

FOREWORD

The forward and CC instructions will be provided at a later date

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

1.0	INTERNATIONAL STANDARDS AND RECOMMENDED PRACTICES PART I. DEFINITIONS When the following terms are used in the Standards for the Airworthiness of Aircraft, they have the following meanings: <i>Aeroplane.</i> A power-driven heavier-than-air aircraft, deriving its lift in flight chiefly from aerodynamic reactions on surfaces which remain fixed under given conditions of flight.	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.3	<i>Aircraft.</i> Any machine that can derive support in the atmosphere from the reactions of the air other than the reactions of the air against the earth's surface. <i>Note.</i> ††— When the word aircraft is used, it includes the remotely piloted aircraft. ----- †† Applicable as of 26 November 2026.	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.4	<i>Airworthy.</i> † The status of an aircraft, engine, propeller or part when it conforms to its approved design and is in a condition for safe operation. -----	Article (10) Reg. (EU) 2018/1139, 21A.165(c), 21A.307(a) Reg. (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>		No definition exists in the Regulation				

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	† Applicable until 25 November 2026.	748/2012								216/2008 or Regulation 748/2012 (Part-21) but the content of the definition is used in those rules in the same meaning.
1.0.5	<i>Airworthy</i> . <i>Error! Bookmark not defined.</i> The status of an aircraft, remote pilot station, engine, propeller or part when it conforms to its approved design and is in a condition for safe operation. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.6	<i>Anticipated operating conditions</i> . <i>Error! Bookmark not defined.</i> Those conditions which are known from experience or which can be reasonably envisaged to occur during the operational life of the aircraft taking into account the operations for which the aircraft is made eligible, the conditions so considered being relative to the meteorological state of the	Annex II, VII, IXReg. (EU) 2018/1139	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		The term is not defined. However, reference is made to 'anticipated

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	<p>atmosphere, to the configuration of terrain, to the functioning of the aircraft, to the efficiency of personnel and to all the factors affecting safety in flight. Anticipated operating conditions do not include:</p> <p>a) those extremes which can be effectively avoided by means of operating procedures; and</p> <p>b) those extremes which occur so infrequently that to require the Standards to be met in such extremes would give a higher level of airworthiness than experience has shown to be necessary and practical.</p> <p>----- † Applicable until 25 November 2026.</p>							operating conditions' and 'anticipated flight conditions for the operational life of the aircraft' in the Annexes which are then further elaborated in the CS and AMC.		
1.0.7	<p>Anticipated operating conditions.<i>Error! Bookmark not defined.</i> Those conditions which are known from experience or which can be reasonably envisaged to occur during the operational life of the aircraft and remote pilot station taking into account the operations for which the aircraft or remote pilot station is made eligible, the conditions so considered being relative to the meteorological state of the atmosphere, to the configuration of terrain, to the functioning of the aircraft and remote pilot station, to the efficiency of personnel and to all the factors affecting safety in flight. Anticipated operating conditions do not include:</p>		<input type="checkbox"/>							

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	<p>a) those extremes which can be effectively avoided by means of operating procedures; and</p> <p>b) those extremes which occur so infrequently that to require the Standards to be met in such extremes would give a higher level of airworthiness than experience has shown to be necessary and practical.</p> <p>-----</p> <p>†† Applicable as of 26 November 2026.</p>									
1.0.8	<p>Appropriate airworthiness requirements.† The comprehensive and detailed airworthiness codes established, adopted or accepted by a Contracting State for the class of aircraft, engine or propeller under consideration.</p> <p>-----</p> <p>† Applicable until 25 November 2026.</p>	21.B.7021.B.75 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.9	<p>Appropriate airworthiness requirements.†† The comprehensive and detailed airworthiness codes established, adopted or accepted by a Contracting State for the class of aircraft, remote pilot station, engine or propeller under consideration.</p> <p>-----</p> <p>†† Applicable as of 26 November 2026.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.10	<p>Approved. Accepted by a Contracting State as suitable for a particular purpose.</p>	21.B.80Reg. (EU) 748/2012Reg.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		(EU) 1321/2014								
1.0.11	C2 Link. <i>Error! Bookmark not defined.</i> The data link between the remotely piloted aircraft and the remote pilot station for the purposes of managing the flight. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.12	C2 Link interruption. <i>Error! Bookmark not defined.</i> Any temporary situation where the C2 Link is unavailable, discontinuous, introduces too much delay, or has inadequate integrity; but where the lost C2 Link decision time has not been exceeded. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.13	C2 Link specification. <i>Error! Bookmark not defined.</i> The minimum performance to be achieved by the C2 Link equipment in conformity with the applicable airworthiness system design requirements. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.14	Category A. With respect to helicopters, means a multi-engine helicopter designed with engine and system isolation features specified in Part IVB of Annex 8 and capable of operations using take-off and landing data scheduled under a critical engine failure concept which assures adequate designated surface area and	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	adequate performance capability for continued safe flight or safe rejected take-off.									
1.0.15	Category B. With respect to helicopters, means a single-engine or multi-engine helicopter which does not meet Category A standards. Category B helicopters have no guaranteed capability to continue safe flight in the event of an engine failure, and a forced landing is assumed.	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.16	Configuration (as applied to the aeroplane). A particular combination of the positions of the moveable elements, such as wing flaps and landing gear, etc., that affect the aerodynamic characteristics of the aeroplane.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The term is not defined.	
1.0.17	Continuing airworthiness. <i>Error! Bookmark not defined.</i> The set of processes by which an aircraft, engine, propeller or part complies with the applicable airworthiness requirements and remains in a condition for safe operation throughout its operating life. ----- † Applicable until 25 November 2026.	Article 2(d) Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.18	Continuing airworthiness. <i>Error! Bookmark not defined.</i> The set of processes by which an aircraft, remote pilot station, engine, propeller or part complies with the applicable airworthiness requirements and remains in a condition for safe operation throughout its operating life.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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1.0.19	Critical engine(s). Any engine whose failure gives the most adverse effect on the aircraft characteristics relative to the case under consideration. <i>Note.— On some aircraft there may be more than one equally critical engine. In this case, the expression “the critical engine” means one of those critical engines.</i>	CS-Definitions.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.20	Design landing mass. The maximum mass of the aircraft at which, for structural design purposes, it is assumed that it will be planned to land.	CS 23/25/27/29.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.21	Design take-off mass. The maximum mass at which the aircraft, for structural design purposes, is assumed to be planned to be at the start of the take-off run.	CS 23/25/27/29.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.22	Design taxiing mass. The maximum mass of the aircraft at which structural provision is made for load liable to occur during use of the aircraft on the ground prior to the start of take-off.	CS 25.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.0.23	Detect and avoid. †† The capability to see, sense or detect conflicting traffic or other hazards and take the appropriate action. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.24	Discrete source damage. Structural damage of the aeroplane that is likely to result from: impact with a bird, uncontained fan blade failure, uncontained engine failure, uncontained high-energy rotating machinery failure or similar causes.	AMC 25.571	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.25	Engine. A unit used or intended to be used for aircraft propulsion. It consists of at least those components and equipment necessary for functioning and control, but excludes the propeller/rotors (if applicable).	CS-definitions, CS 27/29.901	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.26	Factor of safety. A design factor used to provide for the possibility of loads greater than those assumed, and for uncertainties in design and fabrication.	CS 23/25/27/29.303	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.27	Final approach and take-off area (FATO). A defined area over which the final phase of the approach manoeuvre to hover or landing is completed and from which the take-off manoeuvre is commenced. Where the FATO is to be used by performance Class	Regulation EU 965/2012, Annex 1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	1 helicopters, the defined area includes the rejected take-off area available.									
1.0.28	Fireproof. The capability to withstand the application of heat by a flame for a period of 15 minutes. <i>Note.— The characteristics of an acceptable flame can be found in ISO 2685.</i>	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.29	Fire resistant. The capability to withstand the application of heat by a flame for a period of 5 minutes. <i>Note.— The characteristics of an acceptable flame can be found in ISO 2685.</i>	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.30	Handover. <i>Error! Bookmark not defined.</i> The act of passing piloting control from one remote pilot station to another. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.31	Helicopter. A heavier-than-air aircraft supported in flight chiefly by the reactions of the air on one or more power-driven rotors on substantially vertical axes. <i>Note.— Some States use the term “rotorcraft” as an alternative to “helicopter”.</i>	CS-Definitions. Reg. (EU) 965/2012 Annex I	<input checked="" type="checkbox"/>	<input type="checkbox"/>		Note: there is a difference between the definitio				

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										ns in OPS regulation and CS-Definitions.
1.0.32	Human factors principles. Principles which apply to aeronautical design, certification, training, operations and maintenance and which seek safe interface between the human and other system components by proper consideration to human performance.	Annex II 145.A.30 (e) Reg. (EU) 2018/1139 Reg. (EU) 1321/2014CS-23CS-25CS-27CS-29	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.33	Human performance. Human capabilities and limitations which have an impact on the safety and efficiency of aeronautical operations.	Annex II 145.A.30 (e) Reg. (EU) 2018/1139 Reg. (EU) 1321/2014CS-23CS-25CS-27CS-29	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.34	Landing surface. That part of the surface of an aerodrome which the aerodrome authority has declared available for the normal ground or water run of aircraft landing in a particular direction.	Art. 37 (5) Reg. (EU) 2019/1138 CS-ADR-	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		DSN.A.002								
1.0.35	Limit loads. The maximum loads assumed to occur in the anticipated operating conditions.	CS 23/25/27/29.301	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.36	Load factor. The ratio of a specified load to the weight of the aircraft, the former being expressed in terms of aerodynamic forces, inertia forces, or ground reactions.	CS definitions. CS 23/25/27/29.321	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.37	Lost C2 Link decision time. <i>Error! Bookmark not defined.</i> The maximum length of time permitted before declaring a lost C2 Link state during which the C2 Link performance is not sufficient to allow the remote pilot to actively manage the flight in a safe and timely manner appropriate to the airspace and operational conditions. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.38	Lost C2 Link state. †† The state of the RPAS in which the C2 Link performance has degraded, as a result of a C2 Link interruption that is longer than the lost		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	C2 Link decision time, to a point where it is not sufficient to allow the remote pilot to actively manage the flight in a safe and timely manner. ----- †† Applicable as of 26 November 2026.									
1.0.39	Maintenance. † The performance of tasks on an aircraft, engine, propeller or associated part required to ensure the continuing airworthiness of an aircraft, engine, propeller or associated part including any one or combination of overhaul, inspection, replacement, defect rectification, and the embodiment of a modification or repair. ----- † Applicable until 25 November 2026.	Art. 2 (h)Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The EU definition excludes pre-flight inspections , having a separate definition.	
1.0.40	Maintenance. <i>Error! Bookmark not defined.</i> The performance of tasks on an aircraft, remote pilot station, engine, propeller or associated part required to ensure the continuing airworthiness of an aircraft, remote pilot station, engine, propeller or associated part including any one or combination of overhaul, inspection, replacement, defect rectification, and the embodiment of a modification or repair. ----- †† Applicable as of 26 November 2026.	Art. 2 (h) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The EU definition excludes pre-flight inspections , having a separate definition.	
1.0.41	Maintenance organization's procedures manual. A document endorsed by the head of the maintenance organization which details the maintenance	145.A.70M.A .604 CAO.A.025R	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		Different wording

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	organization's structure and management responsibilities, scope of work, description of facilities, maintenance procedures and quality assurance or inspection systems.	eg. (EU) 1321/2014								: the EU rules use the term "maintenance organisation exposition", or "maintenance organisation manual".
1.0.42	Maintenance records. Records that set out the details of the maintenance carried out on an aircraft, engine, propeller or associated part.	145.A.55M.A.614Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.43	Maintenance release. A document which contains a certification confirming that the maintenance work to which it relates has been completed in a satisfactory manner in accordance with appropriate airworthiness requirements.	145.A.50M.A.612M.A.613 M.A.801M.A.802CAO.A.065CAO.A.070Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	Different wording with the same meaning: certificate of release to service.					

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1.0.44	<p>Modification. A change to the type design of an aircraft, engine or propeller.</p> <p><i>Note.— A modification may also include the embodiment of the modification which is a maintenance task subject to a maintenance release. Further guidance on aircraft maintenance, modification and repair is contained in the Airworthiness Manual (Doc 9760).</i></p>	Subpart DReg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.45	<p>Nominal C2 Link state.<i>Error! Bookmark not defined.</i> The state of the RPAS when the C2 Link performance is sufficient to allow the remote pilot to actively manage the flight of the RPA in a safe and timely manner appropriate to the airspace and operational conditions.</p> <p>----- †† Applicable as of 26 November 2026.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.46	<p>Organization responsible for the type design.<i>Error! Bookmark not defined.</i> The organization that holds the type certificate, or equivalent document, for an aircraft, engine or propeller type, issued by a Contracting State.</p> <p>----- † Applicable until 25 November 2026.</p>	Art. 72 Reg. (EU) 2018/1139 21.A.20 , 21.A.44 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.47	<p>Organization responsible for the type design.<i>Error! Bookmark not defined.</i> The organization that holds the type certificate, or equivalent document, for an aircraft, remote pilot station, engine or propeller type, issued by a Contracting State.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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1.0.48	Orphan aircraft type. An aircraft which has its Type Certificate revoked by the State of Design, and no longer has a designated State of Design in accordance with Annex 8. These aircraft do not meet the Standards of Annex 8.	21.A.701Reg. (EU) 748/2012GM 21.A.701 (a)	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		No formal definition is implemented however the term is used and orphan aircraft are addressed through Regulation (EU) 748/2012
1.0.49	Performance Class 1 helicopter. A helicopter with performance such that, in case of engine failure, it is able to land on the rejected take-off area or safely continue the flight to an appropriate landing area.	Art. 2 of Annex IReg. (EU) 965/2012	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Term is used for operations and not airworthiness. For type-certification	

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									n, performance is related to Category A	
1.0.50	Performance Class 2 helicopter. A helicopter with performance such that, in case of engine failure, it is able to safely continue the flight, except when the failure occurs prior to a defined point after take-off or after a defined point before landing, in which cases a forced landing may be required.	Annex I Reg. (EU) 965/2012	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Term is used for operations and not airworthiness. For type-certification, performance is related to Category A.	
1.0.51	Performance Class 3 helicopter. A helicopter with performance such that, in case of engine failure at any point in the flight profile, a forced landing must be performed.	Annex I Reg. (EU) 965/2012	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Term is used for operations and not airworthiness. For type-certification, performance is	

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									related to Category B	
1.0.52	Powerplant. The system consisting of all the engines, drive system components (if applicable), and propellers (if installed), their accessories, ancillary parts, and fuel and oil systems installed on an aircraft but excluding the rotors for a helicopter.	CS 25.901, CS 23.901, CS 27/29.901	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.53	Pressure-altitude. An atmospheric pressure expressed in terms of altitude which corresponds to that pressure in the standard atmosphere.	Art. 2 (101) Reg. (EU) 923/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.54	Quality of service delivered (QoSD). †† A statement of the QoS achieved or delivered to the RPAS operator by the C2CSP. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.55	Quality of service required (QoS R). <i>Error! Bookmark not defined.</i> A statement of the QoS requirements of the RPAS operator to the C2CSP. <i>Note.— The QoS R may be expressed in descriptive terms (criteria) listed in the order of priority, with preferred performance value for each criterion. The</i>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<i>C2CSP then translates these into parameters and metrics pertinent to the service.</i> ----- †† Applicable as of 26 November 2026.									
1.0.56	Remote pilot station (RPS).Error! Bookmark not defined. The component of the remotely piloted aircraft system containing the equipment used to pilot the remotely piloted aircraft. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>							
1.0.57	Remotely piloted aircraft (RPA).Error! Bookmark not defined. An unmanned aircraft which is piloted from a remote pilot station. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>							
1.0.58	Remotely piloted aircraft system (RPAS).Error! Bookmark not defined. A remotely piloted aircraft, its associated remote pilot station(s), the required C2 Link(s) and any other components as specified in the type design. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>							
1.0.59	Rendering (a Certificate of Airworthiness) valid. The action taken by a Contracting State, as an alternative to issuing its own Certificate of Airworthiness, in		<input type="checkbox"/>	<input checked="" type="checkbox"/>		Reg. (EU) 748/201				

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
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	accepting a Certificate of Airworthiness issued by any other Contracting State as the equivalent of its own Certificate of Airworthiness.									2 Requires that a certificate of airworthiness is issued.
1.0.60	Repair. The restoration of an aircraft, engine, propeller or associated part to an airworthy condition in accordance with the appropriate airworthiness requirements after it has been damaged or subjected to wear.	21.A.431A (c) Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.61	Satisfactory evidence. A set of documents or activities that a Contracting State accepts as sufficient to show compliance with an airworthiness requirement.	21.B.80Reg. (EU) 748/2012Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.0.62	Standard atmosphere. An atmosphere defined as follows: a) the air is a perfect dry gas; b) the physical constants are: — Sea level mean molar mass: $M_0 = 28.964\ 420 \times 10^{-3}$ kg mol ⁻¹	CS-Definitions	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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				Level of implementation of SARPs						
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<p>pressure: — Sea level atmospheric</p> <p>— $P_0 = 1\,013.250\text{ hPa}$</p> <p>— Sea level temperature: $t_0 = 15^\circ\text{C}$ $T_0 = 288.15\text{ K}$</p> <p>— Sea level atmospheric density: $\rho_0 = 1.225\,0\text{ kg m}^{-3}$</p> <p>— Temperature of the ice point: $T_i = 273.15\text{ K}$</p> <p>— Universal gas constant: $R^* = 8.314\,32\text{ JK}^{-1}\text{mol}^{-1}$</p> <p>c) the temperature gradients are:</p>	<i>Geopotential altitude (km)</i>		<i>Temperature gradient (Kelvin per standard geopotential kilometre)</i>						
	<i>From</i>	<i>To</i>							
	-5.0	11.0	-6.5						
	11.0	20.0	0.0						
	20.0	32.0	+1.0						
	32.0	47.0	+2.8						
	47.0	51.0	0.0						
	51.0	71.0	-2.8						
	71.0	80.0	-2.0						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p><i>Note 1.— The standard geopotential metre has the value 9.806 65 m² s⁻².</i></p> <p><i>Note 2.— See Doc 7488 for the relationship between the variables and for tables giving the corresponding values of temperature, pressure, density and geopotential.</i></p> <p><i>Note 3.— Doc 7488 also gives the specific weight, dynamic viscosity, kinematic viscosity and speed of sound at various altitudes.</i></p>									
1.0.63	State of Design. The State having jurisdiction over the organization responsible for the type design.	Regulation (EC) 2018/1139. Regulation (EU) 748/2012.	<input checked="" type="checkbox"/>	<input type="checkbox"/>		Part 21 defines 'principal place of business'. The content of the definition is used in the same meaning.				
1.0.64	State of Design of Modification. The State having jurisdiction over the individual or organization responsible for the design of the modification or repair of an aircraft, engine or propeller.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

1.0.65	<p>State of Manufacture.† The State having jurisdiction over the organization responsible for the final assembly of the aircraft, engine or propeller.</p> <p>-----</p> <p>† Applicable until 25 November 2026.</p>	Regulation (EC) 2018/1139. Regulation (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>	The content of the definition is used in the same meaning					
1.0.66	<p>State of Manufacture.†† The State having jurisdiction over the organization responsible for the final assembly of the aircraft, remote pilot station, engine or propeller.</p> <p>-----</p> <p>†† Applicable as of 26 November 2026.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.67	<p>State of Registry. The State on whose register the aircraft is entered.</p> <p><i>Note.— In the case of the registration of aircraft of an international operating agency on other than a national basis, the States constituting the agency are jointly and severally bound to assume the obligations which, under the Chicago Convention, attach to a State of Registry. See, in this regard, the Council Resolution of 14 December 1967 on Nationality and Registration of Aircraft Operated by International Operating Agencies which can be found in Policy and Guidance Material on the Economic Regulation of International Air Transport (Doc 9587).</i></p>	Regulation (EC) 2018/1139. Regulation (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>		Reliance is placed on the ICAO definition				

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				Level of implementation of SARPs						
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1.0.68	Switchover. <i>Error! Bookmark not defined.</i> The act of transferring the active datalink path between the RPS and the RPA from one of the links or networks that constitutes the C2 Link to another link or network that constitutes the C2 Link. ----- †† Applicable as of 26 November 2026.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.69	Take-off surface. That part of the surface of an aerodrome which the aerodrome authority has declared available for the normal ground or water run of aircraft taking off in a particular direction.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Not defined.	
1.0.70	Type Certificate. † A document issued by a Contracting State to define the design of an aircraft, engine or propeller type and to certify that this design meets the appropriate airworthiness requirements of that State. <i>Note.— In some Contracting States a document equivalent to a Type Certificate may be issued for an engine or propeller type.</i> ----- † Applicable until 25 November 2026.	21.A.41 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.0.71	Type Certificate. †† A document issued by a Contracting State to define the design of an aircraft, remote pilot station, engine or propeller type and to certify that		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>this design meets the appropriate airworthiness requirements of that State.</p> <p><i>Note 1.†††— In some Contracting States a document equivalent to a Type Certificate may be issued for an engine or propeller type.</i></p> <p><i>Note 2.Error! Bookmark not defined.— A document equivalent to a Type Certificate may be issued for a remote pilot station type.</i></p> <p>-----</p> <p>†† Applicable as of 26 November 2026. ††† As of 26 November 2026, this Note becomes Note 1.</p>									
1.0.72	<p>Type design.<i>Error! Bookmark not defined.</i> The set of data and information necessary to define an aircraft, engine or propeller type for the purpose of airworthiness determination.</p> <p>-----</p> <p>† Applicable until 25 November 2026.</p>	(EU) 748/2012 Annex I (Part 21), amended by Reg (EU) 69/2014, Point 21.A.31	<input checked="" type="checkbox"/>	<input type="checkbox"/>	Necessary type design data and information are listed	detailed definition				
1.0.73	<p>Type design.<i>Error! Bookmark not defined.</i> The set of data and information necessary to define an aircraft, remote pilot station, engine or propeller type for the purpose of airworthiness determination.</p> <p>-----</p> <p>†† Applicable as of 26 November 2026.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
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1.0.74	Ultimate load. The limit load multiplied by the appropriate factor of safety.	CS 23/25/27/29.301 and 303	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.1	<p>PART II. PROCEDURES FOR CERTIFICATION AND CONTINUING AIRWORTHINESS</p> <p><i>Note.— Although the Convention on International Civil Aviation allocates to the State of Registry certain functions which that State is entitled to discharge, or obligated to discharge, as the case may be, the Assembly recognized, in Resolution A23-13, that the State of Registry may be unable to fulfil its responsibilities adequately in instances where aircraft are leased, chartered or interchanged — in particular without crew — by an operator of another State and that the Convention may not adequately specify the rights and obligations of the State of the Operator in such instances until such time as Article 83 bis of the Convention enters into force. Accordingly, the Council urged that if, in the above-mentioned instances, the State of Registry finds itself unable to discharge adequately the functions allocated to it by the Convention, it delegate to the State of the Operator, subject to acceptance by the latter State, those functions of the State of Registry that can more adequately be discharged by the State of the Operator. It was understood that pending entry into force of</i></p>	<p>Art. 11, 12, 13, 14Reg. (EU) 2018/1139Subpart B, D, E21.B.80Reg. (EU) 748/2012Art. 13Reg. (EU) 1005/2009 Subpart B, D Annex Ib Reg (EU) 748/2012</p>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Chapter 1 b): Cut-off and end dates are prescribed for the phasing out of halon.	EU legislation establishes cut-off dates for existing aircraft types and end dates for all aircraft. After these dates the use of halon would no longer be permitted.

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				Level of implementation of SARPs						
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	<p><i>Article 83 bis of the Convention, the foregoing action would only be a matter of practical convenience and would not affect either the provisions of the Chicago Convention prescribing the duties of the State of Registry or any third State. However, as Article 83 bis entered into force on 20 June 1997, such transfer agreements will have effect in respect of those Contracting States which have ratified the related Protocol (Doc 9318) upon fulfilment of the conditions established in Article 83 bis.</i></p> <p>CHAPTER 1. TYPE CERTIFICATION</p> <p>1.1 Applicability</p> <p>The Standards of this chapter shall be applicable to all aircraft, and to engines and propellers if type certificated separately, for which the application for certification was submitted to a Contracting State on or after 13 June 1960, except that:</p> <p>a) the provisions of 1.4 of this part shall only be applicable to an aircraft type for which an application for a Type Certificate was submitted to the State of Design on or after 2 March 2004;</p> <p>b) the provisions of 1.4 of this part shall only be applicable to an engine or propeller type for</p>								
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Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference				Not Applicable	Details of Difference	Remarks	
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Sigini- ficant Differ- ence
				Level of implementation of SARPs						
				A) More Exactin- g or Exceed- s	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

<p>which an application for a Type Certificate was submitted to the State of Design on or after 10 November 2016;</p> <p>c) the provisions of 1.2.6 of this part shall only be applicable to an aircraft type for which an application for a Type Certificate was submitted to the State of Design on or after 31 December 2014;</p> <p>d) the provisions of 1.2.7 of this part shall only be applicable to an aircraft type for which an application for a Type Certificate is submitted to the State of Design on or after 28 November 2024; and</p> <p>e) the provisions of 1.4 of this part shall only be applicable to a remotely piloted aircraft and to a remote pilot station if type certificated separately, for which an application for a Type Certificate is submitted to the State of Design on or after 26 November 2026.</p> <p><i>Note 1.— Until 25 November 2026, normally, a request for a Type Certificate is submitted by the manufacturer when the aircraft, engine or propeller is intended for serial production.</i></p> <p><i>Note 2.— For Part VB aeroplanes, guidance material concerning the appropriate airworthiness safety levels commensurate with acceptable risk levels is contained in the Airworthiness Manual (Doc 9760).</i></p> <p><i>Note.3 — As of 26 November 2026, the provisions of this part support remotely piloted aircraft systems operation SARPs in Annex 6.</i></p>									
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Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

1.1	<p style="text-align: center;">1.1 Applicability</p> <p>The Standards of this chapter shall be applicable to all aircraft, and to engines and propellers if type certificated separately, for which the application for certification was submitted to a Contracting State on or after 13 June 1960, except that:</p> <p>a) the provisions of 1.4 of this part shall only be applicable to an aircraft type for which an application for a Type Certificate was submitted to the State of Design on or after 2 March 2004;</p> <p>b) the provisions of 1.4 of this part shall only be applicable to an engine or propeller type for which an application for a Type Certificate was submitted to the State of Design on or after 10 November 2016;</p> <p>c) the provisions of 1.2.6 of this part shall only be applicable to an aircraft type for which an application for a Type Certificate was submitted to the State of Design on or after 31 December 2014;</p> <p>d) the provisions of 1.2.7 of this part shall only be applicable to an aircraft type for which an application for a Type Certificate is submitted to the State of Design on or after 28 November 2024; and</p> <p>e) the provisions of 1.4 of this part shall only be applicable to a remotely piloted aircraft and to a remote pilot station if type certificated separately, for</p>		<input type="checkbox"/>							
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
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	<p>which an application for a Type Certificate is submitted to the State of Design on or after 26 November 2026.</p> <p><i>Note 1.— As of 26 November 2026, normally, a request for a Type Certificate is submitted by the manufacturer when the aircraft, remote pilot station, engine or propeller is intended for serial production.</i></p> <p><i>Note 2.— For Part VB aeroplanes, guidance material concerning the appropriate airworthiness safety levels commensurate with acceptable risk levels is contained in the Airworthiness Manual (Doc 9760).</i></p> <p><i>Note.3 — As of 26 November 2026, the provisions of this part support remotely piloted aircraft systems operation SARPs in Annex 6.</i></p>								
1.2.1	<p>1.2 Design aspects of the appropriate airworthiness requirements</p> <p>1.2.1 Until 25 November 2026, the design aspects of the appropriate airworthiness requirements, used by a Contracting State for type certification of an aircraft, engine or propeller or for any change to such type certification, shall be such that compliance with them will ensure compliance with the Standards of this part and, where applicable, with the Standards of Parts III, IV, V, VI or VII of this Annex.</p>	Reg. (EU) 2018/1139, Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>					
1.2.1	1.2.1 As of 26 November 2026, the design aspects of the appropriate airworthiness requirements,		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	

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	used by a Contracting State for type certification of an aircraft, remote pilot station, engine or propeller or for any change to such type certification, shall be such that compliance with them will ensure compliance with the Standards of this part and, where applicable, with the Standards of Parts III, IV, V, VI, VII, VIII, IX or X of this Annex.									
1.2.2	<p>1.2.2 Recommendation.— <i>When establishing the appropriate airworthiness requirements, a risk-based proportionality approach should be applied.</i></p> <p><i>Note.— For Part VB aeroplanes, guidance material concerning the appropriate airworthiness safety levels commensurate with acceptable risk levels is contained in Doc 9760.</i></p>	21.A.14 21.B.80Reg. (EU) 748/2012CS-22CS-23CS-25CS-26CS-27CS-29CS-31GBCS-31HBCS-31TGBCS-34CS-36CS-APUCS-AWOCSECS-PCSETSOCS-LSACSVLACSVLR 21L.A.23 21L.B.43 AnnexIb Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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1.2.3	1.2.3 The design shall not have any features or characteristics that render it unsafe under the anticipated operating conditions.	21L.A.27 Annex Ib 21.A.21 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.4	1.2.4 Until 25 November 2026, where the design features of a particular aircraft, engine or propeller render any of the design aspects of the appropriate airworthiness requirements or the Standards in Parts III, IV, V, VI or VII inappropriate, the Contracting State shall apply appropriate requirements that will give at least an equivalent level of safety.	21.A.21, 21.A.16 Reg. (EU) 748/2012 21L.A.27, 21L.B.43 Annex Ib Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.4	1.2.4 As of 26 November 2026, where the design features of a particular aircraft, remote pilot station, engine or propeller render any of the design aspects of the appropriate airworthiness requirements or the Standards in Parts III, IV, V, VI, VII, VIII, IX or X inappropriate, the Contracting State shall apply appropriate requirements that will give at least an equivalent level of safety.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.2.5	1.2.5 Until 25 November 2026, where the design features of a particular aircraft, engine or propeller render any of the design aspects of the appropriate airworthiness requirements or the Standards in Parts III, IV, V, VI or VII inadequate, additional requirements that are	21L.A.27, 21L.B.43 Annex Ib Reg (EU) 748/2012 21.A.21,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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	considered by the Contracting State to give at least an equivalent level of safety shall be applied. <i>Note.— An Airworthiness Manual (Doc 9760) containing guidance material has been published by ICAO.</i>	21.A.16 Reg. (EU) 748/2012								
1.2.5	<i>1.2.5</i> As of 26 November 2026, where the design features of a particular aircraft, remote pilot station, engine or propeller render any of the design aspects of the appropriate airworthiness requirements or the Standards in Parts III, IV, V, VI, VII, VIII, IX or X inadequate, additional requirements that are considered by the Contracting State to give at least an equivalent level of safety shall be applied. <i>Note 1.— An Airworthiness Manual (Doc 9760) containing guidance material has been published by ICAO.</i> <i>Note 2.— As of 26 November 2026, a Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019) containing guidance material has been published by ICAO.</i> ----- 1. As of 26 November 2026, this Note becomes Note 1.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.2.6	1.2.6 Until 25 November 2026, the approved design of an aircraft under Parts IIIB, IVB, VA and VB of this Annex shall use extinguishing agents that are not	Reg. (EC) 1005/2009	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Cut-off dates and end dates	

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>listed in the 1987 <i>Montreal Protocol on Substances that Deplete the Ozone Layer</i> as it appears in the Eighth Edition of the <i>Handbook for the Montreal Protocol on Substances that Deplete the Ozone Layer</i>, Annex A, Group II, in the aircraft fire suppression or extinguishing systems in the lavatories, engines and auxiliary power unit.</p> <p><i>Note.— Information concerning extinguishing agents is contained in the UNEP Halons Technical Options Committee Technical Note No. 1 — New Technology Halon Alternatives and FAA Report No. DOT/FAA/AR-99-63, Options to the Use of Halons for Aircraft Fire Suppression Systems.</i></p>								are prescribed by Regulation No 1005/2009 for the phasing out of Halons. For cargo compartment, Regulation No 1005/2009 provides a cut-off date of end 2018 against 28 November 2024 (chapter 1.1 of this Annex).	
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1.2.6	1.2.6 As of 26 November 2026, the approved design of an aircraft under Parts IIIB, IVB, V, VIII and IX of this Annex shall use extinguishing agents that are not listed in the 1987 <i>Montreal Protocol on Substances that Deplete the Ozone Layer</i> as it appears in the Eighth		<input type="checkbox"/>							
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>Edition of the <i>Handbook for the Montreal Protocol on Substances that Deplete the Ozone Layer</i>, Annex A, Group II, in the aircraft fire suppression or extinguishing systems in the lavatories, engines and auxiliary power unit.</p> <p><i>Note.— Information concerning extinguishing agents is contained in the UNEP Halons Technical Options Committee Technical Note No. 1 — New Technology Halon Alternatives and FAA Report No. DOT/FAA/AR-99-63, Options to the Use of Halons for Aircraft Fire Suppression Systems.</i></p>									
1.2.7	<p>1.2.7 The approved design of an aircraft under Part IIIB of this Annex shall use extinguishing agents that are not listed in the 1987 <i>Montreal Protocol on Substances that Deplete the Ozone Layer</i> as it appears in the Tenth Edition of the <i>Handbook for the Montreal Protocol on Substances that Deplete the Ozone Layer</i>, Annex A, Group II, in the aircraft fire suppression or extinguishing systems for the cargo compartment.</p> <p><i>Note.— Information concerning acceptable agents is contained in the report of the UNEP Halons Technical Options Committee Technical Note No. 1 — New Technology Halon Alternatives and FAA Report No. DOT/FAA/AR-11-31, Options to the Use of Halons for Aircraft Fire Suppression Systems.</i></p>	Reg. (EC) 1005/2009	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Cut-off dates and end dates are prescribed by Regulation No 1005/2009 for the phasing out of Halons.For cargo compartment, Regulation No 1005/2009 provides a	Certification Specifications have been amended in 2012 to refer to the above mentioned legislation.

