7.1.4 Diameter:	2 000 mm
	7.1.5 Sense of Rotation:	Anticlockwise in flight direction
	or	
	7.2.1 Model:	L 26.1.8100.5

7.2.2 Type Certificate: ---

7.2.3 Number of blades: 2

7.2.4 Diameter: 2 000 mm

7.2.5 Sense of Rotation: Anticlockwise in flight direction.

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline, with min. 72 octanes.
Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.
LBZ 72
LBZ 78
LBE 80
LBE 87
Shell 80
ESSO 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.
MS20
Aeroshell W100
Aeroshell W120 (in tropical climates)

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: 80 litres
Usable: 78 litres

2 x 35 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never exceed speed limit	V_{NE}	300 km/h IAS
Maximum speed limit near the ground	V_H	205 km/h IAS
Maximum flaps extended	V_{FE}	

	speed limit	160 km/h IAS
11. Maximum Operating Altitude:	4 750 m	
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.	
13. Maximum Weights:	Max. Take-off and Landing weight:	815 kg
	Max. Variable Load:	190 kg
14. Centre of Gravity Range:	18 % – 30.5 % M.A.C. M.A.C. is 1 532 mm; 0 % M.A.C. is 621 mm aft reference datum.	
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.	
16. Control surface deflections:	Elevator deflection up down	25° ± 2° 20° ± 2°
	Elevator trim up down	23° ± 2° 40° ± 2°
	Rudder deflection left an right	30° ± 2°
	Ailerons deflection up down	108 mm (+5, -3) mm 98 mm (+5, -3) mm
	Wing flaps position: retracted take-off landing	0° 15° 43° (+ 2°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.	
18. Minimum Flight Crew:	1 (Pilot)	
19. Maximum Passenger Seating Capacity:	2 (including crew)	
20. (Reserved)		
21. Baggage/Cargo Compartments:	Maximum 10 kg	
22. Wheels and Tyres:	The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL; The wheels K 34-990 or K 13-0000.00 of tail gear	

with tyre BARUM or MITAS 260x85-4 with tube 260x85.

BIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Směrnice pro pilota letounu Z 126
   - In English language — Instruction for Pilot on the use and Handling of the Z 126 Aircraft

2. Flight manual supplement:
   - In Czech language — Dodatek k letové příručce Z 126 – modifikace T

3. Maintenance and operating manual:
   - In Czech language — Návod k obsluze a ošetřování letounu Z 126, date of issue 1954
   - In German language — Flugzeug Z 126 Bedienungsanleitung, date of issue 1956

4. Repair manual:
   - In Czech language — Opravárenská příručka Z 126, Z 226, date of issue 1994

5. Illustrated parts catalogue:
   - In Czech language — Seznam náhradních součástí draku letounu Z 126 Trenér, date of issue 1954

Note: Revisions are issued in Czech and English languages only.

BV. Notes:

Following Z 126 aircraft has been converted to the model
Z 226 S/N: 780
by the aicraft manufacturer.

SECTION C: Z 226 B

CI. General

1. a) Type:	Z 26
b) Model:	Z 226 B
2. Airworthiness category:	Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan n.p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: 06-08 – 41-09; 54-09; 246 – 285
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	October 25,1956

7. EASA Type Certificate replaces CAA CZ Type Certificate No. 4-5150-1956.

CII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K4, P3 British Civil Airworthiness Requirements, Cat D-4
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None
9. EASA Environmental	ICAO Annex 16, Volume I, Chapter 10

Standards:

CIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of aircraft Z 226 B, No. S-B 226.000.
2. Description: The Z 226 B aircraft is two-seat, single-engine monoplane. Control system is installed in rear pilot compartment only.
3. Equipment: Approved equipment list is stated in document Maintenance Manual Z 226 B, Chapter V.
4. Dimensions:
   Wing Span: 10.280 m
   Length: 7.800 m
   Height: 2.778 m
   Wing Area: 14.900 m^2
5. Engine:
   5.1 Model: Walter Minor 6-III
   5.2 Type Certificate: No. 132/2-L/6A-a.i.-1947 Ministry of Transport, CZ
   5.3 Limitations:

Max. Continuous power	
Max. Power	118 kW (160 HP)
Max. Engine speed	2 500 RPM
Max. Consumption	53 l/h
Max. Cruising power	
Max. Power	92 kW (125 HP)
Max. Engine speed	2 300 RPM
Max. Consumption	38 l/h

6. Load factors: + 3.8 g, - 1.5 g
7. Propellers:
   7.1.1 Model: V-226
   7.1.2 Type Certificate: ---
   7.1.3 Number of blades: 2
   7.1.4 Diameter: 2 050 mm
   7.1.5 Sense of Rotation: Anticlockwise in flight direction

   or

   7.2.1 Model: Z-226.640
   7.2.2 Type Certificate: ---

7.2.3 Number of blades: 2

7.2.4 Diameter: 2 050 mm

7.2.5 Sense of Rotation: Anticlockwise in flight direction

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline, with min. 72 octanes.
Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.
LBZ 72
LBZ 78
LBE 80
LBE 87
Shell 80
ESSO 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.
MS20
Aeroshell W100
Aeroshell W120 (in tropical climates)

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: 125 litres
Usable: 123 litres

2 x 35 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank
45 litres in auxiliary fuselage tank

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never exceed speed limit	V_{NE}	300 km/h IAS
Maximum speed limit near the ground	V_H	200 km/h IAS
Maximum flaps extended speed limit	V_{FE}	130 km/h IAS

11. Maximum Operating Altitude:	7 150 m
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.
13. Maximum Weights:	Max. Take-off and Landing weight: In towing 680 kg In flight without towing 770 kg Max. Variable Load: In towing 103 kg In flight without towing 193 kg
14. Centre of Gravity Range:	20.3 % – 24.4 % M.A.C. M.A.C. is 1 532 mm; 0 % M.A.C. is 621 mm aft reference datum.
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.
16. Control surface deflections:	Elevator deflection up 25° ± 2° down 20° ± 2° Elevator trim up 23° ± 2° down 40° ± 2° Rudder deflection left an right 30° ± 2° Ailerons deflection up 108 mm (+5, -3) mm down 98 mm (+5, -3) mm Wing flaps position: retracted 0° take-off 15° landing 43° (+ 2°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.
18. Minimum Flight Crew:	1 (Pilot)
19. Maximum Passenger Seating Capacity:	2 (including crew)
20. (Reserved)	
21. Baggage/Cargo Compartments:	Maximum 23 kg (except for towing operation)
22. Wheels and Tyres:	The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150;

The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL;

The wheels K 34-990 or K 13-0000.00 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

CIV. Operating and Service Instructions

1. Flight manual

–	In Czech language	Směrnice pro pilota letounu Z 226 B, T
-	In English language	Instruction for Pilot on the Use and Handling of the Z 226 Aircraft
–	In German language	Richtlinien Fuer den Flugzeugfuhrer des Flugzeuges Z 226

2. Maintenance manual

–	In Czech language	Návod k obsluze letounu Z 226 B, date of issue 1957

3. Overhaul manual

–	In Czech language	Opravárenská příručka Z 126, Z 226, date of issue 1994

4. Catalog of spare parts:

–	In Czech language	Seznam náhradních součástí draku letounu Z 226 B, T

Note: Revisions are issued in Czech and English languages only.

CV. Notes

Following Z 226 B aircraft have been converted to the models:

Z 126 T	S/N: 14-08, 15-08;
Z 226 T	S/N: 12-08, 25-09;
Z 226 M	S/N: 21-09, 28-09, 31-09, 37-09, 39-09, 40-09, 253;
Z 226 MS	S/N: 07-08,10-08,11-08, 16-08, 17-08, 19-08, 22-09, 26-09, 29-09, 33-09, 34-09, 36-09, 41-09, 246, 247, 250, 252, 256, 260, 261, 264, 265, 266, 268, 270, 273, 274, 275, 276, 279, 281, 282, 284, 285;
Z 226 LS	S/N: 280

by the aircraft manufacturer.

SECTION D: Z 226 T

DI. General

1. a) Type:	Z 26
b) Model:	Z 226 T
2. Airworthiness category:	Aerobatic (A) Utility (U) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n.p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: 42 – 53; 55 – 245; 286 – 370
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	October 25, 1957

7. EASA Type Certificate replaces CAA CZ Type Certificate No. 55.470 – 1957.

DII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K4, K5
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None

9. EASA Environmental Standards: ICAO Annex 16, Vol. I, Chapter 10

DIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of aircraft Z 226 T No. T-S 226.000.

2. Description: The Z-226 T aircraft is two-seat, low wing, single-engine monoplane. Control system is installed in both of pilot compartments.

3. Equipment: Approved equipment list is stated in document Maintenance Manual Z 226 T, Chapter V.

4. Dimensions:
   Wing Span: 10.282 m
   Length: 7.800 m
   Height: 2.060 m
   Wing Area: 14.900 m^2

5. Engine:

   5.1 Model: Walter Minor 6-III

   5.2 Type Certificate: No. 132/2-L/6A-a.i.-1947 Ministry of Transport, CZ

   5.3 Limitations:
   Max. Continuous power
   Max. Power 118 kW (160 HP)
   Max. Engine speed 2 500 RPM
   Max. Consumption 53 l/h

   Max. Cruising power
   Max. Power 92 kW (125 HP)
   Max. Engine speed 2 300 RPM
   Max. Consumption 38 l/h

6. Load factors:
   For category Aerobatic (A) +6.0 g, -3.0 g
   For category Utility (U) +4.0 g, -2.0 g
   For category Normal (N) +3.5 g, -1.0 g

7. Propellers:

   7.1.1 Model: T-226

   7.1.2 Type Certificate: ---

   7.1.3 Number of blades: 2

   7.1.4 Diameter: 2 050 mm

   7.1.5 Sense of Rotation: Anticlockwise in flight direction.

   or

7.2.1 Model:	Z-226.641
7.2.2 Type Certificate:	---
7.2.3 Number of blades:	2
7.2.4 Diameter:	2 000 mm
7.2.5 Sense of Rotation:	Anticlockwise in flight direction.

8. Fluids:

8.1 Fuel:	Non-ethylated aviation gasoline, with min. 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06% vol. LBZ 72 LBZ 78 LBE 80 LBE 87 Shell 80 ESSO 80 AVGAS 100 LL (DEFENCE STANDARD 91/90, ASTM D910)
8.2 Oil:	For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4. MS 20 Aeroshell W100 Aeroshell W120 (in tropical climates)
8.3 Coolant:	None

9. Fluid capacities:

9.1 Fuel:	Total: 80 litres Usable: 78 litres 2 x 35 litres in main tanks 3 litres in connecting tank 7 litres in gravity tank
9.2 Oil:	Minimum 7 litres – Maximum 11 litres
9.3 Coolant system capacity:	None

10. Air Speeds:

Never exceed speed limit	V_{NE}	
category A, U, N		300 km/h IAS
Maximum speed limit near the ground		
category A, U, N	V_H	230 km/h IAS
Maximum flaps extended speed limit		

	category A, U, N	V_{FE}	130 km/h IAS

11. Maximum Operating Altitude: 5 300 m

12. Allweather Operations Capability: The aircraft is approved for VFR Day flights.

13. Maximum Weights:

Max. Take-off and Landing weight:

- For category Aerobatic (A)	730 kg
- For category Utility (U)	820 kg
- For category Normal (N)	820 kg

Max. Variable Load:

- For category Aerobatic (A)	92 kg
- For category Utility (U)	182 kg
- For category Normal (N)	182 kg

14. Centre of Gravity Range: 18.6 % – 23.0 % M.A.C.
M.A.C. is 1 532 mm;
0 % M.A.C. is 621 mm aft reference datum.

15. Datum: The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.

16. Control surface deflections:

Elevator deflection	up	25° ± 2°
	down	20° ± 2°
Elevator trim	up	23° ± 2°
	down	40° ± 2°
Rudder deflection	left an right	30° ± 2°
Ailerons deflection	up	108 mm (+5, -3) mm
	down	98 mm (+5, -3) mm
Wing flaps position:	retracted	0°
	take-off	15°
	landing	43° (+ 2°, - 3°)

17. Levelling Means: Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.

18. Minimum Flight Crew: 1 (Pilot)

19. Maximum Passenger Seating Capacity: 2 (including crew)

20. (Reserved)

21. Baggage/Cargo Compartments: None

22. Wheels and Tyres: The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS

420x150-6.5 TL with tube 420x150;

The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL;

The wheels K 34-990 or K 13-0000.00 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

DIV. Operating and Service Instructions

1.	Flight manual	
	- In Czech language	Směrnice pro pilota letounu Z 226 B, T
	- In English language	Instruction for Pilot on the Use and Handling of the Z 226 Aircraft
	- In German language	Richtlinien Fuer den Flugzeugfuhrer des Flugzeuges Z 226
2.	Maintenance manual	
	- In Czech language	Návod k obsluze letounu Z 226 T, 1957
3.	Overhaul manual	
	- In Czech language	Opravárenská příručka Z 126, Z 226, 1994
4.	List of spare parts	
	- In Czech language	Seznam náhradních součástí draku letounu Z 226 B, T

Note: Revisions are issued in Czech and English versions languages only.

DV. Notes:

Following Z 226 T aircraft have been converted to the models:

Z 126	S/N: 163
Z 126 M	S/N: 103;
Z 126 T	S/N: 123, 169, 177, 179, 180;
Z 226 A	S/N: 108, 164, 353;
Z 226 AS	S/N: 154, 200;
Z 226 M	S/N: 47, 0104, 120, 122, 127 – 129, 132, 134, 137, 140, 155,167, 188, 191, 192, 196;
Z 226 MS	S/N: 104, 105, 0105,106, 107, 119, 121, 124, 130, 131, 133, 135, 138, 139, 149, 158, 159, 161, 168, 170, 175, 181 – 184, 186, 189, 190, 193, 194, 198, 202, 204, 243, 322, 350, 368, 370

by the aircraft manufacturer.