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				Level of implementation of SARPs						
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									cut-off date of end 2018 against 28 November 2024 (chapter 1.1 of this Annex).	
1.3.1	<p>1.3 Proof of compliance with the appropriate airworthiness requirements</p> <p>1.3.1 Until 25 November 2026, there shall be an approved design consisting of such drawings, specifications, reports and documentary evidence as are necessary to define the design of the aircraft, engine or propeller and to show compliance with the design aspects of the appropriate airworthiness requirements.</p> <p><i>Note.— The approval of the design is facilitated, in some States, by approving the design organization.</i></p>	21L.A.25, 21L.A.26, 21L.A.27, 21L.B.43 Annex Ib Reg (EU) 748/2012 21.A.20, 21.A.33, 21.A.35, 21.A.31 21.A.21, 21.A.16 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.1	1.3.1 As of 26 November 2026, there shall be an approved design consisting of such drawings, specifications, reports and documentary evidence as are necessary to define the design of the aircraft, remote pilot station, engine or propeller and to show compliance with		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
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	the design aspects of the appropriate airworthiness requirements. <i>Note.— The approval of the design is facilitated, in some States, by approving the design organization.</i>									
1.3.2	1.3.2 Recommendation. — <i>Contracting States should balance risks and rigor in the determination of compliance based on the acceptable level of risk determined for the product.</i> <i>Note.— For the type certification of Part VB aeroplanes, guidance material addressing how States may balance risks and rigor in the determination of compliance is contained in Doc 9760.</i>	21L.B.46 Annex Ib Reg (EU) 748/2012 21.B.100 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.3	1.3.3 Until 25 November 2026, the aircraft, engine or propeller shall be subjected to such inspections and ground and flight tests as are deemed necessary by the State to show compliance with the design aspects of the appropriate airworthiness requirements.	21L.A.25, 21L.A.26, 21L.A.27, Annex Ib Reg (EU) 748/2012 21.A.33, 21.A.35 21.A.21, 21.A.16 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.3	1.3.3 As of 26 November 2026, the aircraft, remote pilot station, engine or propeller shall be subjected to such inspections and ground and flight tests as are		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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				Level of implementation of SARPs						
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	deemed necessary by the State to show compliance with the design aspects of the appropriate airworthiness requirements.									
1.3.4	1.3.4 Until 25 November 2026, in addition to determining compliance with the design aspects of the appropriate airworthiness requirements for an aircraft, engine or propeller, Contracting States shall take whatever other steps they deem necessary to ensure that the design approval is withheld if the aircraft , engine or propeller is known or suspected to have dangerous features not specifically guarded against by those requirements.	21L.A.3, 21L.A.27, 21L.B.47, 21L.B.48 Annex Ib Reg (EU) 748/2012 21.A.21, 21.A.3 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.4	1.3.4 As of 26 November 2026, in addition to determining compliance with the design aspects of the appropriate airworthiness requirements for an aircraft, remote pilot station, engine or propeller, Contracting States shall take whatever other steps they deem necessary to ensure that the design approval is withheld if the aircraft, remote pilot station, engine or propeller is known or suspected to have dangerous features not specifically guarded against by those requirements.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.3.5	1.3.5 Until 25 November 2026, a Contracting State issuing an approval for the design of a modification, of a repair or of a replacement part shall do so on the basis of satisfactory evidence that the aircraft, engine or propeller is in compliance with the airworthiness requirements used for the issuance of the Type	21L.A.67, 21L.A.68, 21L.B.81, 21L.A.86, Annex Ib Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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				Level of implementation of SARPs						
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	<p>Certificate, its amendments or later requirements when determined by the State.</p> <p><i>Note 1.— While a repair may be completed and shown to be in compliance with the set of requirements that had been selected for the original type certification of the aircraft, engine or propeller, some repairs may need to be shown to comply with the latest applicable certification requirements. In such cases, States may issue a repair design approval against the latest set of requirements for that aircraft, engine or propeller type.</i></p> <p><i>Note 2.— The approval of the design of a modification to an aircraft, engine or propeller is signified, in some States, by the issuance of a supplemental Type Certificate or amended Type Certificate.</i></p>	748/2012 21.A.95, 97, 101, 103, 115, 437 Reg. (EU) 748/2012							
1.3.5	<p>1.3.5 As of 26 November 2026, a Contracting State issuing an approval for the design of a modification, of a repair or of a replacement part shall do so on the basis of satisfactory evidence that the aircraft, remote pilot station, engine or propeller is in compliance with the airworthiness requirements used for the issuance of the Type Certificate, its amendments or later requirements when determined by the State.</p> <p><i>Note 1.— While a repair may be completed and shown to be in compliance with the set of requirements that had been selected for the original type certification of the aircraft, remote pilot station, engine or propeller, some repairs may need to be shown to comply with the latest applicable certification requirements. In such</i></p>		<input type="checkbox"/>						

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	<p><i>cases, States may issue a repair design approval against the latest set of requirements for that aircraft, remote pilot station, engine or propeller type.</i></p> <p><i>Note 2.— The approval of the design of a modification to an aircraft, remote pilot station, engine or propeller is signified, in some States, by the issuance of a supplemental Type Certificate or amended Type Certificate.</i></p>									
1.4.1	<p>1.4 Issuance of Type Certificate</p> <p>1.4.1 Until 25 November 2026, the State of Design, upon receipt of satisfactory evidence that the aircraft, engine or propeller type if certificated separately is in compliance with the design aspects of the appropriate airworthiness requirements, shall issue a Type Certificate to define the type design and to signify its approval of the design of the aircraft type.</p>	21L.A.27 Annex Ib Reg (EU) 748/2012 21.A.21Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4.1	<p><i>1.4.1</i> As of 26 November 2026, the State of Design, upon receipt of satisfactory evidence that the aircraft, remote pilot station, engine or propeller type if certificated separately is in compliance with the design aspects of the appropriate airworthiness requirements, shall issue a Type Certificate to define the type design and to signify its approval of the design of the aircraft type.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.4.2	<p>1.4.2 Until 25 November 2026, when a Contracting State, other than the State of Design, issues a</p>	21L.A.27 21L.B.43,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	Type Certificate for an aircraft, engine or propeller type, it shall do so on the basis of satisfactory evidence that the aircraft, engine or propeller type is in compliance with the design aspects of the appropriate airworthiness requirements.	21.B.47 Annex Ib Reg (EU) 748/2012 21.A.17, 21 Reg. (EU) 748/2012								
1.4.2	1.4.2 As of 26 November 2026, when a Contracting State, other than the State of Design, issues a Type Certificate for an aircraft, remote pilot station, engine or propeller type, it shall do so on the basis of satisfactory evidence that the aircraft, remote pilot station, engine or propeller type is in compliance with the design aspects of the appropriate airworthiness requirements.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.4.3	1.4.3 As of 26 November 2026, the Type Certification of the remotely piloted aircraft shall include the remote pilot station and the C2 Link as defined in the appropriate parts.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
1.5.1	1.5 Suspension of Type Certificate 1.5.1 When the State of Design takes action in accordance with its established procedures to suspend in whole or in part a Type Certificate for an aircraft, engine or propeller type, it shall immediately: a) notify Contracting States of the suspension; the time period, if known, that the suspension	PR.SLC.0001	<input checked="" type="checkbox"/>	<input type="checkbox"/>		EASA procedure PR.SLC .0001 requires to inform all				

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	is in force; the cause of the suspension; and any recommended action to be undertaken if the nature of the suspension affects the airworthiness of the affected aircraft, engine or propeller type; and b) establish with the State of Manufacture, if other than the State of Design, any actions necessary to address their respective airworthiness responsibilities under the agreement or arrangement established in accordance with 2.4.5 of this part.									affected States via issuing an AD.
1.5.2	1.5.2 A Contracting State that issued a Type Certificate for an aircraft, engine or propeller type under 1.4.2 of this part, on the basis of the Type Certificate issued by the State of Design, shall immediately notify the State of Design of a suspension originated in respect of its equivalent Type Certificate.	PR.SLC.0001	<input checked="" type="checkbox"/>	<input type="checkbox"/>		EASA procedure PR.SLC.0001 requires to inform all affected States via issuing an AD.				
1.5.3	1.5.3 During the period of suspension notified in 1.5.1 and 1.5.2, the State of Design shall continue to fulfil its assigned obligations on continuing airworthiness under Chapter 4 of this part.	21L.A.4 Annex Ib Reg (EU) 748/2012 21.A.3BReg.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		(EU) 748/2012								
1.5.4	1.5.4 The State of Design shall notify Contracting States and the State of Manufacture, if other than the State of Design, on a regular basis the status of the suspension and reinstatement of the suspended Type Certificate.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Not implemented Process is not established	
1.6.1	<p>1.6 Revocation of Type Certificate</p> <p>1.6.1 The State of Design shall establish procedures for the revocation of a Type Certificate when the organization responsible for the type design surrenders or abandons the Type Certificate, or ceases to exist, and as a result the continuing airworthiness responsibilities established under Chapter 4 of this part can no longer be fulfilled for the affected aircraft type in service. The procedures, at a minimum, shall include:</p> <p>a) notification to all Contracting States of an intent to revoke a Type Certificate and the proposed termination of the production approval under 2.4 of this part; and</p> <p>b) consultation with States of Registry for the collection, identification and establishment of supplemental airworthiness requirements considered necessary for the continued airworthiness of the candidate orphan aircraft type.</p>	PR.SLC.0001	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EASA procedure PR.SLC.0001 requires to inform all affected States via issuing an AD.	

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1.6.2	1.6.2 Except for reasons concerning the immediate safety of an aircraft type, the State of Design shall not unduly revoke a Type Certificate without providing ample notice and guidance to States of Registry that will be assuming ultimate responsibility for the continued airworthiness of orphaned aircraft on their civil register.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		Process is not established.
1.6.3	1.6.3 The State of Design shall notify Contracting States, including the State of Manufacture if other than the State of Design, of the revocation of a Type Certificate and the effective date on which it ceases to be the designated State of Design under Annex 8.	PR.SLC.0001	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		EASA procedure PR.SLC.0001 requires to inform all affected States via issuing an AD.
1.7.1	1.7 Transfer of Type Certificate 1.7.1 The State of Design shall establish procedures for the transfer of a Type Certificate that ensures continued compliance of the approved design of the aircraft, engine or propeller type with the appropriate airworthiness requirements	21L.A.29 Annex Ib Reg (EU) 748/2012 21.A.47Reg. (EU) 748/2012Article 68	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
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	<p>a) for a transfer in which the State of Design remains the same; and</p> <p>b) for a transfer in which the State of Design changes to another Contracting State.</p>	(1)Article 90 (2)Reg. (EU) 2018/1139								
1.7.2	1.7.2 The State of Design shall, upon completion of the transfer, issue or reissue its Type Certificate in accordance with 1.4.1 of this part.	21L.A.29 Annex Ib Reg (EU) 748/2012 21.A.47Reg. (EU) 748/2012Article 68 (1)Article 90 (2)Reg. (EU) 2018/1139	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.7.3	1.7.3 Where the State of Manufacture of an aircraft, engine or propeller is not the State of Design, there shall be an agreement or arrangement in accordance with 2.4.5 and 4.2.3 of this part.	21L.A.5 Annex Ib Reg (EU) 748/2012 21.A.4 Reg. (EU) 748/2012Article 68 (1)Article 90 (2)Reg. (EU) 2018/1139	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.7.4	1.7.4 The State of Design shall notify all Contracting States of the transfer and the organization responsible for the type design for purposes of the	EASA Certification Handbook	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	continuing airworthiness reporting requirements under Chapter 4 of this part. <i>Note.— Guidance material on the process for transfer of a Type Certificate is contained in Doc 9760.</i>	14.2								
2.1	CHAPTER 2. PRODUCTION 2.1 Applicability Until 25 November 2026, the Standards of this chapter are applicable to the production of all aircraft, engines, propellers and associated parts.	Reg. (EU) 2018/1139. Subparts F, G, Reg. (EU) 748/2012 Subpart G Annex Ib Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1	2.1 Applicability As of 26 November 2026, the Standards of this chapter are applicable to the production of all aircraft, remote pilot stations, engines, propellers and associated parts.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.2	2.2 Aircraft, engine and propeller production1 Until 25 November 2026, the State of Manufacture shall ensure that each aircraft, engine or propeller, including	Subparts F, G Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	associated parts manufactured by sub-contractors and/or suppliers, is airworthy at the time of release.	Subpart G Annex Ib Reg (EU) 748/2012								
2.2	2.2 Aircraft, engine and propeller production1 As of 26 November 2026, the State of Manufacture shall ensure that each aircraft, remote pilot station, engine or propeller, including associated parts manufactured by sub-contractors and/or suppliers, is airworthy at the time of release.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.3	2.3 Aircraft parts production The Contracting State taking responsibility for the production of aircraft parts manufactured under the design approval referred to in 1.3.5 of this part shall ensure that the aircraft parts are airworthy.	Subparts F, G Reg. (EU) 748/2012 Subpart G Annex Ib Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.1	2.4 Production approval 2.4.1 Until 25 November 2026, when approving production of an aircraft, engine, propeller or associated part, the Contracting State having jurisdiction over the organization responsible for production shall: a) examine the supporting data and inspect the production facilities and processes so as to determine that the manufacturing organization is in compliance with the appropriate production requirements; and	Subpart G Annex Ib Reg (EU) 748/2012 Subparts F, G Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) ensure that the manufacturing organization has established and can maintain a quality system or a production inspection system such as to guarantee that each aircraft, engine, propeller or associated part produced by the organization or by sub-contractors and/or suppliers is airworthy at the time of release.</p> <p><i>Note 1.— Normally, the oversight of production is facilitated by approving the manufacturing organization.</i></p> <p><i>Note 2.— Where the State of Manufacture is a State other than the Contracting State where the associated parts are produced, there may be an agreement or arrangement acceptable to both States to support the oversight responsibilities of the State of Manufacture over the organizations manufacturing the associated parts.</i></p> <p>-----</p> <p>1. As of 26 November 2026, paragraph 2.2 will be titled “Aircraft, remote pilot station, engine and propeller production”.</p>								
2.4.1	<p>2.4.1 As of 26 November 2026, when approving production of an aircraft, remote pilot station, engine, propeller or associated part, the Contracting State having jurisdiction over the organization responsible for production shall:</p>		<input type="checkbox"/>						

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				Level of implementation of SARPs						
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	<p>a) examine the supporting data and inspect the production facilities and processes so as to determine that the manufacturing organization is in compliance with the appropriate production requirements; and</p> <p>b) ensure that the manufacturing organization has established and can maintain a quality system or a production inspection system such as to guarantee that each aircraft, remote pilot station, engine, propeller or associated part produced by the organization or by sub-contractors and/or suppliers is airworthy at the time of release.</p> <p><i>Note 1.— Normally, the oversight of production is facilitated by approving the manufacturing organization.</i></p> <p><i>Note 2.— Where the State of Manufacture is a State other than the Contracting State where the associated parts are produced, there may be an agreement or arrangement acceptable to both States to support the oversight responsibilities of the State of Manufacture over the organizations manufacturing the associated parts.</i></p>									
2.4.2	<p>2.4.2 Recommendation.— <i>A Contracting State should balance risks and rigor when approving production of aircraft or aircraft parts based on the acceptable level of risk determined for the product as specified by the State of Design.</i></p> <p><i>Note.— For the production approval of Part VB aeroplanes and their parts, guidance material addressing</i></p>	21.B.230Reg. (EU) 748/2012 21L.B.143, 21L.B.144 Annex Ib Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>how States may balance risks and rigor in the determination of compliance is contained in the Airworthiness Manual (Doc 9760).</i>	748/2012								
2.4.3	2.4.3 Until 25 November 2026, the manufacturing organization shall hold, for each aircraft, engine, propeller or associated part, a design approval as referred to in 1.3 of this part, or the right of access under an agreement or arrangement to the approved design data relevant for production purposes.	21.A.4, 122(b), 133(c), 165(c)(2) Reg. (EU) 748/2012 21L.A.5 21L.A.122 Annex Ib Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3	2.4.3 As of 26 November 2026, the manufacturing organization shall hold, for each aircraft, remote pilot station, engine, propeller or associated part, a design approval as referred to in 1.3 of this part, or the right of access under an agreement or arrangement to the approved design data relevant for production purposes.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.4.4	2.4.4 Until 25 November 2026, records shall be maintained such that the origin of each aircraft, engine, propeller and associated part, and its identification with the approved design and production data, can be established. <i>Note.— The origin of an aircraft, engine, propeller and associated part refers to the manufacturer.</i>	Subparts F, G, Q Reg. (EU) 748/2012 21L.A.7 Annex Ib Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>the date of manufacture, the serial number or other information that can be tracked to its production record.</i>	748/2012								
2.4.4	<p>2.4.4 As of 26 November 2026, records shall be maintained such that the origin of each aircraft, remote pilot station, engine, propeller and associated part, and its identification with the approved design and production data, can be established.</p> <p><i>Note.— The origin of an aircraft, remote pilot station, engine, propeller and associated part refers to the manufacturer, the date of manufacture, the serial number or other information that can be tracked to its production record.</i></p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.4.5	<p>2.4.5 Until 25 November 2026, where the State of Manufacture is not the State of Design, there shall be an agreement or arrangement acceptable to both States to:</p> <p>a) ensure that the manufacturing organization has the right of access to the approved design data relevant for production purposes;</p> <p>b) address the responsibilities of each State with regard to design, manufacture and continuing airworthiness of the aircraft, engine or propeller during the period of the agreement or arrangement, including such period when the State of Design takes action to suspend in whole or in part the Type Certificate of the affected aircraft type; and</p>	<p>21.A.4Reg. (EU) 748/2012Art. 68 (1)Article 90 (2)Reg. (EU) 2018/1139</p> <p>21L.A.5 Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	c) terminate the production approval under this part when the State of Design revokes the Type Certificate corresponding to that aircraft type.									
2.4.5	<p>2.4.5 As of 26 November 2026, where the State of Manufacture is not the State of Design, there shall be an agreement or arrangement acceptable to both States to:</p> <p>a) ensure that the manufacturing organization has the right of access to the approved design data relevant for production purposes;</p> <p>b) address the responsibilities of each State with regard to design, manufacture and continuing airworthiness of the aircraft, remote pilot station, engine or propeller during the period of the agreement or arrangement, including such period when the State of Design takes action to suspend in whole or in part the Type Certificate of the affected aircraft type; and</p> <p>c) terminate the production approval under this part when the State of Design revokes the Type Certificate corresponding to that aircraft type.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
3.1	CHAPTER 3. CERTIFICATE OF AIRWORTHINESS	Reg. (EU) 2018/1139, Reg. (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p><i>Note.— The Certificate of Airworthiness as used in these Standards is the Certificate of Airworthiness referred to in Article 31 of the Convention.</i></p> <p>3.1 Applicability</p> <p>The Standards of this chapter are applicable in respect of all aircraft, except 3.3 and 3.4 which are not applicable in respect of all aircraft that are of a type of which the prototype was submitted to appropriate national authorities for certification before 13 June 1960.</p>	748/2012								
3.2.1	<p>3.2 Eligibility, issuance and continued validity of a Certificate of Airworthiness</p> <p>3.2.1 A Certificate of Airworthiness shall be issued by a Contracting State on the basis of satisfactory evidence that the aircraft complies with the design aspects of the appropriate airworthiness requirements.</p>	<p>Subpart H, Reg. (EU) 748/2012</p> <p>Subpart H Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.2.2	<p>3.2.2 As of 26 November 2026, the Certificate of Airworthiness issued to a remotely piloted aircraft shall convey evidence of the airworthy status of the remotely piloted aircraft system (RPAS), as a complete system, to ensure it conforms to the type design and is in a condition for safe operation.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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3.2.2	<p>3.2.21 A Contracting State shall issue or render valid a Certificate of Airworthiness for which it intends to claim recognition pursuant to Article 33 of the Convention on International Civil Aviation when it has satisfactory evidence that the aircraft complies with the applicable Standards of this Annex through compliance with appropriate airworthiness requirements.</p> <p><i>Note.— Some Contracting States facilitate the issuance of a “Special Certificate of Airworthiness” or similar to denote that an aircraft does not meet the Standards of Annex 8. While not valid for the purpose of international flight, such a document provides conditions and limitations that may be required by other Contracting States for the purpose of granting approvals to fly within or through their jurisdiction.</i></p> <p>-----</p> <p>1. As of 26 November 2026, paragraphs 3.2.2, 3.2.3, 3.2.4 and 3.2.5 are renumbered.</p>	<p>Subpart HReg. (EU) 748/2012</p> <p>Subpart H Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.2.3	<p>3.2.3^{Error! Bookmark not defined.} A Certificate of Airworthiness shall be renewed or shall remain valid, subject to the laws of the State of Registry, provided that the State of Registry shall require that the continuing airworthiness of the aircraft shall be determined by a periodical inspection at appropriate intervals having regard to lapse of time and type of service or, alternatively, by means of a system of inspection, approved by the State, that will produce at least an equivalent result.</p>	<p>Part M, Subpart I, and Part ML Subpart I, Reg. (EU) 1321/2014</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	----- 1. As of 26 November 2026, paragraphs 3.2.2, 3.2.3, 3.2.4 and 3.2.5 are renumbered.										
3.2.4	<p>3.2.4^{Error! Bookmark not defined.} When an aircraft possessing a valid Certificate of Airworthiness issued by a Contracting State is entered on the register of another Contracting State, the new State of Registry, when issuing its Certificate of Airworthiness may consider the previous Certificate of Airworthiness as satisfactory evidence, in whole or in part, that the aircraft complies with the applicable Standards of this Annex through compliance with the appropriate airworthiness requirements.</p> <p><i>Note.— Some Contracting States facilitate the transfer of aircraft onto the register of another State by the issue of an “Export Certificate of Airworthiness” or similarly titled document. While not valid for the purpose of flight, such a document provides confirmation by the exporting State of a recent satisfactory review of the airworthiness status of the aircraft. Guidance material on the issue of an “Export Certificate of Airworthiness” is contained in the Airworthiness Manual (Doc 9760).</i></p> <p>----- 1. As of 26 November 2026, paragraphs 3.2.2, 3.2.3, 3.2.4 and 3.2.5 are renumbered.</p>	<p>Subpart H, Reg. (EU) 748/2012</p> <p>Subpart H Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>							
3.2.5	3.2.5 ² When a State of Registry renders valid a Certificate of Airworthiness issued by another Contracting State, as an alternative to issuance of its own		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>		There is no mechanism of	

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	<p>Certificate of Airworthiness, it shall establish validity by suitable authorization to be carried with the former Certificate of Airworthiness accepting it as the equivalent of the latter. The validity of the authorization shall not extend beyond the period of validity of the Certificate of Airworthiness being rendered valid. The State of Registry shall ensure that the continuing airworthiness of the aircraft is determined in accordance with 3.2.3^{Error!} <small>Bookmark not defined.</small></p> <p>-----</p> <p>2. As of 26 November 2026, paragraphs 3.2.2, 3.2.3, 3.2.4 and 3.2.5 are renumbered.</p>								rendering the CofA valid.	
3.3.1	<p>3.3 Standard form of Certificate of Airworthiness</p> <p>3.3.1 Until 25 November 2026, the Certificate of Airworthiness shall contain the information shown in Figure 1 and shall be generally similar to it.</p>	Part 21 Appendix VI (EASA Form 25)21.B.80Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.3.1	<p>3.3.1 As of 26 November 2026, the Certificate of Airworthiness for all aircraft except remotely piloted aircraft (RPA) shall contain the information shown in Figure 1 and shall be generally similar to it.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
3.3.1.1	<p>3.3.1.1 As of 26 November 2026, the Certificate of Airworthiness for all RPA shall contain the</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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				Level of implementation of SARPs						
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	information shown in Figure 2 and shall be generally similar to it.									
3.3.2	<p>3.3.2 When Certificates of Airworthiness are issued in a language other than English, they shall include an English translation.</p> <p><i>Note.— Article 29 of the Convention on International Civil Aviation requires that the Certificate of Airworthiness be carried on board every aircraft engaged in international air navigation.</i></p>	<p>21.A.175 Reg. (EU) 748/2012 and Appendix VI</p> <p>Appendices to Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>		Introduction to Part 21 Appendices requires to have at least an English translation.				
3.4	<p>3.4 Aircraft limitations and information</p> <p>Each aircraft shall be provided with a flight manual, placards or other documents stating the approved limitations within which the aircraft is considered airworthy as defined by the appropriate airworthiness requirements and additional instructions and information necessary for the safe operation of the aircraft.</p> <p><i>Note. — As of 26 November 2026, information necessary for the safe operation of the RPA include those applicable to remote pilot stations (RPS) and C2 Link.</i></p>	<p>CS 23/27/29. 1501-1589. CS 25.1501-1591</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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3.5	<p>3.5 Temporary loss of airworthiness</p> <p>Any failure to maintain an aircraft in an airworthy condition as defined by the appropriate airworthiness requirements shall render the aircraft ineligible for operation until the aircraft is restored to an airworthy condition.</p> <p><i>Note. — As of 26 November 2026, for remotely piloted aircraft, that which must be restored to an airworthy condition includes the RPS controlling the RPA, the required C2 Link(s) or any other components defined by the appropriate airworthiness requirements.</i></p>	M.A.902 and ML.A.902 Reg. (EU) 1321/2014.	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
3.6.1	<p>3.6 Damage to aircraft</p> <p>3.6.1 When an aircraft has sustained damage, the State of Registry shall judge whether the damage is of a nature such that the aircraft is no longer airworthy as defined by the appropriate airworthiness requirements.</p>	<p>21.A.445 Reg. (EU) 748/2012, M.A.403 and ML.A.403 Reg. (EU) 1321/2014.</p> <p>21L.A.211 Annex Ib Reg (EU) 748/2012</p>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Assessment also allowed by DOA under procedure agreed with the Agency.	Safety is assured through the DOA concept.
3.6.2	<p>3.6.2 If the damage is sustained or ascertained when the aircraft is in the territory of another Contracting State, the authorities of the other Contracting State shall be entitled to prevent the aircraft from resuming its flight on the condition that they shall advise the State of</p>	Art. 62.2(c) Reg. (EU) 2018/1139	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	Registry immediately, communicating to it all details necessary to formulate the judgement referred to in 3.6.1.																
3.6.3	3.6.3 When the State of Registry considers that the damage sustained is of a nature such that the aircraft is no longer airworthy, it shall prohibit the aircraft from resuming flight until it is restored to an airworthy condition. The State of Registry may, however, in exceptional circumstances, prescribe particular limiting conditions to permit the aircraft to fly a non-commercial air transport operation to an aerodrome at which it will be restored to an airworthy condition. In prescribing particular limiting conditions, the State of Registry shall consider all limitations proposed by the Contracting State that had originally, in accordance with 3.6.2, prevented the aircraft from resuming its flight. That Contracting State shall permit such flight or flights within the prescribed limitations.	21.A.445 Subpart P Reg. (EU) 748/2012 21L.A.211 Annex Ib Reg (EU) 748/2012	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EASA Permit to Fly (including flight conditions) may be issued by an approved DOA.								
3.6.4	3.6.4 When the State of Registry considers that the damage sustained is of a nature such that the aircraft is still airworthy, the aircraft shall be allowed to resume its flight.	21.A.445 Reg. (EU) 748/2012 21L.A.211 Annex Ib Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>									
	<table border="1" style="width: 100%; border-collapse: collapse;"> <tr> <td style="width: 15%; text-align: center;">*</td> <td style="width: 60%; text-align: center;"><i>State of Registry Issuing Authority</i></td> <td style="width: 25%;"></td> </tr> <tr> <td colspan="3" style="text-align: center;">CERTIFICATE OF AIRWORTHINESS</td> </tr> <tr> <td style="text-align: center;">1. Nationality and</td> <td style="text-align: center;">2. Manufacturer and manufacturer's</td> <td style="text-align: center;">3. Aircraft serial number</td> </tr> </table>	*	<i>State of Registry Issuing Authority</i>		CERTIFICATE OF AIRWORTHINESS			1. Nationality and	2. Manufacturer and manufacturer's	3. Aircraft serial number							
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<p>registration marks</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p>	<p>designa tion of aircraft**</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p>	<p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p> <p>.....</p>								
4. Categories and/or operation***										
<p>5. This Certificate of Airworthiness is issued pursuant to the Convention on International Civil Aviation dated 7 December 1944 and †..... in respect of the above-mentioned aircraft which is considered to be airworthy when maintained and operated in accordance with the foregoing and the pertinent operating limitations.</p> <p>Date of issue Signature</p> <p>† Insert reference to appropriate Airworthiness Code.</p>										
6. ****										
* For use of the State of Registry.										

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	<p>** Manufacturer's designation of aircraft should contain the aircraft type and model.</p> <p>*** This space is normally used to indicate the certification basis, i.e. certification code, with which the particular aircraft complies and/or its permitted operational category, e.g. commercial air transportation, aerial work or private.</p> <p>**** This space shall be used either for periodic endorsement (giving date of expiry) or for a statement that the aircraft is being maintained under a system of continuous inspection.</p> <p>Figure 1</p> <table border="1" style="width: 100%; border-collapse: collapse;"> <tr> <td style="width: 15%; text-align: center;">*</td> <td style="text-align: center;"><i>State of Registry Issuing Authority</i></td> <td style="width: 15%;"></td> </tr> <tr> <td colspan="3" style="text-align: center;">CERTIFICATE OF AIRWORTHINESS — RPA</td> </tr> <tr> <td style="text-align: center;">1. Nationality and</td> <td style="text-align: center;">2. Manufacturer and manufacturer's designation of remotely</td> <td style="text-align: center;">3. Remotely piloted aircraft</td> </tr> </table>	*	<i>State of Registry Issuing Authority</i>		CERTIFICATE OF AIRWORTHINESS — RPA			1. Nationality and	2. Manufacturer and manufacturer's designation of remotely	3. Remotely piloted aircraft							
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<p>considered to be airworthy when maintained and operated in accordance with the foregoing and the pertinent operating limitations.</p> <p>Date of issue Signature</p> <p>† Insert reference to appropriate Airworthiness Code.</p>										
<p>8. *****</p>										
<p>* For use of the State of Registry.</p> <p>** Manufacturer's designation of remotely piloted aircraft (RPA) should contain the RPA type and model.</p> <p>*** This space should contain identification of the C2 Link(s) permitted for the command and control of the RPA to comply with the certification code and to operate in accordance with specified operating limitations.</p> <p>**** This space is normally used to indicate the certification basis, i.e. certification code, with which the particular RPA complies and/or its permitted operational category, e.g. commercial air transportation, aerial work or private.</p> <p>***** This space shall be used either for periodic endorsement (giving date of expiry) or for a statement that the RPA</p>										

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	<p>is being maintained under a system of continuous inspection.</p> <p style="text-align: center;">Figure 23</p> <p style="text-align: center;">-----</p> <p>3 As of 26 November 2026, <i>insert</i> new Figure 2 in Annex 8, Part II, end of Chapter 3.</p>									
4.1	<p>CHAPTER 4. CONTINUING AIRWORTHINESS</p> <p style="text-align: center;">4.1 Applicability</p> <p>Until 25 November 2026, the Standards of this chapter are applicable to all aircraft, engines, propellers and associated parts.</p>	Aviation Regulations «Continuous Airworthiness process». paragraph 1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1	<p style="text-align: center;">4.1 Applicability</p> <p>As of 26 November 2026, the Standards of this chapter are applicable to all aircraft, remote pilot stations, engines, propellers and associated parts.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.2.1.1	<p style="text-align: center;">4.2.1 State of Design</p> <p>4.2.1.1 Until 25 November 2026, the State of Design of an aircraft shall:</p>	21L.A.4, 21L.A.5, 21L.B.43 Annex Ib Reg	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>a) transmit to every Contracting State which has in accordance with 4.2.4.1 a) advised the State of Design of the aircraft that it has entered the aircraft on its register, and to any other Contracting State upon request, any generally applicable information which it has found necessary for the continuing airworthiness and safe operation of the aircraft, including any engines and propellers (hereinafter called mandatory continuing airworthiness information), and notification of the suspension or revocation of a Type Certificate;</p> <p><i>Note 1.— The term “mandatory continuing airworthiness information” is intended to include mandatory requirements for modification, replacement of parts or inspection of aircraft and amendment of operating limitations and procedures. Among such information is that issued by Contracting States in the form of airworthiness directives.</i></p> <p><i>Note 2.— The Continuing Airworthiness of Aircraft in Service (Cir 95) provides the necessary information to assist Contracting States in establishing contact with competent authorities of other Contracting States for the purpose of maintaining continuing airworthiness of aircraft in service.</i></p> <p><i>Note 3.— If the State of Design of the aircraft is satisfied that mandatory continuing airworthiness information previously issued by the State of Design of the engine or propeller under 4.2.1.2 fully addresses a continuing airworthiness issue, then the State of Design of the aircraft need not retransmit that</i></p>	<p>(EU) 748/2012 Art. 77.1 (h)Reg. (EU) 2018/113921. A.3B, 4 Reg. (EU) 748/2012Reg. (EU) 2015/64021.B .80Reg. (EU) 748/2012 CS-23CS-25CS-27CS 29.1529 CS-29 Appendix ACS 25.571</p>							
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	<p><i>information to Contracting States that have already been informed.</i></p> <p>b) ensure that, in respect of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass, there exists a system for:</p> <p>i) receiving information submitted in accordance with 4.2.4.1 g);</p> <p>ii) deciding if and when airworthiness action is needed;</p> <p>iii) developing the necessary airworthiness actions; and</p> <p>iv) promulgating the information on those actions including that required in 4.2.1.1 a);</p> <p>c) ensure that, in respect of aeroplanes over 5 700 kg maximum certificated take-off mass, there exists a continuing structural integrity programme to ensure the airworthiness of the aeroplane. The programme shall include specific information concerning corrosion prevention and control.</p>								
4.2.1.1	<p><i>4.2.1.1</i> As of 26 November 2026, the State of Design of an aircraft shall:</p> <p>a) transmit to every Contracting State which has in accordance with 4.2.4.1 a) advised the State of Design of the aircraft that it has entered the aircraft on its register, and to any other Contracting State upon</p>		<input type="checkbox"/>						

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	<p>request, any generally applicable information which it has found necessary for the continuing airworthiness and safe operation of the aircraft, including any remote pilot stations, engines and propellers (hereinafter called mandatory continuing airworthiness information);</p> <p><i>Note 1.— The term “mandatory continuing airworthiness information” is intended to include mandatory requirements for modification, replacement of parts or inspection of aircraft and amendment of operating limitations and procedures. Among such information is that issued by Contracting States in the form of airworthiness directives.</i></p> <p><i>Note 2.— The Continuing Airworthiness of Aircraft in Service (Cir 95) provides the necessary information to assist Contracting States in establishing contact with competent authorities of other Contracting States for the purpose of maintaining continuing airworthiness of aircraft in service.</i></p> <p><i>Note 3.— If the State of Design of the aircraft is satisfied that mandatory continuing airworthiness information previously issued by the State of Design of the remote pilot station, engine or propeller under 4.2.1.2 fully addresses a continuing airworthiness issue, then the State of Design of the aircraft need not retransmit that information to Contracting States that have already been informed.</i></p> <p>b) ensure that, in respect of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass, there exists a system for:</p>								
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Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference				Not Applicable	Details of Difference	Remarks	
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>i) receiving information submitted in accordance with 4.2.4.1 g);</p> <p>ii) deciding if and when airworthiness action is needed;</p> <p>iii) developing the necessary airworthiness actions; and</p> <p>iv) promulgating the information on those actions including that required in 4.2.1.1 a);</p> <p>c) ensure that, in respect of aeroplanes over 5 700 kg maximum certificated take-off mass, there exists a continuing structural integrity programme to ensure the airworthiness of the aeroplane. The programme shall include specific information concerning corrosion prevention and control.</p> <p>d) as of 26 November 2026, ensure that, in respect of remotely piloted aeroplanes and remotely piloted helicopters, there exists a system for:</p> <p>i) receiving information submitted in accordance with 4.2.4.1 g);</p> <p>ii) deciding if and when airworthiness action is needed;</p> <p>iii) developing the necessary airworthiness actions; and</p>							
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Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>iv) promulgating the information on those actions including that required in 4.2.1.1 a)</p> <p>e) as of 26 November 2026, ensure that, in respect of remotely piloted aeroplanes and remotely piloted helicopters, there exists a continuing structural integrity programme appropriate to its mass and operational category to ensure the airworthiness of the remotely piloted aeroplane or the remotely piloted helicopter. The programme, when appropriate, shall include specific information concerning corrosion prevention and control.</p>									
4.2.1.2	<p>4.2.1.2 Until 25 November 2026, the State of Design of an engine or a propeller, where it is different from the State of Design of the aircraft, shall:</p> <p>a) transmit any continuing airworthiness information to the State of Design of the aircraft and to any other Contracting State upon request;</p> <p><i>Note.— While the overall responsibility for the transmission of mandatory continuing airworthiness information rests with the State of Design of the aircraft, it is recognized that some States of Design of the engine or propeller transmit mandatory continuing airworthiness information directly to States of Registry and other Contracting States. This practice has the benefit of speeding up the availability of mandatory continuing airworthiness information and processing this information in the normal way in accordance</i></p>	<p>Art. 77.1(h) Reg.(EU)2018/1139 and 21.B.45Reg.(EU) 748/2012</p> <p>21L.B.11, 21L.B.12 Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	exchange of information required for all data relevant for safety of the products, parts and appliances					

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Sigini ficant Difference			
				Level of implementation of SARPs						
				A) More Exactin g or Exceed s	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p><i>with 4.2.4.1 d). However, if the State of Design of the aircraft subsequently transmits additional mandatory continuing airworthiness information to that of the State of Design of the engine or propeller, then the mandatory continuing airworthiness information originating from the State of Design of the aircraft must take precedence in case of incompatibility.</i></p> <p>b) ensure that, in respect of engines and propellers installed on aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass, there exists a system for:</p> <p>i) receiving information submitted in accordance with 4.2.4.1 g);</p> <p>ii) deciding if and when airworthiness action is needed; and</p> <p>iii) developing the necessary airworthiness actions.</p>								
4.2.1.2	<p><i>4.2.1.2 As of 26 November 2026, the State of Design of a remote pilot station, engine or a propeller, where it is different from the State of Design of the aircraft, shall:</i></p> <p>a) transmit any continuing airworthiness information to the State of Design of the aircraft and to any other Contracting State upon request;</p>		<input type="checkbox"/>						

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference				Not Applicable	Details of Difference	Remarks	
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p><i>Note.— While the overall responsibility for the transmission of mandatory continuing airworthiness information rests with the State of Design of the aircraft, it is recognized that some States of Design of the remote pilot station, engine or propeller transmit mandatory continuing airworthiness information directly to States of Registry and other Contracting States. This practice has the benefit of speeding up the availability of mandatory continuing airworthiness information and processing this information in the normal way in accordance with 4.2.4.1 d). However, if the State of Design of the aircraft subsequently transmits additional mandatory continuing airworthiness information to that of the State of Design of the remote pilot station, engine or propeller, then the mandatory continuing airworthiness information originating from the State of Design of the aircraft must take precedence in case of incompatibility.</i></p> <p>b) ensure that, in respect of engines and propellers installed on remotely piloted aeroplanes, remotely piloted helicopters, aeroplanes over 5 700 kg, helicopters over 3 175 kg maximum certificated take-off mass and remote pilot stations integrated with remotely piloted aeroplanes and remotely piloted helicopters, there exists a system for:</p>								
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>i) receiving information submitted in accordance with 4.2.4.1 g);</p> <p>ii) deciding if and when airworthiness action is needed; and</p> <p>iii) developing the necessary airworthiness actions.</p>									
4.2.1.3	4.2.1.3 As of 26 November 2026, where the State of Design of a modification is different from the State of Design of the aircraft, remote pilot station, engine or propeller being modified, the State of Design of the modification shall transmit the mandatory continuing airworthiness information to all States that have the modified aircraft on their registries.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.2.1.4	<p>4.2.1.4 Until 25 November 2026, where, for a given aircraft, engine or propeller, the State of Manufacture is not the State of Design, then the State of Design shall ensure that there is an agreement acceptable to both States to ensure that the manufacturing organization cooperates with the organization responsible for the type design in assessing information on the design, manufacture and operation of the aircraft, engine or propeller.</p> <p><i>Note.— Guidance material on the interpretation of “the organization responsible for type design” is contained in Doc 9760.</i></p>	<p>21.A.4, 21.A.122, 21.A.129, 21.A.133 Reg. (EU) 748/2012.</p> <p>21L.A.5 21L.A.122 21L.A.127 Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.2.1.4	4.2.1.4 As of 26 November 2026, where, for a given aircraft, remote pilot station, engine or propeller,		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>the State of Manufacture is not the State of Design, then the State of Design shall ensure that there is an agreement acceptable to both States to ensure that the manufacturing organization cooperates with the organization responsible for the type design in assessing information on the design, manufacture and operation of the aircraft, remote pilot station, engine or propeller.</p> <p><i>Note.— Guidance material on the interpretation of “the organization responsible for type design” is contained in Doc 9760.</i></p>									
4.2.1.5	<p>4.2.1.5 The State of Design shall ensure that sensitive aviation security information is not transmitted when distributing mandatory continuing airworthiness information.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Process is not established.	
4.2.1.6	<p>4.2.1.6 The State of Design shall ensure that sensitive aviation security information is securely transmitted to the appropriate authority in the States of Registry in accordance with Annex 17 — <i>Aviation Security — Safeguarding International Civil Aviation against Acts of Unlawful Interference.</i></p> <p><i>Note.— Guidance material on the secure transmission of sensitive aviation security information is contained in Doc 9760.</i></p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Process is not established.	

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

4.2.2.1	<p>4.2.2 State of Design of Modification</p> <p>4.2.2.1 Where the State of Design of Modification is the same as the State of Design of the aircraft, engine or propeller, the State shall follow the instruction in 4.2.1.</p>	See references for Section 4.2.1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.2.2.2	<p>4.2.2.2 Where the State of Design of Modification is different from the State of Design of the aircraft, engine or propeller, the State of Design of Modification shall:</p> <ol style="list-style-type: none"> 1. have a system to make available to every Contracting State, any mandatory continuing airworthiness information related to the modification or repair; 2. ensure that, in respect of the modification or repair of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass, there exists a system for: <ol style="list-style-type: none"> 1. receiving information submitted in accordance with 4.2.4.1 g); 2. deciding if and when airworthiness action is needed; 3. developing the necessary airworthiness actions; 4. promulgating the information on 	Reg. (EU) 748/2012 - Annex I (Part 21): 21.A.3A, 21.A.3B, Annex Ib: 21L.A.3, 21L.A.4, (21L.B.23)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	5. those actions, including that required in 4.2.2.2 a); and ensure that all mandatory continuing airworthiness information which it, as the State of Design of Modification, originated in respect of that aircraft, engine or propeller is transmitted to the appropriate State of Design.									
4.2.2.3	4.2.2.3 Where the State of Design of Modification is different from the Contracting State having jurisdiction over the organization responsible for manufacturing of the modification or repair parts, the State of Design of Modification shall ensure that there is an agreement acceptable to both States to ensure that the manufacturing organization cooperates with the organization responsible for the design of the modification or repair in assessing information received on experience with operating the aircraft.	Reg(EU)No748/2012 – Annex I: 21.A.4, Annex Ib: 21L.A.5	<input checked="" type="checkbox"/>	<input type="checkbox"/>		All holders of Design approvals issued by EASA are bound by 21.A.4 (and 21L.A.5), which requires the establishment of a DO/PO arrangement, governing the				

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				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

									coordination between the Design Organisation (DO) and the manufacturing organisation (PO), regardless of the location of the manufacturing organisation. The acceptance of the DO/PO arrangement by the State having jurisdiction over the
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

										manufacturing organization is formalized either by the existence of a bilateral agreement, a working arrangement or the unilateral acceptance of EASA designs
4.2.3	4.2.3 State of Manufacture Until 25 November 2026, the State of Manufacture shall ensure that where it is not the State of Design there is an agreement acceptable to both States to ensure that the manufacturing organization cooperates with the organization responsible for the type design in assessing information on the design, manufacture and operation of the aircraft, engine or propeller.	21.A.4, 129, 165, 133(c) Reg. (EU) 748/2012 21L.A.5 21L.A.122 21L.A.127 Annex Ib Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

		748/2012								
4.2.3	<p>4.2.3 State of Manufacture</p> <p>As of 26 November 2026, the State of Manufacture shall ensure that where it is not the State of Design there is an agreement acceptable to both States to ensure that the manufacturing organization cooperates with the organization responsible for the type design in assessing information on the design, manufacture and operation of the aircraft, remote pilot station, engine or propeller.</p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.2.4.1	<p>4.2.4 State of Registry</p> <p>4.2.4.1 Until 25 November 2026, the State of Registry shall:</p> <p>a) ensure that, when it first enters on its register an aircraft of a particular type for which it is not the State of Design and issues or validates a Certificate of Airworthiness in accordance with 3.2 of this part, it shall advise the State of Design that it has entered such an aircraft on its register;</p> <p>b) determine the continuing airworthiness of an aircraft in relation to the appropriate airworthiness requirements in force for that aircraft;</p> <p>c) develop or adopt requirements to ensure the continuing airworthiness of the aircraft during its service life, including requirements to ensure that the aircraft:</p>	<p>21.A.3, 21.B.60, Reg. (EU) 748/2012. M.A.202 145.A.60 Reg. (EU) 1321/2014</p> <p>21L.A.3 21L.B.23 Annex Ib Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
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	<p>i) continues to comply with the appropriate airworthiness requirements after a modification, a repair or the installation of a replacement part; and</p> <p>ii) is maintained in an airworthy condition and in compliance with the maintenance requirements of Annex 6 — <i>Operation of Aircraft</i>, and where applicable, Parts III, IV, V, VI and VII of this Annex;</p> <p>d) upon receipt of mandatory continuing airworthiness information from the State of Design, adopt the mandatory information directly or assess the information received and take appropriate action;</p> <p>e) have a system to monitor and obtain mandatory continuing airworthiness information from the State of Design of Modification, where the State of Design of Modification is different from the State of Registry, and adopt the mandatory information directly or assess the information received and take appropriate action;</p> <p>f) ensure that all mandatory continuing airworthiness information which it, as the State of Registry, originated in respect of that aircraft, is transmitted to the appropriate State of Design and State of Design of Modification; and</p> <p>g) ensure that, in respect of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum</p>								
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	certificated take-off mass, there exists a system whereby information on faults, malfunctions, defects and other occurrences that cause or might cause adverse effects on the continuing airworthiness of the aircraft is transmitted to the organization responsible for the type design of that aircraft. Whenever this information relates to an engine or propeller, such information shall be transmitted to both the organization responsible for engine or propeller type design and the organization responsible for aircraft type design. Where a continuing airworthiness safety issue is associated with a modification or repair, the State of Registry shall ensure that there exists a system whereby the above information is transmitted to the individual or organization responsible for the design of the modification or repair.									
4.2.4.1	<p>4.2.4.1 As of 26 November 2026, the State of Registry shall:</p> <p>a) ensure that, when it first enters on its register an aircraft of a particular type for which it is not the State of Design and issues or validates a Certificate of Airworthiness in accordance with 3.2 of this part, it shall advise the State of Design that it has entered such an aircraft on its register;</p> <p>b) determine the continuing airworthiness of an aircraft in relation to the appropriate airworthiness requirements in force for that aircraft;</p> <p>c) develop or adopt requirements to ensure the continuing airworthiness of the aircraft during its</p>		<input type="checkbox"/>							

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
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	<p>service life, including requirements to ensure that the aircraft:</p> <p>i) continues to comply with the appropriate airworthiness requirements after a modification, a repair or the installation of a replacement part; and</p> <p>ii) is maintained in an airworthy condition and in compliance with the maintenance requirements of Annex 6 — <i>Operation of Aircraft</i>, and where applicable, Parts III, IV, V, VI, VII, VIII, IX and X of this Annex;</p> <p>d) upon receipt of mandatory continuing airworthiness information from the State of Design, adopt the mandatory information directly or assess the information received and take appropriate action;</p> <p>e) ensure that all mandatory continuing airworthiness information which it, as the State of Registry, originated in respect of that aircraft, is transmitted to the appropriate State of Design; and</p> <p>f) ensure that, in respect of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass, remotely piloted aeroplanes and remotely piloted helicopters there exists a system whereby information on faults, malfunctions, defects and other occurrences that cause or might cause adverse effects on the continuing airworthiness of the aircraft is transmitted to the organization responsible for the type design of that aircraft. Whenever this information relates</p>								
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Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
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	to a remote pilot station, an engine or propeller, such information shall be transmitted to both the organization responsible for remote pilot station, engine or propeller type design and the organization responsible for aircraft type design. Where a continuing airworthiness safety issue is associated with a modification, the State of Registry shall ensure that there exists a system whereby the above information is transmitted to the organization responsible for the design of the modification.									
4.2.4.2	<p>4.2.4.2 When approving a maintenance organization or accepting the approval of a maintenance organization issued by another Contracting State, the State of Registry shall ensure compliance with the Standards of Chapter 6 of this part.</p> <p><i>Note.— Chapter 6 provides requirements for accepting the approval of a maintenance organization issued by another Contracting State.</i></p>	145.A.10M.A.601CAO.A.015Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.2.4.3	4.2.4.3 The State of Registry shall ensure that sensitive aviation security information is not transmitted when distributing mandatory continuing airworthiness information.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Process is not regulated.	
4.2.4.4	4.2.4.4 The State of Registry shall ensure that sensitive aviation security information is securely transmitted to the appropriate authority in the State of Design in accordance with Annex 17.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Process is not established.	

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<i>Note.— Guidance material on the transmission of sensitive aviation security information is contained in Doc 9760.</i>									
4.2.5	4.2.5 All Contracting States Until 25 November 2026, each Contracting State shall establish, in respect of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass, the type of information that is to be reported to its airworthiness authority by operators, organizations responsible for type design and maintenance organizations. Procedures for reporting this information shall also be established. _____	21.A.3 Reg. (EU) 748/2012. 145.A.60 Reg. (EU) 1321/2014 R Eg. (EU) 376/2014 21L.A.3 Annex Ib Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.2.5	4.2.5 All Contracting States As of 26 November 2026, each Contracting State shall establish, in respect of aeroplanes over 5 700 kg and helicopters over 3 175 kg maximum certificated take-off mass and remotely piloted aircraft systems, the type of information that is to be reported to its airworthiness authority by operators, organizations responsible for type design and maintenance organizations. Procedures for reporting this information shall also be established. _____		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

6.1	<p>CHAPTER 6. MAINTENANCE ORGANIZATION APPROVAL</p> <p style="text-align: center;">6.1 Applicability</p> <p>The Standards of this chapter are applicable to the approval of organizations involved in the maintenance of aircraft, engines, propellers and associated parts. Approval certificates issued before 5 November 2020 shall be amended before 5 November 2022 to ensure compliance with the requirements in 6.2.3.</p>	145.A.10, CAO.A.015 Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.2.1	<p>6.2.1 The Contracting State concerned shall define appropriate requirements for the approval of a maintenance organization in accordance with the Standards of this chapter.</p> <p><i>Note.— Guidance material on the approval of an approved maintenance organization is contained in the Airworthiness Manual (Doc 9760).</i></p>	Art 4 (1)Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.2.2	<p>6.2.2 The issuance of a maintenance organization approval by a Contracting State shall be dependent upon the applicant demonstrating compliance with the applicable Standards of this chapter through compliance with appropriate requirements defined in accordance with 6.2.1 and relevant provisions contained in Annex 19 for such organizations.</p>	145.B.310, CAO.B.050 Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	SMS is applicable for all Part-145 approved maintenance organisations	Regulation (EU) 2021/1963 has been published on 8 Nov				

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

									ns. SMS is not applicable for Part-CAO approved maintenance organisations as it is not a relevant provision of Annex 19.	2021 which requires SMS in Part-145 approved maintenance organisations as of 02 December 2022. Part CAO: no SMS requirements.
6.2.3	6.2.3 The approval certificate shall contain at least the following information: a) the issuing authority and the name, title and signature of the person issuing the certificate; b) the maintenance organization's name and registered address; c) the maintenance organization approval reference number;	145.B.310, CAO.B.050 Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<p>d) the date of current issue;</p> <p>e) in the case of certificates of limited duration, the expiration date;</p> <p>f) the scope of approval, in relation to aircraft, component and/or specialized maintenance, and to the type of aircraft and components covered by the approval; and</p> <p>g) the locations of the maintenance facilities, unless the information is included in a separate document referred to in the approval certificate.</p> <p><i>Note.— Guidance material on the content of the approval certificate is contained in Doc 9760.</i></p>									
6.2.3.1	<p>6.2.3.1 Recommendation.— <i>The approval certificate should follow the template in the Appendix and contain the date of original issue if different from the date of current issue.</i></p>	145.B.310, CAO.B.050 Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The maintenance organisation approval certificate contains a general reference to the approved Exposition (MOE or CAE). The Exposition shall	ICAO Annex 8 Appendix (APP-1-1) point (10) as well as Doc 9760 Attachment E to Chapter 10 recommends to

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								contain specific chapter(s) which contains the the locations of the maintenance facilities.	include a specific reference on the approval certificate to the appropriate section/c hapter and paragraph of the MOE where the facilities are listed. The EASA Form 3-145 (Appendix III to Annex II Part-145 of Reg. 1341/2014) and
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
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									EASA Form 3-CAO (Appendix I to Annex Vd Part-CAO of Re. 1321/2014) do not explicitly require the listing of all maintenance facilities nor a specific reference to the MOE resp. CAE chapter and paragraph.
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
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									These facilities are referenced in MOE as per 145.A.70(a)(15), including aircraft base maintenance facilities and line stations. Following AMC1 145.A.70(a), these are typically in MOE 1.8 and 5.3. For CAO, article CAO.A.025(a)(9)
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) applies and, following the AMC1 of CAO.A. 025, the locations must be included in the manual CAE chapter A.3.
6.2.4	6.2.4 The continued validity of the approval shall depend upon the organization remaining in compliance with the appropriate requirements of 6.2.1 and 6.2.2.	145.A.90 (a)(1), CAO.A.110 (a) (1) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	SMS not yet implemented for all maintenance organisations (SMS for all Part 145 AMO shall be in place by 2 Dec 2024).	Regulation (EU) 2021/1963 has been published on 8 Nov 2021 which includes SMS in Part 145 organisation

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				Level of implementation of SARPs						
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										Part CAO: no SMS requirements.
6.2.5	6.2.5 The maintenance organization shall notify the Contracting State which issued the maintenance organization approval of any changes to the organization's scope of work, location or personnel nominated in accordance with this chapter.	145.A.85, CAO.A.105 (a) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EU Regulation also considers small changes controlled by the organisation through procedures approved by the competent authority.	
6.2.6	6.2.6 Where a Contracting State accepts, in whole or in part, a maintenance organization approval issued by another Contracting State, it shall establish a process for the recognition of such approval and successive changes. In such a case, the recognizing Contracting State shall build an adequate liaison with the Contracting State that initially issued the maintenance organization approval.	Art. 68 (1)Reg. (EU) 2018/1139	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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6.3.1	<p>6.3 Maintenance organization's procedures manual</p> <p>6.3.1 The maintenance organization shall provide for the use and guidance of maintenance personnel concerned a procedures manual which may be issued in separate parts containing the following information:</p> <p>a) a general description of the scope of work authorized under the organization's terms of approval;</p> <p>b) a description of the organization's procedures and quality or inspection system in accordance with 6.4;</p> <p>c) a general description of the organization's facilities;</p> <p>d) names and duties of the person or persons required by 6.6.1 and 6.6.2;</p> <p>e) a description of the procedures used to establish the competence of the maintenance personnel required by 6.6.4;</p> <p>f) a description of the method used for the completion and retention of the maintenance records required by 6.7;</p>	145.A.70 (a), CAO.A.025 (a) Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
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	<p>g) a description of the procedures for preparing the maintenance release and the circumstances under which the release is to be signed;</p> <p>h) the personnel authorized to sign the maintenance release and the scope of their authorization;</p> <p>i) a description, when applicable, of contracted activities;</p> <p>j) a description, when applicable, of the additional procedures for complying with an operator's maintenance procedures and requirements;</p> <p>k) a description of the procedures for complying with the information reporting requirements of 4.2.4.1 f) and 4.2.5 of this part;</p> <p>l) a description of the procedure for receiving, assessing, amending and distributing within the maintenance organization all necessary airworthiness data from the organization responsible for the type design; and</p> <p>m) a description of the procedures for implementing changes affecting the approval of the maintenance organization.</p>									
6.3.2	6.3.2 The maintenance organization shall ensure that the procedures manual is amended as necessary to keep the information contained therein up to date.	145.A.70(b) & (c), CAO.A.025 (a), CAO.A.040 (d) Reg. (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		1321/2014								
6.3.3	<p>6.3.3 The maintenance organization shall furnish copies of all amendments to the procedures manual promptly to all organizations or persons to whom the manual has been issued.</p> <p><i>Note.— Guidance material on the content of a maintenance organization's procedures manual is contained in Doc 9760.</i></p>	<p>145.A.70 (b), CAO.A.025, CAO.A.105 (b) Reg. (EU) 1321/2014 AMC1 145.A.70, AMC2 CAO.A.025</p>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Part-145 does not provide for a direct requirement for distribution of the manual to the end users, however the paragraphs 145.A.70 (b) and AMC1 145.A.70 have that objective. Same for CAO.A.025 and AMC2 CAO.A.025.	
6.4.1	6.4 Maintenance procedures and quality assurance system	<p>145.A.65(b), CAO.A.060 Reg. (EU)</p>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Maintenance organisations are	

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	6.4.1 The maintenance organization shall establish procedures acceptable to the Contracting State granting the approval which ensure good maintenance practices and compliance with all relevant Standards prescribed in 6.2.1 and 6.2.2.	1321/2014							additionally required to control specialized services and to ensure procedures to minimize the risk of multiple errors and capture errors on multiple systems.	
6.4.2	6.4.2 The maintenance organization shall ensure compliance with 6.4.1 by either establishing an independent quality assurance system to monitor compliance with, and adequacy of, the procedures, or by providing a system of inspection to ensure that all maintenance is properly performed.	145.A.200(a)(6), CAO.A.100 Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	No Difference in Part-145. CAO have independent quality assurance system except if it is considered small CAO, then	

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									an organisational review is enough.	
6.5.1	<p align="center">6.5 Facilities</p> <p>6.5.1 The maintenance organization shall provide the appropriate facilities and working environments for the tasks to be performed.</p> <p><i>Note.— Guidance material on requirements for approved maintenance organization facilities is contained in Doc 9760.</i></p>	145.A.25 (a), (b), (c), (d), CAO.A.030 Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
6.5.2	6.5.2 The maintenance organization shall have the necessary technical data, equipment, tools and material to perform the work for which it is approved.	145.A.45, 145.A.40, CAO.A.50, CAO.A.55 Reg. (EU) 1321/2014	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EU Regulation adds that the maintenance data has to be current and tools and equipment controlled and calibrated.	
6.5.3	6.5.3 The maintenance organization shall ensure that storage conditions provide adequate security and prevent deterioration of, and damage to, stored items such as parts, equipment, tools and material.	145.A.25 (a) and (d), CAO.A.030 (c),	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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		CAO.A.050, CAO.A.055 Reg. (EU) 1321/2014								
6.6.1	<p align="center">6.6 Personnel</p> <p>6.6.1 The maintenance organization shall nominate an accountable executive who, irrespective of other functions, is accountable on behalf of the organization.</p> <p><i>Note.— Guidance material on the responsibilities of an accountable executive is contained in Doc 9760 and the Safety Management Manual (SMM) (Doc 9859).</i></p>	145.A.30 (a), CAO.A.035 (a) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EU Regulation adds details of his/her responsibilities.	
6.6.2	6.6.2 The maintenance organization's accountable executive shall nominate a person or group of persons whose responsibilities include ensuring that the maintenance organization is in compliance with the requirements of 6.2.1 and 6.2.2.	145.A.30(b), (c) and (ca), CAO.A.035 (b) Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
6.6.3	6.6.3 The maintenance organization shall employ the necessary personnel to plan, perform, supervise, inspect and release the maintenance work to be performed.	145.A.30(d), (f), (g), (h), (i) and (j), CAO.A.035 (d) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The regulation has different levels of detail in regards to the different maintainan	

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									ce organisations. Part-145 is very detailed, Part-CAO is less detailed, but the process covers the different aspects of the standard.	
6.6.4	6.6.4 The maintenance organization shall establish the competence of maintenance personnel in accordance with procedures and to a level acceptable to the Contracting State granting the approval. If the person signing the maintenance release is a non-licensed person, the person shall meet the qualification requirements specified in Annex 1 — <i>Personnel Licensing</i> to sign a maintenance release.	145.A.30(cc), (e), (g), (h), (i) and (j). 145.A.35, CAO.A.035 (d), (e) and (f), CAO.A.040 (a), (b), (c) Art 5 (6) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The qualification in accordance with Annex 1 is not required for component certifying staff, specialized services certifying staff. In	

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									accordance with Art 5(6)(ii) of Reg. 1321/2014 the national requirements of the Member State for the component certifying staff apply.	
6.6.5	<p>6.6.5 The maintenance organization shall ensure that all maintenance personnel receive initial and continuation training appropriate to their assigned tasks and responsibilities. The training programme established by the maintenance organization shall include training in knowledge and skills related to human performance, including coordination with other maintenance personnel and flight crew.</p> <p><i>Note.— Guidance material to design training programmes to develop knowledge and skills in human performance can be found in the Human Factors Training Manual (Doc 9683).</i></p>	145.A.30 (e) 145.A.35 (d), (e), CAO.A.35(c), (d), and (e), AMC1 CAO.A.035(c), AMC1 CAO.A.035(e) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Human performance not directly mentioned in Part-CAO.	

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6.7.1	6.7 Records 6.7.1 The maintenance organization shall retain detailed maintenance records to show that all requirements for the signing of a maintenance release have been met.	145.A.55 (a), CAO.A.090 (a) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Part-145 requires to keep also subcontractor's release documents.	
6.7.2	6.7.2 The records required by 6.7.1 shall be kept for a minimum period of one year after the signing of the maintenance release.	145.A.55(a), CAO.A.090 (b) Reg. (EU) 1321/2014	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EU Regulation requires 3 years.	
6.7.3	6.7.3 Records kept in accordance with 6.7 shall be maintained in a form and format that ensures readability, security and integrity of the records at all times. <i>Note 1.— The form and format of the records may include, for example, paper records, film records, electronic records or any combination thereof.</i> <i>Note 2.— Guidance material regarding electronic aircraft maintenance records is contained in Doc 9760.</i>	145.A.55 (c), CAO.A.090 (d), (e) Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
6.8.1	6.8 Maintenance release	145.A.50, CAO.A.065,	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	6.8.1 A maintenance release shall be completed and signed to certify that the maintenance work performed has been completed satisfactorily and in accordance with approved data and the procedure described in the maintenance organization's procedures manual.	CAO.A.070 Reg. (EU) 1321/2014								
6.8.2	6.8.2 A maintenance release shall be signed and include the following: a) basic details of the maintenance carried out including detailed reference to the data used; b) the date such maintenance was completed; c) the identity of the approved maintenance organization; and d) the identity of the person or persons signing the release.	145.A.50 (a) AMC 145.A.50 (b), Appendix II to Part M, CAO.A.065 Reg. (EU) 1321/2014	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	EU Regulation includes the limitations to airworthiness or operations, if any. For components a specific form is required (EASA Form 1).	
1.1.1	PART III. LARGE AEROPLANES PART IIIA. AEROPLANES OVER 5 700 KG FOR WHICH APPLICATION	Reg. (EU) 2018/1139. Reg. (EU) 748/2012. CS-23, CS-	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	<p>FOR CERTIFICATION WAS SUBMITTED ON OR AFTER 13 JUNE 1960 BUT BEFORE 2 MARCH 2004</p> <p><i>Note.— The provisions of Part IIIA are the same as those contained in Part III of Annex 8, Ninth Edition (including Amendment 99), except for modified applicability clauses and cross-references.</i></p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part, except for those specified in 8.4, are applicable in respect of all aeroplanes designated in 1.1.3 that are of types of which the prototype was submitted to the appropriate national authorities for certification on or after 13 June 1960, but before 2 March 2004.</p>	25									
1.1.2	1.1.2 The Standards specified in 8.4 are applicable in respect of all aeroplanes designated in 1.1.3 that are of types of which the prototype was submitted to the appropriate national authorities for certification on or after 22 March 1985, but before 2 March 2004.	CS 23.1383-1403, CS 25.1383-1403	<input checked="" type="checkbox"/>	<input type="checkbox"/>							

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1.1.3	<p>1.1.3 Except for those Standards and Recommended Practices which specify a different applicability, the Standards and Recommended Practices of this part shall apply to aeroplanes with a maximum certificated take-off mass greater than 5 700 kg and intended for the carriage of passengers or cargo or mail in international air navigation.</p> <p><i>Note.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p>	CS 25.1, CS 23.1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.4	<p>1.1.4 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the aeroplanes designated in 1.1.3 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.</p>	CS-25, CS-23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.5	<p>1.1.5 Unless otherwise stated, the Standards apply to the complete aeroplane including its powerplant, systems and equipment.</p>	CS-25.1; CS 23.1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2	<p>1.2 Number of engines</p> <p>The aeroplane shall have not less than two engines.</p>	CS 25.121(a), CS 23.1(a)(2)	<input checked="" type="checkbox"/>	<input type="checkbox"/>		CS-25 does not specify 2 or more				

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										engines as a limitation. Practically a single engine aeroplane could not comply with CS-25.
1.3.1	<p>1.3 Operating limitations</p> <p>1.3.1 Limiting conditions shall be established for the aeroplane, its powerplant and its equipment (see 9.2). Compliance with the Standards of this part shall be established assuming that the aeroplane is operated within the limitations specified. The limitations shall be sufficiently removed from any condition(s) prejudicial to the safety of the aeroplane to render the likelihood of accidents arising therefrom extremely remote.</p> <p><i>Note.— Guidance material concerning the expression “extremely remote” is contained in the Airworthiness Manual (Doc 9760).</i></p>	CS 25. 21, 23, 25, 27, 33, 101, 1583. CS 23.21, 23, 25, 33, 1501	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.3.2	<p>1.3.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the aeroplane, e.g. mass, centre of gravity location, load distribution, speeds, and altitude or pressure-altitude, shall be established within which compliance with all the pertinent Standards in this part is shown, except that combinations of conditions which are fundamentally impossible to achieve need not be considered.</p> <p><i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each separate operating condition, e.g. take-off, en route, landing.</i></p> <p><i>Note 2.— The following items, for instance, may be considered as basic aeroplane limitations:</i></p> <ul style="list-style-type: none"> — <i>maximum certificated take-off mass;</i> — <i>maximum certificated taxiing mass;</i> — <i>maximum certificated landing mass;</i> — <i>maximum certificated zero fuel mass;</i> <p><i>and</i></p> <ul style="list-style-type: none"> — <i>most forward and rearward centre of gravity positions in various configurations (take-off, en route, landing).</i> <p><i>Note 3.— Maximum operating mass may be limited by the application of noise certification Standards (see Annex 16 — Environmental Protection, Volume I —</i></p>	CS 25.23, 25, 27, 29, 1527. CS 23. 23, 25, 29, 1501, 1519	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
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	Aircraft Noise, and Annex 6 — Operation of Aircraft, Part I — International Commercial Air Transport — Aeroplanes and Part II — International General Aviation — Aeroplanes).									
1.4	1.4 Unsafe features and characteristics Under all anticipated operating conditions, the aeroplane shall not possess any feature or characteristic that renders it unsafe.	21.A.21, 16 Reg. (EU) 748/2012 CS 25.143, CS23.143, 601	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.5.1	1.5 Proof of compliance 1.5.1 Compliance with the appropriate airworthiness requirements shall be based on evidence from tests, calculations, or calculations based on tests, provided that in each case the accuracy achieved will ensure a level of airworthiness equal to that which would be achieved were direct tests conducted.	21.A.20, 33, 35 Reg. (EU) 748/2012. CS 25.21, 307, CS 23.21, 307	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.5.2	1.5.2 The tests of 1.5.1 shall be such as to provide reasonable assurance that the aeroplane, and its components, systems and equipment, are reliable and function correctly under the anticipated operating conditions.	21.A.20, 21.A.33, 21.A.35 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.1.1	CHAPTER 2. FLIGHT 2.1 General 2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon an aeroplane or aeroplanes of the type for which a Certificate of Airworthiness is sought, or by calculations based on such tests, provided that the results obtained by calculations are equal in accuracy to, or conservatively represent, the results of direct testing.	CS 27 Subpart B	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.2	2.1.2 Compliance with each Standard shall be established for all applicable combinations of aeroplane mass and centre of gravity position, within the range of loading conditions for which certification is sought.	CS 25.21, 23, 25, 27. CS 23.21, 23, 25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.3	2.1.3 Where necessary, appropriate aeroplane configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the aeroplane's flying qualities.	CS 25.101 (d), (e), (f). CS 23.45	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1.1	2.2 Performance 2.2.1 General	CS 25.1587. CS 23.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	2.2.1.1 Sufficient data on the performance of the aeroplane shall be determined and scheduled in the flight manual to provide operators with the necessary information for the purpose of determining the total mass of the aeroplane on the basis of the values, peculiar to the proposed flight, of the relevant operational parameters, in order that the flight may be made with reasonable assurance that a safe minimum performance for that flight will be achieved.									
2.2.1.2	2.2.1.2 Achieving the performance scheduled for the aeroplane shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the flight crew. <i>Note.— Guidance material on human performance can be found in the Human Factors Training Manual (Doc 9683).</i>	CS 25.101 (h). CS 23.45(f)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1.3	2.2.1.3 The scheduled performance of the aeroplane shall be consistent with compliance with 1.3.1 and with the operation in logical combinations of those of the aeroplane's systems and equipment, the operation of which may affect performance.	CS 25.101 (f). CS 23.45(h)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.2	2.2.2 Minimum performance At the maximum masses scheduled (see 2.2.3) for take-off and for landing as functions of the aerodrome elevation or pressure-altitude either in the standard	CS 25 Subpart B. CS 23 Subpart	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>atmosphere or in specified still air atmospheric conditions, and, for seaplanes, in specified conditions of smooth water, the aeroplane shall be capable of accomplishing the minimum performances specified in 2.2.2.1 and 2.2.2.2, respectively, not considering obstacles or runway or water run length.</p> <p><i>Note.— This Standard permits the maximum take-off mass and maximum landing mass to be scheduled in the aeroplane flight manual against, for example:</i></p> <ul style="list-style-type: none"> — <i>aerodrome elevation, or</i> — <i>pressure-altitude at aerodrome level, or</i> — <i>pressure-altitude and atmospheric temperature at aerodrome level,</i> <p><i>so as to be readily usable when applying the national code on aeroplane performance operating limitations.</i></p>	B								
2.2.2.1	<p>2.2.2.1 <i>Take-off</i></p> <p>a) The aeroplane shall be capable of taking off assuming the critical engine to fail (see 2.2.3), the remaining engines being operated within their take-off power limitations.</p> <p>b) After the end of the period during which the take-off power may be used, the aeroplane shall be capable of continuing to climb, with the critical engine inoperative and the remaining engine(s) operated within their maximum continuous power limitations, up</p>	CS 25.20, CS 25.25, 105, 107, 109, 115, 113, 119, 121. CS 23. 51(c), 57, 63, 67, 69, 45(h).	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	to a height that it can maintain and at which it can carry out a circuit of the aerodrome. c) The minimum performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is scheduled (see 2.2.3), the departure from the scheduled values is not disproportionate.									
2.2.2.2	2.2.2.2 <i>Landing</i> a) Starting from the approach configuration and with the critical engine inoperative, the aeroplane shall be capable, in the event of a missed approach, of continuing the flight to a point from which a fresh approach can be made. b) Starting from the landing configuration, the aeroplane shall be capable, in the event of a balked landing, of making a climb-out, with all engines operating.	CS 25. 119, 121 (d). CS 23.67 (c)(4), 77 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.2.3	2.2.3 Scheduling of performance Performance data shall be determined and scheduled in the flight manual so that its application by means of the operating rules to which the aeroplane is to be operated in accordance with 5.2 of Annex 6, Part I, will provide a safe relationship between the performance of the aeroplane and the aerodromes and routes on which it is capable of being operated. Performance data shall be	21.B.80Reg. (EU) 748/2012CS-25 subpart BCS 25.1587CS-23 subpart BCS 23.237CS	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Scheduling of landing distance with runway slope is not required. Performance is not	CS-23 and related AMC complies except that performance is