SECTION E: Z 226 A

EI. General

1. a) Type:	Z 26
b) Model:	Z 226 A
2. Airworthiness category:	Aerobatic (A) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n.p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: 01-08 – 04-08;
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	April 27, 1963

7. EASA Type Certificate replaces CAA CZ Type Certificate No. 63 001 – Z 226 A.

EII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K5 British Civil Airworthiness Requirements, Cat D
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None
9. EASA Environmental	ICAO Annex 16, Volume I, Chapter 10

Standards:

EIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of aircraft Z 226 A, No. S-A 226.000.

2. Description: The Z 226 A aircraft is one-seat, low wing, single-engine monoplane.

3. Equipment: Approved equipment list is stated in document Flight manual.

4. Dimensions:
   Wing Span: 10.280 m
   Length: 7.820 m
   Height: 2.060 m
   Wing Area: 14.900 m^2

5. Engine:

   5.1 Model: Walter Minor 6-III

   5.2 Type Certificate: No. 132/2-L/6A-a.j.-1947, Issued by Czech Ministry of Transport

   5.3 Limitations:

Max. Continuous power	
Max. Power	118 kW (160 HP)
Max. Engine speed	2 500 RPM
Max. Consumption	53 l/h
Max. Cruising power	
Max. Power	92 kW (125 HP)
Max. Engine speed	2 300 RPM
Max. Consumption	38 l/h

6. Load factors:

For category Aerobatic (A)	+6.0 g,	-3.0 g
For category Normal (N)	+3.5 g,	-1.0 g

7. Propellers:

   7.1.1 Model: Z-226.641.2

   7.1.2 Type Certificate: ---

   7.1.3 Number of blades: 2

   7.1.4 Diameter: 2 050 mm

   7.1.5 Sense of Rotation: Anticlockwise in flight direction.

   or

   7.2.1 Model: Z-326.641.1

   7.2.2 Type Certificate: ---

7.2.3 Number of blades: 2

7.2.4 Diameter: 2 000 mm

7.2.5 Sense of Rotation: Anticlockwise in flight direction.

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline, with min. 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.
LBZ 72
LBZ 78
LBE 80
LBE 87
Shell 80
ESSO 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.
MS 20
Aeroshell W100
Aeroshell W120 (in tropical climates)

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: for category A: 80 litres
for category N: 120 litres

Usable: for category A: 78 litres
for category N: 118 litres

2 x 35 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank
40 litres in fuselage auxiliary tank

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds: Never exceed speed limit V_{NE}
(category A, N) 300 km/h IAS

	Maximum speed limit near the ground (category A, N) V_H 215 km/h IAS Maximum flaps extended speed limit (category A, N) V_{FE} 130 km/h IAS
11. Maximum Operating Altitude:	6 450 m
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.
13. Maximum Weights:	Max. Take-off and Landing weight: - For category Aerobatic (A) 715 kg - For category Normal (N) 745 kg Max. Variable Load: - For category Aerobatic (A) 90 kg - For category Normal (N) 187 kg
14. Centre of Gravity Range:	19 % - 26 % MAC M.A.C. is 1 532 mm; 0 % M.A.C. is 621 mm aft reference datum.
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.
16. Control surface deflections:	Elevator deflection up 25° ± 2° down 20° ± 2° Elevator trim up 23° ± 2° down 40° ± 2° Rudder deflection left an right 30° ± 2° Ailerons deflection up 108 mm (+5, -3) mm down 98 mm (+5, -3) mm Wing flaps position: retracted 0° take-off 15° landing 43° (+ 2°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.
18. Minimum Flight Crew:	1 (Pilot)
19. Maximum Passenger Seating Capacity:	1 (including crew)
20. (Reserved)	

21. Baggage/Cargo Compartments:	13 kg (for category Normal only)
22. Wheels and Tyres:	The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL; The wheels K 34-990 or K 13-0000.00 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

EIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Směrnice pro pilota o použití a technice pilotáže letounu Z 226 B, T (upravené pro Z 226 A)

2. Maintenance and operating manual:
   - In Czech language — Návod k obsluze a ošetřování letounu Z – 226 T (upravený pro Z 226 A)

EV. Notes

Following Z 226 A aircraft have been converted to the models:

Z 226 AS S/N: 01-08, 02-08

by the aircraft manufacturer.

SECTION F: Z 226 M

FI. General

1. a) Type:	Z 26
b) Model:	Z 226 M
2. Airworthiness category:	Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n.p. Letiště 1578 765 81 Otrokovice CZECHOSLOVAKIA S/N: see Z 226 B, Z 226 T aircraft
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	November 8, 1979

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 4 – 5150 – 1956, Supplement No. 1.

FII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K4, P3 British Civil Airworthiness Requirements, Cat D-4
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None
9. EASA Environmental	ICAO Annex 16, Volume I, Chapter 10

Standards:

FIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of aircraft Z 226 M – conversion Z 226 T to Z 226 M aircraft, No. S-M 226.000.

2. Description: The Z 226 M aircraft is two-seat, low wing, single-engine monoplane. Control system is installed in rear pilot compartment only.

3. Equipment: Approved minimum equipment list is stated in document Maintenance Manual Z 226 MS (M), Chapter 12.

4. Dimensions:
   Wing Span: 10.280 m
   Length: 7.860 m
   Height: 2.778 m
   Wing Area: 14.900 m^2

5. Engine:

   5.1 Model: M 137 A

   5.2 Type Certificate: 96-02 issued by CAA CZ

   5.3 Limitations:

Max. Take off power (max. 5 min.)	
Max. Power	133 kW (180 HP)
Max. Engine speed	2 750 RPM
Max. Consumption	61 l/h
Max. Manifold pressure	100±2 kPa
Max. Continuous power	
Max. Power	118 kW (160 HP)
Max. Engine speed	2 680 RPM
Max. Consumption	53 l/h
Max. Manifold pressure	95±2 kPa
Max. Cruising power	
Max. Power	103 kW (140 HP)
Max. Engine speed	2 580 RPM
Max. Consumption	48 l/h
Max. Manifold pressure	87 kPa

6. Load factors: + 3.8 g; - 1.5 g

7. Propellers:

   7.1.1 Model: Z 42.6411

   7.1.2 Type Certificate: 70-06 issued by CAA CZ

   7.1.3 Number of blades: 2

7.1.4 Diameter: 2 000 mm

7.1.5 Sense of Rotation: Anticlockwise in flight direction

or

7.2.1 Model: Z 42.6413

7.2.2 Type Certificate: 70-07 issued by CAA CZ

7.2.3 Number of blades: 2

7.2.4 Diameter: 2 050 mm

7.2.5 Sense of Rotation: Anticlockwise in flight direction

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline, with min. 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.
LBZ 72
LBZ 78
LBE 80
LBE 87
Shell 80
ESSO 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.
MS 20
Aeroshell W100
Aeroshell W120

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: 125 litres
Usable: 123 litres

2 x 35 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank
45 litres in fuselage auxiliary tank

9.2 Oil: Minimum 9 litres – Maximum 11 litres

9.3 Coolant system capacity:	None		
10. Air Speeds:	Never exceed speed limit	V_{NE}	290 km/h IAS
	Normal operating speed limit	V_{NO}	216 km/h IAS
	Design manoeuvring speed limit	V_A	183 km/h IAS
	Maximum flaps extended speed limit	V_{FE}	130 km/h IAS
11. Maximum Operating Altitude:	7 150 m		
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.		
13. Maximum Weights:	Max. Take-off and Landing weight: - at towing with Z 42.6411 propeller - at towing with Z 42.6413 propeller		 780 kg 860 kg
	Max. Variable Load: - at towing with Z 42.6411 propeller - at towing with Z 42.6413 propeller		 106 kg 186 kg
14. Centre of Gravity Range:	18.6 % - 24.4 % MAC M.A.C. is 1 532 mm; 0 % M.A.C. is 621 mm aft reference datum.		
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up down	25° ± 1° 20° ± 1°
	Elevator trim	up down	23° ± 2° 40° ± 2°
	Rudder deflection	left an right	30° ± 2°
	Ailerons deflection	up down	108 mm (+5, -3) mm 98 mm (+5, -3) mm
	Wing flaps position:	retracted take-off landing	0° 15° 43° (+ 2°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.		
18. Minimum Flight Crew:	1 (Pilot)		
19. Maximum Passenger	2 (including crew)		

Seating Capacity:

20. (Reserved)

21. Baggage/Cargo Compartments:	Maximum 23 kg
22. Wheels and Tyres:	The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL; The wheels K 34-990 or K 13-0000.00 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

FIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Směrnice pro pilota letounu Z 226 M, date of issue 1978

2. Maintenance and operating manual:
   - In Czech language — Popis – obsluha – údržba ZLIN 226 MS (M), date of issue 1992

3. Overhaul manual:
   - In Czech language — Opravárenská příručka Z 126, Z 226, date of issue 1994

FV. Notes

None

SECTION G: Z 226 MS

GI. General

1. a) Type:	Z 26
b) Model:	Z 226 MS
2. Airworthiness category:	Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n.p. Letiště 1578 765 81 Otrokovice CZECHOSLOVAKIA S/N: see Z 226 B, Z 226 T aircraft
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	October 21, 1986

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 4 – 5150 – 1956, Supplement No. 1, Modification.

GII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschriften für Flugzeuge, K4, P3 British Civil Airworthiness Requirements, Cat D-4
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None

9. EASA Enviromental Standards:	ICAO Annex 16, Volume I, Chapter 10

GIII. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of aircraft Z 226 MS, No. MS 226.000.
2. Description:	The Z 226 MS aircraft is two-seat, low wing, single-engine, monoplane. Control system is installed in rear pilot compartment only.
3. Equipment:	Approved minimum equipment list is stated in document Maintenance Manual Z 226 MS (M), Chapter 12.
4. Dimensions:	Wing Span: 10.280 m Length: 7.860 m Height: 2.778 m Wing Area: 14.900 m^2
5. Engine:	
5.1 Model:	M 137 A
5.2 Type Certificate:	96-02 issued by CAA CZ
5.3 Limitations:	Max. Take off power (max. 5 min.) Max. Power 133 kW (180 HP) Max. Engine speed 2 750 RPM Max. Consumption 61 l/h Max. Manifold pressure 100±2 kPa Max. Continuous power Max. Power 118 kW (160 HP) Max. Engine speed 2 680 RPM Max. Consumption 53 l/h Max. Manifold pressure 95±2 kPa Max. Cruising power Max. Power 103 kW (140 HP) Max. Engine speed 2 580 RPM Max. Consumption 48 l/h Max. Manifold pressure 87 kPa
6. Load factors:	+ 3.8 g; - 1.5 g
7. Propeller:	
7.1.1 Model:	V 503
7.1.2 Type Certificate:	64 002 issued by CAA CZ
7.1.3 Number of blades:	2

7.1.4	Diameter:	1 950 mm
7.1.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.2.1	Model:	V 503 A
7.2.2	Type Certificate:	69-02 issued by SLI CZ
7.2.3	Number of blades:	2
7.2.4	Diameter:	2 000 mm
7.2.5	Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1	Fuel:	Non-ethylated aviation gasoline, with min. 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol. LBZ 72 LBZ 78 LBE 80 LBE 87 Shell 80 ESSO 80 AVGAS 100 LL (DEFENCE STANDARD 91/90, ASTM D910)
8.2	Oil:	For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4. MS 20 Aeroshell W100 Aeroshell W120 (in tropical climates)
8.3	Coolant:	None

9. Fluid capacities:

9.1	Fuel:	Total: 125 litres Usable: 123 litres 2 x 35 litres in main tanks 3 litres in connecting tank 7 litres in gravity tank 45 litres in fuselage auxiliary tank
9.2	Oil:	Minimum 9 litres – Maximum 11 litres

9.3 Coolant system capacity:	None		
10. Air Speeds:	Never exceed speed limit	V_{NE}	290 km/h IAS
	Normal operating speed limit	V_{NO}	216 km/h IAS
	Design manoeuvring speed limit	V_A	183 km/h IAS
	Maximum flaps extended speed limit	V_{FE}	130 km/h IAS
11. Maximum Operating Altitude:	6 000 m		
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.		
13. Maximum Weights:	Max. Take-off and Landing weight:		890 kg
	Max. Variable Load:		191 kg
14. Centre of Gravity Range:	17 % - 24.5 % MAC M.A.C. is 1 532 mm; 0 % M.A.C. is 621 mm aft reference datum.		
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up down	25° ± 1° 20° ± 1°
	Elevator trim	up down	23° ± 2° 40° ± 2°
	Rudder deflection	left an right	30° ± 2°
	Ailerons deflection	up down	108 mm (+5, -3) mm 98 mm (+5, -3) mm
	Wing flaps position:	retracted take-off landing	0° 15° 43° (+ 2°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.		
18. Minimum Flight Crew:	1 (Pilot)		
19. Maximum Passenger Seating Capacity:	2 (including crew)		
20. (Reserved)			

21. Baggage/Cargo Compartments:	Maximum 23 kg
22. Wheels and Tyres:	The wheel K 420.00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheel K 56-1100.00 of main gear with tubeless tyre MITAS 420x150-6.5 TL; The wheels K 34-990 or K 13-0000.00 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

GIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language: Letová příručka Z 226 MS, date of issue 1986

2. Maintenance and operating manual:
   - In Czech language: Popis – obsluha – údržba ZLIN 226 MS (M), date of issue 1992

3. Overhaul manual:
   - In Czech language: Opravárenská příručka Z 126, Z 226, date of issue 1994

GV. Notes

None

SECTION H: Z 326

HI. General

1. a) Type:	Z 26
b) Model:	Z 326
2. Airworthiness category:	Aerobatic (A) Utility (U) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n. p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: 301 – 304; 501 – 523 Strojírny první pětiletky, n. p. Kunovice, závod Moravan Otrokovice CZECHOSLOVAKIA S/N: 524 – 530; 532 – 550; 556 – 580; 587 – 595; 600 – 907 Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 908 – 933
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	October 13, 1959

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. č.j. 2417/59.