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	determined and scheduled for the following stages for the ranges of mass, altitude or pressure-altitude, wind velocity, gradient of the take-off and landing surface for landplanes; water surface conditions, density of water and strength of current for seaplanes; and for any other operational variables for which the aeroplane is to be certificated.	23.1587CS 23.2155CS 23.2620						scheduled for variations in water surface conditions, density of water and strength of current.	not scheduled for variations in water surface conditions, density of water and strength of current. CS/JAR 23.237/CS 23.2155 requires that the allowable water surface conditions and any necessary water handling procedures for seaplane
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										to be established. However, factors on landing distance are applied by operational rules, where appropriate.
2.2.3.1	2.2.3.1 <i>Take-off.</i> The take-off performance data shall include the accelerate-stop distance and the take-off path.	CS 25.109, 111, 115, 1587. CS 23.55, 1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.3.1.1	2.2.3.1.1 <i>Accelerate-stop distance.</i> The accelerate-stop distance shall be the distance required to accelerate and stop, or, for a seaplane to accelerate and come to a satisfactorily low speed, assuming the critical engine to fail suddenly at a point not nearer to the start of the take-off than that assumed when determining the take-off path (see 2.2.3.1.2).	CS 25.109. CS 23.55	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.2.3.1.2	2.2.3.1.2 <i>Take-off path.</i> The take-off path shall comprise the ground or water run, initial climb and climb-out, assuming the critical engine to fail suddenly during the take-off (see 2.2.3.1.1). The take-off path shall be scheduled up to a height that the aeroplane can maintain and at which it can carry out a circuit of the aerodrome. The climb-out shall be made at a speed not less than the take-off safety speed as determined in accordance with 2.3.1.3.	CS 25.111, 115. CS 23.57, 61	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.3.2	2.2.3.2 <i>En route.</i> The en-route climb performance shall be the climb (or descent) performance with the aeroplane in the en-route configuration with: a) the critical engine inoperative; and b) the two critical engines inoperative in the case of aeroplanes having three or more engines. The operating engine(s) shall not exceed maximum continuous power.	CS 25.123. CS 23.69	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.3.3	2.2.3.3 <i>Landing.</i> The landing distance shall be the horizontal distance traversed by the aeroplane from a point on the approach flight path at a selected height above the landing surface to the point on the landing surface at which the aeroplane comes to a complete stop, or, for a seaplane, comes to a satisfactorily low speed. The selected height above the landing surface and the approach speed shall be appropriately related to operating practices. This distance may be supplemented by such distance margin as may be necessary; if so, the selected	CS 25.125. CS 23.75	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>height above the landing surface, the approach speed and the distance margin shall be appropriately interrelated and shall make provision for both normal operating practices and reasonable variations therefrom.</p> <p><i>Note.— If the landing distance includes the distance margin specified in this Standard, it is not necessary to allow for the expected variations in the approach and landing techniques in applying 5.2.11 of Annex 6, Part I.</i></p>									
2.3	<p>2.3 Flying qualities</p> <p>The aeroplane shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the aeroplane is approved.</p>	CS 25.21 (c). CS 23.45 (b) (1), (3), 141	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.1	<p>2.3.1 Controllability</p> <p>The aeroplane shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power, changes of aeroplane configurations) without requiring exceptional skill, alertness or strength on the part of the pilot even in the event of failure of any engine. A technique for safely controlling the aeroplane</p>	CS 25.143, 145, 147, 149. CS 23. 143, 145, 147, 149.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	shall be established for all stages of flight and aeroplane configurations for which performance is scheduled. <i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i>									
2.3.1.1	2.3.1.1 <i>Controllability on the ground (or water).</i> The aeroplane shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.	CS 25.143, CS 25. CS 23.231, 233, CS 23.235, CS 23.237	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.1.2	2.3.1.2 <i>Controllability during take-off.</i> The aeroplane shall be controllable in the event of sudden failure of the critical engine at any point in the take-off, when the aeroplane is handled in the manner associated with the scheduling of take-off paths and accelerate-stop distances.	CS 25.149, CS 23.149.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.1.3	2.3.1.3 <i>Take-off safety speed.</i> The take-off safety speeds assumed when the performance of the aeroplane (after leaving the ground or water) during the take-off is determined shall provide an adequate margin above the stall and above the minimum speed at which the aeroplane remains controllable after sudden failure of the critical engine.	CS 25.107. CS 23.51 (c)(4)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.3.2	2.3.2 Trim The aeroplane shall have such trim and other characteristics as to ensure that the demands made on the pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. This shall apply both in normal operation and in the conditions associated with the failure of one or more engines for which performance characteristics are established.	CS 25.161. CS 23.161	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.3	2.3.3 Stability The aeroplane shall have such stability in relation to its other flight characteristics, performance, structural strength and most probable operating conditions (e.g. aeroplane configurations and speed ranges) as to ensure that demands made on the pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the aeroplane shall not, however, be such that excessive demands are made on the pilot's strength or that the safety of the aeroplane is prejudiced by lack of manoeuvrability in emergency conditions.	CS 25.171, 173, 175, 177, 181. 672 CS 23. 171, 173, 175, 177, 181. 672	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.4.1	2.3.4 Stalling 2.3.4.1 <i>Stall warning.</i> When the aeroplane approaches a stall both in straight and turning flight with all engines operating and with one engine inoperative, a	CS 23.207. CS 25.207	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	clear and distinctive stall warning shall be apparent to the pilot with the aeroplane in all permissible configurations and powers, except those which are not considered to be essential for safe flying. The stall warning and other characteristics of the aeroplane shall be such as to enable the pilot to arrest the development of the stall after the warning begins and, without altering the engine power, to maintain full control of the aeroplane.									
2.3.4.2	2.3.4.2 <i>Behaviour following a stall.</i> In any configuration and power in which it is considered that the ability to recover from a stall is essential, the behaviour of the aeroplane following a stall shall not be so extreme as to make difficult a prompt recovery without exceeding the airspeed or strength limitations of the aeroplane. It shall be acceptable to throttle back the operating engines during recovery from the stall.	CS 25.201, 203. CS 23.201, 203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.4.3	2.3.4.3 <i>Stalling speeds.</i> The stalling speeds or minimum steady flight speeds in configurations appropriate for each stage of flight (e.g. take-off, en route, landing) shall be established. One of the values of the power used in establishing the stalling speeds shall be not more than that necessary to give zero thrust at a speed just above the stall.	CS 25.103. CS 23.49	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.5	2.3.5 Flutter and vibration It shall be demonstrated by suitable tests that all parts of the aeroplane are free from flutter and excessive vibration in all aeroplane configurations under all speed conditions within the operating limitations of the aeroplane (see	CS 25.251, 253, 629. CS 23.251, 253, 629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>1.3.2). There shall be no buffeting severe enough to interfere with control of the aeroplane, to cause structural damage or to cause excessive fatigue to the flight crew.</p> <p><i>Note.— Buffeting as a stall warning is considered desirable and discouragement of this type of buffeting is not intended.</i></p>									
3.1	<p>CHAPTER 3. STRUCTURE</p> <p>3.1 General</p> <p>The Standards of this chapter apply to the aeroplane structure consisting of all portions of the aeroplane, the failure of which would seriously endanger the aeroplane.</p>	CS-23 Subpart C. CS-25 Subpart C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.1.1	<p>3.1.1 Mass and mass distribution</p> <p>Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.</p>	CS 25.321, CS 23.321	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.1.2	<p>3.1.2 Limit loads</p> <p>Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.3, 3.4 and 3.5 shall be considered as limit loads.</p>	CS 25.301, CS 23.301	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.1.3	<p>3.1.3 Strength and deformation</p> <p>In the various loading conditions prescribed in 3.3, 3.4 and 3.5, no part of the aeroplane structure shall sustain detrimental deformation at any load up to and including the limit load, and the aeroplane structure shall be capable of supporting the ultimate load.</p>	CS 25.305, CS 23.305	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.2.1	<p>3.2 Airspeeds</p> <p>3.2.1 Design airspeeds</p> <p>Design airspeeds shall be established for which the aeroplane structure is designed to withstand the corresponding manoeuvring and gust loads in accordance with 3.3. In establishing the design airspeeds, consideration shall be given to the following speeds:</p> <p>a) V_A, the design manoeuvring speed;</p>	CS 25.335, CS 23.335	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) V_B, the speed at which the maximum vertical gust velocity assumed in accordance with 3.3.2 can be withstood;</p> <p>c) V_C, a speed not expected to be exceeded in normal cruising flight taking into account possible effects of upsets when flying in turbulent conditions;</p> <p>d) V_D, maximum dive speed, sufficiently greater than the speed in c), to make it unlikely that such a design speed would be exceeded as a result of inadvertent speed increases in the anticipated operating conditions, taking into account the flying qualities and other characteristics of the aeroplane;</p> <p>e) V_{E1} to V_{En}, maximum speeds at which flaps and landing gears may be extended or other configuration changes be made.</p> <p>The speeds V_A, V_B, V_C, and V_E in a), b), c) and e) shall be sufficiently greater than the stalling speed of the aeroplane to safeguard against loss of control in turbulent air.</p>								
3.2.2	<p>3.2.2 Limiting airspeeds</p> <p>Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.3.1, shall be included in the aeroplane</p>	CS 25.1583. CS 23.1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	flight manual as part of the operating limitations (see 9.2.2).									
3.3	3.3 Flight loads The flight loading conditions of 3.3.1, 3.3.2 and 3.5 shall be considered for the range of mass and mass distributions prescribed in 3.1.1 and at airspeeds established in accordance with 3.2.1. Asymmetrical as well as symmetrical loading shall be taken into account. The air, inertia and other loads resulting from the specified loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively.	CS 25.301, 321, 333. CS 23.301, 321, 333.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.3.1	3.3.1 Manoeuvring loads Manoeuvring loads shall be computed on the basis of manoeuvring load factors appropriate to the manoeuvres permitted by the operating limitations. They shall not be less than values that experience indicates will be adequate for the anticipated operating conditions.	CS 25. 331, 333, 337, 349, 351, 421, 423, 427. CS 23. 331, 333, 337, 349, 351, 421, 423, 427, 441	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.3.2	3.3.2 Gust loads Gust loads shall be computed for vertical and horizontal gust velocities and gradients that statistics or other	CS 25. 341, 427. CS 23. 341, 333, 425,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	evidence indicates will be adequate for the anticipated operating conditions.	443								
3.4	<p>3.4 Ground and water loads</p> <p>The structure shall be able to withstand all the loads due to the reactions of the ground and water surface that are likely to arise during taxiing, take-off and landing.</p>	21.B.80Reg. (EU) 748/2012CS 25.253CS 25.471CS 25.519CS 23.471CS 23.537CS 23.2210CS 23.2220	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS 25 and CS 23 do not contain specifications for water loads.	Large flying-boats are not under development. Would this happen, EASA would develop the necessary special conditions in accordance with Part-21.
3.4.1	<p>3.4.1 Landing conditions</p> <p>The landing conditions at the design take-off mass and at the design landing mass shall include such symmetrical</p>	CS 25. 473-487. CS 23. 477-	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	and asymmetrical attitudes of the aeroplane at ground or water contact, such velocities of descent, and such other factors affecting the loads imposed upon the structure as might be present in the anticipated operating conditions.	505								
3.5	3.5 Miscellaneous loads In addition to or in conjunction with the manoeuvring and gust loads and with the ground and water loads, consideration shall be given to all other loads (flight control loads, cabin pressures, effects of engine operation, loads due to changes of configuration, etc.) that are likely to occur in the anticipated operating conditions.	CS 25. 361-459. CS 23 361-459	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.6	3.6 Flutter, divergence and vibration The aeroplane structure shall be designed to be free from flutter, structural divergence (i.e. unstable structural distortion due to aerodynamic loading), and loss of control due to structural deformation, at speeds within and sufficiently beyond the operating limitations to comply with 1.3.1. Adequate strength shall be provided to withstand the vibration and buffeting that might occur in the anticipated operating conditions.	CS 25.252, 629. CS 23.251, 629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.7	3.7 Fatigue strength	CS 25.571. CS 23. 571-575,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>The strength and fabrication of the aeroplane shall be such as to ensure that the probability of disastrous fatigue failure of the aeroplane's structure under repeated loads and vibratory loads in the anticipated operating conditions is extremely remote.</p> <p><i>Note.— Guidance material concerning the expression "extremely remote" is contained in the Airworthiness Manual (Doc 9760).</i></p>	627								
4.1	<p>CHAPTER 4. DESIGN AND CONSTRUCTION</p> <p>4.1 General</p> <p>Details of design and construction shall be such as to give reasonable assurance that all aeroplane parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	21.B.80Reg. (EU) 748/2012CS 25.601CS 23.601CS 23.2250	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The added sentence "They shall also observe human factors principles" is not fully complied with.	CS 25.1302 and AMC 25.1302 (Installed Systems and Equipment for Use by the Flight Crew), created at Amdt 3 of CS-

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									25 effective on 19/02/2007 take into account potential design related human factors issues during the compliance demonstration. EASA may require, as part of the certification planning process, that a specific evaluation, analysis,
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										or assessment of a human factors issue to become part of the demonstration that the design is in compliance with requirements.
4.1.1	4.1.1 Substantiating tests The functioning of all moving parts essential to the safe operation of the aeroplane shall be demonstrated by suitable tests in order to ensure that they will function correctly under all operating conditions for such parts.	CS 25.601, 683, 685. CS 23.601, 683, 685	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.2	4.1.2 Materials All materials used in parts of the aeroplane essential for its safe operation shall conform to approved specifications. The approved specifications shall be such	CS 25.603, 613. CS 23.603, 613.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	that materials accepted as complying with the specifications will have the essential properties assumed in the design.									
4.1.3	4.1.3 Manufacturing methods The methods of manufacturing and assembly shall be such as to produce a consistently sound structure which shall be reliable with respect to maintenance of strength in service.	CS 25.605, CS 23.605	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.1.4	4.1.4 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the aeroplane will receive.	CS 25.609, 571. CS 23.609, 571-575.	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.1.5	4.1.5 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the aeroplane that require such attention, either periodically or after unusually severe operations.	CS 25.611, CS 23.611	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.1.6	4.1.6 Systems design features	21.B.80Reg. (EU)	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Less protective	The differen

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	<p>Special consideration shall be given to design features that affect the ability of the flight crew to maintain controlled flight. This shall include at least the following:</p> <p>a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such as to minimize the possibility of jamming, inadvertent operation and unintentional engagement of control surface locking devices.</p> <p>b) <i>System survivability.</i></p> <p>1) For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60 and for which the application for certification was submitted on or after 12 March 2000, aeroplane systems shall be designed, arranged and physically separated to maximize the potential for continued safe flight and landing after any event resulting in damage to the aeroplane structure or systems.</p> <p>2) Recommendation.— <i>For aeroplanes of a maximum certificated take-off mass in excess of 5 700 kg but not exceeding 45 500 kg and for which the application for certification was submitted on or after 12 March 2000, aeroplane systems should be designed, arranged and physically separated to maximize the potential for continued safe flight and landing after any event resulting in damage to the aeroplane structure or systems.</i></p>	<p>748/2012CS 25.685CS 25.671CS 23.685CS 23.671CS 25.777CS 25.779CS 25.781CS 23.777CS 23.779CS 23.781CS 25.1302CS 25.773CS 23.773CS 25.795CS 25.1301CS 25.1309CS 23.1301CS 23.1309CS 25.851-869CS 23.851-865CS 25.856 (a)CS 25.855-858CS 23.855CS 25.831CS 25.831CS 25.841CS 23.831CS 23.841CS-25 Appendix F</p>						<p>for paragraphs (b), (g), (h) and (i). Protection against explosive and incendiary devices was not requested in the applicable airworthiness codes (JAR-25, CS-25) effective within the time span of the applicability of this provision of Part IIIA (from 12 March 2000 until 2 March 2004).</p>	<p>ces related to security standards have been removed by the amendment of CS 25.795 introduced by Amendment 9 to CS-25 effective 12 August 2010. After this date the new security provisions are applicable to new applicati</p>
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	<p>c) <i>Crew environment.</i> The design of the flight crew compartment shall be such as to minimize the possibility of incorrect or restricted operation of the controls by the crew, due to fatigue, confusion or interference. Consideration shall be given at least to the following: layout and identification of controls and instruments, rapid identification of emergency situations, sense of controls, ventilation, heating and noise.</p> <p>d) <i>Pilot vision.</i> The arrangement of the pilot compartment shall be such as to afford a sufficiently extensive, clear and undistorted field of vision for the safe operation of the aeroplane, and to prevent glare and reflections that would interfere with the pilot's vision. The design features of the pilot windshield shall permit, under precipitation conditions, sufficient vision for the normal conduct of flight and for the execution of approaches and landings.</p> <p>e) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the flight crew to deal with, emergencies resulting from foreseeable failures of equipment and systems, the failure of which would endanger the aeroplane. Reasonable provisions shall be made for continuation of essential services following engine or system failures to the extent that such failures are catered for in the performance and operating limitations specified in the Standards in this Annex and in Annex 6, Parts I and II.</p> <p>f) <i>Fire precautions.</i> The design of the aeroplane and the materials used in its manufacture, including cabin interior furnishing materials replaced</p>	Part V							ons for type certification as well as already certificated types subjected to certification of significant changes to TC (application of changed product rule Part 21A.101).
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	<p>during major refurbishing, shall be such as to minimize the possibility of in-flight and ground fires and also to minimize the production of smoke and toxic gases in the event of a fire. Means shall be provided to contain or to detect and extinguish such fires as might occur in such a way that no additional danger to the aeroplane is caused.</p> <p>g) <i>Fire suppression.</i> For aeroplanes for which the application for certification was submitted on or after 12 March 2000, cargo compartment fire suppression systems, including their extinguishing agents, shall be designed so as to take into account a sudden and extensive fire such as could be caused by an explosive or incendiary device or dangerous goods.</p> <p>h) <i>Incapacitation of occupants.</i></p> <p>1) For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60 and for which the application for certification was submitted on or after 12 March 2000, design precautions shall be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases, including those caused by explosive or incendiary devices or dangerous goods, which could incapacitate the occupants of the aeroplane.</p> <p>2) Recommendation.— <i>For aeroplanes of a maximum certificated take-off mass in excess of 5 700 kg but not exceeding 45 500 kg and for which the application for certification was submitted on</i></p>									
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	<p><i>or after 12 March 2000, design precautions should be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases, including those caused by explosive or incendiary devices or dangerous goods, which could incapacitate the occupants of the aeroplane.</i></p> <p>i) <i>Protection of the flight crew compartment from smoke and fumes.</i></p> <p>1) For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60 and for which the application for certification was submitted on or after 12 March 2000, means shall be provided to minimize entry into the flight crew compartment of smoke, fumes and noxious vapours generated by an explosion or fire on the aeroplane.</p> <p>2) Recommendation.— <i>For aeroplanes of a maximum certificated take-off mass in excess of 5 700 kg but not exceeding 45 500 kg and for which the application for certification was submitted on or after 12 March 2000, means should be provided to minimize entry into the flight crew compartment of smoke, fumes and noxious vapours generated by an explosion or fire on the aeroplane.</i></p>								
4.1.7.1	<p>4.1.7 Emergency landing provisions</p> <p>4.1.7.1 Provisions shall be made in the design of the aeroplane to protect the occupants, in the event of</p>	CS 25.561, 562. CS 23.561,	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	an emergency landing, from fire and from the direct effects of deceleration forces as well as from injuries arising from the effect of deceleration forces on the aeroplane's interior equipment.	562								
4.1.7.2	4.1.7.2 Facilities shall be provided for the rapid evacuation of the aeroplane in conditions likely to occur following an emergency landing. Such facilities shall be related to the passenger and crew capacity of the aeroplane.	CS 25.803-819. CS 23.803-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.7.3	4.1.7.3 The interior layout of the cabin and the position and number of emergency exits, including the means of locating and illuminating the escape paths and exits, shall be such as to facilitate rapid evacuation of the aeroplane in conditions likely to occur following an emergency landing.	CS 25.791, 807-817. CS 23.791, 807-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.7.4	4.1.7.4 On aeroplanes certificated for ditching conditions, provisions shall be made in the design to give maximum practicable assurance that safe evacuation from the aeroplane of passengers and crew can be executed in case of ditching.	CS 25.801, 563, 807(i). CS 23.807(e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.8	4.1.8 Ground handling Adequate provisions shall be made in the design to minimize the risk that ground-handling operations (e.g. towing, jacking) may cause damage, which could pass unnoticed, to the parts of the aeroplane essential for its	CS 25.509, 489, 519. CS 23.509, 507.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	safe operation. The protection that any limitations and instructions for such operations might provide may be taken into account.									
5.1	<p>CHAPTER 5. ENGINES</p> <p>5.1 Scope</p> <p>The Standards of this chapter shall apply to engines of all types that are used on the aeroplane as primary propulsion units.</p>	CS-E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2	<p>5.2 Design, construction and functioning</p> <p>The engine complete with accessories shall be designed and constructed so as to function reliably within its operating limitations under the anticipated operating conditions when properly installed in the aeroplane in accordance with Chapter 7 and, if applicable, fitted with a suitable propeller.</p>	CS E.20	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.3	<p>5.3 Declared ratings, conditions and limitations</p> <p>The power ratings and the conditions of the atmosphere upon which they are based and all operating conditions and limitations, which are intended to govern the operation of the engine shall be declared.</p>	CS E.40, 320, 620	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.4	<p>5.4 Tests</p> <p>An engine of the type shall complete satisfactorily such tests as are necessary to verify the validity of the declared ratings, conditions and limitations and to ensure that it will operate satisfactorily and reliably. The tests shall include at least the following:</p> <p>a) <i>Power calibration.</i> Tests shall be conducted to establish the power or thrust characteristics of the engine when new and also after the tests in b) and c). There shall be no excessive decrease in power at the conclusion of all the tests specified.</p> <p>b) <i>Operation.</i> Tests shall be conducted to ensure that starting, idling, acceleration, vibration, overspeeding and other characteristics are satisfactory and to demonstrate adequate margins of freedom from detonation, surge or other detrimental conditions as may be appropriate to the particular type engine.</p> <p>c) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, thrust, speeds and other operating conditions as are necessary to</p>	21.A.20 Reg. (EU) 748/2012. CS E 350, 730, 340, 360-430, 640, 650, 700,710, 720, 750, 780-840, 860-890, 740, 440.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	demonstrate reliability and durability of the engine. They shall also include operation under conditions in excess of the declared limits to the extent that such limitations might be exceeded in actual service.									
6.1	<p>CHAPTER 6. PROPELLERS</p> <p style="text-align: center;">6.1 Scope</p> <p>The Standards of this chapter shall apply to propellers of all types.</p>	CS-P	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.2	<p>6.2 Design, construction and functioning</p> <p>The propeller assembly complete with accessories shall be designed and constructed so as to function reliably within its operating limitations under the anticipated operating conditions when properly fitted to the engine and installed in the aeroplane in accordance with Chapter 7.</p>	CS P.30	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.3	<p>6.3 Declared ratings, conditions and limitations</p> <p>The power ratings and all operating conditions and limitations which are intended to govern the operation of the propeller shall be declared.</p>	CS P.50	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.4	<p>6.4 Tests</p> <p>A propeller of the type shall complete satisfactorily such tests as are necessary to ensure that it will operate satisfactorily and reliably within the declared ratings, conditions and limitations. The tests shall include at least the following:</p> <p>a) <i>Operation.</i> Tests shall be conducted to ensure that strength vibration and overspeeding characteristics are satisfactory and to demonstrate proper and reliable functioning of pitch changing and control mechanisms.</p> <p>b) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, speeds and other operating conditions as are necessary to demonstrate reliability and durability of the propeller.</p>	CS P 330-440	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.1.1	CHAPTER 7. POWERPLANT INSTALLATION 7.1 General 7.1.1 Applicable Standards The powerplant installation shall comply with the Standards of Chapter 4 and with the Standards of this chapter.	CS-23 Subpart E. CS-25 Subpart E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.1.2	7.1.2 Compliance with engine and propeller limitations The powerplant installation shall be so designed that the engines and propellers (if applicable) are capable of being used in the anticipated operating conditions. In conditions established in the aeroplane flight manual, the aeroplane shall be capable of being operated without exceeding the limitations established for the engines and propellers in accordance with Chapters 5, 6 and this chapter.	CS 25.1581. CS 23.1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.1.3	7.1.3 Control of engine rotation In those installations where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided for the crew to stop the rotation of the engine in flight or to reduce it to a safe level.	CS 25.903 (c). CS 23.903 (d) and (e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.1.4	7.1.4 Engine restarting Means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.	CS 25.903 (e). CS 23.903 (g)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.1	7.2 Arrangement and functioning 7.2.1 Independence of engines The powerplant shall be arranged and installed so that each engine together with its associated systems is capable of being controlled and operated independently from the others and so that there is at least one arrangement of the powerplant and systems in which any failure, unless the probability of its occurrence is extremely remote, cannot result in a loss of more power than that resulting from complete failure of the critical engine.	CS 25.903 (b). CS 23.903 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.2	7.2.2 Propeller vibration The propeller vibration stresses shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the aeroplane.	CS 25.907. CS 23.907	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.2.3	7.2.3 Cooling The cooling system shall be capable of maintaining powerplant temperatures within the established limits (see 7.1.2) at ambient air temperatures up to the maximum air temperature appropriate to the intended operation of the aeroplane. The maximum and, if necessary, minimum ambient air temperature for which the powerplant has been established as being suitable shall be scheduled in the aeroplane flight manual.	CS 25.1041. CS 25.1521(d), 1583(b), CS 23.1041	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.4	7.2.4 Associated systems The fuel, oil, air induction and other systems associated with the powerplant shall be capable of supplying each engine in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power, aeroplane attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.	CS 25. 951-1001, CS 25. 1011-1027, CS 25.1091-1103, CS 25.1121-1123. CS 23. 951-1001, CS 23. 1011-1027, CS 23.1091-1111, CS 23.1121-1125	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.5	7.2.5 Fire protection For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the	CS 25.1181-1207. CS 23.1181-1203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>following shall apply in addition to the general Standard of 4.1.6 e).</p> <p>a) <i>Isolation.</i> Such regions shall be isolated by fire-resisting material from other regions of the aeroplane where the presence of fire would jeopardize continued flight, taking into account the probable points of origin and paths of propagation of fire.</p> <p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be capable of containing the fluid when exposed to fire conditions. Means shall be provided for the crew to shut off the flow of flammable fluids into such regions if a fire occurs.</p> <p>c) <i>Fire detection.</i> A sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions.</p> <p>d) <i>Fire extinguishment.</i> Such regions shall be provided with a fire extinguisher system capable of extinguishing any fire likely to occur therein, unless the degree of isolation, quantity of combustibles, fire resistance of the structure and other factors are such that any fire likely to occur in the region would not jeopardize the safety of the aeroplane.</p>								
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Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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8.1	<p>CHAPTER 8. INSTRUMENTS AND EQUIPMENT</p> <p>8.1 Required instruments and equipment</p> <p>The aeroplane shall be provided with approved instruments and equipment necessary for the safe operation of the aeroplane in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the crew to operate the aeroplane within its operating limitations. Instruments and equipment design shall observe human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Parts I and II, for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	21.B.80Reg. (EU) 748/2012CS-23 Subpart FCS-25 Subpart FCS 25.1303CS 25.1305CS 25.1307CS 25.1327CS 25.1337CS 23.1303CS 23.1305CS 23.1327CS 23.1337CS 23.2500CS 23.2510	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The sentence 'shall observe Human Factors principles' is not fully complied with.	CS 25.1302 and AMC 25.1302 (Installed Systems and Equipment for Use by the Flight Crew), created at Amdt 3 of CS-25 effective on 19/02/2007 take into account potential design related human factors issues during
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									the compliance demonstration. EASA may require, as part of the certification planning process, that a specific evaluation, analysis, or assessment of a human factors issue becomes part of the demonstration that the design is in
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										compliance with requirements.
8.2	8.2 Installation Instrument and equipment installations shall comply with the Standards of Chapter 4.	CS 25.601. CS 23.601	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.3	8.3 Safety and survival equipment Prescribed safety and survival equipment that the crew or passengers are expected to use or operate at the time of an emergency shall be reliable, readily accessible and easily identified, and its method of operation shall be plainly marked.	CS 25. 1411, 1415, 1421, 1423. CS 23.1411, 1415	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.4.1	8.4 Navigation lights and anti-collision lights 8.4.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by aeroplanes in flight or operating on the movement area of an aerodrome shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft or personnel on the ground with as much time as possible for interpretation and for subsequent manoeuvre necessary to avoid a collision. In the design of such	CS 25.1385-1401. CS 23.1385-1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note 1.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry sky, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i></p> <p><i>Note 2.— Detailed technical specifications for exterior lights for aeroplanes can be found in the Airworthiness Manual (Doc 9760).</i></p> <p>----- 1 Please refer to 1.1.2 of this part.</p>									
8.4.2	<p>8.4.2 Lights shall be installed in aeroplanes so as to minimize the possibility that they will:</p> <p>a) adversely affect the satisfactory performance of the flight crews' duties; or</p> <p>b) subject an outside observer to harmful dazzle.</p> <p><i>Note.— In order to avoid the effects mentioned in 8.4.2, it will be necessary in some cases to provide</i></p>	CS 25.1383, 1401, 1403. CS 23.1383, 1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>means whereby the pilot can switch off or reduce the intensity of the flashing lights.</i>									
9.1	<p>CHAPTER 9. OPERATING LIMITATIONS AND INFORMATION</p> <p style="text-align: center;">9.1 General</p> <p>The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the aeroplane, shall be made available by means of an aeroplane flight manual, markings and placards, and such other means as may effectively accomplish the purpose. The limitations and information shall include at least those prescribed in 9.2, 9.3 and 9.4.</p>	<p>CS-23 Subpart G. CS-25 Subpart G. CS 25.1541, 1563, 1581. CS 23. 1541, 1563, 1581.</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2	<p style="text-align: center;">9.2 Operating limitations</p> <p>Limitations which there is a risk of exceeding in flight and which are defined quantitatively shall be expressed in suitable units and corrected if necessary for errors in measurements so that the flight crew can, by reference to</p>	<p>CS 25.1541. CS 23.1541</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	the instruments available to them, readily determine when the limitations are reached.									
9.2.1	9.2.1 Loading limitations The loading limitations shall include all limiting masses, centre of gravity positions, mass distributions and floor loadings (see 1.3.2).	CS 25.1519, 1533	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.2	9.2.2 Airspeed limitations The airspeed limitations shall include all speeds (see 3.2) that are limiting from the standpoint of structural integrity or flying qualities of the aeroplane, or from other considerations. These speeds shall be identified with respect to the appropriate aeroplane configurations and other pertinent factors.	CS 25.1501-1517. CS 23.1501-1513	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.3	9.2.3 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the aeroplane (see 7.1.2 and 7.2.3).	CS 25.1521, 1522. CS 23.1521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.4	9.2.4 Limitations on equipment and systems	CS 23.1501. CS 25.1501	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the aeroplane.									
9.2.5	9.2.5 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the aeroplane (see 1.3.1).	CS 25.1501, 1533. CS 23.1501	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.6	9.2.6 Flight crew limitations The flight crew limitations shall include the minimum number of flight crew personnel necessary to operate the aeroplane, having regard, among other things, to the accessibility to the appropriate crew members of all necessary controls and instruments and to the execution of the established emergency procedures. <i>Note.— The circumstances in which the flight crew shall include members in addition to the minimum flight crew are defined in Annex 6, Parts I and II.</i>	CS 25.1523. CS 23.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.7	9.2.7 Flying time limitation after system or engine failure The systems limitations shall include the maximum flying time for which system reliability has been established in relation to the approval of operations by aeroplanes with two turbine engines beyond the threshold time established in accordance with 4.7 of Annex 6, Part I.	AMC 20-6	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>Note.— The maximum time established in accordance with 4.7 of Annex 6, Part I, for a particular route may be less than that determined in accordance with 9.2.7 because of the operational considerations involved.</i>									
9.3.1	<p>9.3 Operating information and procedures</p> <p>9.3.1 Types of eligible operations</p> <p>There shall be listed the particular types of operations, as may be defined in Annex 6, Parts I and II, or be generally recognized, for which the aeroplane has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements.</p>	CS 25.1525. CS 23.1525	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.3.2	<p>9.3.2 Loading information</p> <p>The loading information shall include the empty mass of the aeroplane, together with a definition of the condition of the aeroplane at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the crew and payload, the usable fuel supply and the drainable oil; it includes the mass of all fixed ballast,</i></p>	CS 25.1519. CS 23.1519	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>unusable fuel supply, undrainable oil, total quantity of engine coolant and total quantity of hydraulic fluid.</i>									
9.3.3	<p>9.3.3 Operating procedures</p> <p>A description shall be given of normal and emergency operating procedures which are peculiar to the particular aeroplane and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.</p>	CS 25.1583, 1585. CS 23.1583, 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
9.3.4	<p>9.3.4 Handling information</p> <p>Sufficient information shall be given on any significant or unusual features of the aeroplane characteristics. Those stalling speeds or minimum steady flight speeds required to be established by 2.3.4.3 shall be scheduled.</p>	CS 25.1581, 1583, 1545, 1557, 1587. CS 23.1545, 1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
9.3.5	<p>9.3.5 Least-risk bomb location</p> <p>For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60 and for which the application for certification was submitted on or after 12 March 2000, a least-risk location on the aeroplane shall be identified where a bomb or other explosive device may be placed to minimize the effects on the aeroplane in the case of detonation.</p>	21.B.80 Reg. (EU) 748/2012 CS 25.795	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Implemented in CS-25 Amdt 9 in 2003. TC after 2003 are compliant with this provision.	The difference concerns only the applicability date.

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9.4	9.4 Performance information The performance of the aeroplane shall be scheduled in accordance with 2.2. There shall be included information regarding the various aeroplane configurations and powers involved and the relevant speeds, together with information that would assist the flight crew in attaining the performance as scheduled.	CS 25.1587, 1591. CS 23.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.5	9.5 Aeroplane flight manual A flight manual shall be made available. It shall identify clearly the specific aeroplane or series of aeroplanes to which it is related. The flight manual shall include at least the limitations, information and procedures specified in this chapter.	CS 25.1581, CS 23.1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.6.1	9.6 Markings and placards 9.6.1 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the flight crew during flight.	CS 25.1541-1563. CS 23.1541-1567	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.6.2	9.6.2 Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in ground servicing (e.g. towing, refuelling) that	CS 25.1557 and Appendix H. CS 23.1557 and Appendix	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	could pass unnoticed and that could jeopardize the safety of the aeroplane in subsequent flights.	G								
10.1	CHAPTER 10. CONTINUING AIRWORTHINESS — MAINTENANCE INFORMATION 10.1 General Information for use in developing procedures for maintaining the aeroplane in an airworthy condition shall be made available. The information shall include that described in 10.2, 10.3 and 10.4.	CS-23-1529 abd Subpart G, CS-25-1529 and Appendix H. Part M, Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
10.2	10.2 Maintenance information Maintenance information shall include a description of the aeroplane and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.	CS 25.1529 and Appendix H. CS 23.1529 and Appendix G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
10.3	10.3 Maintenance programme information	M.A.302 Reg. (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed.	1321/2014. CS 25.1529 Appendix H. CS 23.1529 Appendix G								
10.4	10.4 Maintenance information resulting from the type design approval Maintenance tasks and frequencies that have been specified as mandatory by the State of Design in approval of the type design shall be identified as such.	CS 25 Appendix H4. CS 23 Appendix G4	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
11.1.1	CHAPTER 11. SECURITY 11.1 Aeroplanes used for domestic commercial operations Recommendation. — <i>International Standards and Recommended Practices set forth in this chapter should be applied by all Contracting States for aeroplanes engaged in domestic commercial operations (air services).</i>	21.B.80 Reg. (EU) 48/2012 CS 25.772 CS 25.795	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Not covered (except for pilots compartment doors) by the applicable airworthiness codes (JAR-25, CS-25) effective within the	New security provisions of CS 25.795 introduced by Amendment 9 to CS-25 effective 12 August

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									time span of applicability of this provision of Part IIIA (from 12 March 2000 until 2 March 2004).	2010.
11.2	<p>11.2 Least-risk bomb location</p> <p>For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60 and for which the application for certification was submitted on or after 12 March 2000, consideration shall be given during the design of the aeroplane to the provision of a least-risk bomb location so as to minimize the effects of a bomb on the aeroplane and its occupants.</p>	21.B.80Reg. (EU) 748/2012CS 25.795	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Implemented in 2010 instead of 2000.	New security provisions of CS 25.795 introduced by Amendment 9 to CS25 effective 12 August 2010. This includes 'least-risk bomb location'

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11.3.1	<p>11.3 Protection of the flight crew compartment</p> <p>Recommendation.— <i>In all aeroplanes which are required by Annex 6, Part I, Chapter 13 to have an approved flight crew compartment door and for which an application for amending the Type Certificate to include a derivative type design was submitted to the appropriate national authority, consideration should be given to reinforcing the flight crew compartment bulkheads, floors and ceilings so as to resist penetration by small arms fire and grenade shrapnel and to resist forcible intrusions, if these areas are accessible in flight to passengers and cabin crew.</i></p> <p><i>Note.</i>— <i>Standards and Recommended Practices concerning the requirements for the flight crew compartment door in all commercial passenger-carrying aeroplanes are contained in Annex 6, Part I, Chapter 13.</i></p>	CS 25.772. CS 25.795	<input checked="" type="checkbox"/>	<input type="checkbox"/>		However CS 25.795 become part of the certification basis established for certification of a significant change to TC i.a.w. 21.A.101 (Changed Product Rule). The security provisions introduced through				
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										the amendment of CS 25.795 by Amendment 9 to CS-25 effective 12 August 2010.
11.4	<p>11.4 Interior design</p> <p>For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60 and for which the application for certification was submitted on or after 12 March 2000, consideration shall be given to design features that will deter the easy concealment of weapons, explosives or other dangerous objects on board aircraft and that will facilitate search procedures for such objects.</p>	21.B.80Reg. (EU) 748/2012CS 25.795	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Implemented in 2010 instead of 2000.	