HII. Certification Basis

1. Reference Date for determining the applicable requirements:	---

2. (Reserved)

3. (Reserved)

4.	Airworthiness Requirements:	Bauvorschriften für Flugzeuge, Kat. K 5 British Civil Airworthiness Requirements, Cat. D
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemptions:	None
8.	EASA Equivalent Safety Findings:	None
9.	EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

HIII. Technical Characteristics and Operational Limitations

1.	Type Design Definition:	The specification list of aircraft Z 326, No. S-Z 326.000.
2.	Description:	The Z 326 aircraft is two-seat, low wing, single-engine, monoplane
3.	Equipment:	Approved equipment list is stated in document Flight Manual Z 326.
4.	Dimensions:	Wing Span: 10.596 m without auxiliary tanks 10.845 m with auxiliary tanks 7.820 m Length: 2.060 m Height: 15.451 m^2 Wing Area:
5.	Engine:	
	5.1 Model:	Walter Minor 6-III.
	5.2 Type Certificate:	č.j. 132/2-L/6A-a.i.-1947
	5.3 Limitations:	Max. Continuous power Max. Power 118 kW (160 HP) Max. Engine speed 2 500 RPM Max. Consumption 54 l/h Max. Cruising power Max. Power 92 kW (125 HP) Max. Engine speed 2 300 RPM Max. Consumption 36 l/h
6.	Load factors:	For category Aerobatic (A) +6.0 g; -3.0 g For category Utility (U) +4.5 g; -1.8 g For category Normal (N) +3.5 g; -1.0 g

7. Propellers:

7.1.1	Model:	Z – 326.641
7.1.2	Type Certificate:	č.j. 2417/59 from date October 13, 1959
7.1.3	Number of blades:	2
7.1.4	Diameter:	2 000 mm
7.1.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.2.1	Model:	Z – 226.640
7.2.2	Type Certificate:	č.j. 6312/61 from date September 26, 1961
7.2.3	Number of blades:	2
7.2.4	Diameter:	2 050 mm
7.2.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.3.1	Model:	Z – 326.641.1
7.3.2	Type Certificate:	č.j. 2417/59 from date October 13, 1959
7.3.3	Number of blades:	2
7.3.4	Diameter:	2 000 mm
7.3.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.4.1	Model:	Z – 42.6413
7.4.2	Type Certificate:	No. 70-07 issued by CAA CZ
7.4.3	Number of blades:	2
7.4.4	Diameter:	2 050 mm
7.4.5	Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1	Fuel:	Non-ethylated aviation gasoline with min. 72 ÷ 87 octanes. Application of ethylated fuels is only

permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.

Recommended kinds of fuel:
LBZ 72
LBZ 83

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.
Recomended kinds of oil:
MS 20
AERO-SHELL W100
AERO-SHELL W120 in tropical climates.

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel:

Total:	for category A, U:	100 litres
	for category N:	170 litres
Usable:	for category A, U:	98 litres
	for category N:	168 litres

2 x 45 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank
2 x 35 litres in auxiliary wing tip tanks

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never Exceed Speed Limit	V_{NE}	320 km/h IAS
Normal Operationg Speed Limit	V_{NO}	212 km/h IAS

11. Maximum Operating Altitude: 4 750 m

12. Allweather Operations Capability: The aircraft is approved for VFR Day flights.

13. Maximum Weights: Max. Take-off and Landing weight:

- For category Aerobatic (A) 910 kg
- For category Utility (U) 935 kg
- For category Normal (N) 975 kg

Maximum Variable Load:

	- For category Aerobatic (A)		177 kg
	- For category Utility (U)		202 kg
	- For category Normal (N)		182 kg
14. Centre of Gravity Range:	18 ÷ 30 % MAC M.A.C. is 1 545 mm; 0 % M.A.C. is 621 mm aft reference datum.		
15. Datum:	The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up	25° ± 1°
		down	20° ± 1°
	Elevator trim	up	25° ± 2°
		down	40° ± 2°
	Rudder deflection	left an right	28° ± 2°
	Rudder trim	left	5° ± 1°
		right	30° ± 2°
	Ailerons deflection	up	108 mm (+5, -3) mm
		down	98 mm (+5, -3) mm
	Wing flaps position:	retracted	0°
		take-off	15°
		landing	40° (+ 5°, - 3°)
17. Leveling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.		
18. Minimum Flight Crew:	1 (Pilot)		
19. Maximum Passenger Seating Capacity:	2 (including crew)		
20. (Reserved)			
21. Baggage/Cargo Compartments:	---		
22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6,5 or MITAS 420x150-6,5 TL with tube 420x150; The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85–4 with tube 260x85.		

HIV. Operating and Service Instructions

1. Flight Manual:
   - In Czech language: Letová příručka letounu s vrtulí Z 326, date of issue 1964 Směrnice pro pilota letounu Z 326 s vrtulí V 503
   - In English language: Instruction for Pilot on the Use and Handling of the Z 326 Aircraft
   - In German language: Anweisungen für den Flugzeug Führer über die Verwendung und Technik des Fliegens mit dem Flugzeug Z 326

2. Technical Manual:
   In Czech language: Technický popis a návod k obsluze letounu Z 326 – Speciál s vrtulí V 503
   - In English language: Description of Aircraft Z 326 and Direction for its Operation

3. Repair Manual:
   - In Czech language: Příručka pro generální opravu letounu Z 326, date of issue 1961
   - In English language: Instruction Manual for Major Overhaul of Z 326 Aircraft, date of issue 1963 Instruction for Complete Overhauls and Minor Repairs of the Z 326 Tail Landing Gear.

4. Catalogue of spare parts:
   - In Czech language: Trener Master Z 326 – Seznam náhradních součástí, date of issue 1963

HV. Notes:

Following Z 326 aircraft have been converted to the models:

Z 326 M	S/N: 301, 604, 606, 608 – 610, 612, 833;
Z 326 MF	S/N: 605, 861, 893;
Z 526	S/N: 869, 886, 890, 901, 916, 930, 932;
Z 526 A	S/N: 922;
Z 526 M	S/N: 909;
Z 526 ML	S/N: 921;
Z 526 F	S/N: 838

by the aircraft manufacturer.

SECTION I: Z 326 A

I.I. General

1. a) Type:	Z 26
b) Model:	Z 326 A
2. Airworthiness category:	Aerobatic (A) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Strojírny první pětiletky, n. p. Kunovice, závod Moravan Otrokovice CZECHOSLOVAKIA S/N: 531, 551 – 555, 581 – 586, 596 – 599
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	December 30, 1960

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. č.j. 8061/60.

I.II. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschrifted für Flugzeuge, Kat. K 5 British Civil Airworthiness Requirements, Cat. D
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety Findings:	None

9. EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10	

I.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of aircraft Z 326 A, No. S-A 326.000.	
2. Description:	The Z 326 A aircraft is one-seat, low wing, single-engine monoplane.	
3. Equipment:	Approved equipment list is stated in document Flight Manual Z 326 A.	
4. Dimensions:	Wing Span:	10.596 m without auxiliary tanks
		10.850 m with auxiliary tank
	Length:	7.820 m
	Height:	2.060 m
	Wing Area:	15.451 m^2
5. Engine:		
5.1 Model:	Walter Minor 6-III.	
5.1 Type Certificate:	č.j. 132/2-L/6A-a.i.-1947	
5.1 Limitations:	Max. Continuous power	
	Max. Power	118 kW (160 HP)
	Max. Engine speed	2 500 RPM
	Max. Consumption	54 l/h
	Max. Cruising power	
	Max. Power	92 kW (125 HP)
	Max. Engine speed	2 300 RPM
	Max. Consumption	36 l/h
6. Load factors:	For category Aerobatic (A)	+6.0 g; -3.0 g
	For category Normal (N)	+3.5 g; -1.0 g
7. Propeller:		
7.1 Model:	Z – 326.641	
7.2 Type Certificate:	č.j. 8061/60, December 30, 1960	
7.3 Number of blades:	2	
7.4 Diameter:	2 000 mm	
7.5 Sense of Rotation:	Anticlockwise in flight direction	
8. Fluids:		
8.1 Fuel:	Non-ethylated aviation gasoline with min. 72 ÷ 87 octanes. Application of ethylated fuels is only	

permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.

Recomended kinds of oil:
MS 20
AERO-SHELL W100
AERO-SHELL W120 in tropical climates.

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: 80 litres
Usable: 78 litres

2 x 35 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds: Never Exceed Speed Limit V_{NE} 320 km/h CAS

Normal Operationg Speed Limit V_{NO} 212 km/h CAS

11. Maximum Operating Altitude: 5 200 m

12. Allweather Oparations Capability: The aircraft is approved for VFR Day flights.

13. Maximum Weights: Max. Take-off and Landing weight:

- For category Aerobatic (A) 790 kg
- For category Normal (N) 850 kg

Maximum Variable Load:
- For category A, N 111 kg

14. Centre of Gravity Range: 18.6 ÷ 24.5 % MAC
M.A.C. is 1 545 mm;
0 % M.A.C. is 621 mm aft reference datum.

15. Datum: The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.

16. Control surface deflections: Elevator deflection up 25° ± 1°

		down	20° ± 1°
	Elevator trim	up	25° ± 2°
		down	40° ± 2°
	Rudder deflection	left an right	28° ± 2°
	Rudder trim	left	5° ± 1°
		right	30° ± 2°
	Ailerons deflection	up	108 mm (+5, -3) mm
		down	98 mm (+5, -3) mm
	Wing flaps position:	retracted	0°
		take-off	15°
		landing	40° (+ 5°, - 3°)

17. Levelling Means: Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.

18. Minimum Flight Crew: 1 (Pilot)

19. Maximum Passenger Seating Capacity: 1 (including crew)

20. (Reserved)

21. Baggage/Cargo Compartments: Maximum is 34 kg

22. Wheels and Tyres: The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6,5 or MITAS 420x150-6,5 TL with tube 420x150;

The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85–4 with tube 260x85.

I.IV. Operating and Service Instructions

1. Flight Manual:

- In Czech language	Letová příručka letounu s vrtulí Z 326, date of issue 1964 Směrnice pro pilota letounu Z 326 s vrtulí V 503
- In English language	Instruction for Pilot on the Use and Handling of the Z 326 Aircraft
- In German language	Anweisungen für den Flugzeug Führer über die Verwendung und Technik des Fliegens mit dem Flugzeug Z 326

2. Technical Manual:

- In Czech language	Technický popis a návod k obsluze letounu Z 326 – Speciál s vrtulí V 503
- In English language	Description of Aircraft Z 326 and Direction for its Operation

3. Repair Manual:

- In Czech language	Příručka pro generální opravu letounu Z 326, date of issue 1961
- In English language	Instruction Manual for Major Overhaul of Z 326 Aircraft,date of issue 1963Instruction for Complete Overhauls and Minor Repairs of the Z 326 Tail Landing Gear.

4. Catalogue of spare parts:

- In Czech language	Trener Master Z 326 – Seznam náhradních součástí, date of issue 1963

I.V. Notes:

None

SECTION J: Z 326 M

JI. General

1. a) Type:	Z 26
b) Model:	Z 326 M
2. Airworthiness category:	Utility (U) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n. p. Gottwaldov – Otrokovice CZECHOSLOVAKIA S/N: see Z 326 aircraft
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	February 5, 1976

7. The EASA Type Certificate replaces the CAA CZ Type Certificate Supplement No. 4 from date 13.10.1959 (č.j. 2417/59)

JII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	Bauvorschrifted für Flugzeuge, Kat. K 5 British Civil Airworthiness Requirements, Cat. D
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None

8. EASA Equivalent Safety Findings:	None
9. EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

JIII. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of aircraft Z 326 M – conversion Z 326 to Z 326 M aircraft, No. S-M 326.000.
2. Description:	The Z 326 M aircraft is two-seat, low wing, single-engine monoplane.
3. Equipment:	Approved equipment list is stated in document Maintenance Manual ZLIN 326 M, MS.
4. Dimensions:	Wing Span: 10.596 m Length: 7.820 m Height: 2.060 m Wing Area: 15.451 m^2
5. Engine:	
5.1 Model:	M 137 A
5.2 Type Certificate:	No. 96-02 issued by CAA CZ
5.3 Limitations:	Max. Take off power (max. 5 min.) Max. Power 132 kW (180 HP) Max. Engine speed 2 750 RPM Max. Consumption 61 l/h Max. Manifold pressure 100 ±2 kPa Max. Continuous power Max. Power 118 kW (160 HP) Max. Engine speed 2 680 RPM Max. Consumption 52 l/h Max. Manifold pressure 95 kPa ±2 kPa Max. Cruising power Max. Power 103 kW (140 HP) Max. Engine speed 2 580 RPM Max. Consumption 45 l/h Max. Manifold pressure 87 kPa
6. Load factors:	For category Utility (U) +5.5 g; -3.0 g For category Normal (N) +3.5 g; -1.0 g
7. Propellers:	

7.1.1	Model:	Z 42.6411
7.1.2	Type Certificate:	No. 70-06 issued by CAA CZ
7.1.3	Number of blades:	2
7.1.4	Diameter:	2 050 mm
7.1.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.2.1	Model:	Z 42.6413
7.2.2	Type Certificate:	No. 70-07 issued by CAA CZ
7.2.3	Number of blades:	2
7.2.4	Diameter:	2 050 mm
7.2.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.3.1	Model:	V 503
7.3.2	Type Certificate:	No. 64 002 issued by CAA CZ
7.3.3	Number of blades:	2
7.3.4	Diameter:	1 950 mm
7.3.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.4.1	Model:	V 503 A
7.4.2	Type Certificate:	No. 69-02 issued by CAA CZ
7.4.3	Number of blades:	2
7.4.4	Diameter:	2 000 mm
7.4.5	Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1	Fuel:	Non-ethylated aviation gasoline with min. 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not

exceed the value of 0.06 % vol.
Recommended kinds of fuel:
LBZ 72
LBZ 78
LBE 80
LBE 87

SHELL 80
ESSO 80
100L
Grade 100/130.

8.2 Oil: For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4.
Recomended kinds of oil:
MS 20
AERO-SHELL W100
AERO-SHELL W120
ELF AD-100.

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: for category U: 100 litres
for category N: 170 litres

Usable: for category U: 98 litres
for category N: 168 litres

2 x 45 litres in main tanks
3 litres in connecting tank
7 litres in gravity tank
2 x 35 litres in auxiliary wing tip tanks

9.2 Oil: Minimum 7 litres – Maximum 11 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never Exceed Speed Limit V_{NE}
308 km/h IAS

Normal Operating Speed Limit V_{NO}
222 km/h IAS

Design Manoeuvring Speed Limit V_A
222 km/h IAS

Maximum Flaps Extended Speed Limit V_{FE}
140 km/h IAS

11. Maximum Operating Altitude: 4 750 m

12. Allweather Operations Capability: The aircraft is approved for VFR Day flights.

13. Maximum Weights:

Max. Take-off and Landing weight:
- For category Utility (U) 910 kg
- For category Normal (N) 975 kg

Maximum Variable Load:
- For category Utility (U): pilot 77 kg
  person, cargo 85 kg
- For category Normal (N): pilot 77 kg
  person, cargo 90 kg

14. Centre of Gravity Range: 18 ÷ 30 % MAC
M.A.C. is 1 545 mm; 0 % M.A.C. is 616 mm aft reference datum.

15. Datum: The rear part of firewall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.

16. Control surface deflections:

Elevator deflection	up	25° ± 1°
	down	20° ± 1°
Elevator trim	up	25° ± 2°
	down	40° ± 2°
Rudder deflection	left an right	28° ± 2°
Rudder trim	left	5° ± 1°
	right	30° ± 2°
Ailerons deflection	up	108 mm (+5, -3) mm
	down	98 mm (+5, -3) mm
Wing flaps position:	retracted	0°
	take-off	15°
	landing	40° (+ 5°, - 3°)

17. Levelling Means: Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.

18. Minimum Flight Crew: 1 (Pilot)

19. Maximum Passenger Seating Capacity: 2 (including crew)

20. (Reserved)

21. Baggage/Cargo Compartments: For category Utility (U): 85 kg
For category Normal (N): 90 kg

22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6,5 or MITAS 420x150-6,5 TL with tube 420x150; The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85–4 with tube 260x85.

JIV. Operating and Service Instructions

1. Flight Manual:	
– In Czech language	Z 326 Dodatek č. 1 k letové příručce – Z 326 M, date of issue 1976
2. Technical Manual:	
– In Czech language	Popis – obsluha – údržba ZLIN 326 M, MS, date of issue 1994

JV. Notes:

None

SECTION K: Z 526

KI. General

1.	a) Type:	Z 26
	b) Model:	Z 526
2.	Airworthiness category:	Aerobatic (A) Normal (N)
3.	Type Certificate Holder:	ZLIN AIRCRAFT a.s. Olivova 4/2096 110 00 Praha 1 Czech Republic
4.	Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1006-1015, 1021-1025, 1031-1035, 1046-1066, 1068-1074, 1077-1088
5.	Certification Application Date:	---
6.	CAA CZ Type Certificate Date:	April 26, 1966

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 1617/66, Supplement No. 2.

KII. Certification Basis

1.	Reference Date for determining the applicable requirements:	---
2.	(Reserved)	
3.	(Reserved)	
4.	Airworthiness Requirements:	Bauvorschriften für Flugzeuge, Kat. K; British Civil Airworthiness Requirements, Cat. D, AIR 2052, effective November 7, 1968
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemptions:	None

8. EASA Equivalent Safety Findings:	None
9. EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

KIII. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of Z 526 aircraft, No. S-Z 526.000.		
2. Description:	The Z 526 aircraft is two-seat, low wing, single-engine, monoplane.		
3. Equipment:	Approved equipment list is stated in document Technical Description, Operation Instruction for Z 526 Aircraft.		
4. Dimensions:	Wing Span:	10.845 m	with auxiliary tanks
		10.596 m	without auxiliary tanks
	Length:	8.000 m	
	Height:	2.060 m	
	Wing Area:	15.450 m^2	
5. Engine:			
5.1 Model:	Walter Minor 6-III		
5.2 Type Certificate:	Č.j. 132/2-L/6A-a.i.-1947		
5.3 Limitations:	Max. Continuous power		
	Max. Power	118 kW (160 HP)	
	Max. Engine speed	2 500 RPM	
	Max. Consumption	53 l/h	
	Max. Cruising power		
	Max. Power	92 kW (125 HP)	
	Max. Engine speed	2 300 RPM	
	Max. Consumption	38 l/h	
6. Load factors:	For category Aerobatic (A)	+6 g,	-3 g
	For category Normal (N)	+3.5 g,	-1 g
7. Propeller:			
7.1 Model:	V 503		
7.2 Type Certificate:	No. 64 002 issued by CAA CZ		
7.3 Number of blades:	2		

7.4 Diameter:	1 950 mm
7.5 Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1 Fuel:	Non-ethylated aviation gasoline, with min. 72 octanes (for example LBE 72 or LBE 83). Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06 % vol.
8.2 Oil:	For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C. Recommended kinds of oil: MS20-GOST 1013-49 ELF AD-100 AERO-SHELL W100 AERO-SHELL W120
8.3 Coolant:	None

9. Fluid capacities:

9.1 Fuel:	Total: for category A: 100 litres for category N: 170 litres Usable: for category A: 98 litres for category N: 168 litres 2 x 45 litres in main tanks 3 litres in connecting tank 7 litres in gravity tank 2 x 35 litres in auxiliary wing tip tanks
9.2 Oil:	Minimum 9 litres – Maximum 11 litres
9.3 Coolant system capacity:	None

10. Air Speeds:

Never Exceed Speed Limit	V_{NE} 292 km/h IAS
Normal Operating Speed Limit	V_{NO} 234 km/h IAS
Design Manoeuvring Speed Limit	V_A 230 km/h IAS
Maximum Flaps Extended Speed Limit	V_{FE} 140 km/h IAS

11. Maximum Operating 5 000 m

Altitude:

12. Allweather Operations Capability: The aircraft is approved for VFR Day flights.

13. Maximum Weights: Max. Take-off and Landing weight:

- For category Aerobatic	(A)	940 kg
- For category Normal	(N)	975 kg

Maximum Variable Load:

- For category Aerobatic	(A)	192 kg
- For category Normal	(N)	166 kg

14. Centre of Gravity Range: 17 % ÷ 27.4 % MAC
M.A.C. is 1 545 mm; 0 % M.A.C. is 616 mm aft reference datum.

15. Datum: The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.

16. Control surface deflections:

Elevator deflection	up	25° ± 1°
	down	20° ± 1°
Elevator trim	up	25° ± 2°
	down	40° ± 2°
Rudder deflection	left an right	28° ± 2°
Rudder trim	left	5° ± 1°
	right	30° ± 2°
Ailerons deflection	up	108 mm (+5, -3) mm
	down	98 mm (+5, -3) mm
Wing flaps position:	retracted	0°
	take-off	15°
	landing	40° (+ 5°, - 3°)

17. Levelling Means: Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.

18. Minimum Flight Crew: 1 (Pilot)

19. Maximum Passenger Seating Capacity: 2 (including crew)

20. (Reserved)

21. Baggage/Cargo Compartments: Maximum 17 kg

22. Wheels and Tyres: The wheels K 12-0100.00 or K 420.1-00 of main

gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150;

The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

KIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Letová příručka školního a akrobatického letounu Z 526 – Z 526 A.
     Směrnice pro pilota. Použití a technika pilotáže letounů TRENER typů Z 526 a Z 526 A.
   - In English language — Instruction for Pilot on the Use and Handling of the Training and Acrobatic Z 526 and Z 526 A Aircraft, date of issue 1966
   - In German language — Handbuch für den Flugzeugführer zum Schulungs und kunstflugzeuges Z 526 – Z 526 A, date of issue 1966

2. Descriprion – Operation – Maintenance:
   - In Czech language — Technický popis a návod k obsluze letounu Z 526 – Z 526 A
   - In English language — Technical Description, Operation Instruction for Z 526 – Z 526 A Aircrafts, date of issue 1966
   - In German language — Technische Beschreibung und Bedienungsanleitung zum Flugzeug Z 526 – Z 526 A, date of issue 1966

3. Overhaul Manual:
   - In English language — Major Overhaul of Z 526 – Z 526 A Aircraft, date of issue 1969

4. Catalogue of spare parts:
   - In Czech, English, German language
     Trener Master Z 526 Katalog, date of issue 1967

KV. Notes:

Following Z 526 aircraft have been converted to the model:

Z 326	S/N: 909, 1008
Z 526 F	S/N: 1071
Z 526 M	S/N: 1021, 1023, 1031-1033, 1053, 1068, 1077, 1088
Z 726	S/N: 1069

by the aircraft manufacturer.

SECTION L: Z 526 A

LI. General

1.	a) Type:	Z 26
	b) Model:	Z 526 A
2.	Airworthiness category:	Aerobatic (A) Normal (N)
3.	Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4.	Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1001-1005, 1016-1020, 1036-1045, 1067
5.	Certification Application Date:	---
6.	CAA CZ Type Certificate Date:	April 26, 1966

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 1617/66, Supplement No. 2.

LII. Certification Basis

1.	Reference Date for determining the applicable requirements:	---
2.	(Reserved)	
3.	(Reserved)	
4.	Airworthiness Requirements:	Bauvorschriften für Flugzeuge, Kat. K; British Civil Airworthiness Requirements, Cat. D
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemptions:	None
8.	EASA Equivalent Safety	None

Findings:

9. EASA Environmental Standards: ICAO Annex 16, Volume I, Chapter 10

LIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of Z 526 A aircraft, No. S-A 526.000.

2. Description: The Z 526 A aircraft is one-seat, low wing, single-engine, monoplane.

3. Equipment: Approved equipment list is stated in document Technical Description, Operation Instruction for Z 526 – Z 526 A Aircraft.

4. Dimensions:

Wing Span:	10.845 m	with auxiliary tanks
	10.596 m	without auxiliary tanks
Length:	8.000 m	
Height:	1.950 m	
Wing Area:	15.450 m^2	

5. Engine:

5.1 Model: Walter Minor 6-III

5.2 Type Certificate: Č.j. 132/2-L/6A-a.i.-1947

5.3 Limitations:

Max. Continuous power	
Max. Power	118 kW (160 HP)
Max. Engine speed	2 500 RPM
Max. Consumption	53 l/h
Max. Cruising power	
Max. Power	92 kW (125 HP)
Max. Engine speed	2 300 RPM
Max. Consumption	38 l/h

6. Load factors:

For category Aerobatic (A)	+ 6.0 g,	- 3.0 g
For category Normal (N)	+ 3.5 g,	- 1.0 g

7. Propeller:

7.1 Model: V 503

7.2 Type Certificate: No. 64002 issued by CAA CZ

7.3 Number of blades: 2

7.4 Diameter: 1 950 mm

7.5 Sense of Rotation:	Anticlockwise in flight direction
8. Fluids:	
8.1 Fuel:	Non-ethylated aviation gasoline, with min. 72 octanes (for example LBE 72 or LBE 83). Application of ethylated fuels is only permitted in case, the T.E.L. content does not exceed the value of 0.06 % vol.
8.2 Oil:	For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C. Recommended kinds of oil: MS20-GOST 1013-49 ELF AD-100 AERO-SHELL W100 AERO-SHELL W120
8.3 Coolant:	None
9. Fluid capacities:	
9.1 Fuel:	Total: for category A: 80 litres for category N: 150 litres Usable: for category A: 78 litres for category N: 148 litres 2 x 35 litres in main tanks 3 litres in connecting tank 7 litres in gravity tank 2 x 35 litres in auxiliary wing tip tanks
9.2 Oil:	Minimum 9 litres – Maximum 11 litres
9.3 Coolant system capacity:	None
10. Air Speeds:	Never Exceed Speed Limit V_{NE} 292 km/h IAS Normal Operating Speed Limit V_{NO} 236 km/h IAS Design Manoeuvring Speed Limit V_A 230 km/h IAS Maximum Flaps Extended Speed Limit V_{FE} 140 km/h IAS
11. Maximum Operating Altitude:	6 000 m
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.

13. Maximum Weights:

Max. Take-off and Landing weight:

- For category Aerobatic (A)	850 kg
- For category Normal (N)	910 kg

Maximum Variable Load:

- For category Aerobatic (A)	142 kg
- For category Normal (N)	131 kg

14. Centre of Gravity Range:

17 % ÷ 29 % MAC
M.A.C. is 1 545 mm; 0 % M.A.C. is 621 mm aft reference datum.

15. Datum:

The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.

16. Control surface deflections:

Elevator deflection	up	28° ± 1°
	down	24° ± 1°
Elevator trim	up	25° ± 2°
	down	35° ± 2°
Rudder deflection	left an right	30° ± 2°
Rudder trim	left	5° ± 1°
	right	30° ± 2°
Ailerons deflection	up	112 mm (+5, -3) mm
	down	108 mm (+5, -3) mm
Wing flaps position:	retracted	0°
	take-off	15°
	landing	40° (+ 5°, - 3°)

17. Levelling Means:

Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.

18. Minimum Flight Crew:

1 (Pilot)

19. Maximum Passenger Seating Capacity:

1 (including crew)

20. (Reserved)

21. Baggage/Cargo Compartments:

Maximum 17 kg

22. Wheels and Tyres:

The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150;

The wheels K 13-0000.00 or K 34-990 of tail gear

with tyre BARUM or MITAS 260x85-4 with tube 260x85.

LIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Letová příručka školního a akrobatického letounu Z 526 – Z 526 A
     Směrnice pro pilota. Použití a technika pilotáže letounů TRENER typů Z 526 a Z 526 A.
   - In English language — Instruction for Pilot on the Use and Handling of the Training and Acrobatic Z 526 and Z 526 A Aircraft, date of issue 1966
   - In German language — Handbuch für den Flugzeugführer zum Schulungs und kunstflugzeuges Z 526 – Z 526 A, date of issue 1966

2. Description – Operation – Maintenance:
   - In Czech language — Technický popis a návod k obsluze letounu Z 526 – Z 526 A
   - In English language — Technical Description, Operation Instruction for Z 526 – Z 526 A Aircrafts, date of issue 1966
   - In German language — Technische Beschreibung und Bedienungsanleitung zum Flugzeug Z 526 – Z 526 A, date of issue 1966

3. Overhaul Manual:
   - In English language — Major Overhaul of Z 526 – Z 526 A Aircraft, date of issue 1969

4. Catalogue of spare parts:
   - In Czech, English, German language
     Trener Master Z 526 Katalog, date of issue 1967

LV. Notes:

Following Z 526 A aircraft has been converted to the model:

Z 526 AM S/N: 1019
by the aircraft manufacturer.