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1.1.1	<p>PART IIIB. AEROPLANES OVER 5 700 KG FOR WHICH APPLICATION FOR CERTIFICATION WAS SUBMITTED ON OR AFTER 2 MARCH 2004</p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all aeroplanes designated in 1.1.2 for which an application for the issue of a Type Certificate was submitted to the appropriate national authorities on or after 2 March 2004.</p>	Reg. (EU) 2018/1139. Part-21 Reg. (EU) 748/2012. CS-25, CS-23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.2	<p>1.1.2 Except for those Standards and Recommended Practices which specify a different applicability, the Standards and Recommended Practices of this part shall apply to all aeroplanes with a maximum certificated take-off mass greater than 5 700 kg and intended for the carriage of passengers or cargo or mail in international air navigation.</p> <p><i>Note 1.— The aeroplanes described in 1.1.2 are known in some States as transport category aeroplanes.</i></p> <p><i>Note 2.— The following Standards do not include quantitative specifications comparable to those</i></p>	CS 25.1, CS 23.1.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i>									
1.1.3	1.1.3 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the aeroplanes designated in 1.1.2 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.	CS 25, CS 23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.4	1.1.4 Unless otherwise stated, the Standards apply to the complete aeroplane including its powerplant, systems and equipment.	CS 25.1, CS 23.1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2	1.2 Number of engines The aeroplane shall have not less than two engines.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>		Aviation rules for civil aircraft registration 12.02.2008 № 5 (as amended by resolution Ministry

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										of transportation and communications The Republic of Belarus May 29, 2020 No. 29) paragraph 112
1.3.1	1.3 Operating limitations 1.3.1 Limiting conditions shall be established for the aeroplane, its powerplant, systems and equipment (see 7.2). Compliance with the Standards of this part shall be established assuming that the aeroplane is operated within the limitations specified. The limitations shall include a margin of safety to render the likelihood of accidents arising therefrom extremely remote.	CS 25.21 (a), 23, 25, 27, 101, 1583. CS 23. 21(a), 23, 25, 33, 45, 1501	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.2	1.3.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the aeroplane, e.g. mass, centre of gravity location, load distribution, speeds, ambient air temperature and altitude, shall be established within which compliance with all the pertinent Standards in this part is shown.	CS 25.23, 25, 27, 29. 1519, 1527. CS 23.23, 25, 29, 1501,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each separate operating condition, e.g. take-off, en route, landing.</i></p> <p><i>Note 2.— Maximum operating mass may be limited by the application of noise certification Standards (see Annex 16 — Environmental Protection, Volume I — Aircraft Noise, and Annex 6 — Operation of Aircraft, Part I — International Commercial Air Transport — Aeroplanes and Part II — International General Aviation — Aeroplanes).</i></p>	15198								
1.4	<p>1.4 Unsafe features and characteristics</p> <p>Under all anticipated operating conditions, the aeroplane shall not possess any feature or characteristic that renders it unsafe.</p>	21.A.21, 16 Reg. (EU) 748/2012. CS 25.143, 601. CS 23.143,601	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.5	<p>1.5 Proof of compliance</p> <p>The means by which compliance with the appropriate airworthiness requirements is demonstrated shall ensure that in each case the accuracy achieved will be such as to provide reasonable assurance that the aeroplane, its components and equipment comply with the requirements and are reliable and function correctly under the anticipated operating conditions.</p>	21.A.20, 33, 35 Reg. (EU) 748/2012. CS 25.21. 307. CS 23.21, 307	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.1.1	<p>CHAPTER 2. FLIGHT</p> <p>2.1 General</p> <p>2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon an aeroplane or aeroplanes of the type for which a Type Certificate is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.</p>	Annex II Reg (EU) 2018/1139, Reg (EU) 748/2012 CS-23 1383, 1403. CS-25.1383, 1403	<input checked="" type="checkbox"/>	<input type="checkbox"/>					
2.1.2	<p>2.1.2 Compliance with each Standard shall be established for all applicable combinations of aeroplane mass and centre of gravity position, within the range of loading conditions for which certification is sought.</p>	CS 25.21, 23, 25, 27. CS 23.21, 23, 25	<input checked="" type="checkbox"/>	<input type="checkbox"/>					
2.1.3	<p>2.1.3 Where necessary, appropriate aeroplane configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the aeroplane's flying qualities.</p>	CS 25. 101 (d), (e) (f). CS 23.45	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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2.2.1	<p>2.2 Performance</p> <p>2.2.1 Sufficient data on the performance of the aeroplane shall be determined and furnished in the flight manual to provide operators with the necessary information for the purpose of determining the maximum total mass of the aeroplane at the time of take-off that would allow the flight to be made with reasonable assurance that a safe minimum performance for that flight will be achieved considering the values of the operational parameter peculiar to the proposed flight.</p>	<p>21.B.80 Reg. (EU) 748/2012 CS 25.1519CS 25.1583CS 25.1587CS 23.1519CS 23.1583CS 23.1587CS 23.2620</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.2	<p>2.2.2 Achieving the performance furnished in the flight manual for the aeroplane shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the flight crew.</p> <p><i>Note.— Guidance material on human performance can be found in the Human Factors Training Manual (Doc 9683).</i></p>	<p>21.B.80Reg. (EU) 748/2012 CS 25.101 (h)CS 23.45 (f)CS 23.2105</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.2	<p>2.2.2 Until 3 November 2021, achieving the performance scheduled for the aeroplane shall take into consideration human performance and in particular shall</p>	<p>21.B.80Reg. (EU) 748/2012CS 25.101 (h)CS</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	not require exceptional skill or alertness on the part of the flight crew.	23.45 (f)CS 23.2105								
2.2.3	2.2.3 The performance data in the flight manual of the aeroplane shall be consistent with compliance with 1.3.1 and with the operation in logical combinations of those of the aeroplane's systems and equipment, the operation of which may affect performance.	21.B.80Reg. (EU) 748/2012CS 25.21 (a)CS 25.23CS 25.25CS 25.27CS 25.101CS 25.1583CS 23.21 (a)CS 23.23CS 23.25CS 23.33CS 23.45CS 23.1583CS 23.2100CS 23.2400CS 23.2105CS 23.2620	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.3	2.2.3 Until 3 November 2021, the scheduled performance of the aeroplane shall be consistent with compliance with 1.3.1 and with the operation in logical combinations of those of the aeroplane's systems and equipment, the operation of which may affect performance.	21.B.80Reg. (EU) 748/2012CS 25.21 (a)CS 25.23CS 25.25CS 25.27CS 25.101CS 25.1583CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		23.21 (a)CS 23.23CS 23.25CS 23.33CS 23.45CS 23.1583CS 23.2105							
2.2.4.1	<p>2.2.4 Minimum performance</p> <p>2.2.4.1 For aeroplanes for which application for certification was submitted before 2 March 2019, at the maximum masses scheduled for take-off and for landing permitted by the performance data in the flight manual (see 2.2.7.2) as functions of the aerodrome elevation or pressure-altitude either in the standard atmosphere or in specified still air atmospheric conditions, and, for seaplanes, in specified conditions of smooth water, the aeroplane shall be capable of accomplishing the minimum performances specified in 2.2.5 and 2.2.6, respectively, not considering obstacles, or runway or water run length.</p> <p><i>Note.— This Standard permits the maximum take-off mass and maximum landing mass to be scheduled in the flight manual against, for example:</i></p> <p>— aerodrome elevation, or</p> <p>— pressure-altitude at aerodrome level, or</p>	21.B.80Reg. (EU) 748/2012CS-25 Subpart BCS-23 Subpart BCS 25.101 (a)CS 23.45 (a)CS 23.2105	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	— <i>pressure-altitude and atmospheric temperature at aerodrome level,</i> <i>so as to be readily usable when applying the national code on aeroplane performance operating limitations.</i>									
2.2.4.2	2.2.4.2 For aeroplanes for which application for certification was submitted on or after 2 March 2019, at the maximum masses scheduled for take-off and for landing permitted by the performance data in the flight manual (see 2.2.7.3) as functions of the aerodrome elevation or pressure-altitude either in the standard atmosphere or in specified still air atmospheric conditions, and, for seaplanes, in specified conditions of smooth water, the aeroplane shall be capable of accomplishing the minimum performances specified in 2.2.5 and 2.2.6, respectively, not considering obstacles, or runway or water run length.	21.B.80Reg. (EU) 748/2012CS-25 Subpart BCS 25.101 (a)CS-23 Subpart BCS 23.45 (a)CS 23.2105	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.5	2.2.5 Take-off a) The aeroplane shall be capable of taking off assuming the critical engine to fail (see 2.2.7), the remaining engine(s) being operated within their take-off power or thrust limitations. b) After the end of the period during which the take-off power or thrust may be used, the aeroplane shall be capable of continuing to climb, with the critical engine inoperative and the remaining engine(s) operated within their maximum continuous power or thrust limitations, up to a height that it can	21.B.80Reg. (EU) 748/2012CS 25.25CS 25.105CS 25.107CS 25.109CS 25.113CS 25.115CS 25.119CS 25.121CS 23.51 (c)CS 23.53CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	maintain and at which it can continue safe flight and landing. c) The minimum performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is furnished (see 2.2.7), the departure from the furnished values is not disproportionate.	23.57CS 23.63CS 23.67 (c)CS 23.69CS 23.45 (h)CS 23.2105CS 23.2115CS 23.2120CS 23.2125								
2.2.6	2.2.6 Landing a) Starting from the approach configuration and with the critical engine inoperative, the aeroplane shall be capable, in the event of a missed approach, of continuing the flight to a point from which another approach can be made. b) Starting from the landing configuration, the aeroplane shall be capable, in the event of a balked landing, of making a climb-out, with all engines operating.	CS 25.119, 121 (d). CS 23.67 (c)(4), 77(c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.2.7.1	2.2.7 Performance data 2.2.7.1 The following stages are considered, as applicable:	21.B.80Reg. (EU) 748/2012 CS-25 Subpart B CS 25.1587 CS 25.1592 CS-23	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Scheduling of landing distance with runway slope is not required.	CS 23 performance is not scheduled for variation

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	<p>a) <i>Take-off.</i> The take-off performance data shall include the accelerate-stop distance and the take-off path.</p> <p>b) <i>Accelerate-stop distance.</i> The accelerate-stop distance shall be the distance required to accelerate and stop, or, for a seaplane to accelerate and come to a satisfactorily low speed, assuming the critical engine to fail suddenly at a point not nearer to the start of the take-off than that assumed when determining the take-off path (see 2.2.7.1 c)). Additionally, for landplanes, the distance shall be based on operations with all the wheel brake assemblies at the fully worn limit of their allowable wear range.</p> <p>c) <i>Take-off path.</i> The take-off path shall comprise the ground or water run, initial climb and climb-out, assuming the critical engine to fail suddenly during the take-off (see 2.2.7.1 b)). The take-off path shall be scheduled up to a height from which the aeroplane can continue safe flight and landing. The climb-out shall be made at a speed not less than the take-off safety speed as determined in accordance with 2.3.2.4.</p> <p>d) <i>En-route.</i> The en-route climb performance shall be the climb (or descent) performance with the aeroplane in the en-route configuration with:</p> <p>1) the critical engine inoperative;</p> <p>and</p>	Subpart BCS 23.1587 CS 23.2620						Performance is not scheduled for variations in water surface conditions, density of water and strength of current. CS 23.237/CS 23.2155 requires that the allowable water surface conditions and any necessary water handling procedures for seaplanes be established. However
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	<p>2) the two critical engines inoperative in the case of aeroplanes having three or more engines.</p> <p>The operating engine(s) shall not exceed maximum continuous power or thrust.</p> <p>e) <i>Landing. Landing performance data at the time of take-off.</i> The landing distance shall be the horizontal distance traversed by the aeroplane from a point on the approach flight path at a selected height above the landing surface to the point on the landing surface at which the aeroplane comes to a complete stop, or, for a seaplane, comes to a satisfactorily low speed. The selected height above the landing surface and the approach speed shall be appropriately related to operating practices. This distance may be supplemented by such distance margin as may be necessary; if so, the selected height above the landing surface, the approach speed and the distance margin shall be appropriately interrelated and shall make provision for both normal operating practices and reasonable variations therefrom. For landplanes, this distance shall be based on operations with all the wheel brake assemblies at the fully worn limit of their allowable wear range.</p> <p><i>Note.— If at time of take-off landing performance data includes the distance margin specified in this Standard, it is not necessary to allow for the expected variations in the approach and landing techniques in applying 5.2.11 of Annex 6, Part I.</i></p>								<p>r, factors on landing distance are applied by operational rules, where appropriate. CS 25.1592 at Amendment 27 (applicable from 7 Dec 2021) introduced specifications addressing landing performance at the time of</p>
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	<i>f) Landing. At time of landing performance data.</i> The landing distance shall be the horizontal distance traversed by the aeroplane from a point on the approach flight path to the point on the landing surface at which the aeroplane comes to a complete stop, or, for a seaplane, comes to a satisfactorily low speed. The approach speed, use of deceleration devices, and airborne portion of the landing distance shall be in accordance with and reflect directly actual normal operating practices. This distance may be supplemented by such distance margin as may be necessary. For landplanes, this distance shall be based on operations with all the wheel brake assemblies at the fully worn limit of their allowable wear range.									arrival.
2.2.7.2	2.2.7.2 For aeroplanes for which application for certification was submitted before 2 March 2019, performance data shall be determined and furnished in the flight manual so that its application by means of the operating rules to which the aeroplane is to be operated in accordance with 5.2 of Annex 6, Part I, will provide a safe relationship between the performance of the aeroplane and the aerodromes and routes on which it is capable of being operated. Performance data shall be determined and furnished for the stages in 2.2.7.1 a) to e) for the ranges of mass, altitude or pressure-altitude, wind velocity, gradient of the take-off and landing surface for landplanes; water surface conditions, density of water and strength of current for seaplanes; and for any other operational variables for which the aeroplane is to be certificated.	21.B.80Reg. (EU) 748/2012 CS 25.101 (e), (d), (f)CS 25.123CS 25.1591 CS 23.45 (f), (h)CS 23.69CS 23.237CS 23.1587CS 23.2105CS 23.2125CS 23.2155CS 23.2620	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Scheduling of landing distance with runway slope is not required. Performance is not scheduled for variations in water surface conditions, density of water and strength of	CS 23 performance is not scheduled for variations in water surface conditions, density of water and strength of current.

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								current. For CS-25 aeroplanes, supplementary take-off and landing performance information for operation on runways contaminated with standing water, slush, snow or ice may be provided, but this is not mandatory (see CS and AMC 25.1591).	CS 23.237/ CS 23.2155 requires that the allowable water surface conditions and any necessary water handling procedures for seaplanes be established. However, factors on landing distance are applied by operational rules, where
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										appropriate.
2.2.7.3	2.2.7.3 For aeroplanes for which application for certification was submitted on or after 2 March 2019, performance data shall be determined and furnished in the flight manual. Such performance data shall be so that its application by means of the operating rules to which the aeroplane is to be operated in accordance with 5.2 of Annex 6, Part I, will provide a safe relationship between the performance of the aeroplane and the aerodromes and routes on which it is capable of being operated. Performance data shall be determined and furnished for the stages in 2.2.7.1 a) to f) for the ranges of mass, pressure-altitude, ambient temperature, wind velocity, and for any other operational variables for which the aeroplane is to be certificated. Additionally, the take-off performance data and the at time of landing performance data shall include the effect of the gradient and conditions (dry, wet or contaminated) of the take-off or landing surface as appropriate for landplanes, and water surface conditions, density of water and strength of current for seaplanes. The at time of take-off landing performance data need only to be determined with standard day temperature and level, dry landing surfaces for landplanes, but shall include the effect of water surface conditions, density of water, and strength of current for seaplanes.	21.B.80Reg. (EU) 748/2012 CS 25.101 (e), (d), (f)CS 25.1587CS 25.1591 CS 25.1592 CS-23 Subpart BCS 23.1587CS 23.2620	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Scheduling of landing distance with runway slope is not required. Performance is not scheduled for variations in water surface conditions, density of water and strength of current. For CS-25 aeroplanes, supplementary take-off and landing performance information for	CS 23 performance is not scheduled for variations in water surface conditions, density of water and strength of current. CS 23.237/CS 23.2155 requires that the allowable water surface conditions and

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								operation on runways contaminated with standing water, slush, snow or ice may be provided, but this is not mandatory (see CS and AMC 25.1591).	any necessary water handling procedures for seaplanes be established. However, factors on landing distance are applied by operational rules, where appropriate. CS 25.1592 at Amendment 27 (applicable from 7 Dec 2021) introduced
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									ed specifications requiring to provide landing-distance information for assessing the landing performance at the time of arrival on dry, wet, slippery wet runways , and runways contaminated with standing water, slush, snow, or ice.
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2.3.1	2.3 Flying qualities 2.3.1 The aeroplane shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the aeroplane is approved.	CS 25.21 (c). CS 23.45(b)(1),(3), 141.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.1	2.3.2 Controllability 2.3.2.1 The aeroplane shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power or thrust, changes of aeroplane configurations) without requiring exceptional skill, alertness or strength on the part of the pilot even in the event of failure of any engine. A technique for safely controlling the aeroplane shall be established for all stages of flight and aeroplane configurations for which performance is scheduled. <i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i>	CS 25.145,147, 149, 143. CS 23. 143, 145, 147, 149.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.2	2.3.2.2 <i>Controllability on the ground (or water).</i> The aeroplane shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.	CS-25.143, 231, 233, 235. CS 23.143, 231,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.3.2.3	2.3.2.3 <i>Controllability during take-off.</i> The aeroplane shall be controllable in the event of sudden failure of the critical engine at any point in the take-off, when the aeroplane is handled in the manner associated with the scheduling of take-off paths and accelerate-stop distances.	CS 25.149. CS 23.149.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.4	2.3.2.4 <i>Take-off safety speed.</i> The take-off safety speeds assumed when the performance of the aeroplane (after leaving the ground or water) during the take-off is determined shall provide an adequate margin above the stall and above the minimum speed at which the aeroplane remains controllable after sudden failure of the critical engine.	CS 25.107. CS 23.51(c)(4)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.3	2.3.3 Trim The aeroplane shall have such trim characteristics as to ensure that the demands made on the pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. This shall apply both in normal operation and in the conditions associated with the failure of one or more engines for which performance characteristics are established.	CS 25.161. CS 23.161	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.4.1	<p>2.4 Stability and control</p> <p>2.4.1 Stability</p> <p>The aeroplane shall have such stability in relation to its other flight characteristics, performance, structural strength and most probable operating conditions (e.g. aeroplane configurations and speed ranges) as to ensure that demands made on the pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the aeroplane shall not, however, be such that excessive demands are made on the pilot's strength or that the safety of the aeroplane is prejudiced by lack of manoeuvrability in emergency conditions. It shall be shown that any combination of failures or conditions that would result in the need for exceptional piloting skills is extremely improbable. The stability may be achieved by natural or artificial means, or a combination of both. If compliance with the flight characteristics requirements is dependent upon a stability augmentation system or upon any other automatic or power-operated system, compliance shall be shown with 4.2 of this part.</p>	CS 25.171-181; 672. CS 23.171-181; 672	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2.1	<p>2.4.2 Stalling</p> <p>2.4.2.1 <i>Stall warning.</i> When the aeroplane approaches a stall both in straight and turning flight, a clear and distinctive stall warning shall be apparent to the pilot with the aeroplane in all permissible configurations</p>	21.B.80Reg. (EU) 748/2012CS-25 CS 25.201 CS 25.203 CS 25.207CS-23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	and powers or thrusts, except those which are not considered to be essential for safe flying. The stall warning and other characteristics of the aeroplane shall be such as to enable the pilot to arrest the development of the stall after the warning begins and, without altering the engine power or thrust, to maintain full control of the aeroplane.	CS 23.201CS 23.203CS 23.207CS 23.2150								
2.4.2.2	2.4.2.2 <i>Behaviour following a stall.</i> In any configuration and at any level of power or thrust in which it is considered that the ability to recover from a stall is essential, the behaviour of the aeroplane following a stall shall not be so extreme as to make difficult a prompt recovery without exceeding the airspeed or strength limitations of the aeroplane.	CS 25.201, 203. CS 23.201- 203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2.3	2.4.2.3 <i>Stalling speeds.</i> The stalling speeds or minimum steady flight speeds in configurations appropriate for each stage of flight (e.g. take-off, en route, landing) shall be established. One of the values of the power or thrust used in establishing the stalling speeds shall be not more than that necessary to give zero thrust at a speed just above the stall.	CS 25.103. CS 23.49	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3.1	2.4.3 Flutter and vibration 2.4.3.1 It shall be demonstrated by suitable tests, analyses or any acceptable combination of tests and analyses that all parts of the aeroplane are free from flutter and excessive vibration in all aeroplane configurations under all speed conditions within the operating limitations of the aeroplane (see 1.3.2). There	CS 25.251, 629. CS 23.251, 629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	shall be no vibration or buffeting severe enough to cause structural damage.									
2.4.3.2	<p>2.4.3.2 There shall be no vibration or buffeting severe enough to interfere with control of the aeroplane or to cause excessive fatigue to the flight crew.</p> <p><i>Note.— Buffeting as a stall warning is considered desirable and discouragement of this type of buffeting is not intended.</i></p>	CS 25.143, 251. CS 23.143, 251	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
3.1.1	<p>CHAPTER 3. STRUCTURE</p> <p>3.1 General</p> <p>3.1.1 For aeroplanes for which application for certification was submitted before 24 February 2013, the aeroplane structure shall be designed, manufactured and provided with instructions for its maintenance and repair with the objective of avoiding catastrophic failure throughout its operational life.</p>	21.B.80Reg. (EU) 748/2012CS-23 Subpart CCS-25 Subpart C	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Current CS 25/23 does not mandate the provision of structural repair manuals.	Provision of structural repair manuals is a normal industry practice.
3.1.2	3.1.2 For aeroplanes for which application for certification was submitted on or after 24 February 2013, the aeroplane structure shall be designed, manufactured	Part-21, Subpart M21.B.80Reg	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Hazardous not specificall	

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	and provided with instructions for its maintenance and repair with the objective of avoiding hazardous and catastrophic failure throughout its operational life.	. (EU) 748/2012CS-23 Subparts C, D CS-25 Subparts C, D							y addressed in relation to fatigue.	
3.2	3.2 Mass and mass distribution Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.	CS 25.321, CS 23.321	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.3	3.3 Limit loads Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.6 shall be considered as limit loads.	CS 25.301, CS 23.301	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.4	3.4 Strength and deformation In the various loading conditions prescribed in 3.6, no part of the aeroplane structure shall sustain detrimental deformation at any load up to and including the limit load, and the aeroplane structure shall be capable of supporting the ultimate load.	CS 25.305, CS 23.305	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.5.1	<p>3.5 Airspeeds</p> <p>3.5.1 Design airspeeds</p> <p>Design airspeeds shall be established for which the aeroplane structure is designed to withstand the corresponding manoeuvring and gust loads. To avoid inadvertent exceedances due to upsets or atmospheric variations, the design airspeeds shall provide sufficient margin for the establishment of practical operational limiting airspeeds. In addition, the design air-speeds shall be sufficiently greater than the stalling speed of the aeroplane to safeguard against loss of control in turbulent air. Consideration shall be given to a design manoeuvring speed, a design cruising speed, a design dive speed and any other design airspeeds necessary for configurations with high lift or other special devices.</p>	CS 25.335, CS 23.335	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.5.2	<p>3.5.2 Limiting airspeeds</p> <p>Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the flight manual as part of the operating limitations (see 7.2).</p>	CS 25.1583, CS 23.1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.6.1	<p>3.6 Strength</p> <p>3.6.1 All structural elements shall be designed to withstand the maximum loads expected in service under all anticipated operating conditions without failure, permanent distortion or loss of functionality. In determining these loads, account shall be taken of:</p> <p>a) the expected operational life of the aeroplane;</p> <p>b) the vertical and horizontal gust environment, taking into consideration the expected variations in mission profile and loading configurations;</p> <p>c) the manoeuvre spectrum, taking into account variations in mission profile and loading configurations;</p> <p>d) asymmetrical as well as symmetrical loading;</p> <p>e) the ground and water loads, including taxi, landing and take-off loads, and ground/water handling loads;</p> <p>f) the speed range of the aeroplane, taking into account the aeroplane characteristics and operation limitations;</p> <p>g) vibration and buffeting loads;</p>	CS-25 Subpart C, CS 25.609, CS-23 Subpart C, CS 23.609	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
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	<p>h) corrosion or other degradation, given the maintenance specified, and various operating environments; and</p> <p>i) any other loads, such as flight control loads, cabin pressurization loads, engine loads, or dynamic loads due to changes to the steady state configuration.</p>									
3.6.2	3.6.2 The air, inertia and other loads resulting from the specific loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively.	CS 25.301, CS 23.301	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
3.7	<p>3.7 Survivability</p> <p>The aeroplane shall be designed so as to provide the occupants with the maximum practicable protection in the event of structural failure, or in the event of damage due to ground, water or object impact. Consideration shall be given to at least the following:</p> <p>a) likely impact with birds;</p> <p>b) energy absorption by the airframe, occupant seats and restraints;</p> <p>c) the probable behaviour of the aeroplane in ditching; and</p>	21.B.80Reg. (EU) 748/2012CS 25.561CS 25.562CS 25.563 CS 25.631CS 23.561CS 23.562CS 23.775 (h) (1)CS 23.807CS 23.2250CS 23.2270CS 23.2320	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Only bird impact on windshield is required for CS-23 Commuter. Certification with ditching provisions is not required per CS-23 and CS-25. Some ditching design	Note that the current CS 25.807(e) requires provision of ditching emergency exits for passengers whether or not

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	d) allowing egress in the shortest practicable time.							provisions are provided in CS-25 (25.801), which include investigating the probable behaviour of the aeroplane in a water landing. However these provisions are applicable only under request if the applicant seeks certification for ditching. CS-23 does not include equivalent ditching	certification with ditching provisions is requested. Similarly, CS 23.807(e)/CS 23.2250 also requires some ditching emergency exits for twin-engined aeroplanes (which include Commuters).CS-25 Amendment 24 introduces an amendment of
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									provisions.	CS 25.671; paragraph(d) requires that after the failure of all engines at any time of the flight, a flare to a landing and a flare to a ditching can be achieved , thereby minimizing the risk of damage and injuries during contact with the ground or
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										water.
3.8.1	<p>3.8 Structural durability</p> <p>3.8.1 For aeroplanes for which application for certification was submitted before 24 February 2013, the design and construction of the aeroplane shall, wherever practicable, conform to damage tolerance principles and shall be such as to ensure that the probability of catastrophic failure during the operational life is extremely remote, taking into account:</p> <p>a) the expected environment;</p> <p>b) the expected repeated loads applied in service;</p> <p>c) expected vibrations from aerodynamic interaction or internal sources;</p> <p>d) thermal cycles;</p> <p>e) accidental and discrete source damage;</p> <p>f) likely corrosion or other degradation;</p> <p>g) specified maintenance; and</p> <p>h) likely structural repairs.</p>	CS 25.571, CS 23.571-575	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.8.2	3.8.2 For aeroplanes for which application for certification was submitted on or after 24 February 2013,	CS 25.571, 627CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>the design and construction of the aeroplane shall, wherever practicable, conform to damage tolerance and failsafe principles and shall be such as to avoid catastrophic failure during the operational life, taking into account:</p> <ul style="list-style-type: none"> a) the expected environment; b) the expected repeated loads applied in service; c) expected vibrations from aerodynamic interaction or internal sources; d) thermal cycles; e) accidental and discrete source damage; f) likely corrosion or other degradation; g) widespread fatigue damage; h) specified maintenance; and i) likely structural repairs. <p><i>Note.— The expression “wherever practicable” is introduced to ensure that when an effective damage-tolerant structure cannot be achieved within the limitations of geometry, inspectability or good design practice, the structure can be designed to the fatigue evaluation (safe-life) principles. Typical examples of structures that might not be amenable to damage-tolerant</i></p>	23.571-575, 627							
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	<i>design are landing gear, engine mounts and their attachments.</i>									
3.9	3.9 Special factors For aeroplanes for which application for certification was submitted on or after 24 February 2013, the design features (e.g. castings, bearings or fittings), the strength of which is subject to variability in manufacturing processes, deterioration in service or any other cause, shall be accounted for by a suitable factor.	CS-25.619-625CS 23.619, 625	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.1.1	CHAPTER 4. DESIGN AND CONSTRUCTION 4.1 General 4.1.1 Details of design and construction shall be such as to give reasonable assurance that all aeroplane parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by	21.B.80Reg. (EU) 748/2012CS 25.601CS 25.1302CS 23.601CS 23.2250	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	The sentence 'consider Human Factors principles' is not fully complied with.	CS 25.1302 and AMC 25.1302 (Installed Systems and Equipment for Use by

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	<p>other appropriate investigations or both. They shall also consider human factors principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>								the Flight Crew), created at Amdt 3 of CS-25 effective on 19/02/2007 take into account potential design related human factors issues during the compliance demonstration. EASA may require, as part of the certification planning
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										process, that a specific evaluation, analysis, or assessment of a human factors issue becomes part of the demonstration that the design is in compliance with requirements.
4.1.2	4.1.2 Substantiation of moving parts The functioning of all moving parts essential to the safe operation of the aeroplane shall be demonstrated by suitable tests in order to ensure that they will function correctly under all operating conditions for such parts.	CS 25.601, 683, 685. CS 23.601, 683, 685	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.1.3	4.1.3 Materials All materials used in parts of the aeroplane essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design. The effect of the materials on the occupants of the aeroplane and other persons on the ground, and the environment in general, in normal and emergency situations, shall be taken into account.	CS 25.603; 613. CS 23.603; 613. Reg. (EU) 1907/2006	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.4	4.1.4 Manufacturing methods The methods of manufacturing and assembly shall be such as to produce a consistently sound structure which shall be reliable with respect to maintenance of strength in service.	CS 25.605, CS 23.605	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.5	4.1.5 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the aeroplane will receive.	CS 25.609, 571. CS 23.609, 571-575	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.1.6	4.1.6 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the aeroplane that require such attention, either periodically or after unusually severe operations.	CS 25.611, CS 23.611	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.2	4.2 Systems design features Special consideration shall be given to design features that affect the ability of the flight crew to maintain controlled flight. This shall include at least the following: a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such that: 1) each control and control system shall operate with the ease, smoothness and precision appropriate to its function; 2) continued safe flight and landing of the aeroplane shall not be prevented by: i) any single failure not shown to be extremely improbable in the control system; or	21.B.80Reg. (EU) 748/2012CS 25.685CS 25.671CS 23.685 CS 23.2300CS 25.777CS 25.779CS 25.781CS 23.777CS 23.779CS 23.781CS 23.2600CS 25.1302CS 25.773CS 23.773CS 23.2600CS 25.795CS 25.1301CS 25.1309CS 23.1301CS 23.1309CS	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Less protective for paragraphs (b), (g), (h) and (i). Protection against explosive and incendiary devices was not requested in the CS 25 amendments up to and including amendment 8.	Linked to Amdt109: The aim is to complete the transposition of the standard in paragraph (g)(4) before it becomes applicable to individual aeroplanes issued