SECTION M: Z 526 F

MI. General

1. a) Type:	Z 26
b) Model:	Z 526 F
2. Airworthiness category:	Aerobatic (A) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1075-1076, 1089-1100, 1102-1117, 1119, 1121-1124, 1128, 1146-1159, 1161-1185, 1231-1300, 1311-1325
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	October 14, 1969

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 69-04.

MII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requirements:	FAR PART 23 effective March 14, 1969
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None

7. EASA Exemptions: None

8. EASA Equivalent Safety Findings:

§ 23.177(a)(2), (3) – In aerobatic (without auxiliary wing tip tanks) there is no tendency to raise the low wing in a slip. This deviation is admitted with regard to the purpose of the aircraft (aerobatic flying) and to the operational experience. It is possible to ensure the fulfilment of the requirement by the installation of the spring device, which the manufacturer delivers on request. In normal category, there is no deviation the requirement is met.

§ 23.207 – The requirement of “clear and distinct stall warning” is not met in aerobatic category in case of wing stalling with power and in turning flight stalls in normal category in all conditions. The deviation is admitted with regard to the operational experience.

§ 23.613(c) – In the design and the construction of the airplane the Czechoslovak material standards (ČSN) and the specifications being in force for the Czechoslovak aircraft industry have been used. The deviation is admitted because the intent of the requirement is fulfilled.

§ 23.781 – The shape of the flaps and landing gear control knobs does not meet the requirement. The deviation is admitted with regard to the location and the sense of movement of these controls, which ensure the same level of safety.

§ 23.955 – The requirement of the fuel flow rate delivered by the fuel pump system to the engine is not met. The deviation is admitted with regard to the fact that the fuel flow is throttled by the valve LUN 7520.02 and is by 50 % higher than the take-off consumption of the engine.

§ 23.991(b) – The aircraft is not equipped with emergency pump for fuel supply resume into the engine in case of main fuel pump failure. It is admitted with regard to the fact that: The engine is equipped with high-pressure injector, which is joined with low-pressure supply fuel pump into one aggregate. Any pertinent failure of this aggregate would cause contemporaneous breakdown of supply pump and injector. In that case, no emergency pump could ensure fuel supply and distribution to finish the flight without excessive efforts and attention distraction of the pilot. No failure of the low-pressure supply fuel pump has

occurred yet and its occurrence is extremely improbable.

§ 23.1145 – The requirement of means to prevent the inadvertent operation of ignition switches is not met. The deviation is admitted with regard to the location and the shape of the switch.

§ 23.1183(a) – The requirement of the fire resistancy of the lines (hoses) is not met.

§ 23.1191(g) – The requirement of the resistance of fittings against the flame penetration is not met. The deviation is admitted with regard to the operation experience.

§ 23.1337(b) – The LUN 1600 indicator does not meet the requirement of the calibration of the fuel quantity indicator in gallons or pounds, where the calibration in litres is used. The deviation is admitted because the intent of the requirement is fulfilled and safety level is not affected.

§ 23.1357(d) – The requirement of the battery circuit braker resetting is not met. The deviation is admitted with regard to the operational experience.

§ 23.1389, § 23.1391, § 23.1393, § 23.1395 and §23.1397 – Requirements concerning the location and intensities of position lights are not met. This is admitted because the airplane is certified for VFR-DAY flights and the position lights are fitted only for the facilitation of the manoeuvering the plane on the ground.

9. EASA Environmental Standards: ICAO Annex 16, Volume I, Chapter 10

MIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of aircraft Z 526 F
No. S-F 526.000 (up to 22nd series including)
No. S-F 526.000.1 (from 23rd series)

2. Description: The Z 526 F aircraft is two-seat, low wing, single-engine, cantilever monoplane.

3. Equipment: Approved equipment list is stated in document Description, Operation, Maintenance ZLIN 526 F, Chapter 10.

4. Dimensions:
Wing Span: 10.596 m
Length: 8.000 m
Height: 2.060 m
Wing Area: 15.450 m^2

5. Engine:

5.1 Model:	M 137 A	
5.2 Type Certificate:	No. 96-02 issued by CAA CZ	
5.3 Limitations:	Max. Take-off power	
	Max. Power	132 kW (180 k)
	Max. Engine speed	2 750 RPM
	Max. Consumption	59 l/h
	Max. Manifold pressure	100 kPa
	Max. Continuous power	
	Max. Power	118 kW (160 k)
	Max. Engine speed	2 680 RPM
	Max. Consumption	52 l/h
	Max. Manifold pressure	95 kPa
	Max. Cruising power	
	Max. Power	103 kW (140 k)
	Max. Engine speed	2 580 RPM
	Max. Consumption	44 l/h
	Max. Manifold pressure	88 kPa
6. Load factors:	For category Aerobatic (A)	+6.0 g, -3.0 g
	For category Normal (N)	+3.8 g, -1.5 g

7. Propeller:

7.1 Model:	V 503 A
7.2 Type Certificate:	No. 69-02 issued by CAA CZ
7.3 Number of blades:	2
7.4 Diameter:	2 000 mm
7.5 Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1 Fuel:	LBZ 78 SHELL 80 ESSO 80 (TEO max. 0.06 % objemu) Grade 100/130 (TEO max. 0.06% objemu) AVGAS 100 LL (DEFENCE STANDARD 91/90ASTM D910). AVGAS 100 L AVGAS 80 (See service instruction of Engine manufacturer)
8.2 Oil:	AEROSHELL Oil W 100 AEROSHELL Oil W 120

ELF Aviation AD 100
MOBIL Aero Oil 100

BP Aero D 100
CASTROL Aero AD 100
TOTAL Aero D 100

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: for category A: 95.5 litres
for category N: 165.5 litres

Usable: for category A: 93 litres
for category N: 163 litres

2 x 45 litres in main tanks
5.5 litres in connecting tank
2 x 35 litres in auxiliary wing tip tanks

9.2 Oil: Minimum 9 litres – Maximum 14 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never Exceed Speed Limit	V_{NE}	305 km/h IAS
Normal Operating Speed Limit	V_{NO}	230 km/h IAS
Design Manoeuvring Speed Limit	V_A	230 km/h IAS
Maximum Flaps Extended Speed Limit	V_{FE}	152 km/h IAS
Maximum Landing Gear Operating Speed	V_{LO}	140 km/h IAS

11. Maximum Operating Altitude: 5 200 m

12. Allweather Operations Capability: The aircraft is approved for VFR Day flights.

13. Maximum Weights:

Max. Take-off and Landing weight:
- For category Aerobatic (A) 940 kg
- For category Normal (N) 975 kg

Max. Variable Load:
- For category Aerobatic (A) 192 kg
- For category Normal (N) 167 kg

14. Centre of Gravity Range:	20.4 % ÷ 27.4 % MAC M.A.C. is 1 545 mm; 0 % M.A.C. is 616 mm aft reference datum.		
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up	25° ± 1°
		down	20° ± 1°
	Elevator trim	up	25° ± 2°
		down	40° ± 2°
	Rudder deflection	left an right	28° ± 2°
	Rudder trim	left	5° ± 1°
		right	30° ± 2°
	Ailerons deflection	up	108 mm (+5, -3) mm
		down	98 mm (+5, -3) mm
	Wing flaps position:	retracted	0°
		take-off	15°
		landing	40° (+ 5°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.		
18. Minimum Flight Crew:	1 (Pilot)		
19. Maximum Passenger Seating Capacity:	2 (including crew)		
20. (Reserved)			
21. Baggage/Cargo Compartments:	None		
22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.		

MIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Letová příručka ZLÍN 526 F, date of issue 1971
   - In English language — Pilot´s Handbook for the Z 526 F Aircraft, date of issue 1971
   - In German language — Z 526 F Flughandbuch, date of issue 1969

2. Description – Operation – Maintenance:
   - In Czech language — Popis – obsluha – údržba ZLIN 526 F, date of issue 1972
   - In English language — Description – Operation – Maintenance ZLIN 526 F, date of issue November 1971
   - In German language — Beschreibung – Bedienung – Instalhaltung ZLIN 526 F, date of issue 1972

3. Overhaul Manual:
   - In Czech language — Opravárenská příručka ZLIN 526 F, date of issue 1971
   - In English language — Overhaul Manual ZLIN 526 F, date of issue 1971
   - In German language — Reparatur Handbuch ZLIN 526 F, date of issue 1971

4. Catalogue of spare parts:
   - In Czech, English, German language

     Katalog ZLIN 526 F Trener, date of issue 1971

MV. Notes:

Note 1: Following Z 526 F aircraft have been converted to the model:

Z 526 L	S/N: 1155-1159
Z 526 FI	S/N: 1170-1173
Z 526 AFS-V	S/N: 1119
Z 726	S/N: 1075

by the aircraft Manufacturer.

Note 2: S/N 1026 and S/N 1027 have been converted to Zlin Z 526 ASM (for definition of Z 526 ASM see CAA Cz STC No. STC-001-97).

SECTION N: Z 526 L

NI. General

1.	a) Type:	Z 26
	b) Model:	Z 526 L
2.	Airworthiness category:	Aerobatic (A) Normal (N)
3.	Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4.	Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1160
5.	Certification Application Date:	---
6.	CAA CZ Type Certificate Date:	May 10, 1971

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 71-06.

NII. Certification Basis

1.	Reference Date for determining the applicable requirements:	---
2.	(Reserved)	
3.	(Reserved)	
4.	Airworthiness Requirements:	FAR PART 23 effective March 14, 1969
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemptions:	None
8.	EASA Equivalent Safety Findings:	§ 23.177(a)(2)(3) – Good controllability of the aircraft.

§ 23.613(c), § 23.615 – The sense of requirement is met.

§ 23.781 – The same safety level is ensured.

§ 23.1183(a) – Operation experience.

§ 23.1337(b) – Fuel marking in litres does not affect the degree of safety.

§ 23.1357 – Good operation experience.

9. EASA Environmental Standards: ICAO Annex 16, Volume I, Chapter 10

NIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of Z 526 L aircraft, No. S-L 526.000.

2. Description: The Z 526 L aircraft is two-seat, low wing, single-engine, monoplane.

3. Equipment: Approved equipment list is stated in document Technical description and Maintenance instructions Z 526 L aircraft, Chapter 10.

4. Dimensions:

Wing Span:	10.596 m
Length:	7.650 m
Height:	2.060 m
Wing Area:	15.450 m^2

5. Engine:

5.1 Model: TEXTRON Lycoming AIO – 360 – B1B

5.2 Type Certificate: No. 1E10 issued by FAA

5.3 Limitations:

Max. Take-off power (MT)

Max. Power	149 kW (202 HP)
Max. Engine speed	2 750 RPM
Max. Consumption	61 l/h
Max. Manifold pressure	101 kPa

Max. Continuous Cruising power (75 % MC)

Max. Power	112 kW (152 HP)
Max. Engine speed	2 450 RPM
Max. Consumption	46 l/h
Max. Manifold pressure	82 kPa

	Max. Economic Cruising power (65 % MC)	
	Max. Power	97 kW (132 HP)
	Max. Engine speed	2 350 RPM
	Max. Consumption	36 l/h
	Max. Manifold pressure	78 kP
6. Load factors:	For category Aerobatic (A)	+6.0 g, -3.0 g
	For category Normal (N)	+3.8 g, -1.5 g

7. Propeller:

7.1 Model:	Hartzell HC-C2YK-4
7.2 Type Certificate:	No. P-920 issued by FAA
7.3 Number of blades:	2
7.4 Diameter:	1 880 mm
7.5 Sense of Rotation:	Clockwise in flight direction

8. Fluids:

8.1 Fuel:	Aviation gasoline with min. 100/130 octanes.
8.2 Oil:	For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residue does not exceed the value of 0.4. Recommended kinds of oil: MS20-GOST 1013-49 ELF AD-100 AERO-SHELL W100 AERO-SHELL W120
8.3 Coolant:	None

9. Fluid capacities:

9.1 Fuel:	Total: for category A: 95.5 litres for category N: 165.5 litres Usable: for category A: 93 litres for category N: 163 litres 2 x 45 litres in main tanks 5.5 litres in connecting tank 2 x 35 litres in auxiliary wing tip tanks
9.2 Oil:	Minimum 9 litres – Maximum 11 litres

9.3 Coolant system capacity:	None		
10. Air Speeds:	Never Exceed Speed Limit V_{NE}		
	category A		305 km/h IAS
	category N		269 km/h IAS
	Normal Operating Speed Limit V_{NO}		
	category A, N		230 km/h IAS
	Design Manoeuvring Speed Limit V_A		
	category A		230 km/h IAS
	category N		189 km/h IAS
	Maximum Flaps Extended Speed Limit V_{FE}		
	category A, N		148 km/h IAS
11. Maximum Operating Altitude:	6 500 m		
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.		
13. Maximum Weights:	Max. Take-off and Landing weight:		
	- For category Aerobatic (A)		940 kg
	- For category Normal (N)		975 kg
	Max. Variable Load:		
	- For category Aerobatic (A)		170 kg
	- For category Normal (N)		144 kg
14. Centre of Gravity Range:	20.6 % ÷ 26.5 % MAC M.A.C. is 1 545 mm; 0 % M.A.C. is 616 mm aft reference datum.		
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up	25° ± 1°
		down	20° ± 1°
	Elevator trim tab	up	25° ± 2°
		down	40° ± 2°
	Rudder deflection	right and left	28° ± 2°
	Rudder trim tab	left	5° ± 1°
		right	30° ± 2°
	Ailerons deflection	up	108 mm (+ 5/-3) mm
		down	98 mm (+ 5/-3) mm

	Wing flaps positions retracted 0° take-off 15° landing 40° (+ 5°, -3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.
18. Minimum Flight Crew:	1 (Pilot)
19. Maximum Passenger Seating Capacity:	2 (including crew)
20. (Reserved)	
21. Baggage/Cargo Compartments:	Maximum 25 kg
22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

NIV. Operating and Service Instructions

1. Flight manual:
   – In English language — Flight Manual Z 526 L

2. Description – Operation – Maintenance:
   – In English language — Technical Description and Maintenance Instruction, ZLIN 526 L Aircraft, date of issue 1970

3. Overhaul Manual:
   – In English language — Overhaul Manual Z 526 L, date of issue 1972

4. Catalogue of spare parts:
   – In Czech language — Katalog ZLIN 526 L (Dodatek ke katalogu Z 526 F), date of issue 1972
   – English language — Spare Parts Catalogue Z 526 L (Supplement of the Z 526 F Aircraft Catalogue), date of issue 1972
   – German language — Ersatzeteilkatalog Z 526 L (Nachtrag zum Katalog des Flugzeugs Z 526 F), date of issue 1972

NV. Notes:

None

SECTION O: Z 526 AFS

OI. General

1. a) Type:	Z 26
b) Model:	Z 526 AFS
2. Airworthiness category:	Aerobatic (A) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1126, 1201-1230, 1301-1310, 1326-1330
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	June 30,1972

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 72-04.