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	<p>ii) any event that results in a jam of a flight control in any normally encountered position of the flight controls;</p> <p>3) the possibility of jamming, inadvertent operation and unintentional engagement of control surface locking devices is minimized; and</p> <p>4) each element of each flight control system is designed, or distinctively and permanently marked, to minimize the probability of any incorrect assembly that could result in the malfunction of the system.</p> <p>b) <i>System survivability.</i></p> <p>1) For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60, aeroplane systems shall be designed, arranged and physically separated to maximize the potential for continued safe flight and landing after any event resulting in damage to the aeroplane structure or systems.</p> <p>2) Recommendation.— <i>For aeroplanes of a maximum certificated take-off mass in excess of 5 700 kg but not exceeding 45 500 kg, aeroplane systems should be designed, arranged and physically separated to maximize the potential for continued safe flight and landing after any event resulting in damage to the aeroplane structure or systems.</i></p>	<p>23.2605CS 25.851-869CS 23.851-865CS 23.325CS 25.856 (a)CS 25.855-858CS 23.855CS 23.2325CS 25.831CS 25.856 (a)CS 25.831CS 25.841CS 23.831CS 23.841CS 23.2510CS-25 Appendix F Part V</p>						<p>Linked to Amdt109: Paragraph(4) has not been implemented</p>	<p>with certificate or airworthiness on or after 1 January 2025.</p>
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	<p>c) <i>Crew environment.</i> The design of the flight crew compartment shall be such as to minimize the possibility of incorrect or restricted operation of the controls by the crew, due to fatigue, confusion or interference. Consideration shall be given at least to the following: layout and identification of controls and instruments, rapid identification of emergency situations, sense of controls, ventilation, heating and noise.</p> <p>d) <i>Pilot vision.</i> The arrangement of the flight crew compartment shall be such as to afford a sufficiently extensive, clear and undistorted field of vision for the safe operation of the aeroplane, and to prevent glare and reflections that would interfere with the pilot's vision. The design features of the windshield shall permit, under precipitation conditions, sufficient vision for the normal conduct of flight and for the execution of approaches and landings.</p> <p>e) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the flight crew to deal with, emergencies resulting from foreseeable failures of equipment and systems, the failure of which would endanger the aeroplane. Reasonable provisions shall be made for continuation of essential services following engine or system failures to the extent that such failures are catered for in the performance and operating limitations specified in the Standards in this Annex and in Annex 6, Parts I and II.</p> <p>f) <i>Fire precautions.</i></p>								
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	<p>1) The design of the aeroplane and the materials used in its manufacture shall be such so as to minimize the risk of in-flight and ground fires, to minimize the production of smoke and toxic gases in the event of a fire and to delay the occurrence of flashover resulting from heat release in the cabin. Means shall be provided to contain or to detect and extinguish such fires as might occur in such a way that no additional danger to the aeroplane is caused. Lavatories installed in aeroplanes shall be equipped with a smoke detection system and a built-in fire extinguisher system for each receptacle intended for the disposal of towels, paper or waste.</p> <p>2) For aeroplanes for which application for certification was submitted on or after 24 February 2013, design precautions shall be taken to minimize the risk of an uncontained fire initiating in areas of the aeroplane that contain high concentrations of wiring or equipment that are not normally accessible in flight.</p> <p><i>Note.— Design precautions may include the selection of appropriate materials and types of equipment installed in these areas, as well as the reduction of possible ignition sources, typically by preventing the ingress of fuel or fuel vapour, upgrading the flammability requirements of aircraft wiring or improving the detection of overheating or smoke and the indication of its presence to the flight crew, etc.</i></p> <p>g) <i>Cargo compartment protection.</i></p>								
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	<p>1) Each cargo compartment shall be equipped with a built-in fire detection system, and a means to suppress a fire, except when the presence of a fire would be easily discovered by a crew member while at their station and the crew member has a means to extinguish it rapidly.</p> <p>2) The means to suppress a fire for each cargo compartment not accessible to a crew member shall include a built-in fire suppression system.</p> <p>3) For aeroplanes of a maximum certificated take-off mass in excess of 45 000 kg or with a passenger seating capacity greater than 60, cargo compartment fire suppression systems, including their extinguishing agents, shall be designed so as to take into account a sudden and extensive fire such as could be caused by an explosive or incendiary device.</p> <p>4) For those aeroplanes for which the individual certificate of airworthiness is first issued on or after 1 January 2025, the elements of the aeroplane design associated with cargo compartment fire protection, and a summary of the demonstrated standards that were considered in the process of aeroplane certification shall be included in the required aeroplane documentation and made available to the operator.</p> <p><i>Note.— Guidance material on the elements of cargo compartment fire protection and associated demonstrated standards is contained in the Guidance for</i></p>								
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	<p>Safe Operations Involving Aeroplane Cargo Compartments (Doc 10102).</p> <p>h) <i>Incapacitation of occupants.</i></p> <p>1) For aeroplanes for which application for certification was submitted on or after 24 February 2013, design precautions shall be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases that could incapacitate the occupants of the aeroplane.</p> <p>2) In addition, for aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60, design precautions shall be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases caused by explosive or incendiary devices or dangerous goods which could incapacitate the occupants of the aeroplane.</p> <p>3) Recommendation.— <i>For aeroplanes of a maximum certificated take-off mass in excess of 5 700 kg but not exceeding 45 500 kg, design precautions should be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases, including those caused by explosive or incendiary devices or dangerous goods, which could incapacitate the occupants of the aeroplane.</i></p>								
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	<p>i) <i>Protection of the flight crew compartment from smoke and fumes.</i></p> <p>1) For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60, means shall be provided to minimize entry into the flight crew compartment of smoke, fumes and noxious vapours generated by an explosion or fire on the aeroplane.</p> <p>2) Recommendation.— <i>For aeroplanes of a maximum certificated take-off mass in excess of 5 700 kg but not exceeding 45 500 kg, means should be provided to minimize entry into the flight crew compartment of smoke, fumes and noxious vapours generated by an explosion or fire on the aeroplane.</i></p>									
4.3	<p>4.3 Aeroelasticity</p> <p>The aeroplane shall be free from flutter, structural divergence, and loss of control due to structural deformation and aeroelastic effects, at all speeds within and sufficiently beyond the design envelope to comply with 1.3.1. Account shall be taken of the characteristics of the aeroplane and variations in pilot skill and workload. Allowable limits for aerodynamic control surfaces and how those limits are to be monitored shall be specified so as to ensure that the aeroplane remains free from aeroelastic problems during its operational life.</p>	CS 25-629, CS 23-629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.4.1	<p>4.4 Occupants accommodation features</p> <p>4.4.1 Seating and restraints</p> <p>Adequate seating and restraints shall be provided for the occupants, taking account of the likely flight and emergency landing loads to be encountered. Attention shall be paid to minimizing injury to occupants due to contact with surrounding structures during the operation of the aeroplane.</p>	CS 25.561, 562, 785, CS 23.561, 562, 785	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.4.2	<p>4.4.2 Cabin environment</p> <p>Ventilation, heating and, where applicable, pressurization systems shall be designed to provide the cabin with an adequate environment during the anticipated flight and ground or water operating conditions. The systems design shall also consider likely emergency conditions.</p>	CS 25.831, 832, 841, 1309. CS 23.831, 841, 1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.5.1	<p>4.5 Electrical bonding and protection against lightning and static electricity</p> <p>4.5.1 Electrical bonding and protection against lightning and static electricity shall be such as to:</p> <p>a) protect the aeroplane, its systems, its occupants and those who come in contact with the</p>	CS 25-581, 899, 1316. CS-23 867	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	aeroplane on the ground or water from the dangerous effects of lightning discharge and electrical shock; and b) prevent dangerous accumulation of electrostatic charge.									
4.5.2	4.5.2 The aeroplane shall also be protected against catastrophic effects of lightning. Due account shall be taken of the material used in the construction of the aeroplane.	CS 25.581;899, 1316CS 23.867	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.1	4.6 Emergency landing provisions 4.6.1 Provisions shall be made in the design of the aeroplane to protect the occupants, in the event of an emergency landing, from fire and from the direct effects of deceleration forces as well as from injuries arising from the effect of deceleration forces on the aeroplane's interior equipment.	CS 25.561, 562. CS 23.561,562	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.2	4.6.2 Facilities shall be provided for the rapid evacuation of the aeroplane in conditions likely to occur following an emergency landing. Such facilities shall be related to the passenger and crew capacity of the aeroplane and shall be shown to be suitable for their intended purpose.	CS 25 801-819. CS 23 803-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.3	4.6.3 The interior layout of the cabin and the position and number of emergency exits, including the means of locating and illuminating the escape paths and exits, shall be such as to facilitate rapid evacuation of the	CS 25.791, 803-819. CS 23. 791-	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	aeroplane in conditions likely to occur following an emergency landing.	815								
4.6.4	4.6.4 On aeroplanes certificated for ditching conditions, provisions shall be made in the design to give maximum practicable assurance that safe evacuation from the aeroplane of passengers and crew can be executed in case of ditching.	CS 25.801, 563, 807(i), CS 23.807(e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.7	4.7 Ground handling Adequate provisions shall be made to minimize the risk that normal ground handling operations (e.g. towing, jacking) may cause damage, which could pass unnoticed, to the parts of the aeroplane essential for its safe operation. The protection that any limitations and instructions for such operations might provide may be taken into account.	CS 25.489, 509, 519. CS 23.507,509	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.1	CHAPTER 5. POWERPLANT 5.1 Engines	CS-E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	The Standards of Part VI of this Annex shall apply to each engine that is used on the aeroplane as a primary propulsion unit.									
5.2	5.2 Propellers The Standards of Part VII of this Annex shall apply to each propeller that is used on the aeroplane.	CS-P	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.1	5.3 Powerplant installation 5.3.1 Compliance with engine and propeller limitations The powerplant installation shall be so designed that the engines and propellers (if applicable) are capable of functioning reliably in the anticipated operating conditions. In conditions established in the flight manual, the aeroplane shall be capable of being operated without exceeding the limitations established for the engines and propellers in accordance with this chapter and with Parts VI and VII.	CS E-20. CS 25.1581. CS 23.1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.2	5.3.2 Control of engine rotation In those installations where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided for the crew to	CS 25.903 (c). CS 23.903 (d), (e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	stop the rotation of the failed engine in flight or to reduce it to a safe level.									
5.3.3	<p>5.3.3 Turbine engine installation</p> <p>For a turbine engine installation:</p> <p>a) the design shall minimize the hazards to the aeroplane in the event of failure of engine rotating parts, or an engine fire which burns through the engine case; and</p> <p>b) the powerplant installation shall be designed to give reasonable assurance that those engine operating limitations that adversely affect the structural integrity of rotating parts shall not be exceeded in service.</p>	CS25.903 (d), CS 23.903(b), CS E 130, 510, 590, 810, 840, 850. CS E.50, 830, 860, 870, 920	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.4	<p>5.3.4 Engine restarting</p> <p>Means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.</p>	CS-25.903 (e), CS 23.903 (g)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.1	<p>5.3.5 Arrangement and functioning</p> <p>5.3.5.1 <i>Independence of engines.</i> For aeroplanes for which application for certification was submitted before 24 February 2013, the powerplant shall be arranged and installed so that each engine together with its associated systems is capable of being controlled</p>	CS 25.903 (b), CS 23.903 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	and operated independently from the others and so that there is at least one arrangement of the powerplant and systems in which any failure, unless the probability of its occurrence is extremely remote, cannot result in a loss of more power than that resulting from complete failure of the critical engine.									
5.3.5.2	<p>5.3.5.2 <i>Independence of engines and associated systems.</i> For aeroplanes for which application for certification was submitted on or after 24 February 2013, the engines together with their associated systems shall be arranged and isolated from each other to allow operation, in at least one configuration, so that the failure or malfunction of any engine, or of any system that can affect the engine, will not:</p> <p>a) prevent the continued safe operation of the remaining engine(s); or</p> <p>b) require immediate action by any crew member for continued safe operation of the remaining engine(s).</p>	CS 25.903 (b). CS 23.903 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.3	<p>5.3.5.3 <i>Propeller vibration.</i> The propeller vibration stresses shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the aeroplane.</p>	CS 25.907, CS 23.907	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.4	<p>5.3.5.4 <i>Cooling.</i> The cooling system shall be capable of maintaining the temperature of powerplant components and fluids within the established limits (see 5.3.1) at ambient air temperatures up to the maximum air</p>	CS 25.1041 25.1521(d),1583(b). CS 23.1041	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	temperature appropriate to the intended operation of the aeroplane. The maximum and, if necessary, minimum ambient air temperature for which the powerplant has been established as being suitable shall be scheduled in the flight manual.									
5.3.5.5	5.3.5.5 <i>Associated systems.</i> The fuel, oil, air induction and other systems associated with the powerplant shall be capable of supplying each engine in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power or thrust, aeroplane attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.	CS 25.951-1001. CS 25.1011-1027. CS 25.1091-1103. CS 25.1121-1123. CS 23.951-1001. CS 23.1011-1027. CS 23.1091-1111. CS 23.1121-1125	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.6	5.3.5.6 <i>Fire protection.</i> For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.2 f): a) <i>Isolation.</i> Such regions shall be isolated by fireproof material from other regions of the aeroplane where the presence of fire would jeopardize continued flight, taking into account the probable points of origin and paths of propagation of fire.	CS 25.1181-1207. CS 23.1181-1203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be fire resistant. Drainage of each region shall be provided to minimize hazards resulting from the failure of any component containing flammable fluids. Means shall be provided for the crew to shut off the flow of flammable fluids into such regions if a fire occurs. Where sources of flammable fluid exist in such regions, the whole of the related system within the region, including supporting structure, shall be fireproof or shielded from the effects of the fire.</p> <p>c) <i>Fire detection.</i> A sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions.</p> <p>d) <i>Fire extinguishment.</i> Such regions shall be provided with a fire extinguisher system capable of extinguishing any fire likely to occur therein, unless the degree of isolation, quantity of combustibles, fire resistance of the structure and other factors are such that any fire likely to occur in the region would not jeopardize the safety of the aeroplane.</p>									
6.1.1	CHAPTER 6. SYSTEMS AND EQUIPMENT 6.1 General	CS 25/23.1303. CS.25.1302 25/23.1305 25/23.1307	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>6.1.1 The aeroplane shall be provided with approved instruments, equipment and systems, including guidance and flight management systems necessary for the safe operation of the aeroplane in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the crew to operate the aeroplane within its operating limitations. Instruments and equipment design shall observe human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Parts I and II, for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	25/23.1327 25/23.1337								
6.1.2	<p>6.1.2 The design of the instruments, equipment and systems required by 6.1.1 and their installation shall be such that:</p> <p>a) an inverse relationship exists between the probability of a failure condition and the severity of its effect on the aircraft and its occupants, as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p>	CS 25.1301; .1309; .1353; .1431. CS 23.1301; .1309; .1353; .1431.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	c) electromagnetic interference between them is minimized.									
6.1.3	6.1.3 Means shall be provided to warn the crew of unsafe system operating conditions and to enable them to take corrective action.	CS 23.1309(b,3) CS 25.1309(c), 1322	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.1.4	6.1.4 Electrical power supply The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations of the aeroplane and essential power loads after failures that affect the electrical generating system and under expected environmental conditions.	CS 25.1309 (c), CS 25.1322 1310, 1351, 1353, 1355, 1357, 1362, CS 23.1309, 1351, 1361	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.1.5	6.1.5 Development assurance of complex electronic hardware and system software For aeroplanes for which application for certification was submitted on or after 24 February 2013, complex electronic hardware and system software shall be developed, verified and validated such as to ensure that the systems in which they are used perform their intended functions at a level of safety that complies with the requirements of this section, notably those of 6.1.2 a) and 6.1.2 b).	AMC 25.1309, AMC 20-115, AMC 20-152 AMC1 23.2510	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>Note.— Some States accept the use of national or international industry standards for the development assurance (development, verification and validation) of complex electronic hardware and systems software.</i>									
6.2	6.2 Installation Instrument and equipment installations shall comply with the Standards of Chapter 4.	CS 25.601, 1301, 1309 CS 23.60, 1301, 1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.3	6.3 Safety and survival equipment Prescribed safety and survival equipment that the crew or passengers are expected to use or operate at the time of an emergency shall be reliable, readily accessible and easily identified, and its method of operation shall be plainly marked.	CS 25.1411, 1415, 1421, 1423, CS 23.1411, 1415.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.4.1	6.4 Navigation lights and anti-collision lights 6.4.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by aeroplanes in flight or operating on the movement area of an aerodrome shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft or personnel on the ground with as much time as possible for interpretation and for subsequent manoeuvre	CS 25.1385-1401, CS 23 1385-1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions. <i>Note.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry sky, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i>									
6.4.2	6.4.2 Lights shall be installed in aeroplanes so as to minimize the possibility that they will adversely affect the satisfactory performance of the flight crews' duties. <i>Note.— In order to avoid the effects mentioned in 6.4.2, it will be necessary in some cases to provide means whereby the pilot can adjust the intensity of the flashing lights.</i>	CS 25. 1383, 1401, 1403. CS 23.1383, 1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.5	6.5 Electromagnetic interference protection Aeroplane electronic systems, particularly flight-critical and flight-essential systems, shall be protected against electromagnetic interference from both internal and external sources.	CS 25 Amdt 17 : CS 25.1316, CS 25.1317, Appendix R; CS 23 Amdt 4 : CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		23.1306, CS 23.1308, Appendix K; CS 23 Amdt 5 : CS 23.2515, CS 23.2520, AMC 20-136, AMC 20-158.								
6.6	6.6 Ice protection If certification for flight in icing conditions is requested, the aeroplane shall be shown to be able to operate safely in icing conditions likely to be encountered in all anticipated operating environments.	CS 25.1419, CS 23.1419	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.1	CHAPTER 7. OPERATING LIMITATIONS AND INFORMATION 7.1 General The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the aeroplane, shall be made available by means of a	CS-23 Subpart G. CS-25 Subpart G. CS 25.1541, 1563, 1581. CS 23. 1541, 1563, 1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	flight manual, markings and placards, and such other means as may effectively accomplish the purpose.									
7.2.1	<p>7.2 Operating limitations</p> <p>7.2.1 Limitations which might be exceeded in flight and which are defined quantitatively shall be expressed in suitable units. These limitations shall be corrected if necessary for errors in measurements so that the flight crew can, by reference to the instruments available to them, readily determine when the limitations are reached.</p>	CS 25.1541. CS 23.1541	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.2	<p>7.2.2 Loading limitations</p> <p>The loading limitations shall include all limiting masses, centre of gravity positions, mass distributions and floor loadings (see 1.3.2).</p>	CS 25.1519, 1533 CS 23.1519, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.3	<p>7.2.3 Airspeed limitations</p> <p>The airspeed limitations shall include all speeds (see 3.5) that are limiting from the standpoint of structural integrity or flying qualities of the aeroplane, or from other considerations. These speeds shall be identified with respect to the appropriate aeroplane configurations and other pertinent factors.</p>	CS 25.1503-1517. CS 23.1501-1513	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.2.4	7.2.4 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the aeroplane (see 5.3.1 and 5.3.5.4).	CS 25.1521, CS 23.1521, 1522	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.5	7.2.5 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the aeroplane.	CS 23.1501, CS 25.1501	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.6	7.2.6 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the aeroplane (see 1.3.1).	CS 25.1533, 1501 CS 23.1501	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.7	7.2.7 Flight crew limitations The flight crew limitations shall include the minimum number of flight crew personnel necessary to operate the aeroplane, having regard, among other things, to the accessibility to the appropriate crew members of all necessary controls and instruments and to the execution of the established emergency procedures. <i>Note.— The circumstances in which the flight crew shall include members in addition to the minimum flight crew are defined in Annex 6, Parts I and II.</i>	CS 25.1523, CS 23.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.2.8	<p>7.2.8 Flying time limitation after system or engine failure</p> <p>The systems limitations shall include the maximum flying time for which system reliability has been established in relation to the approval of operations by aeroplanes with two turbine engines beyond the threshold time established in accordance with 4.7 of Annex 6, Part I.</p> <p><i>Note.— The maximum time established in accordance with 4.7 of Annex 6, Part I, for a particular route may be less than that determined in accordance with 7.2.8 because of the operational considerations involved.</i></p>	AMC 20	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.1	<p>7.3 Operating information and procedures</p> <p>7.3.1 Types of eligible operations</p> <p>The particular types of operations for which the aeroplane has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.</p>	CS 25.1525, CS 23.1525	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.2	7.3.2 Loading information	CS 25.1519. CS 23.1519	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
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	<p>The loading information shall include the empty mass of the aeroplane, together with a definition of the condition of the aeroplane at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the crew and payload, the usable fuel supply and the drainable oil; it includes the mass of all fixed ballast, unusable fuel supply, undrainable oil, total quantity of engine coolant and total quantity of hydraulic fluid.</i></p>									
7.3.3	<p>7.3.3 Operating procedures</p> <p>A description shall be given of normal and emergency operating procedures which are peculiar to the particular aeroplane and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.</p>	CS 25.1583, 1585. CS 23.1583, 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.4	<p>7.3.4 Handling information</p> <p>Sufficient information shall be given on any significant or unusual features of the aeroplane characteristics. Those stalling speeds or minimum steady flight speeds required to be established by 2.4.2.3 shall be scheduled.</p>	CS 25. 1581, 1583, 1545, 1557. CS 23. 1581, 1583, 1545, 1557	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.5	<p>7.3.5 Least-risk bomb location</p>	CS 25.795 (c)(1)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60, a least-risk location on the aeroplane shall be identified where a bomb or other explosive device may be placed to minimize the effects on the aeroplane in the case of detonation.									
7.4	7.4 Performance information The performance of the aeroplane shall be scheduled in accordance with 2.2. There shall be included information regarding the various aeroplane configurations and powers or thrusts involved and the relevant speeds, together with information that would assist the flight crew in attaining the performance as scheduled.	CS 25.1587, 1591. CS 23.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.5	7.5 Flight manual A flight manual shall be made available. It shall identify clearly the specific aeroplane or series of aeroplanes to which it is related. The flight manual shall include at least the limitations, information and procedures specified in 7.2, 7.3, 7.4 and 7.6.1.	CS 25.1581. CS 23.1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.1	7.6 Markings and placards	CS 25.1541-1563. CS 23.1541-	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	7.6.1 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the flight crew during flight.	1567								
7.6.2	7.6.2 Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in ground servicing (e.g. towing, refuelling) that could pass unnoticed and that could jeopardize the safety of the aeroplane in subsequent flights.	CS 25.1557 and Appendix H. CS 23. 1557 and Appendix G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.1	7.7 Continuing airworthiness — maintenance information 7.7.1 General Information for use in developing procedures for maintaining the aeroplane in an airworthy condition shall be made available. The information shall include that described in 7.7.2, 7.7.3 and 7.7.4.	CS 25 1529 and Appendix H. CS 23. 1529 and Appendix G. Part M, Reg. (EU) 1321/2014.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.2	7.7.2 Maintenance information Maintenance information shall include a description of the aeroplane and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.	CS 25 Appendix H. CS 23 Appendix GCS 25.1529, CS 23.1529	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.7.3	<p>7.7.3 Maintenance programme information</p> <p>Maintenance programme information shall include the maintenance tasks and recommended intervals at which these tasks are to be performed.</p> <p><i>Note.— The development of initial maintenance programme information at the time of aeroplane type certification is sometimes referred to as the Maintenance Review Board (MRB) process.</i></p>	<p>M.A.302 Reg. (EU) 1321/2014. CS 25 Appendix H. CS 23 Appendix G CS 25.1529, CS 23.1529</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.4	<p>7.7.4 Mandatory maintenance requirements resulting from the type design approval</p> <p>Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 7.7.3.</p> <p><i>Note.— Mandatory requirements identified as part of the type design approval are often referred to as Certification Maintenance Requirements (CMR) and/or airworthiness limitations.</i></p>	<p>CS 25 Appendix H 4. CS 23 Appendix G 4</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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8.1	<p>CHAPTER 8. CRASHWORTHINESS AND CABIN SAFETY</p> <p style="text-align: center;">8.1 General</p> <p>Crashworthiness shall be taken into account in the design of aeroplanes to improve the probability of occupant survival.</p>	CS 25. 561, 562, 721, 783, 785, 789, 801. CS 23. 561, 562, 721, 783, 785.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.2.1	<p>8.2 Design emergency landing loads</p> <p>8.2.1 For aeroplanes for which application for certification was submitted before 24 February 2013, emergency landing (crash) loads shall be determined for all categories of aeroplanes so that the interiors, furnishings, support structure and safety equipment can be designed to maximize survivability for the occupants. Items to be considered shall include:</p> <p>a) dynamic effects;</p> <p>b) restraint criteria for items that could cause a hazard;</p> <p>c) distortion of the fuselage in the areas of emergency exits;</p> <p>d) fuel cell integrity and position; and</p>	CS 25. 561, 562, 721, 785, 789, 801, 787, 807, 1185, 1351. CS 23. 561, 562, 721, 785 787, 807(b), 967, 1351.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	e) integrity of electrical systems to avoid sources of ignition.									
8.2.2	<p>8.2.2 For aeroplanes for which application for certification was submitted on or after 24 February 2013, emergency landing (crash) loads shall be determined so that the interiors, furnishings, support structure and safety equipment can be designed to protect the occupants under emergency landing conditions. Items to be considered shall include:</p> <p>a) dynamic effects;</p> <p>b) restraint criteria for items that could cause a hazard;</p> <p>c) deformation of the fuselage in the areas of emergency exits;</p> <p>d) fuel cell integrity and position; and</p> <p>e) integrity of electrical systems to avoid sources of ignition.</p>	CS 25. 561, 562, 721, 785, 789, 801, 787, 807, 1185, 1351. CS 23. 561, 562, 721, 785 787, 807(b), 967, 1351.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.3	<p>8.3 Cabin fire protection</p> <p>The cabin shall be so designed as to provide fire protection to the occupants in the event of airborne systems failures or a crash situation. Items to be considered shall include:</p>	CS 25.851-869. CS 23.851-865	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Sigini- ficant Differ- ence			
				Level of implementation of SARPs						
				A) More Exactin- g or Exceed- s	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	<ul style="list-style-type: none"> a) flammability of cabin interior materials; b) fire resistance and the generation of smoke and toxic fumes; c) provision of safety features to allow for safe evacuation; and d) fire detection and suppression equipment. 									
8.4	<p style="text-align: center;">8.4 Evacuation</p> <p>The aeroplane shall be equipped with sufficient emergency exits to allow maximum opportunity for cabin evacuation within an appropriate time period. Items to be considered shall include:</p> <ul style="list-style-type: none"> a) number of seats and seating configuration; b) number, location and size of exits; c) marking of exits and provision of instructions for use; d) likely blockages of exits; e) operation of exits; and 	CS 25-801-819. CS 23.803-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	f) positioning and weight of evacuation equipment at exits, e.g. slides and rafts.									
8.5	<p>8.5 Lighting and marking</p> <p>Emergency lighting shall be provided and shall have the following characteristics:</p> <p>a) independence from main electrical supply;</p> <p>b) automatic activation upon loss of normal power/impact;</p> <p>c) visual indication of the path to emergency exits in smoke-filled cabin conditions;</p> <p>d) illumination both inside and outside the aeroplane during evacuation; and</p> <p>e) no additional hazard in the event of fuel spillage.</p>	CS 25.811-812. CS 25.811-812	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.6	<p>8.6 Survival equipment</p> <p>The aeroplane shall be so equipped as to provide the crew and occupants with the maximum opportunity to survive in the expected external environment for a reasonable time span. Items to be considered shall include:</p>	CAT.IDE.A.3 05 Reg. EU 965/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>		Only installation of such equipment is				

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	<ul style="list-style-type: none"> a) number of life rafts/life jackets; b) survival equipment suited to the likely environment; c) emergency radios and pyrotechnical distress signalling equipment; and d) automatic emergency radio beacons. 									covered into CS.
9.1	<p>CHAPTER 9. OPERATING ENVIRONMENT AND HUMAN FACTORS</p> <p style="text-align: center;">9.1 General</p> <p>The aeroplane shall be designed to allow safe operation within the performance limitations of its passengers and those who operate, maintain and service it.</p> <p><i>Note.— The human/machine interface is often the weak link in an operating environment; so, it is necessary to ensure that the aeroplane is capable of being controlled at all phases of the flight (including any degradation due to failures) and that neither the crew nor</i></p>	All Subpart B of CS-25 101, 143, 149, 161, 171, 203, 207, 251 and 25-611, 697, 771, 773, 779, 781, 783, 831, 832, 833, 841, 843, 901, 1141, A1141 1301, 1309, 1581, Appendix D.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>passengers are harmed by the environment in which they have been placed for the duration of the flight.</i>	All Subpart B of CS-23 45, 143, 149, 161, 171, 203, 207, 251 and 23-611, 697, 771, 773, 779, 781, 831, 841, 843, 901, 995, 1141, 1301, 1309, 1523, 1581							
9.2.1	9.2 Flight crew 9.2.1 The aeroplane shall be designed in such a way as to allow safe and efficient control by the flight crew. The design shall allow for variations in flight crew skill and physiology commensurate with flight crew licensing limits. Account shall be taken of the different expected operating conditions of the aeroplane in its environment, including operations degraded by failures.	All Subpart B of CS-25 101, 143, 149, 161, 171, 203, 207, 251 and 25-771, 773, 779, 781, 1141, A1141, 1301, 1309, 1581, Appendix D. All Subpart B of CS-23 23.45, 143, 149, 161, 171, 201, 203, 207, 251 and 23-771,	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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		773, 779, 781, 1141, 1301, 1309, 1501, 1523, 1581								
9.2.2	<p>9.2.2 The workload imposed on the flight crew by the design of the aeroplane shall be reasonable at all stages of flight. Particular consideration shall be given to critical stages of flight and critical events which may reasonably be expected to occur during the service life of the aeroplane, such as a contained engine failure or windshear encounter.</p> <p><i>Note.— Workload can be affected by both cognitive and physiological factors.</i></p>	<p>All Subpart B of CS-25 101, 143, 149, 161, 171, 203, 207, 251 and 25-771, 773, 779, 781, 1141, A1141, 1301, 1309, 1581, Appendix D. All Subpart B of CS-23 23.45, 143, 149, 161, 171, 201, 203, 207, 251 and 23-771, 773, 779, 781, 1141, 1301, 1309, 1501, 1523, 1581</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.3	9.3 Ergonomics	CS 25.143, 611, 697, 771, 773,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
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	<p>During design of the aeroplane, account shall be taken of ergonomic factors including:</p> <p>a) ease of use and prevention of inadvertent misuse;</p> <p>b) accessibility;</p> <p>c) flight crew working environment;</p> <p>d) cockpit standardization; and</p> <p>e) maintainability.</p>	779, 781, 783, 901, 1141, A1141, 1302. CS 23.143, 611, 697, 771, 773, 777, 779, 781, 785, 791, 901, 1141, 1155, 1367, 1401, 841, 1321, 1322							
9.4	<p>9.4 Operating environmental factors</p> <p>The design of the aeroplane shall take into consideration the flight crew operating environment including:</p> <p>a) effect of aeromedical factors such as level of oxygen, temperature, humidity, noise and vibration;</p> <p>b) effect of physical forces during normal flight;</p> <p>c) effect of prolonged operation at high altitude; and</p> <p>d) physical comfort.</p>	All Subpart B of CS-25 25.101, 143, 149, 161, 171, 203, 251 and 25.771, 773, 779, 781, 831, 832, 833, 841, 843, 1141, A1141, 1301, 1309, 1581, Appendix D. All Subpart B of CS-23	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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		23.45, 143, 149, 161, 171, 203, 251 and 23.771, 773, 779, 781, 831, 841, 843, 1141, 1301, 1309, 1523, 1581								
10.1.1	CHAPTER 10. SECURITY 10.1 Aeroplanes used for domestic commercial operations Recommendation. — <i>International Standards and Recommended Practices set forth in this chapter should be applied by all Contracting States for aeroplanes engaged in domestic commercial operations (air services).</i>	CS 25.772, 795.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
10.2	10.2 Least-risk bomb location For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60, consideration shall be given during the design of the aeroplane to the provision of a	CS 25.795	<input checked="" type="checkbox"/>	<input type="checkbox"/>	New security provisions of CS 25.795 introduced by	Implemented in 2010 instead of 2004.				

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	least-risk bomb location so as to minimize the effects of a bomb on the aeroplane and its occupants.								Amendment 9 to CS25 effective 12 August 2010. This includes 'least-risk bomb location'..	
10.3.1	<p>10.3 Protection of the flight crew compartment</p> <p>10.3.1 In all aeroplanes, which are required by Annex 6, Part I, Chapter 13 to have an approved flight crew compartment door, and for which an application for the issue of a Type Certificate was submitted to the appropriate national authority on or after 20 May 2006, the flight crew compartment bulkheads, floors and ceilings shall be designed to resist penetration by small arms fire and grenade shrapnel and to resist forcible intrusions, if these areas are accessible in flight to passengers and cabin crew.</p>	CS 25.795 (a)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
10.3.2	<p>10.3.2 Recommendation.— <i>In all aeroplanes, which are required by Annex 6, Part I, Chapter 13 to have an approved flight crew compartment door, and for which an application for amending the Type Certificate to include a derivative type design was submitted to the appropriate national authority on or after 20 May 2006, consideration should be given to reinforcing the flight crew compartment bulkheads, floors and ceilings so as to</i></p>	CS 25.795 (a). 21.A.101 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>resist penetration by small arms fire and grenade shrapnel and to resist forcible intrusions, if these areas are accessible in flight to passengers and cabin crew.</i></p> <p><i>Note.— Standards and Recommended Practices concerning the requirements for the flight crew compartment door in all commercial passenger-carrying aeroplanes are contained in Annex 6, Part I, Chapter 13.</i></p>									
10.4	<p>10.4 Interior design</p> <p>For aeroplanes of a maximum certificated take-off mass in excess of 45 500 kg or with a passenger seating capacity greater than 60, consideration shall be given to design features that will deter the easy concealment of weapons, explosives or other dangerous objects on board aircraft and that will facilitate search procedures for such objects.</p>	CS 25.795 (c)(3)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.1	<p>PART IV. HELICOPTERS</p> <p>PART IVA. HELICOPTERS FOR WHICH APPLICATION FOR CERTIFICATION WAS SUBMITTED ON OR AFTER 22 MARCH 1991 BUT BEFORE 13 DECEMBER 2007</p>	Reg. (EU) 2018/1139Reg. (EU) 748/2012. CS-27, CS-29	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>Note.— The provisions of Part IVA are the same as those contained in Part IV of Annex 8, Ninth Edition except for modified applicability clauses and cross references.</i></p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all helicopters designated in 1.1.2 that are of types of which the prototype was submitted to the appropriate national authorities for certification on or after 22 March 1991 but before 13 December 2007.</p>									
1.1.2	<p>1.1.2 The Standards of this part shall apply to helicopters intended for the carriage of passengers or cargo or mail in international air navigation.</p> <p><i>Note.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p>	CS 29.1. CS 27.1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.1.3	1.1.3 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the helicopters designated in 1.1.2 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.	CS 29. CS 27748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.4	1.1.4 Unless otherwise stated, the Standards apply to the complete helicopter including its powerplant, systems and equipment.	CS 29.1. CS 27.1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.1	1.2 Limitations 1.2.1 Limiting conditions shall be established for the helicopter, its powerplant and its equipment (see 9.2). Compliance with the Standards of this part shall be established assuming that the helicopter is operated within the limitations specified. The limitations shall be sufficiently removed from any conditions prejudicial to the safety of the helicopter to render the likelihood of accidents arising therefrom extremely remote.	CS 29.25, 27, 33, 1501, 1529, CS 27.25, 27, 33, 1501, 1529	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.2	1.2.2 Limiting ranges of mass, centre of gravity location, load distribution, speeds and ambient conditions shall be established within which compliance with all the pertinent Standards in this part is shown, except that combinations of conditions which are fundamentally impossible to achieve need not be considered.	CS 29.25,27,29. CS 27.25, 27, 29	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each separate operating condition, e.g. take-off, en route, landing.</i></p> <p><i>Note 2.— The following items, for instance, may be considered as basic helicopter limitations:</i></p> <ul style="list-style-type: none"> — <i>maximum certificated take-off (including lift-off) mass;</i> — <i>maximum certificated ground-taxiing mass;</i> — <i>maximum certificated landing mass;</i> — <i>most forward, rearward and lateral centre of gravity positions in various configurations; and</i> — <i>maximum certificated cargo sling mass.</i> <p><i>Note 3.— Maximum operating mass may be limited by the application of Noise Certification Standards (see Annex 16 — Environmental Protection, Volume I — Aircraft Noise, and Annex 6 — Operation of Aircraft, Part III — International Operations — Helicopters).</i></p>									
1.3	1.3 Unsafe features and characteristics	21.A.21, 16 Reg. (EU) 748/2012. CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Sigini- ficant Differ- ence			
				Level of implementation of SARPs						
				A) More Exactin- g or Exceed- s	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

	Under all anticipated operating conditions, the helicopter shall not possess any feature or characteristic that renders it unsafe.	29.601. CS 27.601								
1.4.1	1.4 Proof of compliance 1.4.1 Compliance with the appropriate airworthiness requirements shall be based on evidence either from tests, calculations, calculations based on tests, or other methods, provided that in each case the accuracy achieved will ensure a level of airworthiness equal to that which would be achieved were direct tests conducted.	21.A.20, 33, 35 Reg. (EU) 748/2012. CS 29.21, 307, 601. CS 27.21, 307, 601	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4.2	1.4.2 The tests of 1.4.1 shall be such as to provide reasonable assurance that the helicopter, its components and equipment are reliable and function correctly under the anticipated operating conditions.	21.A.20, 33, 35 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.1	CHAPTER 2. FLIGHT 2.1 General 2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or	CS-27 Subpart B. CS-29 Subpart B. CS 29.21. CS 27.21	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	other tests conducted upon a helicopter or helicopters of the type for which a Certificate of Airworthiness is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.									
2.1.2	2.1.2 Compliance with each Standard shall be established for all applicable combinations of helicopter mass and centre of gravity position, within the range of loading conditions for which certification is sought.	CS 29.21. CS 27.21	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.3	2.1.3 Where necessary, appropriate helicopter configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the helicopter's flying qualities.	CS 29.21. CS 27.21	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1.1	<p>2.2 Performance</p> <p>2.2.1 General</p> <p>2.2.1.1 Sufficient data on the performance of the helicopter shall be determined and scheduled in the flight manual to provide operators with the necessary information for the purpose of determining the total mass of the helicopter on the basis of the values, peculiar to the proposed flight, of the relevant operational parameters, in order that the flight may be made with reasonable</p>	CS 29.1581, 1587. CS 27.1581, 1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	assurance that a safe minimum performance for that flight will be achieved.									
2.2.1.2	<p>2.2.1.2 Achieving the performance scheduled for the helicopter shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the pilot.</p> <p><i>Note.— Guidance material on human performance can be found in the Human Factors Training Manual (Doc 9683).</i></p>	CS 29.45, 51, 141(b). CS 27.51, 75. Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1.3	2.2.1.3 The scheduled performance of the helicopter shall be consistent with compliance with 1.2.1 and with the operation in logical combinations of those of the helicopter's systems and equipment, the operation of which may affect performance.	CS 29.45, 1309, 1587. CS 27.45, 1309, 1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.2	<p>2.2.2 Minimum performance</p> <p>At the maximum mass scheduled (see 2.2.3) for take-off and for landing as functions of the take-off or landing site elevation or pressure-altitude either in the standard atmosphere or in specified still air atmospheric conditions, and, for water operations, in specified conditions of smooth water, the helicopter shall be capable of accomplishing the minimum performances specified in 2.2.2.1 and 2.2.2.2, respectively, not considering obstacles, or final approach and take-off area length.</p>	CS 29 Subpart B. CS 27 Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>Note.— This Standard permits the maximum take-off mass and maximum landing mass to be scheduled in the helicopter flight manual against, for example at the take-off or landing site:</i></p> <ul style="list-style-type: none"> — <i>elevation, or</i> — <i>pressure-altitude, or</i> — <i>pressure-altitude and atmospheric temperature,</i> <p><i>so as to be readily usable when applying the national code on helicopter performance operating limitations.</i></p>									
2.2.2.1	<p>2.2.2.1 <i>Take-off</i></p> <p>a) In the event of critical engine failure, at or after the take-off decision point (for performance Class 1) or the defined point after take-off (for performance Class 2), performance Classes 1 and 2 helicopters shall be capable of continuing safe flight, the remaining engine(s) being operated within the approved limitations.</p> <p>b) The minimum performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is scheduled (see 2.2.3), the departure from the scheduled values is not disproportionate.</p>	21.B.80Reg. (EU) 748/2012 CS 29.53CS 27.67CS-27 Appendix C CS 29.45 (a) (2) CS 27.51 (a) (1)	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address Category A and Category B Helicopters and not class 1, 2 and 3.	