OII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. (Reserved)	
4. Airworthiness Requiremenst:	FAR PART 23, Amdt. 23-9 included
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None
8. EASA Equivalent Safety	§ 23.177 – Requirements are met with except for flight characteristics at sideslips when aileron and

Findings:	rudder control forces are inexpressive and, in some cases, the tendency to raise the low wing is not demonstrated according to regulation requirement. It is admitted with regard to very good aircraft controllability, to the fact that uncontrollable tendencies do not occur and to the fact that the aircraft is aerobatic for which higher manoeuverability is required. § 23.207 – Stall warning is inexpressive. It is admitted with regard to good flight characteristics at stall, to very good aircraft controllability and to the fact that dangerous tendencies do not occur. § 23.613(c), § 23.615 – Materials and design values used for aircraft design and construction comply with the Czechoslovak State Standard and specifications valid for the Czechoslovak aviation industry. It is admitted with regard to the fact that the requirement sense is met. § 23.729(f)(1) – A warning device is not used. It is admitted with regard to the fact that the aircraft is intended for aerobatic flying only. § 23.991(b) – The aircraft is not equipped with emergency pump for fuel supply recovery in case of main fuel pump failure. It is admitted with regard to these reasons: - The engine is equipped with high-pressure pump, which is joined with low-pressure pump to a single aggregate. A failure of this aggregate could cause contemporaneous break of fuel supply by both supply and injection pumps. In such case no emergency pump could ensure sufficient fuel supply to finish the flight without abnormal pilot's skills or effort. A failure of low-pressure pump has not been occurred yet and its occurrence is extremely improbable. § 23.1183(a) – Requirement for hoses fire resistance is not met. § 23.1357(d) – Requirement for battery circuit breaker during flight is not met. It is admitted with regard to operation experiences.
9. EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

OIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of Z 526 AFS aircraft, No. S-AFS 526.000

2. Description: The Z 526 AFS Aircraft is one-seat, low wing, single-engine, cantilever monoplane.

3. Equipment: Approved equipment list is stated in document Description, Operation, Maintenance ZLIN 526 AFS, Section 10.

4. Dimensions:
   Wing Span: 8.840 m
   Length: 7.806 m
   Height: 1.900 m
   Wing Area: 13.810 m^2

5. Engine:

   5.1 Model: M 137 A

   5.2 Type Certificate: No. 96-02 issued by CAA CZ

   5.3 Limitations:

Max. Take-off power	
Max. Power	132 kW (180 HP)
Max. Engine speed	2 750 RPM
Max. Consumption	59 l/h
Max. Manifold pressure	100 kPa
Max. Continuous power	
Max. Power	118 kW (160 HP)
Max. Engine speed	2 680 RPM
Max. Consumption	52 l/h
Max. Manifold pressure	95 kPa
Max. Cruising power	
Max. Power	103 kW (140 HP)
Max. Engine speed	2 580 RPM
Max. Consumption	44 l/h
Max. Manifold pressure	88 kPa

6. Load factors:

For category Aerobatic (A)	+ 7.0 g,	- 4.5 g
For category Normal (N)	+ 3.8 g,	- 1.5 g

7. Propeller:

   7.1 Model: V 503 A

   7.2 Type Certificate: No. 69-02 issued by CAA CZ

   7.3 Number of blades: 2

   7.4 Diameter: 2 000 mm

7.5 Sense of Rotation: Anticlockwise in flight direction

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline with minimum 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06% vol.
BL 78
BP 100L
AVGAS 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with minimal kinematic viscosity of 20 $mm^2 s^{-1}$ at 100°C, which percentual carbon residue does not exceed the value of 0.29 %.
MS 20 – Running in
AEROSHELL Oil 100 – Running in
Aeroshell W100
Aeroshell W120 (in tropical climates)
ELF Aviation AD 100
BP Aero D 100
TOTAL Aero D 100

8.3 Coolant: None

9. Fluids capacities:

9.1 Fuel: Total: for category A: 75.5 litres
for category N: 145.5 litres

Usable: for category A: 73 litres
for category N: 143 litres

2 x 35 litres in main tanks
5.5 litres in connecting tank
2 x 35 litres in auxiliary wing tip tanks

9.2 Oil: Minimum 7 litres – Maximum 14 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never Exceed Speed Limit	V_{NE}	305 km/h IAS
Normal Operating Speed Limit	V_{NO}	230 km/h IAS
Design Manoeuvring Speed Limit	V_A	238 km/h IAS
Maximum Open Landing Gear Speed	V_{LE}	180 km/h IAS

	Maximum Landing Gear Operating Speed V_{LO}		140 km/h IAS
	Maximum Permissible Snap Maneuver speed		160 km/h IAS
11. Maximum Operating Altitude:	5 800 m		
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.		
13. Maximum Weights:	Max. Take-off and landing weight:		
	- For category Aerobatic (A)		740 kg
	- For category Normal (N)		840 kg
	Max. Variable Load:		
	- For category Aerobatic (A)		70 kg
	- For category Normal (N)		83 kg
14. Centre of Gravity Range:	24.8 % – 31 % MAC M.A.C. is 1 609 mm; 0 % M.A.C. is 380 mm aft reference datum.		
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up	28° ± 2°
		down	24° ± 1°
	Elevator trim tab	up	25° ± 2°
		down	40° ± 2°
	Rudder deflection	right and left	30° ± 2°
	Rudder trim tab	left	5° ± 1°
		right	30° ± 2°
	Outside Aileron deflection		
		up	112 mm; (+ 5; - 3) mm
		down	108 mm; (+ 5; - 3) mm
	Inside aileron deflection		
		up	84 mm; (+ 5; - 3) mm
		down	81 mm; (+ 5; - 3) mm
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.		
18. Minimum Flight Crew:	1 (Pilot)		
19. Maximum Passenger Seating Capacity:	1 (including crew)		

20. (Reserved)

21. Baggage/Cargo Compartments: None

22. Wheels and Tyres: The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150;

The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

OIV. Operating and Service Instructions

1. Flight Manual:
   - In Czech language — Letová příručka ZLIN 526 AFS, date of issue 1971
   - In English language — Flight Manual ZLIN 526 AFS
   - In German language — Flugzeug – Betriebshandbuch ZLIN 526 AFS

2. Maintenance Manual:
   - In Czech language — Popis – obsluha – údržba ZLIN 526 AFS, date of issue 1972
   - In English language — Description – Operation – Maintenance ZLIN 526 AFS, date of issue 1972
   - In Germany language — Beschreibung – Bedienung – Instandhaltung ZLIN 526 AFS, date of issue 1972

3. Overhaul Manual:
   - In Czech language — Opravárenská příručka ZLIN 526 AFS (Dodatek k Opravárenské příručce Z 526 F), date of issue 1972
   - In English language — Overhaul Manual ZLIN 526 AFS (Supplement to Overhaul Manual of the Z 526 F Aircraft), date of issue 1972
   - In German language — Reparaturhandbuch ZLIN 526 AFS (Nachtrag zum Reparaturhanbuch Z 526 F), date of issue 1972

4. Illustrated Parts Catalogue:
   - In Czech, German and English language, date of issue 1972
     Katalog Z 526 AFS (Dodatek ke katalogu Z 526 F)
     Ersatzteil Katalog (Nachtrag zum Katalog für das Fugzeug Z 526 F)
     Spare Parts Catalogue (Catalogue Supplement of the Z 526 F Aircraft)

5. Catalogue Supplement:
   - In Czech, German and English language, issued 1973
     Dodatek ke katalogu pro Z 526 AFS

Nachtrag zum Katalog für das Fugzeug Z 526 AFS
Supplement of the Z 526 AFS Catalogue

OV. Notes:

Following Z 526 AFS aircraft have been converted to the model:
Z 526 AFS-V S/N: 1213, 1218-1220, 1224, 1307
by the aircraft manufacturer.

SECTION P: Z 526 AFS-V

PI. General

1.	a) Type	Z 26
	b) Model:	Z 526 AFS-V
2.	Airworthiness category:	Normal (N)
3.	Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4.	Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1101
5.	Certification Application Date:	---
6.	CAA CZ Type Certificate Date:	September 24, 1982

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 72-04, Supplement No. 2.

PII. Certification Basis

1.	Reference Date for determining the applicable requirements:	---
2.	(Reserved)	
3.	(Reserved)	
4.	Airworthiness Requirements:	FAR PART 23, Amdt. 23-9 included
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemptions:	None
8.	EASA Equivalent Safety Findings:	§ 23.177 – Requirements are met with except for flight characteristics at sideslips when aileron and rudder control forces are inexpressive and, in

some cases, the tendency to raise the low wing is not demonstrated according to regulation requirement. It is admitted with regard to a very good aircraft controllability, to the fact that uncontrollable tendencies do not occur and to the fact that the aircraft is aerobatic, for which higher manoeuverability is required.

§ 23.207 – Stall warning is inexpressive. It is admitted with regard to good flight characteristics at stall, to very good aircraft controllability and to the fact that dangerous tendencies do not occur.

§ 23.613(c), § 23.615 – Materials and design values used for aircraft design and construction comply with the Czechoslovak State Standard and specifications valid for the Czechoslovak aviation industry. It is admitted with regard to the fact that the requirement sense is met.

§ 23.991(b) – The aircraft is not equipped with emergency pump for fuel supply recovery in case of main fuel pump failure. It is admitted with regard to these reasons:

- The engine is equipped with high-pressure pump, which is joined with low-pressure pump to a single aggregate. A failure of this aggregate could cause contemporaneous break of fuel supply by both supply and injection pumps. In such case, no emergency pump could ensure sufficient fuel supply to finish the flight without abnormal pilot's skills or effort. A failure of low-pressure pump has not been occurred yet and its occurrence is extremely improbable.

§ 23.1183(a) – Requirement for hoses fire resistance is not met.

§ 23.1357(d) – Requirement for battery circuit breaker during flight is not met. It is admitted with regard to operation experiences.

9. EASA Environmental Standards:

ICAO Annex 16, Volume I, Chapter 10

PIII. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of Z 526 AFS-V aircraft, No. S-AFS 526.000, AFN 526	
2. Description:	The Z 526 AFS-V aircraft is one-seat, low wing, single-engine, cantilever monoplane.	
3. Equipment:	Approved equipment list is stated in document Description, Operation, Maintenance ZLIN 526 AFS, Section 10.	
4. Dimensions:	Wing Span:	8.840 m
	Length:	7.806 m
	Height:	1.900 m
	Wing Area:	13.810 m^2
5. Engine:		
5.1 Model:	M 137 A	
5.2 Type Certificate:	No. 96-02 issued by CAA CZ	
5.3 Limitations:	Max. Take-off power	
	Max. Power	132 kW (180 HP)
	Max. Engine speed	2 750 RPM
	Max. Consumption	59 l/h
	Max. Manifold pressure	100 kPa
	Max. Continuous power	
	Max. Power	118 kW (160 HP)
	Max. Engine speed	2 680 RPM
	Max. Consumption	52 l/h
	Max. Manifold pressure	95 kPa
	Max. Cruising power	
	Max. Power	103 kW (140 HP)
	Max. Engine speed	2 580 RPM
	Max. Consumption	43 l/h
	Max. Manifold pressure	87 kPa
6. Load factors:		+3.8 g, -1.5 g
7. Propeller:		
7.1 Model:	V 503 A	
7.2 Type Certificate:	No. 69-02 issued by CAA CZ	
7.3 Number of blades:	2	
7.4 Diameter:	2 000 mm	
7.5 Sense of Rotation:	Anticlockwise in flight direction	

8. Fluids:

8.1 Fuel:	Non-ethylated aviation gasoline, with minimum 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06% vol. BL 78 BP 100L AVGAS 80 AVGAS 100 LL (DEFENCE STANDARD 91/90, ASTM D910)
8.2 Oil:	For engine operation are recommended mineral oils with minimal kinematic viscosity of 20 $mm^2 s^{-1}$ at 100°C, which percentual carbon residue does not exceed the value of 0.29 %. MS 20 – Running in AEROSHELL Oil 100 – Running in Aeroshell W100 Aeroshell W120 (in tropical climates) ELF Aviation AD 100 BP Aero D 100 TOTAL Aero D 100
8.3 Coolant:	None

9. Fluid capacities:

9.1 Fuel:	Total: 145.5 litres Usable: 143 litres 2 x 35 litres in main tanks 5.5 litres in connecting tank 2 x 35 litres in auxiliary wing tip tanks
9.2 Oil:	Minimum 7 litres – Maximum 14 litres
9.3 Coolant system capacity:	None

10. Air Speeds:

Never Exceed Speed Limit	V_{NE}	305 km/h IAS
Normal Operating Speed Limit	V_{NO}	230 km/h IAS
Design Manoeuvring Speed Limit	V_A	238 km/h IAS
Maximum Open Landing Gear Speed	V_{LE}	180 km/h IAS
Maximum Landing Gear Operating Speed	V_{LO}	140 km/h IAS

11. Maximum Operating 5 800 m

Altitude:

12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.			
13. Maximum Weights:	Max. Take-off and Landing weight:		840 kg	
	Max. Variable Load:		83 kg	
14. Centre of Gravity Range:	24.8 % – 31 % MAC M.A.C. is 1 609 mm; 0 % M.A.C. is 380 mm aft reference datum.			
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.			
16. Control surface deflections:	Elevator deflection	up	28°	± 2°
		down	24°	± 1°
	Elevator trim tab	up	25°	± 2°
		down	35°	± 2°
	Rudder deflection	right and left	30°	± 2°
	Rudder trim tab	left	5°	± 1°
		right	30°	± 2°
	Outside aileron deflection			
		up	112 mm; (+ 5; - 3) mm	
		down	108 mm; (+ 5; - 3) mm	
	Inside aileron deflection			
		up	84 mm; (+ 5; - 3) mm	
		down	81 mm; (+ 5; - 3) mm	
	Wing flaps position:	retracted	0°	
		take-off	15°	
		landing	40°	(+ 5°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.			
18. Minimum Flight Crew:	1 (Pilot)			
19. Maximum Passenger Seating Capacity:	1 (including crew)			
20. (Reserved)				
21. Baggage/Cargo Compartments:	None			
22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150;			

The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

PIV. Operating and Service Instructions

1. Flight Manual:
   - In Czech language — Letová příručka ZLIN 526 AFS, date of issue 1971
     - Dodatek k Letové příručce Z 526 AFS-V, date of issue 1982
   - In English language — Flight Manual ZLIN 526 AFS
   - In German language — Flugzeug – Betriebshandbuch ZLIN 526 AFS

2. Maintenance Manual:
   - In Czech language — Popis – obsluha – údržba ZLIN 526 AFS, date of issue 1972
   - In English language — Description – Operation – Maintenance ZLIN 526 AFS, date of issue 1972
   - In Germany language — Beschreibung – Bedienung – Instandhaltung ZLIN 526 AFS, date of issue 1972

3. Overhaul Manual:
   - In Czech language — Opravárenská příručka ZLIN 526 AFS (Dodatek k Opravárenské příručce Z 526 F), date of issue 1972
   - In English language — Overhaul Manual ZLIN 526 AFS (Supplement to Overhaul Manual of the Z 526 F Aircraft), date of issue 1972
   - In German language — Reparaturhandbuch ZLIN 526 AFS (Nachtrag zum Reparaturhanbuch Z 526 F), date of issue 1972

4. Ilustrated Parts Catalogue:
   - In Czech, German and English language, date of issue 1972
     Katalog Z 526 AFS (Dodatek ke katalogu Z 526 F)
     Ersatzteil Katalog (Nachtrag zum Katalog für das Flugzeug Z 526 F)
     Spare Parts Catalogue (Catalogue Supplement of the Z 526 F Aircraft)

5. Catalogue Supplement:
   - In Czech, German and English language, issued 1973
     Dodatek ke katalogu pro Z 526 AFS
     Nachtrag zum Katalog für das Flugzeug Z 526 AFS
     Supplement of the Z 526 AFS Catalogue

PV. Notes

Z 526 AFS-V

None

SECTION R: Z 526 M

RI. General

1.	a) Type:	Z 26
	b) Model:	Z 526 M
2.	Airworthiness category:	Utility (U) Normal (N)
3.	Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4.	Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: see Z 526 aircraft
5.	Certification Application Date:	---
6.	CAA CZ Type Certificate Date:	February 5, 1976

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 2417/59, Supplement No. 5

RII. Certification Basis

1.	Reference Date for determining the applicable requirements:	---
2.	(Reserved)	
3.	(Reserved)	
4.	Airworthiness Requirements:	Bauvorschriften für Flugzeuge, Kat. K; British Civil Airworthiness Requirements, Cat. D
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemptions:	None

8. EASA Equivalent Safety Findings:	None
9. EASA Environmental Standards:	ICAO Annex 16, Volume I, Chapter 10

RIII. Technical Characteristics and Operational Limitations

1. Type Design Definition:	The specification list of Z 526 M aircraft, No. S-Z526.000.
2. Description:	The Z 526 M aircraft is two-seat, low wing, single-engine, monoplane.
3. Equipment:	Approved equipment list is stated in document Technical Description, Operation Instructions,Chapter 9.
4. Dimensions:	Wing Span: 10.845 m with auxiliary tanks 10.596 m without auxiliary tanks Length: 7.820 m Height: 2.060 m Wing Area: 15.450 m^2
5. Engine:	
5.1 Model:	M 137 A
5.2 Type Certificate:	No. 96-02 issue by CAA CZ
5.3 Limitations:	Max. Take-off power Max. Power 132 kW (180 HP) Max. Engine speed 2 750 RPM Max. Consumption 59 l/h Max. Manifold pressure 100 kPa Max. Continuous power Max. Power 118 kW (160 HP) Max. Engine speed 2 680 RPM Max. Consumption 52 l/h Max. Manifold pressure 95 kPa Max. Cruising power Max. Power 103 kW (140 HP) Max. Engine speed 2 580 RPM Max. Consumption 43 l/h Max. Manifold pressure 87 kPa
6. Load factors:	For category Utility (U) +5.5 g, -3 g For category Normal (N) +3.5 g, -1 g
7. Propellers:	

7.1.1	Model:	V 503
7.1.2	Type Certificate:	No. 64 002 issued by CAA CZ
7.1.3	Number of blades:	2
7.1.4	Diameter:	1 950 mm
7.1.5	Sense of Rotation:	Anticlockwise in flight direction

or

7.2.1	Model:	V 503 A
7.2.2	Type Certificate:	No. 96-02 issued by CAA CZ
7.2.3	Number of blades:	2
7.2.4	Diameter:	2 000 mm
7.2.5	Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1	Fuel:	Non-ethylated aviation gasoline with min. 72÷87 octanes. Application of ethylated fuels is only permitted in case, the T.E.L. content does not exceed the value of 0.06 % vol. Recommended kinds of fuel: LBZ 72 LBZ 83
8.2	Oil:	For engine operation are recommended mineral oils with min. kinematic viscosity of 20 cSt at 100°C, whose percentual carbon residues does not exceed the value of 0.4. Recommended kinds of oil: MS 20 AERO-SHELL W 100 AERO-SHELL W 120 in tropical climates.
8.3	Coolant:	None

9. Fluid capacities:

9.1	Fuel:	Total: for category U: 100 litres for category N: 170 litres Usable: for category U: 97.5 litres for category N: 167.5 litres 2 x 45 litres in main tanks 3 litres in connecting tank

	7 litres in gravity tank 2 x 35 litres in auxiliary wing tip tanks
9.2 Oil:	Minimum 7 litres – Maximum 11 litres
9.3 Coolant system capacity:	None
10. Air Speeds:	Never Exceed Speed Limit V_{NE} 292 km/h IAS Normal Operating Speed Limit V_{NO} 233 km/h IAS Design Manoeuvring Speed Limit V_A 225 km/h IAS Maximum Flaps Extended Speed Limit V_{FE} 140 km/h IAS Maximum Open Landing Gear Speed V_{LE} 292 km/h IAS Maximum Landing Gear Operating V_{LO} 140 km/h IAS Maximum permissible Snap Maneuver Speed Limit for cat. A 160 km/h IAS
11. Maximum Operating Altitude:	5 000 m
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.
13. Maximum Weights:	Max. Take-off and Landing weight: - For category Utility (U) 940 kg - For category Normal (N) 975 kg Max. Variable Load: - For category Utility (U) 195 kg - For category Normal (N) 163 kg
14. Centre of Gravity Range:	17 % ÷ 27.4 % MAC M.A.C. is 1 545 mm; 0 % M.A.C. is 616 mm aft reference datum.
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.
16. Control surface deflections:	Elevator deflection up 25° ± 1° down 20° ± 1° Elevator trim tab up 25° ± 2°

	down	40° ± 2°
Rudder deflection	right and left	28° ± 2°
Rudder trim tab	left	5° ± 1°
	right	30° ± 2°
Ailerons deflection	up	108 mm (+5, -3) mm
	down	98 mm (+5, -3) mm
Wing flaps position:	retracted	0°
	take-off	15°
	landing	40° (+ 5°, - 3°)

17. Levelling Means: Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.

18. Minimum Flight Crew: 1 (Pilot)

19. Maximum Passenger Seating Capacity: 2 (including crew)

20. (Reserved)

21. Baggage/Cargo Compartments: Maximum 17 kg

22. Wheels and Tyres: The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150;

The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

RIV. Operating and Service Instructions

1. Flight manual:
   - In Czech language — Letová příručka školního a akrobatického letounu Z 526 – Z 526 A
     - Dodatek č. 1 k Letové příručce Z 526 M, date of issue 1978
   - In English language — Instruction for Pilot on the Use and Handling of the Training and Acrobatic Z 526 and Z 526 A Aircraft, date of issue 1966
   - In German language — Handbuch für den Flugzeugführer zum Schulungs und kunstflugzeuges Z 526 – Z 526 A, date of issue 1966
     - Nachtrag Nr. 1 zum Flughandbuch Z 526 M, date of issue 1978

2. Descriprion – Operation – Maintenance:
   - In Czech language — Technický popis a návod k obsluze letounu Z 526 – Z 526 A
   - In English language — Technical Description, Operation Instruction for Z 526 – Z 526 A Aircrafts, date of issue 1966
   - In German language — Technische Beschreibung und Bedienungsanleitung zum Flugzeug Z 526 – Z 526 A, date of issue 1966

3. Overhaul Manual:
   - In English language — Major Overhaul of Z 526 – Z 526 A Aircraft, date of issue 1969

4. Catalogue of spare parts:
   - In Czech, English, German language — Trener Master Z 526 Katalog, date of issue 1967

RV. Notes:

None

SECTION S: Z 726

SI. General

1. a) Type:	Z 26
b) Model:	Z 726
2. Airworthiness category:	Utility (U) Normal (N)
3. Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4. Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1331, 1333-1360
5. Certification Application Date:	---
6. CAA CZ Type Certificate Date:	June 10, 1974

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 74-01.

SII. Certification Basis

1. Reference Date for determining the applicable requirements:	---
2. (Reserved)	
3. Reserved)	
4. Airworthiness Requirements:	FAR PART 23, Amdt. 23-13 included
5. Requirements elected to comply:	None
6. EASA Special Conditions:	None
7. EASA Exemptions:	None

8. EASA Equivalent Safety Findings:

§ 23.177(a)(2), (3) – Some requirements for directional and lateral stability and aileron control stick force curve in straight, steady slips are not fully met. It is admitted with regard to very good aircraft controllability and to the fact that dangerous tendencies do not occur and abnormal pilot's skills or effort are not needed.

§ 23.613(c), § 23.615 – Materials and design values used for aircraft design and construction comply with the Czechoslovak State Standard and specifications valid for the Czechoslovak aviation industry. It is admitted with regard to the fact that the requirement sense is met.

§ 23.955(c) – Requirement for flow rate of fuel supplied by fuel pump to the engine is not met. It is admitted with regard to the fact that fuel flow is higher than engine consumption at maximum power.

§ 23.991(b) – The aircraft is not equipped with emergency pump for fuel supply recovery in case of main fuel pump failure. It is admitted with regard to these reasons:

- The engine is equipped with high-pressure pump, which is joined with low-pressure pump to a single aggregate. A failure of this aggregate could cause contemporaneous break of fuel supply by both supply and injection pumps. In such case, no emergency pump could ensure sufficient fuel supply to finish the flight without abnormal pilot's skills or effort. A failure of low-pressure pump has not been occurred yet and its occurrence is extremely improbable.

§ 23.993(d), § 23.1183(a) – Requirement for hoses fire resistance is not met. It is admitted with regard to experiences from operation of the aircraft of former type.

§ 23.1013(e), § 23.1019 – Oil tank outlet is provided with the screen, which does not restrict oil flow. A surface of the screen is multiple bigger than cross section of the outlet fitting, thus safety level is kept.

§ 23.1145(c) – Requirement for protection of magnetos changeover switch against dangerous change over is not met. It is admitted with regard to the shape and location of the changeover switch.

§ 23.1381 to 23.1401 – The aircraft is not equipped

with lighting for night operation.

§ 23.1545(a), (b) – Requirement for a scale for CAS speed is not met. Airspeed indicator scale and its colour markings are done in IAS.

9. EASA Environmental Standards: ICAO Annex 16, Volume I, Chapter 10

SIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of Z 726 aircraft, No. S-Z 726.000.

2. Description: The Z 726 aircraft is two-seat, low wing, single-engine, cantilever monoplane.

3. Equipment: Master equipment list is stated in document Flight Manual Z 726, section 6.

4. Dimensions:
Wing Span: 9.875 m
Length: 7.975 m
Height: 2.060 m
Wing Area: 14.890 m^2

5. Engine:

5.1 Model: M 137 AZ

5.2 Type Certificate: No. 96-02 issued by CAA CZ

5.3 Limitations:

Max. Take-off power	
Max. Power	132 kW (180 HP)
Max. Engine speed	2 750 1/min
Max. Consumption	61 l/hod
Max. Manifold pressure	100 kPa
Max. Continuous power	
Max. Power	118 kW (160 HP)
Max. Engine speed	2 680 1/min
Max. Consumption	52 l/hod
Max. Manifold pressure	95 kPa
Max. Cruising power	
Max. Power	103 kW (140 HP)
Max. Engine speed	2 580 1/min
Max. Consumption	43 l/hod
Max. Manifold pressure	88 kPa

6. Load factors:
For category Utility (U) +4.4 g, -2.2 g
For category Normal (N) +3.8 g, -1.5 g

7. Propeller:

7.1 Model: V 503 A

7.2 Type Certificate: No. 69-02 issued by CAA CZ

7.3 Number of blades: 2

7.4 Diameter: 2 000 mm

7.5 Sense of Rotation: Anticlockwise in flight direction

8. Fluids:

8.1 Fuel: Non-ethylated aviation gasoline, with minimum 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06% vol.
BL 78
BP 100L
AVGAS 80
AVGAS 100 LL
(DEFENCE STANDARD 91/90, ASTM D910)

8.2 Oil: For engine operation are recommended mineral oils with minimal kinematic viscosity of 20 $mm^2 s^{-1}$ at 100°C, which percentual carbon residue does not exceed the value of 0.29 %.
MS 20 – Running in
AEROSHELL Oil 100 – Running in
Aeroshell W100
Aeroshell W120 (in tropical climates)
ELF Aviation AD 100
BP Aero D 100
TOTAL Aero D 100

8.3 Coolant: None

9. Fluid capacities:

9.1 Fuel: Total: for category U: 95.5 litres
for category N: 165.5 litres

Usable: for category U: 93 litres
for category N: 163 litres

2 x 45 litres in main fuel tanks
5.5 litres in connecting tank
2 x 35 litres in auxilitary wing tip tanks

9.2 Oil: Minimum 9 litres – Maximum 14 litres

9.3 Coolant system capacity: None

10. Air Speeds:

Never Exceed Speed Limit	V_{NE} 300 km/h IAS
Normal Operating Speed Limit	V_{NO} 220 km/h IAS
Design Manoeuvring Speed Limit	V_A 203 km/h IAS

	Maximum Flaps Extended Speed Limit	V_{FE} 152 km/h IAS
	Maximum Open Landing Gear Speed	V_{LE} 300 km/h IAS
	Maximum Landing Gear Operating Speed	V_{LO} 140 km/h IAS
	Maximum Permissible Snap Maneuver Speed	160 km/h IAS
11. Maximum Operating Altitude:	4 500 m	
12. Allweather Operations Capability:	The aircraft approved for VFR Day flights.	
13. Maximum Weights:	Max. Take-off weight: - For category Aerobatic (U) - For category Normal (N)	 940 kg 1 000 kg
	Max. Landing weight: - For category Aerobatic (U) - For category Normal (N)	 940 kg 950 kg
	Max. Variable Load: - For category Aerobatic (U) - For category Normal (N)	 250 kg 300 kg
14. Centre of Gravity Range:	17.5 % – 28.5 % MAC M.A.C. is 1 568 mm; 0 % is 602 mm aft reference datum	
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.	
16. Control surface deflections:	Elevator deflection	up 28° ± 1° down 24° ± 1°
	Elevator trim tab	up 25° ± 2° down 40° ± 2°
	Rudder deflection	right and left 28° ± 2°
	Rudder trim tab	left 5° +3° right 30° ± 2°
	Ailerons deflection	up 108 mm (+ 5; - 3) mm down 98 mm (+ 5; - 3) mm
	Wing flaps positions	retracted 0° take-off 15° ± 2° landing 40° (+ 5°, - 3°)
17. Levelling Means:	Levelling points on left and right side of airplane	

	fuselage to be levelled. Measurement plane to be min. 850 mm below.
18. Minimum Flight Crew:	1 (Pilot)
19. Maximum Passenger Seating Capacity:	2 (including crew)
20. (Reserved)	
21. Baggage/Cargo Compartments:	None
22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

SIV. Operating and Service Instructions

1. Flight manual:

– In Czech language	Letová příručka Z 726, date of issue 1974
– In English language	Flight Manual Z 726 ZLIN UNIVERSAL, date of issue 1977

2. Maintenance Manual:

– In Czech language Doc. No.: Do – Z 726 – 2011	Technický popis a návod k obsluze Z 726, date of issue 1974
– In English language Doc. No.: Do – Z 726 – 2011	Technical Manual Z 726 ZLIN UNIVERSAL, date of issue 1974

3. Overhaul Mnual:

– In Czech language	Opravárenská příručka Z 726 ZLIN UNIVERSAL, issued 1978

4. Ilustrated parts catalogue:
   – In Czech, German and English language, issued 1975
   Katalog Z 726 ZLIN UNIVERSAL

SV. Notes

None

SECTION T: Z 726 K

TI. General

1.	a) Type:	Z 26
	b) Model:	Z 726 K
2.	Airworthiness category:	Utility (U) Normal (N)
3.	Type Certificate Holder:	ZLIN AIRCRAFT A.S. Letiště 1578 765 81 Otrokovice Czech Republic
4.	Manufacturer:	Moravan, n. p. Otrokovice CZECHOSLOVAKIA S/N: 1332
5.	Certification Application Date:	---
6.	CAA CZ Type Certificate Date:	June 17, 1974

7. The EASA Type Certificate replaces the CAA CZ Type Certificate No. 74-01.

TII. Certification Basis

1.	Reference Date for determining the applicable requirements:	---
2.	(Reserved)	
3.	(Reserved)	
4.	Airworthiness Requirements:	FAR PART 23, Amdt. 23-13 included
5.	Requirements elected to comply:	None
6.	EASA Special Conditions:	None
7.	EASA Exemtions:	None

8. EASA Equivalent Safety Findings:

§ 23.177(a)(2), (3) – Some requirements for directional and lateral stability and aileron control stick force curve in straight, steady slips are not fully met. It is admitted with regard to very good aircraft controllability and to the fact that dangerous tendencies do not occur and abnormal pilot's skills or effort are not needed.

§ 23.613(c), § 23.615 – Materials and design values used for aircraft design and construction comply with the Czechoslovak State Standard and specifications valid for the Czechoslovak aviation industry. It is admitted with regard to the fact that the requirement sense is met.

§ 23.955(c) – Requirement for flow rate of fuel supplied by fuel pump to the engine is not met. It is admitted with regard to the fact that fuel flow is higher than engine consumption at maximum power.

§ 23.991(b) – The aircraft is not equipped with emergency pump for fuel supply recovery in case of main fuel pump failure. It is admitted with regard to these reasons:

The engine is equipped with high-pressure pump, which is joined with low-pressure pump to a single aggregate. A failure of this aggregate could cause contemporaneous break of fuel supply by both supply and injection pumps. In such case, no emergency pump could ensure sufficient fuel supply to finish the flight without abnormal pilot's skills or effort. A failure of low-pressure pump has not been occurred yet and its occurrence is extremely improbable.

§ 23.993(d), § 23.1183(a) – Requirement for hoses fire resistance is not met. It is admitted with regard to experiences from operation of the aircraft of former type.

§ 23.1013(e), § 23.1019 – Oil tank outlet is provided with the screen, which does not restrict oil flow. A surface of the screen is multiple bigger than cross section of the outlet fitting, thus safety level is kept.

§ 23.1145(c) – Requirement for protection of magnetos changeover switch against dangerous change over is not met. It is admitted with regard to the shape and location of the changeover switch.

§ 23.1381 to 23.1401 – The aircraft is not equipped with lighting for night operation.

§ 23.1545(a), (b) – Requirement for a scale for CAS speed is not met. Airspeed indicator scale and its colour markings are done in IAS.

9. EASA Environmental Standards: ICAO Annex 16, Volume I, Chapter 10

TIII. Technical Characteristics and Operational Limitations

1. Type Design Definition: The specification list of Z 726 aircraft, No. S-K 726.000

2. Description: The Z 726 K aircraft is two-seat, low wing, single-engine, cantilever monoplane.

3. Equipment: Master equipment list is stated in document Flight Manual Z 726 K, section 6.

4. Dimensions:
Wing Span: 9.875 m
Length: 7.975 m
Height: 2.060 m
Wing Area: 14.890 m^2

5. Engine:

5.1. Model: M 337 AK

5.2. Type Certificate: No. 72-08 issue by CAA CZ

5.3. Limitations:

Max. Take-off power	
Max. Power	154 kW (210 HP)
Max. Engine speed	2 750 1/min
Max. Consumption	56 l/hod
Max. Manifold pressure	118 kPa
Max. Continuous power	
Max. Power	125 kW (170 HP)
Max. Engine speed	2 600 1/min
Max. Consumption	52 l/hod
Max. Manifold pressure	98 kPa
Max. Cruising power	
Max. Power	103 kW (140 HP)
Max. Engine speed	2 400 1/min
Max. Consumption	42 l/hod
Max. Manifold pressure	90 kPa

6. Load factors:

For category Utility (U)	+4.4 g, -2.2 g
For category Normal (N)	+3.8 g, -1.5 g

7. Propeller:

7.1 Model:	V 500 A
7.2 Type Certificate:	No 73-03 issued by CAA CZ
7.3 Number of blades:	2
7.4 Diameter:	2 000 mm
7.5 Sense of Rotation:	Anticlockwise in flight direction

8. Fluids:

8.1 Fuel:	Non-ethylated aviation gasoline, with minimum 72 octanes. Application of ethylated fuels is only permitted in case the T.E.L. content does not exceed the value of 0.06% vol. BL 78 BP 100L AVGAS 80 AVGAS 100 LL (DEFENCE STANDARD 91/90, ASTM D910)
8.2 Oil:	For engine operation are recommended mineral oils with minimal kinematic viscosity of 20 $mm^2 s^{-1}$ at 100°C, which percentual carbon residue does not exceed the value of 0.29 %. MS 20 – Running in AEROSHELL Oil 100 – Running in Aeroshell W100 Aeroshell W120 (in tropical climates) ELF Aviation AD 100 BP Aero D 100 TOTAL Aero D 100
8.3 Coolant:	None

9. Fluid capacities:

9.1 Fuel:	Total: for category U: 95.5 litres for category N: 165.5 litres Usable: for category U: 93 litres for category N: 163 litres 2 x 45 litres in main fuel tanks 5.5 litres in connecting tank 2 x 35 litres in auxilitary wing tip tanks
9.2 Oil:	Minimum 9 litres – Maximum 14 litres
9.3 Coolant system capacity:	None

10. Air Speeds:	Never Exceed Speed Limit	V_{NE}	315 km/h IAS
	Normal Operating Speed Limit	V_{NO}	227 km/h IAS
	Design Manoeuvring Speed Limit	V_A	200 km/h IAS
	Maximum Flaps Extended Speed Limit	V_{FE}	152 km/h IAS
	Maximum Open Landing Gear Speed	V_{LE}	300 km/h IAS
	Maximum Landing Gear Operating Speed	V_{LO}	140 km/h IAS
11. Maximum Operating Altitude:	4 500 m		
12. Allweather Operations Capability:	The aircraft is approved for VFR Day flights.		
13. Maximum Weights:	Max. Take-off weight:		
	- For category Utility (U)		940 kg
	- For category Normal (N)		1 000 kg
	Max. Landing weight:		
	- For category Utility (U)		940 kg
	- For category Normal (N)		950 kg
	Max. Variable Load:		
	- For category Utility (U)		250 kg ± 3 %
	- For category Normal (N)		300 kg ± 3 %
14. Centre of Gravity Range:	17.5 % – 28.5 % MAC M.A.C. is 1 568 mm; 0 % M.A.C. is 602 mm aft reference datum.		
15. Datum:	The rear part of fire wall; from it are measured, for purpose of assignation of Gravity Centre, all lateral dimensions.		
16. Control surface deflections:	Elevator deflection	up	28° ± 1°
		down	24° ± 1°
	Elevator trim tab	up	25° ± 2°
		down	40° ± 2°
	Rudder deflection	right and left	28° ± 2°
	Rudder trim tab	left	5° + 3°
		right	30° ± 2°
	Ailerons deflection mm	up	108 mm (+ 5; - 3)
		down	98 mm (+ 5; - 3) mm

	Wing flaps positions retracted 0° take-off 15° ± 2° landing 40° (+ 5°; - 3°)
17. Leveling Means:	Levelling points on left and right side of airplane fuselage to be levelled. Measurement plane to be min. 850 mm below.
18. Minimum Flight Crew:	1 (Pilot)
19. Maximum Passenger Seating Capacity:	2 (including crew)
20. (Reserved)	
21. Baggage/Cargo Compartments:	None
22. Wheels and Tyres:	The wheels K 12-0100.00 or K 420.1-00 of main gear with tyre BARUM or MITAS 420x150-6.5 or MITAS 420x150-6.5 TL with tube 420x150; The wheels K 13-0000.00 or K 34-990 of tail gear with tyre BARUM or MITAS 260x85-4 with tube 260x85.

TIV. Operating and Service Instructions

1. Flight manual:

– In Czech language	Letová příručka Z 726 K, Issue Ref. No. 2264/704/74
– In English language	Flight Manual Z 726 ZLIN UNIVERSAL, date of issue 1977

2. Maintenance Manual:

– In Czech language Doc. No.: Do – Z 726 – 2011	Technický popis a návod k obsluze Z 726, date of issue 1974 - Technický popis Z 726 K, date of issue 1983
– In English language Doc. No.: Do – Z 726 – 2011	Technical Manual Z 726 ZLIN UNIVERSAL, date of issue 1974

3. Overhaul Mnual:

– In Czech language	Opravárenská příručka Z 726 ZLIN UNIVERSAL, issued 1978

4. Ilustrated parts catalogue:

– In Czech, German and English language, issued 1975
Katalog Z 726 ZLIN UNIVERSAL

TV. Notes

None

ADMINISTRATIVE SECTION

I Acronyms

N/A

II Type Certificate Holder Record

Current:
ZLIN AIRCRAFT A.S.
Letiště 1578
765 81 Otrokovice
Czech Republic

Former:

Moravan, n. p.
Gottwaldov – Otrokovice
CZECHOSLOVAKIA

Moravan, n.p.
Letiště 1578
765 81 Otrokovice
CZECHOSLOVAKIA

Moravan, k.p.
Letiště 1578, 765 81 Otrokovice
CZECHOSLOVAKIA

Moravan, a.s.
Letiště 1578, 765 81 Otrokovice
CZECH REPUBLIC

MORAVAN – AEROPLANES, a.s.
Letiště 1578
765 81 Otrokovice
CZECH REPUBLIC

MORAVAN – AVIATION, s.r.o.
Letiště 1578
765 81 Otrokovice
CZECH REPUBLIC

III Change Record

Issue	Date	Changes
Issue 1	27-Mar-2007	Transfer of Z 26 - Series Type Design to EASA
Issue 2	14-Apr-2008	Page1: updated Page 2: updated Page 57, HIII. Tech. Charact. and Oper. Limitation, Item 10: value for v_{NO} corrected Page 92, MV. Notes: Note 2 added Page 139, Change Record: updated All: minor corrections of formatting, different page breake, total page number now 139
Issue 3	24-Aug-2009	Incorporation of changed company name
Issue 4	23 July 2010	Editorial corrections and revision into standard EASA TCDS format
Issue 5	20-Sep-2013	Editorial corrections of typo errors – correction of TC No. for M 137 A engines.