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2.2.2.2	<p>2.2.2.2 <i>Landing</i></p> <p>a) Starting from the approach configuration, in the event of critical engine failure at or before the landing decision point (performance Class 1) or the defined point before landing (performance Class 2), the helicopter shall be capable of continuing safe flight, the remaining engine(s) being operated within the approved limitations.</p> <p>b) Starting from the landing configuration, the helicopter shall be capable, in the event of a balked landing, of making a climb-out, with all engines operating.</p>	21.B.80 Reg. (EU) 748/2012CS 27.65CS 27.75CS-27 Appendix CCS 29.65CS 29.79	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2 and 3.	
2.2.3	<p>2.2.3 Scheduling of performance</p> <p>Performance data shall be determined and scheduled in the flight manual so that its application by means of the operating rules to which the helicopter is to be operated in accordance with 5.1.2 of Annex 6, Part III, will provide a safe relationship between the performance of the helicopter and the aerodromes, heliports and routes on which it is capable of being operated. Performance data shall be determined and scheduled for the following stages for the ranges of mass, altitude or pressure-altitude, wind velocity, and other ambient conditions and any other operational variables for which the helicopter is to be certificated, and additionally for amphibians, water surface conditions and strength of current.</p>	CS 29.1587. CS 27.1587.	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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2.2.3.1	2.2.3.1 <i>Take-off.</i> The take-off performance data shall include the take-off distance required and the take-off path. For performance Class 1 helicopters, it shall also include the rejected take-off distance required.	21.B.80Reg. (EU) 748/2012CS 29.59CS 29.61CS 29.62CS 29.63CS-27 Appendix C	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2 and 3.	
2.2.3.1.1	2.2.3.1.1 <i>Take-off decision point.</i> (For performance Class 1 helicopters only.) The take-off decision point shall be the point in the take-off phase used in determining take-off performance and from which either a rejected take-off may be made or a take-off safely continued, with the critical engine inoperative.	21.B.80Reg. (EU) 748/2012CS 29.55CS-27 Appendix C	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2 and 3.	
2.2.3.1.2	2.2.3.1.2 <i>Take-off distance required.</i> (For performance Class 1 helicopters only.) The take-off distance required shall be the horizontal distance required from the start of the take-off to the point at which the take-off safety speed (V_{TOSS}), a selected height above the take-off surface, and a positive climb gradient are achieved, following failure of the critical engine at the take-off decision point, the remaining engine(s) operating within approved operating limits.	21.B.80 Reg. (EU) 748/2012CS 29.61CS-27 Appendix C	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2	

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									and 3.	
2.2.3.1.3	2.2.3.1.3 <i>Rejected take-off distance required.</i> (For performance Class 1 helicopters only.) The rejected take-off distance required shall be the horizontal distance required from the start of the take-off to the point where the helicopter comes to a complete stop following an engine failure and rejection of the take-off at the take-off decision point.	21.B.80Reg. (EU) 748/2012CS 29.62CS-27 Appendix C	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2 and 3.	
2.2.3.1.4	2.2.3.1.4 <i>Take-off distance required.</i> (For performance Classes 2 and 3 helicopters only.) The take-off distance required shall be the horizontal distance required from the start of take-off to the point where the best rate of climb speed (Vy) or the best angle of climb speed (Vx) or a selected intermediate speed (provided this speed does not involve flight within the avoid areas of the height-velocity diagrams) and a selected height above the take-off surface are achieved, all engines operating at approved take-off power.	21.B.80Reg. (EU) 748/2012CS 29.63CS 29.65CS-27 Appendix CCS 27.65	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2 and 3.	
2.2.3.2	2.2.3.2 <i>En route.</i> The en-route performance shall be the climb, cruise or descent performance with: a) the critical engine inoperative;	21.B.80Reg. (EU) 748/2012CS 29.43CS 29.65CS 29.67CS	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	(b) Not covered by CS-27 and 29	En-route performance is based on climb perform

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	<p>b) the two critical engines inoperative in the case of helicopters having three or more engines; and</p> <p>c) the operating engine(s) not exceeding the power for which they are certificated.</p>	<p>29.141CS 29.143CS 27.67CS-27 Appendix C</p>							<p>ance both for all engines operating and one engine inoperative situations. The case of the two critical power units inoperative for helicopters having three or more engines is not addressed. Concerns only helicopters with 3 or</p>
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									more engines. This standard has been reviewed and found to offer no safety benefit. It has therefore been removed from Part IVA (Amendment 100) and hence the identified difference will also be removed for helicopter certification
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Sigini- ficant Differ- ence			
				Level of implementation of SARPs						
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										ed after Decemb- er 2007.
2.2.3.3	2.2.3.3 <i>Landing.</i> The landing performance data shall include the landing distance required and, for performance Class 1 helicopters, the landing decision point.	CS 29.81, 77, 83. CS 27.75. Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
2.2.3.3.1	2.2.3.3.1 <i>Landing decision point.</i> (For performance Class 1 helicopters only.) The landing decision point shall be the latest point in the approach phase from which either a landing may be made or a rejected landing (go-around) safely initiated, with the critical engine inoperative.	21.B.80Reg. (EU) 748/2012CS 29.77CS-27 Appendix C	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	CS-27 and CS-29 address category A and Category B Helicopters and not class 1, 2 and 3.	
2.2.3.3.2	2.2.3.3.2 <i>Landing distance required.</i> The landing distance required shall be the horizontal distance required to land and come to a complete stop from a point on the approach flight path at a selected height above the landing surface.	CS 29.81, 83. CS 27 Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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2.3	2.3 Flying qualities The helicopter shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the helicopter is approved.	CS 29.141(a). CS 27.141 (a)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.1	2.3.1 Controllability The helicopter shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power, changes of helicopter configurations) without requiring exceptional skill, alertness or strength on the part of the pilot even in the event of failure of any engine. A technique for safely controlling the helicopter shall be established for all stages of flight and helicopter configurations for which performance is scheduled. <i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i>	CS 29.143. CS 27.143	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.1.1	2.3.1.1 Controllability on the ground (or water). The helicopter shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.	CS 25.231, 233, 235, 237. CS 23.231, 233, 235,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.3.1.2	2.3.1.2 <i>Controllability during take-off.</i> The helicopter shall be controllable in the event of sudden failure of the critical engine at any point in the take-off, when the helicopter is handled in the manner associated with the scheduling of the take-off data.	CS29.141 (b), 143. CS 27. 141 (b), 143	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2	2.3.2 Trim The helicopter shall have such trim and handling capabilities as to ensure that the demands made on the pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. In the event of a malfunction of the systems associated with the flight controls, there shall not be any significant deterioration of the handling characteristics.	CS 29.141 (b), 151, 161. CS 27. 141 (b), 151, 161	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.3	2.3.3 Stability The helicopter shall have such stability in relation to its other flight characteristics, performance, structural strength and most probable operating conditions (e.g. helicopter configurations and speed ranges) as to ensure that demands made on the pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the helicopter shall not, however, be such that excessive demands are made on the pilot's strength or that the safety of the helicopter	CS 29. 171-181. CS 27.171-177	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	is prejudiced by lack of manoeuvrability in emergency conditions.									
2.3.4.1	2.3.4 Autorotation 2.3.4.1 <i>Rotor speed control.</i> The autorotation characteristics of the helicopter shall be such as to enable the pilot to control the rotor speed to within prescribed limits and to maintain full control of the helicopter.	CS 29.33 (a), 143, 175(d). CS 27.33(a), 143, 175(c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.4.2	2.3.4.2 <i>Behaviour following a power loss.</i> The behaviour of the helicopter following a power loss shall not be so extreme as to make difficult a prompt recovery of rotor speed without exceeding the airspeed or strength limitations of the helicopter.	CS 29.33(e), 141(b), 143. CS 27.33(e), 141(b), 143	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.4.3	2.3.4.3 <i>Autorotation airspeeds.</i> The autorotation airspeeds recommended for maximum range and minimum rate of descent shall be established.	CS.29.71. CS 27.71	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.5	2.3.5 Flutter and vibration It shall be demonstrated by suitable tests that all parts of the helicopter are free from flutter and excessive vibration in all helicopter configurations under all speed conditions within the operating limitations of the helicopter (see 1.2.2). There shall be no vibration severe enough to	CS 29.241, 251, 629. CS 27.241, 251, 629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	interfere with control of the helicopter, to cause structural damage or to cause excessive fatigue to the flight crew.									
3.1	CHAPTER 3. STRUCTURE 3.1 General The Standards of this chapter apply to the helicopter structure consisting of all portions of the helicopter, the failure of which would seriously endanger the helicopter.	CS-27 Subpart C. CS-29 Subpart C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.1.1	3.1.1 Mass and mass distribution Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.	CS 29. 309, 321. CS 27.309, 321	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.1.2	3.1.2 Limit loads Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads	CS 29.301. CS 27.301	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	obtained for the various loading conditions prescribed in 3.4, 3.5 and 3.6 shall be considered as limit loads.									
3.1.3	<p>3.1.3 Strength and deformation</p> <p>In the various loading conditions prescribed in 3.4, 3.5 and 3.6, no part of the helicopter structure shall sustain detrimental deformation at any load up to and including the limit load, and the helicopter structure shall be capable of supporting the ultimate load.</p>	CS 29.305. CS 27.305	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.2.1	<p>3.2 Airspeeds</p> <p>3.2.1 Design airspeeds</p> <p>Design airspeeds shall be established for which the helicopter structure is designed to withstand the corresponding manoeuvring and gust loads in accordance with 3.4.</p>	CS 29.309, 341. CS 27.309, 341	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.2.2	<p>3.2.2 Limiting airspeeds</p> <p>Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the helicopter flight manual as part of the operating limitations (see 9.2.2). When airspeed limitations are a function of mass, mass distribution, altitude, rotor speed, power or other</p>	CS 29.1583, 1503. CS 27.1583, 1503	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	factors, airspeed limitations based on the critical combination of these factors shall be established.									
3.3	<p>3.3 Main rotor(s) rotational speed limits</p> <p>A range of main rotor(s) speeds shall be established that:</p> <p>a) with power on, provides adequate margin to accommodate the variations in rotor speed occurring in any appropriate manoeuvre and is consistent with the kind of governor or synchronizer used; and</p> <p>b) with power off, allows each appropriate autorotative manoeuvre to be performed throughout the ranges of airspeed and mass for which certification is requested.</p>	CS 29.33. CS 27.33	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.4	<p>3.4 Flight loads</p> <p>The flight loading conditions of 3.4.1, 3.4.2 and 3.6 shall be considered for the range of mass and mass distributions prescribed in 3.1.1 and at airspeeds established in accordance with 3.2.1. Asymmetrical as well as symmetrical loading shall be taken into account. The air, inertia and other loads resulting from the specified loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively.</p>	CS 29.301, 321, 427. CS 27.301, 321, 427	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.4.1	3.4.1 Manoeuvring loads Manoeuvring loads shall be computed on the basis of manoeuvring load factors appropriate to the manoeuvres permitted by the operating limitations. They shall not be less than values that experience indicates will be adequate for the anticipated operating conditions.	CS 29.337, 339, 351. CS 27.337, 339, 351	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.4.2	3.4.2 Gust loads Gust loads shall be computed for vertical and horizontal gust velocities that statistics or other evidence indicates will be adequate for the anticipated operating conditions.	CS 29.341. CS 27.341	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.5	3.5 Ground and water loads The structure shall be able to withstand all the loads due to the reactions of the ground or water surface, as applicable, that are likely to arise during start-up, ground and water taxiing, lift-off, touchdown and rotor braking.	CS 29.471-521. CS 27.471-521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.5.1	3.5.1 Landing conditions The landing conditions at the design take-off mass and at the design landing mass shall include such symmetrical and asymmetrical attitudes of the helicopter at ground or	CS 27/29.473-521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	water contact, such velocities of descent, and such other factors affecting the loads imposed upon the structure as might be present in the anticipated operating conditions.									
3.6	3.6 Miscellaneous loads In addition to or in conjunction with the manoeuvring and gust loads and with the ground and water loads, consideration shall be given to all other loads (flight control loads, cabin pressures, effects of engine operation, loads due to changes of configuration, loads due to external mass, etc.) that are likely to occur in the anticipated operating conditions.	CS 29.361, 391-427. CS 29.547-551. 865. CS 27.361, 391-427. CS 27.547-549, 865	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.7	3.7 Flutter, divergence and vibration Each part of the helicopter structure shall be free from excessive vibration or oscillation (ground resonance, flutter, etc.) under each appropriate speed and power condition.	CS 29.241, 251, 629. CS 27.241, 251, 629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.8	3.8 Fatigue strength The strength and fabrication of the helicopter shall be such as to ensure that the probability of disastrous fatigue failure of the helicopter's structure under repeated loads and vibratory loads in the anticipated operating conditions is extremely remote.	CS 29.571. CS 27.571	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>Note 1.— This Standard can be complied with by the establishment of “safe lives” or “fail safe” characteristics of the structure, having regard to the reasonable expected load magnitudes and frequencies under the anticipated operating conditions and inspection procedures. For some parts of the structure, it might be necessary to establish “fail safe” characteristics as well as “safe lives”.</i></p> <p><i>Note 2.— Guidance material concerning the expression “extremely remote” is contained in the Airworthiness Manual (Doc 9760).</i></p>								
4.1	<p>CHAPTER 4. DESIGN AND CONSTRUCTION</p> <p>4.1 General</p> <p>Details of design and construction shall be such as to give reasonable assurance that all helicopter parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles.</p>	<p>21.B.80Reg. (EU) 748/2012 CS 29.601 CS 27.601 CS 27.1302 CS 29.1302</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	<i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i>									
4.1.1	4.1.1 Substantiating tests The functioning of all moving parts essential to the safe operation of the helicopter shall be demonstrated by suitable tests in order to ensure that they will function correctly under all operating conditions for such parts.	CS 29.601, 683, 685. CS 27.601, 683, 685	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.2	4.1.2 Materials All materials used in parts of the helicopter essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design.	CS 29.603, 613. CS 27.603, 613 + Reg (EU) 1907/2006	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.3	4.1.3 Manufacturing methods The methods of manufacturing and assembly shall be such as to produce a consistently sound structure which shall be reliable with respect to maintenance of strength in service.	CS 29.605. CS 27.605	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.1.4	4.1.4 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the helicopter will receive.	CS 29.609, CS 27.609	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.1.5	4.1.5 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the helicopter that require such attention, either periodically or after unusually severe operations.	CS 29.611, CS 27.611	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
4.1.6	4.1.6 Systems design features Special consideration shall be given to design features that affect the ability of the flight crew to maintain controlled flight. This shall include at least the following: a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such as to minimize the possibility of jamming, inadvertent operation and unintentional engagement of control surface locking devices. i) Each control and control system shall operate with the ease, smoothness and effectiveness appropriate to its function.	21.B.80Reg. (EU) 748/2012CS 29.671CS 29.685CS 27.671CS 27.685CS 29.771CS 29.777CS 29.779CS 29.831CS 27.771CS 27.777CS 27.779CS 27.831CS	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Depressurization not covered	There is no need to address this because there are no pressurised helicopters on the market.

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	<p>ii) Each element of each flight control system shall be designed to minimize the probability of any incorrect assembly that could result in the malfunction of the system.</p> <p>b) <i>Crew environment.</i> The design of the flight crew compartment shall be such as to minimize the possibility of incorrect or restricted operation of the controls by the crew, due to fatigue, confusion or interference. Consideration shall be given at least to the following: layout and identification of controls and instruments, rapid identification of emergency situations, sense of controls, ventilation, heating and noise.</p> <p>c) <i>Pilot vision.</i> The arrangement of the pilot compartment shall be such as to afford a sufficiently extensive, clear and undistorted field of vision for the safe operation of the helicopter, and to prevent glare and reflections that would interfere with the pilot's vision. The design features of the pilot windshield shall permit, under precipitation conditions, sufficient vision for the normal conduct of flight and for the execution of approaches and landings.</p> <p>d) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the flight crew to deal with, emergencies resulting from foreseeable failures of equipment and systems, the failure of which would endanger the helicopter. Reasonable provisions shall be made for continuation of essential services following engine or system failures to the extent that such failures are catered for in the</p>	<p>29.773CS 27.773CS 29.601CS 29.1309CS 27.601CS 27.1309 CS 29.851-863 CS 27.853-863 CS 29.831 CS 27.831</p>							
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	<p>performance and operating limitations specified in the Standards in this Annex and in Annex 6, Part III.</p> <p>e) <i>Fire precautions.</i> The design of the helicopter and the materials used in its manufacture, including cabin interior furnishing materials replaced during major refurbishing, shall be such as to minimize the possibility of in-flight and ground fires and also to minimize the production of smoke and toxic gases in the event of a fire. Means shall be provided to contain or to detect and extinguish, wherever possible, all accessible fires as might occur in such a way that no additional danger to the helicopter is caused.</p> <p>f) <i>Incapacitation of occupants.</i> Design precautions shall be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases that could incapacitate the occupants of the helicopter.</p>								
4.1.7	<p>4.1.7 Emergency landing provisions</p> <p>Provisions shall be made in the design of the helicopter to protect the occupants from fire and effects of deceleration in the event of an emergency landing. Facilities shall be provided for the rapid evacuation of the helicopter in conditions likely to occur following an emergency landing. Such facilities shall be related to the passenger and crew capacity of the helicopter. On helicopters certificated for ditching conditions, provisions shall also be made in the design to give maximum practicable</p>	CS 29. 561, 562, 563, 785, 801, 803, 807-815, 853-863. CS 27. 561, 562, 563, 785, 801, 805, 807, 851, 853-863	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	assurance that safe evacuation from the helicopter of passengers and crew can be executed in case of ditching.									
4.1.8	4.1.8 Ground handling Adequate provisions shall be made in the design to minimize the risk that ground handling operations (e.g. towing, jacking) may cause damage, which could pass unnoticed, to the parts of the helicopter essential for its safe operation. The protection that any limitations and instructions for such operations might provide may be taken into account.	21.B.80 Reg. (EU) 748/2012 CS-27 Appendix AA27.3 (a) (4) CS-29 Appendix AA29.3 (a) (4)	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	No explicit design requirement. Reliance is placed on the Instructions for continued airworthiness	
5.1	CHAPTER 5. ENGINES 5.1 Scope The Standards of this chapter shall apply to engines of all types that are used on the helicopter as primary propulsion units.	CS-E	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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5.2	5.2 Design, construction and functioning The engine complete with accessories shall be designed and constructed so as to function reliably within its operating limitations under the anticipated operating conditions when properly installed in the helicopter in accordance with Chapter 6 and with the suitable rotor and power transmission installed.	CS-E 20	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3	5.3 Declared ratings, conditions and limitations The power ratings and the conditions of the atmosphere upon which they are based and all operating conditions and limitations which are intended to govern the operation of the engine shall be declared.	CS E.40, 320, 620	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.4	5.4 Tests An engine of the type shall complete satisfactorily such tests as are necessary to verify the validity of the declared ratings, conditions and limitations and to ensure that it will operate satisfactorily and reliably. The tests shall include at least the following: a) <i>Power calibration.</i> Tests shall be conducted to establish the power characteristics of the engine when new and also after the tests in b) and c). There shall be no excessive decrease in power at the conclusion of all the tests specified.	CS E.350, 730. CS E.350, 360, 370, 380, 390, 400, 750, 770, 830. CS E 740, 440	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) <i>Operation.</i> Tests shall be conducted to ensure that starting, idling, acceleration, vibration, overspeeding and other characteristics are satisfactory and to demonstrate adequate margins of freedom from detonation, surge or other detrimental conditions as may be appropriate to the particular type engine.</p> <p>c) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, engine and rotor speeds and other operating conditions as are necessary to demonstrate reliability and durability of the engine. They shall also include operation under conditions in excess of the declared limits to the extent that such limitations might be exceeded in actual service.</p>									
6.1	<p>CHAPTER 6. ROTOR AND POWER TRANSMISSION SYSTEMS AND POWERPLANT INSTALLATION</p> <p>6.1 General</p> <p>The powerplant installation, including rotor and power transmission systems, shall comply with the Standards of Chapter 4 and with the Standards of this chapter.</p>	<p>CS-27 Subpart E. CS-29 Subpart E. CS 29.601. CS 27.601</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.2	<p>6.2 Design, construction and functioning</p> <p>The rotor and power transmission systems assembly complete with accessories shall be designed and constructed so as to function reliably within its operating limitations under the anticipated operating conditions when properly fitted to the engine and installed in the helicopter in accordance with this chapter.</p>	CS 29.301-307,547,549, 571,601-625, 675,861,863, 901,908,917, 921,935,1027,1041,1163(d),1301,1305, 1337,1461, 1529,1551, 1585. CS 27.301-307,547,549, 571,601-625, 675,861,863, 901,917, 921,935,1027,1041, 1301,1305, 1337,1461, 1529,1551, 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.3	<p>6.3 Declared ratings, conditions and limitations</p> <p>The power ratings and all operating conditions and limitations which are intended to govern the operation of the rotor and power transmission systems shall be declared.</p>	CS E.040. CS 29.33, 1521, 1527, 1583. CS 27.33, 1521, 1527, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.3.1	6.3.1 Maximum and minimum rotor rotational speed limitations Maximum and minimum speeds for the rotors in both power-on and power-off conditions shall be established. Any operating conditions (e.g. airspeed) that affect such maxima or minima shall be declared.	CS 29.1509, 1521, 1583. CS 27.1509, 1521, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.3.2	6.3.2 Rotor underspeed and overspeed warnings When the helicopter is made to approach a rotor rotational speed limit, with or without engines inoperative, clear and distinctive warnings shall be apparent to the pilot. The warnings and initial characteristics of the condition shall be such as to enable the pilot to arrest the development of the condition after the warning begins and to recover the rotor rotational speed to within prescribed normal limits and to maintain full control of the helicopter.	CS 29.1305 (a)(14). CS 27.1305 (k)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.4	6.4 Tests Rotor and power transmission systems shall complete satisfactorily such tests as are necessary to ensure that they will operate satisfactorily and reliably within the declared ratings, conditions and limitations. The tests shall include at least the following: a) <i>Operation</i> . Tests shall be conducted to ensure that strength, vibration and overspeeding	CS 29.908, 923, 927, 931, 939, 1043-1049, 1461. CS 27. 923, 927, 931, 939, 1043-1049, 1461. CS 29.923. CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>characteristics are satisfactory and to demonstrate proper and reliable functioning of pitch changing and control mechanisms and free wheel mechanisms.</p> <p>b) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, engine and rotor speeds and other operating conditions as are necessary to demonstrate reliability and durability of the rotor and power transmission systems.</p>	27.923								
6.5	<p>6.5 Compliance with engine and rotor and power transmission systems limitations</p> <p>The powerplant installation shall be so designed that the engines and rotor and power transmission systems are capable of being used in the anticipated operating conditions. In conditions established in the helicopter flight manual, the helicopter shall be capable of being operated without exceeding the limitations established for the engines and rotor and power transmission systems in accordance with Chapter 5 and this chapter.</p>	CS 29.309, 901, 1509, 1521. CS 27.309, 901, 1509, 1521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.6	<p>6.6 Control of engine rotation</p> <p>In those installations where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided for the crew to stop the rotation of the engine in flight or to reduce it to a safe level.</p>	CS 29.903, 917. CS 27.903, 917, and Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.7	6.7 Engine restarting Means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.	CS-29 29.903(e). CS-27 27.903(d) and Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.8.1	6.8 Arrangement and functioning 6.8.1 Independence of engines For performance Classes 1 and 2 helicopters, the powerplant shall be arranged and installed so that each engine together with its associated systems is capable of being controlled and operated independently from the others and so that there is at least one arrangement of the powerplant and systems in which any failure, unless the probability of its occurrence is extremely remote, cannot result in a loss of more power than that resulting from complete failure of the critical engine.	CS 29.901 (c), 903 (b), 917 (c). CS 27 901 (c), 903 (b), 917 (c) and Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.8.2	6.8.2 Rotor and power transmission systems vibration The vibration stresses for the rotor and power transmission systems shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the helicopter.	CS 29.251, 907. CS 27.251, 907	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.8.3	6.8.3 Cooling The cooling system shall be capable of maintaining powerplant and power transmission systems temperatures within the established limits (see 6.5) at ambient air temperatures approved for operation of the helicopter. The maximum and minimum air temperatures for which the powerplant and power transmission systems have been established as being suitable shall be scheduled in the helicopter flight manual.	CS 29 1041-1045, 1583. CS 27 1041-1045, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.8.4	6.8.4 Associated systems The fuel, oil, air induction and other systems associated with each engine, each power transmission unit and each rotor shall be capable of supplying the appropriate unit in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power setting, helicopter attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.	CS 29. 951-1001. CS 29.1011-1027. CS 29.1091-1109. CS 29.1121-1125. CS 27. 951-999. CS 27.1011-1027. CS 27.1091-1093. CS 29.1121-1123.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.8.5	6.8.5 Fire protection For designated fire zones where the potential fire hazards are particularly serious because of the proximity of	CS 29.1181-1203. CS 27.1183-	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.1.6 e):</p> <p>a) <i>Isolation.</i> Such zones shall be isolated by fire-resisting material from other zones of the helicopter where the presence of fire would jeopardize continued flight, taking into account the probable points of origin and paths of propagation of fire.</p> <p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such zones shall be capable of containing the fluid when exposed to fire conditions. Means shall be provided for the crew to shut off the flow of hazardous quantities of flammable fluids into such zones if a fire occurs.</p> <p>c) <i>Fire detection.</i> There shall be provided a sufficient number of fire detectors so located as to ensure rapid detection of any fire that might occur in such zones.</p> <p>d) <i>Fire extinguishment.</i> Such zones shall be provided with a fire extinguisher system capable of extinguishing any fire likely to occur therein, unless the degree of isolation, quantity of combustibles, fire resistance of the structure and other factors are such that any fire likely to occur in the zone would not jeopardize the safety of the helicopter.</p>	1195							
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7.1	<p>CHAPTER 7. INSTRUMENTS AND EQUIPMENT</p> <p>7.1 Required instruments and equipment</p> <p>The helicopter shall be provided with approved instruments and equipment necessary for the safe operation of the helicopter in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the crew to operate the helicopter within its operating limitations. Instruments and equipment design shall observe human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Part III, for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	21.B.80Reg. (EU) 748/2012 CS 27.1302 CS 29.1302 CS 27.1303 CS 29.1303 CS 27.1305 CS 29.1305 CS 27.1307 CS 29.1307	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2	<p>7.2 Installation</p> <p>Instrument and equipment installations shall comply with the Standards of Chapter 4.</p>	CS 29.1301, 1309, 1321. CS 27.1301, 1309,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.3	<p>7.3 Safety and survival equipment</p> <p>Prescribed safety and survival equipment that the crew or passengers are expected to use or operate at the time of an emergency shall be reliable, readily accessible and easily identified, and its method of operation shall be plainly marked.</p>	CS 29. 1411, 1413, 1415. CS 27.1411, 1413, 1415	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.4.1	<p>7.4 Navigation lights and anti-collision lights</p> <p>7.4.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by helicopters in flight or operating on the movement area of an aerodrome or a heliport shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft or personnel on the ground with as much time as possible for interpretation and for subsequent manoeuvre necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note 1.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry sky, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring</i></p>	CS 29.1385-1401. CS 27.1385-1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p><i>in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i></p> <p><i>Note 2.— Detailed technical specifications for exterior lights for helicopters can be found in the Airworthiness Manual (Doc 9760).</i></p>									
7.4.2	<p>7.4.2 Lights shall be installed in helicopters so as to minimize the possibility that they will:</p> <p>a) adversely affect the satisfactory performance of the flight crews' duties; or</p> <p>b) subject an outside observer to harmful dazzle.</p> <p><i>Note.— In order to avoid the effects mentioned in 7.4.2, it will be necessary in some cases to provide means whereby the pilot can switch off or reduce the intensity of the flashing lights.</i></p>	CS 29.1381, 1383, 1401. CS 27.1383, 1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8	CHAPTER 8. ELECTRICAL SYSTEMS		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	The electrical system shall be so designed and installed as to ensure that it will perform its intended function under any foreseeable operating conditions.									
9.1	<p>CHAPTER 9. OPERATING LIMITATIONS AND INFORMATION</p> <p style="text-align: center;">9.1 General</p> <p>The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the helicopter, shall be made available by means of a helicopter flight manual, markings and placards, and such other means as may effectively accomplish the purpose. The limitations and information shall include at least those prescribed in 9.2, 9.3 and 9.4.</p>	CS-27 Subpart G. CS-29 subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2	<p style="text-align: center;">9.2 Operating limitations</p> <p>Limitations which there is a risk of exceeding in flight and which are defined quantitatively shall be expressed in suitable units and corrected if necessary for errors in measurements so that the flight crew can, by reference to</p>	CS 29.1503-1523, 1583. CS 27.1503-1523, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	the instruments available to them, readily determine when the limitations are reached.									
9.2.1	9.2.1 Loading limitations The loading limitations shall include all limiting masses, centre of gravity positions, mass distributions and floor loadings (see 1.2.2).	CS 29.1519, CS 27.1519,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.2	9.2.2 Airspeed limitations The airspeed limitations shall include all speeds (see 3.2) that are limiting from the standpoint of structural integrity or flying qualities of the helicopter, or from other considerations. These speeds shall be identified with respect to the appropriate helicopter configurations and other pertinent factors.	CS 29.1503, 1505, 1517. 1519, CS 27.1503, 1505, 1519	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.3	9.2.3 Powerplant and power transmission limitations The powerplant limitations shall include all those established for the various powerplant and transmission components as installed in the helicopter (see 6.5 and 6.6).	CS 29.1521. CS 27.1521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.4	9.2.4 Rotor limitations	CS 29.1509. CS 27.1509	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	Limitations on rotor speeds shall include maximum and minimum rotor speeds for power-off (autorotation) and power-on conditions.									
9.2.5	9.2.5 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the helicopter.	CS 29.1522	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.6	9.2.6 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the helicopter (see 1.2.1).	CS 29.1501(a), 1583, 1587. CS 27.1501(a), 1583, 1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.7	9.2.7 Flight crew limitations The flight crew limitations shall include the minimum number of flight crew personnel necessary to operate the helicopter, having regard, among other things, to the accessibility to the appropriate crew members of all necessary controls and instruments and to the execution of the established emergency procedures. <i>Note.— The circumstances in which the flight crew shall include members in addition to the minimum flight crew are defined in Annex 6, Part III.</i>	CS 29.1523. CS 27.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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9.3.1	<p>9.3 Operating information and procedures</p> <p>9.3.1 Types of eligible operations</p> <p>There shall be listed the particular types of operations, as may be defined in Annex 6, Part III, or be generally recognized, for which the helicopter has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements.</p>	CS 29.1525. CS 27.1525	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.3.2	<p>9.3.2 Loading information</p> <p>The loading information shall include the empty mass of the helicopter, together with a definition of the condition of the helicopter at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the crew and payload, the usable fuel supply and the drainable oil; it includes the mass of all fixed ballast, unusable fuel supply, undrainable oil, total quantity of engine coolant and total quantity of hydraulic fluid.</i></p>	CS 29.1583, 1589. CS 27.1583, 1589	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.3.3	<p>9.3.3 Operating procedures</p>	CS 29.1583, 1585. CS 27.1583,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	A description shall be given of normal and emergency operating procedures which are peculiar to the particular helicopter and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engine(s).	1585								
9.3.4	9.3.4 Handling information Sufficient information shall be given on any significant or unusual features of the helicopter characteristics.	CS 29.1581, 1585. CS 27.1581, 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.4	9.4 Performance information The performance of the helicopter shall be scheduled in accordance with 2.2. There shall be included information regarding the various helicopter configurations and powers involved and the relevant speeds, together with information that would assist the flight crew in attaining the performance as scheduled.	CS 29.1587. CS 27.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.5	9.5 Helicopter flight manual A flight manual shall be made available. It shall identify clearly the specific helicopter or series of helicopters to which it is related. The flight manual shall include at least the limitations, information and procedures specified in this chapter.	CS 29.1581. CS 27.1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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9.6.1	9.6 Markings and placards 9.6.1 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the flight crew during flight.	CS 29.1541-1565. CS 27.1541-1565	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.6.2	9.6.2 Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in ground servicing (e.g. towing, refuelling) that could pass unnoticed and that could jeopardize the safety of the helicopter in subsequent flights.	CS 29.1557, Appendix A. CS 27.1557, Appendix A. CS 29.1541-1565. CS 27.1541-1565	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.1	PART IVB. HELICOPTERS FOR WHICH APPLICATION FOR CERTIFICATION WAS SUBMITTED ON OR AFTER 13 DECEMBER 2007 CHAPTER 1. GENERAL	21.A.17 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all helicopters designated in 1.1.2 for which an application for the issue of a Type Certificate was submitted to the appropriate national authorities on or after 13 December 2007.</p>									
1.1.2	<p>1.1.2 Except for those Standards and Recommended Practices which specify a different applicability, the Standards and Recommended Practices of this part shall apply to helicopters greater than 750 kg maximum certificated take-off mass intended for the carriage of passengers or cargo or mail in international air navigation.</p> <p><i>Note.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p>	CS 27/CS 29	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.3	<p>1.1.3 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the helicopters designated in 1.1.2 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.</p>	21.A.16 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.1.4	1.1.4 Unless otherwise stated, the Standards apply to the complete helicopter including its powerplant, rotors, systems and equipment.	CS 27/29	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.1	<p>1.2 Operating limitations</p> <p>1.2.1 Limiting conditions shall be established for the helicopter, its powerplant, rotors, systems and equipment (see 7.2). Compliance with the Standards of this part shall be established assuming that the helicopter is operated within the limitations specified. The safety implications of exceeding these operating limits shall be considered.</p>	CS 27/29 1501-1529	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.2	<p>1.2.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the helicopter, e.g. mass, centre of gravity location, load distribution, speeds, ambient air temperature and altitude, shall be established within which compliance with all the pertinent Standards of this part is shown.</p> <p><i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each practicably separate operating condition, e.g. take-off, en route, landing.</i></p> <p><i>Note 2.— Maximum operating mass may be limited by the application of noise certification Standards (see Annex 16 — Environmental Protection, Volume I —</i></p>	CS 27/29: 25; 27;29,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	Aircraft Noise, and Annex 6 — Operation of Aircraft, Part III — International Operations — Helicopters).									
1.3	1.3 Unsafe features and characteristics Under all anticipated operating conditions, the helicopter shall not possess any feature or characteristic that renders it unsafe.	21.A.21 Reg. (EU) 748/2012CS 27.601, CS 29.601	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4	1.4 Proof of compliance The means by which compliance with the appropriate airworthiness requirements is demonstrated shall ensure that in each case the accuracy achieved will be such as to provide reasonable assurance that the helicopter, its components and equipment comply with the requirements and are reliable and function correctly under the anticipated operating conditions.	21.A.20 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.1	CHAPTER 2. FLIGHT 2.1 General	CS 27/29.21	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon a helicopter or helicopters of the type for which a Type Certificate is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.									
2.1.2	2.1.2 Compliance with each Standard shall be established for all applicable combinations of helicopter mass and centre of gravity position, within the range of loading conditions for which certification is sought.	CS 27/29.21	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.3	2.1.3 Where necessary, appropriate helicopter configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the helicopter's flying qualities.	CS 27/29.21	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1	2.2 Performance 2.2.1 Sufficient data on the performance of the helicopter shall be determined and scheduled in the flight manual to provide operators with the necessary information for the purpose of determining the total mass of the helicopter on the basis of the values, peculiar to the proposed flight, of the relevant operational parameters, in order that the flight may be made with reasonable	CS 27/29.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	assurance that a safe minimum performance for that flight will be achieved.									
2.2.2	<p>2.2.2 Achieving the performance scheduled for the helicopter shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the flight crew.</p> <p><i>Note.— Guidance material on human performance can be found in the Human Factors Training Manual (Doc 9683).</i></p>	21.B.80Reg. (EU) 748/2012 CS 27.141 CS 29.141 CS 27.1302 CS 29.1302	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.3	2.2.3 The scheduled performance of the helicopter shall be consistent with compliance with 1.2.1 and with the operation in logical combinations of those of the helicopter's systems and equipment, the operation of which may affect performance.	CS 27/29 1587; CS 27/29.45 and 1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.4	<p>2.2.4 Minimum performance</p> <p>At the maximum masses scheduled (see 2.2.7) for take-off and for landing as functions of the take-off and landing site pressure-altitude and temperature in still air conditions, and, for water operations, in specified conditions of smooth water, the helicopter shall be capable of accomplishing the minimum performances specified in 2.2.5 and 2.2.6, respectively, not considering obstacles or final approach and take-off area length.</p>	CS 27/ 29 51. CS 27 Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.2.5	2.2.5 Take-off a) The performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is scheduled (see 2.2.7), the departure from the scheduled values is not disproportionate. b) For Category A helicopters, in the event of critical engine failure at or after the take-off decision point, the helicopter shall be capable of continuing safe flight, the remaining engine(s) being operated within the approved limitations.	CS 27/29.51-67. CS 27 Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.6	2.2.6 Landing a) It shall be possible to make a safe landing on a prepared landing surface after complete power failure occurring during normal cruise. b) For Category A helicopters, starting from the landing configuration in the event of critical engine failure at or before the landing decision point, the helicopter shall be capable of continuing safe flight, the remaining engine(s) being operated within the approved limitations.	CS 27. 75; CS 27 Appendix C. CS 29 53-63 a): CS 29.75, 79, 83(c), 85 b): CS 29.75, 79(a), 85	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.7	2.2.7 Scheduling of performance	CS 29 45-87, .1587. CS 27.73. CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>Performance data shall be determined and scheduled in the flight manual as follows for the ranges of mass, altitude, temperature and other operational variables for which the helicopter is to be certificated, and additionally for amphibians, water surface conditions and strength of current shall be included.</p> <p>a) <i>Hover performance.</i> The hover performance shall be determined for both in-ground effect and out-of-ground effect with all engines operating.</p> <p>b) <i>Climb.</i> The steady rate of climb with the engine(s) operating at or within approved limits shall be established.</p> <p>c) <i>Height-velocity envelope.</i> If there are any combinations of height and forward velocity (including hover) under which a safe landing cannot be made after failure of the critical engine and with the remaining engine(s) (if applicable) operating within approved limits, a height-velocity envelope shall be established.</p> <p>d) <i>Take-off distance — all engines operating.</i> Where required by the operating rules, the take-off distance — all engines operating shall be the horizontal distance required from the start of the take-off to the point where a selected speed up to the best rate of climb speed (Vy) and selected height above the take-off surface are achieved, all engines operating at approved take-off power required.</p> <p>In addition, for Category A helicopters:</p>	<p>27.65-67. CS 27.79. CS 27.65. CS.27 Appendix C. CS 29 53-63; 67; 77-81</p>							
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	<p>e) <i>Minimum performance.</i> The minimum climb performance shall be established for both take-off and landing.</p> <p>f) <i>Take-off decision point.</i> The take-off decision point shall be the point in the take-off phase used in determining take-off performance and from which either a rejected take-off may be made or a take-off safely continued, with the critical engine inoperative.</p> <p>g) <i>Take-off distance required.</i> The take-off distance required shall be the horizontal distance required from the start of the take-off to the point at which the take-off safety speed (V_{TOSS}), a selected height above the take-off surface and a positive climb gradient are achieved, following failure of the critical engine at take-off decision point, the remaining engine(s) operating within approved operating limits. If procedures involve rearward flight, the back-up distance shall be included.</p> <p>h) <i>Rejected take-off distance required.</i> The rejected take-off distance required shall be the horizontal distance required from the start of the take-off to the point where the helicopter comes to a complete stop following engine failure and rejection of the take-off at the take-off decision point.</p> <p>i) <i>Take-off path — climb gradients.</i> The take-off path — climb gradient shall be the steady gradient(s) of climb for the appropriate configuration(s) with the critical engine inoperative from the end of the</p>								
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	<p>take-off distance required to a defined point above the take-off surface.</p> <p>j) <i>Engine inoperative climb.</i> The engine inoperative climb shall be the steady rate of climb/descent with the critical engine inoperative and the operating engine(s) not exceeding the power for which they are certificated.</p> <p>k) <i>Landing decision point.</i> The landing decision point shall be the latest point in the approach phase from which either a landing may be made or a rejected landing (go-around) safely initiated, with the critical engine inoperative.</p> <p>l) <i>Landing distance required.</i> The landing distance required shall be the horizontal distance required to land and come to a complete stop from a point on the approach flight path at a selected height above the landing surface with the critical engine inoperative.</p>									
2.3.1	<p>2.3 Flying qualities</p> <p>2.3.1 The helicopter shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the helicopter is approved.</p>	CS 27/29.141(a)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.3.2.1	<p>2.3.2 Controllability</p> <p>2.3.2.1 The helicopter shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power, changes of helicopter configurations) without requiring exceptional skill, alertness or strength on the part of the pilot even in the event of failure of any engine. A technique for safely controlling the helicopter shall be established for all stages of flight and helicopter configurations for which performance is scheduled.</p> <p><i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i></p>	CS 27/29.143, 1	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.2	<p>2.3.2.2 Controllability on the ground (or water). The helicopter shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.</p>	CS 27/29; 231-241	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.3	<p>2.3.2.3 Controllability during take-off. The helicopter shall be controllable in the event of sudden failure of the critical engine at any point in the take-off, when the helicopter is handled in the manner associated with the scheduling of the take-off data.</p>	CS 27/29.143 (d), .141(b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.3.3	<p>2.3.3 Trim</p> <p>The helicopter shall have such trim and handling capabilities as to ensure that the demands made on the pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. In the event of a malfunction of the systems associated with the flight controls, there shall not be any significant deterioration of the handling characteristics.</p>	CS 27/29.141(b), 151, 161	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.1	<p>2.4 Stability and control</p> <p>2.4.1 Stability</p> <p>The helicopter shall have such stability in relation to its other flight characteristics, performance, structural strength and most probable operating conditions (e.g. helicopter configurations and speed ranges) as to ensure that demands made on the pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the helicopter shall not, however, be such that excessive demands are made on the pilot's strength or that the safety of the helicopter is prejudiced by lack of manoeuvrability in emergency conditions.</p>	CS 27/29.171-177. CS29.181	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.4.2.1	2.4.2 Autorotation 2.4.2.1 <i>Rotor speed control.</i> The autorotation characteristics of the helicopter shall be such as to enable the pilot to control the rotor speed to within prescribed limits and to maintain full control of the helicopter.	CS 27.175 (c). CS 29 175(d). CS 27/29.33(a), 143	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2.2	2.4.2.2 <i>Behaviour following a power loss.</i> The behaviour of the helicopter following a power loss shall not be so extreme as to make difficult a prompt recovery of rotor speed without exceeding the airspeed or strength limitations of the helicopter.	CS 27/29.33 (e), 141(b), 143	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2.3	2.4.2.3 <i>Autorotation airspeeds.</i> For Category A helicopters, airspeeds for autorotative landings shall be established. For other helicopters, the autorotation airspeeds recommended for maximum range and minimum rate of descent shall be established.	CS29.71CS27 .71	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3	2.4.3 Vibration There shall be no vibration or buffeting severe enough to interfere with the control of the helicopter.	CS 27/29.251	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.4	2.4.4 Ground resonance The helicopter shall have no dangerous tendency to oscillate on the ground with the rotor turning.	CS 27/29.241	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.1.1	<p>CHAPTER 3. STRUCTURE</p> <p>3.1 General</p> <p>3.1.1 For helicopters for which application for certification was submitted before 24 February 2013, the helicopter structure shall be designed, manufactured and provided with instructions for its maintenance with the objective of avoiding catastrophic failure throughout its operational life.</p>	CS 27/29 subpart C and CS 27/29.1529 and Appendix A	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
3.1.2	<p>3.1.2 For helicopters for which application for certification was submitted on or after 24 February 2013, the helicopter structure shall be designed, manufactured and provided with instructions for its maintenance and repair with the objective of avoiding hazardous and catastrophic failure throughout its operational life.</p> <p><i>Note.— Structure includes the airframe, undercarriage, control system, blades and rotorhead, rotor pylon and auxiliary lifting surfaces.</i></p>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Current CS 27/29 does not mandate the provision of structural repair manuals.	Provision of structural repair manuals is a normal industry practice. This provision is not implemented.

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3.2	<p>3.2 Mass and mass distribution</p> <p>Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.</p>	CS29.309, 321, CS27.309, 321	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.3	<p>3.3 Limit loads</p> <p>Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.7, 3.8 and 3.9 shall be considered as limit loads.</p>	CS 27/29.301	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.4	<p>3.4 Strength and deformation</p> <p>In the various loading conditions prescribed in 3.7, 3.8 and 3.9, no part of the helicopter structure shall sustain detrimental deformation at any load up to and including the limit load, and the helicopter structure shall be capable of supporting the ultimate load.</p>	CS 27/29.305	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.5.1	<p>3.5 Airspeeds</p> <p>3.5.1 Design airspeeds</p> <p>Design airspeeds shall be established for which the helicopter structure is designed to withstand the corresponding manoeuvring and gust loads in accordance with 3.7.</p>	CS 27/29, 309, 341	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.5.2	<p>3.5.2 Limiting airspeeds</p> <p>Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the flight manual as part of the operating limitations (see 7.2.3). When airspeed limitations are a function of mass, mass distribution, altitude, rotor speed, power or other factors, airspeed limitations based on the critical combination of these factors shall be established.</p>	CS 27/29.1503; 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.6	<p>3.6 Main rotor(s) rotational speed limits</p> <p>A range of main rotor(s) speeds shall be established that:</p> <p>a) with power on, provides adequate margin to accommodate the variations in rotor speed occurring in any appropriate manoeuvre, and is consistent with the kind of governor or synchronizer used; and</p>	CS 27/29.27-33	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	b) with power off, allows each appropriate autorotative manoeuvre to be performed throughout the ranges of airspeed and mass for which certification is requested.									
3.7.1	3.7 Loads 3.7.1 The loading conditions of 3.7, 3.8 and 3.9 shall consider the range of mass and mass distributions prescribed in 3.2, the main rotor rpm ranges established in 3.6, and airspeeds established in accordance with 3.5.1. Asymmetrical as well as symmetrical loading shall be taken into account. The air, inertia and other loads resulting from the specific loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively in consideration of all anticipated operating conditions.	CS 27/29.301, 321, 427	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.7.2	3.7.2 Manoeuvring loads Manoeuvring loads shall be computed on the basis of manoeuvring load factors appropriate to the manoeuvres permitted by the operating limitations. They shall not be less than values that experience indicates will be adequate for the anticipated operating conditions.	CS 27/29 337; 351, 339	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.7.3	3.7.3 Gust loads	CS 27/29.341	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	Gust loads shall be computed for vertical and horizontal gust velocities that statistics or other evidence indicates will be adequate for the anticipated operating conditions.									
3.8.1	3.8 Ground and water loads 3.8.1 The structure shall be able to withstand all the loads due to the reactions of the ground or water surface, as applicable, that arise during start-up, ground and water taxiing, lift-off, touchdown and rotor braking.	CS 27/29.471.521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.8.2	3.8.2 Landing conditions The landing conditions at the maximum certificated take-off mass and at the maximum certificated landing mass shall include such symmetrical and asymmetrical attitudes of the helicopter at ground or water contact, such velocities of descent, and such other factors affecting the loads imposed upon the structure as might be present in the anticipated operating conditions.	CS 27/29.473-521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.9	3.9 Miscellaneous loads In addition to or in conjunction with the manoeuvring and gust loads and with the ground and water loads, consideration shall be given to all other loads (flight control loads, pilot forces, engine torque, loads due to	CS 29.391-427, 547-551; 361; 865CS27.391-427, 547-549, 361,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	changes of configuration, external loads, etc.) that are likely to occur in the anticipated operating conditions.	865								
3.10	3.10 Fatigue strength The strength and fabrication technique of the helicopter structure shall be such as to avoid catastrophic fatigue failure under repeated loads and vibratory loads in the anticipated operating conditions. Environmental degradation, accidental damage and other likely failures shall be considered.	CS 27/29.571	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.11	3.11 Special factors Design features (e.g. castings, bearings or fittings), the strength of which are subject to variability in manufacturing processes, deterioration in service or any other cause, shall be accounted for by a suitable factor.	CS 27/29.619-625	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.1	CHAPTER 4. DESIGN AND CONSTRUCTION 4.1 General	21.B.80Reg. (EU) 748/2012 CS 27.601 CS 29.601	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>4.1.1 Details of design and construction shall be such as to give reasonable assurance that all helicopter parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	CS 27.1302 CS 29.1302								
4.1.2	<p>4.1.2 Substantiation of moving parts</p> <p>The functioning of all moving parts essential to the safe operation of the helicopter shall be demonstrated in order to ensure that they will function correctly under all operating conditions for such parts.</p>	CS 27/29.601; 671; 683; 685	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.3	<p>4.1.3 Materials</p> <p>All materials used in parts of the helicopter essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design.</p>	CS 27/29.603, 613 + Reg.(EU) 1907/2006	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.1.4	4.1.4 Manufacturing methods The methods of manufacturing and assembly shall be such as to produce consistently sound structure which shall be reliable with respect to maintenance of strength in service.	CS 27/29.605,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.5	4.1.5 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the helicopter will receive.	CS 27/29.609	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.6	4.1.6 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the helicopter that require such attention, either periodically or after unusually severe operations.	CS 27/29.611	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.7	4.1.7 Critical parts All critical parts used in the helicopter shall be identified and procedures shall be established to ensure that the required level of integrity for critical parts is controlled during design, manufacture and throughout the service life of those parts.	CS 27/29.602	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.2	<p>4.2 Systems design features</p> <p>Special consideration shall be given to design features that affect the ability of the flight crew to maintain controlled flight. This shall include at least the following:</p> <p>a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such as to minimize the possibility of jamming, inadvertent operation and unintentional engagement of control locking devices.</p> <p>1) Each control and control system shall operate with the ease, smoothness and precision appropriate to its function.</p> <p>2) Each element of each flight control system shall be designed, or distinctively and permanently marked, to minimize the probability of any incorrect assembly that could result in the malfunction of the system.</p> <p>b) <i>Crew environment.</i> The design of the flight crew compartment shall allow operation of the controls by the crew without unreasonable concentration or fatigue.</p> <p>c) <i>Crew vision.</i> The arrangement of the flight crew compartment shall be such as to afford a sufficiently extensive, clear and undistorted field of vision for the safe operation of the helicopter under all foreseeable operating conditions for which certification is requested.</p>	<p>CS 29.671, 685. CS 27.671, 685. CS 29.771, 777, 779, 831. CS 27.771, 777, 779, 831. CS 29.773. CS 27.773. CS 29.601, 1309. CS 27.601, 1309. CS 29.851 863. CS 27.853 863. CS 29.831. CS 27.831 CS 27/29.685; 679; 671. CS 27/29.771. CS 27/29.773; 1321. CS 27/29.1322; 1309. CS 27/29.853 863. CS 27/29.831</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>		<p>Linked to Amdt109: The aim is to complete the transposition of the recommendation in paragraph(g) before it becomes applicable on or after 1 January 2025.</p>				
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	<p>d) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the flight crew to deal with, emergencies resulting from foreseeable failures of equipment and systems which would endanger the helicopter.</p> <p>e) <i>Fire precautions.</i> The helicopter shall have adequate fire protection.</p> <p>f) <i>Incapacitation of crew.</i> Design precautions shall be taken to protect against the presence of toxic gases which under normal operating conditions could incapacitate the flight crew.</p> <p>g) Recommendation.— <i>Cargo compartment protection. As of 7 March 2025, the elements of the helicopter design associated with cargo compartment fire protection, if applicable, and a summary of the demonstrated standards that were considered in the process of helicopter certification should be included in the required helicopter documentation and made available to the operator.</i></p>									
4.3	4.3 Flutter Each aerodynamic surface of the helicopter shall be free from flutter under each appropriate speed and power condition.	CS 27/29.629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.4.1	<p>4.4 Occupant accommodation features</p> <p>4.4.1 Seating and restraints</p> <p>Adequate seating and restraints shall be provided for the occupants, taking account of the likely flight and emergency landing loads to be encountered. Attention shall be paid to minimizing injury to occupants due to contact with surrounding structures during the operation of the helicopter.</p>	CS 27/29.562; 785	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.4.2	<p>4.4.2 Cabin environment</p> <p>Ventilation systems shall be designed to provide the cabin with an adequate environment during the anticipated flight and ground operating conditions.</p>	CS 27/29.831	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.5.1	<p>4.5 Electrical bonding and protection against lightning and static electricity</p> <p>4.5.1 Electrical bonding and protection against lightning and static electricity shall be such as to:</p> <p>a) protect the helicopter, its systems, its occupants and those who come in contact with the</p>	CS 27/29.610	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	helicopter on the ground or water from the dangerous effects of lightning discharge and electrical shock; and b) prevent dangerous accumulation of electrostatic charge.									
4.5.2	4.5.2 The helicopter shall also be protected against catastrophic effects of lightning. Due account shall be taken of the material used in the construction of the helicopter.	CS 27/29.610	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.1	4.6 Emergency landing provisions 4.6.1 Provisions shall be made in the design of the helicopter to protect the occupants from fire and effects of deceleration in the event of an emergency landing.	CS 27/29.561; 562; 952. 853-863; 785. CS 29.851	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.2	4.6.2 For helicopters for which application for certification was submitted before 24 February 2013, facilities shall be provided for rapid evacuation of the helicopter in conditions likely to occur following an emergency landing, and such facilities shall be related to the passenger and crew capacity of the helicopter. On helicopters certificated for ditching conditions, provisions shall also be made in the design to give reasonable assurance that safe evacuation from the helicopter of passengers and crew can be executed in case of ditching.	CS 27.801-807. CS 29.801-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.6.3	4.6.3 For helicopters for which application for certification was submitted on or after 24 February 2013, facilities shall be provided for rapid evacuation of the helicopter in conditions likely to occur following an emergency landing. Such facilities shall be related to the passenger and crew capacity of the helicopter and shall be shown to be suitable for their intended purpose. On helicopters certificated for ditching conditions, provisions shall also be made in the design to give reasonable assurance that safe evacuation from the helicopter of passengers and crew can be executed in case of ditching.	21.B.80Reg. (EU) 748/2012CS 27.801CS 27.807CS 29.801CS 29.815CS 27.801CS 29.801CS 29.815	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	No requirement to show suitability for the intended operation.	
4.7	4.7 Ground handling Adequate provisions shall be made in the design to minimize the risk that normal ground handling operations (e.g. towing, jacking) may cause damage, which could pass unnoticed, to the parts of the helicopter essential for its safe operation. The protection that any limitations and instructions for such operations might provide may be taken into account.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Not implemented.	
5.1	CHAPTER 5. ROTORS AND POWERPLANT	CS E	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	5.1 Engines The Standards of Part VI of this Annex shall apply to each engine that is used on the helicopter as a primary propulsion unit(s).									
5.2.1	5.2 Rotors and powerplant installation 5.2.1 General The powerplant installation and rotors shall comply with the Standards of Chapter 4 and with the Standards of 5.2.	CS 27/29.601; subpart-E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.2	5.2.2 Design, construction and functioning a) The rotors and rotor drive systems assembly complete with accessories shall be designed and constructed so as to function reliably within their operating limitations under the anticipated operating conditions when properly fitted to the engine and installed in the helicopter in accordance with this chapter. b) For helicopters of maximum certificated take-off mass greater than 3 175 kg or helicopters which are certificated to Category A Standard, an assessment shall be conducted for the rotors and rotor drive systems to ensure that they function safely throughout the full range of operating conditions. Where this assessment identifies a failure which could prevent	CS 29.301 307,547,549, 571,601 625, 675,861,863, 901,908,917, 921,935,1027 ,1041,1163(d) ,1301,1305, 1337,1461, 1529,1551, 1585. CS 27.301 307,547,549, 571,601 625, 675,861,863, 901,917,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	continued safe flight or landing of the helicopter, means shall be prescribed to minimize the likelihood of that failure.	921,935,1027,1041,1301,1305,1337,1461,1529,1551,1585 CS 27/29.917 939								
5.2.3	<p>5.2.3 Declared ratings, conditions and limitations</p> <p>The power ratings and all operating conditions and limitations which are intended to govern the operation of the rotors and rotor drive systems shall be declared.</p> <p>a) <i>Maximum and minimum rotor rotational speed limitations.</i> Maximum and minimum speeds for the rotors in both power-on and power-off conditions shall be established. Any operating conditions (e.g. airspeed) that affect such maxima or minima shall be declared.</p> <p>b) <i>Rotor underspeed warnings for single engine helicopters, and for multi-engine helicopters not having an approved device for automatically increasing power when an engine fails.</i> When the helicopter approaches a rotor rotational speed limit, with or without engines inoperative, clear and distinctive warnings shall be apparent to the pilot. The warnings or initial characteristics of the condition shall be such as to enable the pilot to arrest the development of the condition after the warning begins and to recover the rotor rotational</p>	CS 27/29.33 1509; 1521; 1527, 1583. CS 27/29.33 (e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	speed to within prescribed normal limits and to maintain full control of the helicopter.									
5.2.4	<p>5.2.4 Tests</p> <p>Rotors and rotor drive systems shall complete satisfactorily such tests as are necessary to ensure that they will operate satisfactorily and reliably within the declared ratings, conditions and limitations. The tests shall include at least the following:</p> <p>a) <i>Operation.</i> Tests shall be conducted to ensure that strength and vibration characteristics are satisfactory and to demonstrate proper and reliable functioning of pitch changing and control mechanisms and free wheel mechanisms. Overspeed characteristics shall be demonstrated to be satisfactory for helicopters of maximum certificated take-off mass greater than 3 175 kg; and</p> <p>b) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, engine and rotor speeds, and other operating conditions as are necessary to demonstrate reliability and durability of the rotors and rotor drive systems.</p>	CS 27/29.923. CS 29.927 (d)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.5	5.2.5 Compliance with engine, rotor and rotor drive system limitations	CS 27/29.309, 901, 1509,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	The powerplant installation shall be so designed that the engines, rotors and rotor drive systems are capable of functioning reliably in the anticipated operating conditions. In conditions established in the flight manual, the helicopter shall be capable of being operated without exceeding the limitations established for the engines, rotors and rotor drive systems in accordance with this chapter and Part VI.	1521								
5.2.6	5.2.6 Control of engine rotation For helicopters of a maximum certificated take-off mass greater than 3 175 kg and for helicopters which are certificated to Category A Standard, where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided for the crew to stop the rotation of the failed engine in flight or to reduce it to a safe level.	CS 27 Appendix C. CS 29.903(c), 917	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.7	5.2.7 Engine restarting For helicopters of a maximum certificated take-off mass greater than 3 175 kg and for helicopters which are certificated to Category A Standard, means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.	CS 27.903(d) and Appendix C. CS 29.903 (e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.8.1	5.2.8 Arrangement and functioning	CS 29.901 (c), 903 (b), 917 (c). CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	5.2.8.1 <i>Independence of engines.</i> For Category A helicopters for which application for certification was submitted before 24 February 2013, the powerplant shall be arranged and installed so that each engine together with its associated systems is capable of being controlled and operated independently from the others and so that there is at least one arrangement of the powerplant and systems in which any failure, unless the probability of its occurrence is extremely remote, cannot result in a loss of more power than that resulting from complete failure of the critical engine.	27 901 (c), 903 (b), 917 (c) and Appendix C								
5.2.8.2	5.2.8.2 <i>Independence of engines and associated systems.</i> For Category A helicopters for which application for certification was submitted on or after 24 February 2013, the engines together with their associated systems shall be arranged and isolated from each other to allow operation, in at least one configuration, so that the failure or malfunction of any engine, or the failure of any system that can affect any engine, will not: a) prevent the continued safe operation of the remaining engine(s); or b) require immediate action, other than normal pilot action with primary flight controls, by any crew member to maintain safe operation.	CS 27 Appendix C. CS 29.903(b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.8.3	5.2.8.3 <i>Rotors and rotor drive systems vibration.</i> The vibration stresses for the rotors and rotor drive systems shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the helicopter.	CS 27/29.251, 907	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.2.8.4	5.2.8.4 <i>Cooling.</i> The cooling system shall be capable of maintaining the temperature of powerplant components and fluids within the established limits (see 5.2.5) at all ambient temperatures approved for operation of the helicopter. The maximum and minimum ambient air temperatures for which the powerplant has been established as being suitable shall be scheduled in the flight manual.	CS 27.1041-1045, 1583. CS 29.1041-1049; 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.8.5	5.2.8.5 <i>Associated systems.</i> The fuel, oil, air induction and other systems associated with the powerplant and the rotor(s), shall be capable of supplying the appropriate unit in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power setting, helicopter attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.	CS 29. 951 1001. CS 29.1011 1027. CS 29.1091 1109. CS 29.1121 1125. CS 27. 951 999. CS 27.1011 1027. CS 27.1091 1093. CS 29.1121 1123	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2.8.6	5.2.8.6 <i>Fire protection.</i> For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.2 e). a) <i>Isolation.</i> Such regions shall be isolated by fire resistant material from other regions of the	CS 27.1183-1195. CS 27 Appendix C. CS 29.1181-1203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>helicopter where the presence of fire would jeopardize continued flight and landing (helicopters of a maximum certificated take-off mass greater than 3 175 kg or Category A) or would jeopardize safe landing (other helicopters), taking into account the probable points of origin and paths of propagation of fire.</p> <p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be fire resistant. Drainage of each region shall be provided to minimize hazards resulting from the failure of any component containing flammable fluids. Means shall be provided for the crew to shut off the flow of flammable fluids into such regions if a fire occurs. Where sources of flammable fluid exist in such regions, the whole of the related system within the region, including supporting structure, shall be fireproof or shielded from the effects of fires.</p> <p>c) <i>Fire detection.</i> For turbine engine installations, a sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions, unless the fire can be readily observed in flight by the pilot in the cockpit.</p> <p>d) <i>Fire extinguishment.</i> For turbine engine helicopters of a maximum certificated take-off mass greater than 3 175 kg, such regions shall be provided with a fire extinguisher system capable of extinguishing any fire likely to occur therein, unless the degree of isolation, quantity of combustibles, fire resistance of the structure and other factors are such that any fire likely to occur in</p>								
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	the region would not jeopardize the safety of the helicopter.									
6.1.1	<p>CHAPTER 6. SYSTEMS AND EQUIPMENT</p> <p>6.1 General</p> <p>6.1.1 The helicopter shall be provided with approved instruments, equipment and systems necessary for the safe operation of the helicopter in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the crew to operate the helicopter within its operating limitations. Instruments and equipment design shall consider human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Part III , for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	21.B.80Reg. (EU) 748/2012 CS 27.1302 CS 27.1303 CS 29.1303 CS 27.1305 CS 29.1305 CS 27.1307 CS 29.1307 CS 27.1309 CS 29.1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.1.2	<p>6.1.2 The design of the instruments, equipment and systems required by 6.1.1 and their installation shall be such that:</p> <p>a) for a Category A helicopter, an inverse relationship exists between the probability of a failure condition and the severity of its effect on the helicopter and its occupants, as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p> <p>c) electromagnetic interference between them is minimized.</p>	CS 27/29.1309; 1301; 1351; 1431. CS 27 Appendix C	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.1.3	<p>6.1.3 Means shall be provided to warn the crew of unsafe system operating conditions and to enable them to take corrective action.</p>	CS 27/29.1322	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.1.4	<p>6.1.4 Electrical power supply</p> <p>The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations and shall also be such that no single failure or malfunction could impair the ability of the system to supply essential loads for safe operation.</p>	CS 27/29.1351	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.1.5	<p>6.1.5 Development assurance of complex electronic hardware and system software</p> <p>For helicopters for which application for certification was submitted on or after 24 February 2013, complex electronic hardware and system software shall be developed, verified and validated such as to ensure that the systems in which they are used perform their intended functions at a level of safety that complies with the requirements of this part, notably those of 6.1.2 a) and 6.1.2 b).</p> <p><i>Note.— Some States accept the use of national or international industry standards for the development assurance (development, verification and validation) of complex electronic hardware and systems software.</i></p>	CM-SWCEH-002	<input checked="" type="checkbox"/>	<input type="checkbox"/>		To be replaced by an AMC.				
6.2	<p>6.2 Installation</p> <p>Instrument and equipment installations shall comply with the Standards of Chapter 4.</p>	CS 27/29 1301; 1309, 1321	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.3	<p>6.3 Safety and survival equipment</p> <p>Prescribed safety and survival equipment that the crew or passengers are expected to use or operate at the time of an emergency shall be reliable, readily accessible and easily identified, and its method of operation shall be plainly marked.</p>	CS 27/29.1411, 1413,1415	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.4.1	<p>6.4 Navigation lights and anti-collision lights</p> <p>6.4.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by helicopters in flight or operating on the movement area of an aerodrome or a heliport shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft or personnel on the ground with as much time as possible for interpretation and for subsequent manoeuvre necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry sky, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i></p>	CS 27/29.1385-1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.4.2	<p>6.4.2 Lights shall be installed in helicopters so as to minimize the possibility that they will adversely affect the satisfactory performance of the flight crews' duties.</p> <p><i>Note.— In order to avoid the effects mentioned in 6.4.2, it will be necessary in some cases to provide</i></p>	CS 27/29 1381, 1383, 1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>means whereby the pilot can adjust the intensity of the flashing lights.</i>									
6.5	6.5 Electromagnetic interference protection Aircraft electronic systems, particularly flight-critical and flight-essential systems, shall be protected as appropriate against electromagnetic interference from both internal and external sources.	CS 27/29.1301, 1309, 1317, Appendix D; CS 29.1301, 1309, 1317, Appendix E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.6	6.6 Ice protection If certification for flight in icing conditions is required, the helicopter shall be shown to be able to operate safely in all icing conditions likely to be encountered in all anticipated operating environments.	CS 27/29.1419	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.1	CHAPTER 7. OPERATING LIMITATIONS AND INFORMATION 7.1 General	CS 27/29 Subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the helicopter, shall be made available by means of a flight manual, markings and placards, and such other means as may effectively accomplish the purpose.									
7.2.1	7.2 Operating limitations 7.2.1 Limitations which might be exceeded in flight and which are defined quantitatively shall be expressed in suitable units. These limitations shall be corrected if necessary for errors in measurements so that the flight crew can, by reference to the instruments available to them, readily determine when the limitations are reached.	CS 27/29 1503-1523, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.2	7.2.2 Loading limitations The loading limitations shall include all limiting masses, centre of gravity positions, mass distributions and floor loadings (see 1.2.2).	CS 27/29.1519	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.3	7.2.3 Airspeed limitations The airspeed limitations shall include all speeds (see 3.5.2) that are limiting from the standpoint of structural integrity or flying qualities of the helicopter, or from	CS 27.1503-1505, 1519. CS 29 1503-1505,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	other considerations. These speeds shall be identified with respect to the appropriate helicopter configurations and other pertinent factors.	1517								
7.2.4	7.2.4 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the helicopter (see 5.2.5 and 5.2.8.4).	CS 27/29.1521	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.5	7.2.5 Rotor limitations Limitations on rotor speeds shall include maximum and minimum rotor speeds for power-off (autorotation) and power-on conditions.	CS 27/29.1509	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.6	7.2.6 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the helicopter.	CS 27/29.1501 (a). 1583, 1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.7	7.2.7 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the helicopter (see 1.2.1).	CS 27/29.1501(a)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.2.8	<p>7.2.8 Flight crew limitations</p> <p>The flight crew limitations shall include the minimum number of flight crew personnel necessary to operate the helicopter, having regard, among other things, to the accessibility to the appropriate crew members of all necessary controls and instruments and to the execution of the established emergency procedures.</p> <p><i>Note.— The circumstances in which the flight crew shall include members in addition to the minimum flight crew are defined in Annex 6, Part III.</i></p>	CS 27/29.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.1	<p>7.3 Operating information and procedures</p> <p>7.3.1 Types of eligible operations</p> <p>The particular types of operations for which the helicopter has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.</p>	CS 27/29.1525	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.2	<p>7.3.2 Loading information</p> <p>The loading information shall include the empty mass of the helicopter, together with a definition of the condition of the helicopter at the time of weighing, the</p>	CS 27/29.1583 (c); 1589	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the crew and payload, and the usable fuel supply; it includes the mass of all fixed ballast, unusable fuel supply and total quantity of oil, engine coolant and hydraulic fluid.</i></p>									
7.3.3	<p>7.3.3 Operating procedures</p> <p>A description shall be given of normal and emergency operating procedures which are peculiar to the particular helicopter and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.</p>	CS 27/29.1583, 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.4	<p>7.3.4 Handling information</p> <p>Sufficient information shall be given on any significant or unusual features of the helicopter characteristics.</p>	CS 27/29.1581(a), 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.4	<p>7.4 Performance information</p> <p>The performance of the helicopter shall be scheduled in accordance with 2.2. There shall be included information regarding the various helicopter configurations and</p>	CS 27/29.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	powers involved and the relevant speeds, together with information which will assist the flight crew in attaining the performance as scheduled.									
7.5	7.5 Flight manual A flight manual shall be made available. It shall identify clearly the specific helicopter or series of helicopters to which it is related. The flight manual shall include at least the limitations, information and procedures specified in 7.2, 7.3, 7.4 and 7.6.1.	CS 27/29.1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.1	7.6 Markings and placards 7.6.1 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the flight crew during flight.	CS 27/29 1541-1565	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.2	7.6.2 Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in ground servicing (towing, refuelling, etc.) that could pass unnoticed and that could jeopardize the safety of the helicopter in subsequent flights.	CS27/29 . Appendix ACS 29.1541-1565. CS 27.1541-1565	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.7.1	<p>7.7 Continuing airworthiness — maintenance information</p> <p>7.7.1 General</p> <p>Information for use in developing procedures for maintaining the helicopter in an airworthy condition shall be made available. The information shall include that described in 7.7.2, 7.7.3 and 7.7.4.</p>	CS 27/29.1529 and Appendix A. Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.2	<p>7.7.2 Maintenance information</p> <p>Maintenance information shall include a description of the helicopter and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.</p>	CS 27/29.1529 and Appendix A. Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.3	<p>7.7.3 Maintenance programme information</p> <p>Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed.</p> <p><i>Note.— The development of initial maintenance programme information at the time of helicopter type certification is sometimes referred to as the Maintenance Review Board (MRB) process or the process of developing instructions for continued airworthiness.</i></p>	CS 27/29.1529 and Appendix A. M.A.302 Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.7.4	<p>7.7.4 Mandatory maintenance requirements resulting from the type design approval</p> <p>Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 7.7.3.</p> <p><i>Note.— Mandatory requirements identified as part of the type design approval are often referred to as Certification Maintenance Requirements (CMR) and/or airworthiness limitations.</i></p>	CS 27/29.1529 and Appendix A	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.1	<p>CHAPTER 8. CRASHWORTHINESS AND CABIN SAFETY</p> <p>8.1 General</p> <p>Crashworthiness shall be taken into account in the design of helicopters to improve the probability of occupant survival.</p>	CS 27/29.561; 562; 952	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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8.2	<p>8.2 Design emergency landing loads</p> <p>Emergency landing (crash) loads shall be determined so that the interiors, furnishings, support structure and safety equipment can be designed to reasonably protect occupants under emergency landing conditions. Items to be considered shall include:</p> <p>a) dynamic effects;</p> <p>b) restraint criteria for items that could cause a hazard;</p> <p>c) deformation of the fuselage in the areas of emergency exits;</p> <p>d) fuel cell integrity and position; and</p> <p>e) integrity of electrical systems to avoid sources of ignition in the area of fuel components.</p>	<p>CS 27/29.562.</p> <p>CS 27/29.561(c).</p> <p>CS 27/29.807.</p> <p>CS 27/29.952.</p> <p>CS 27/29.952;</p> <p>863</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.3	<p>8.3 Cabin fire protection</p> <p>The cabin shall be so designed as to provide fire protection to the occupants in the event of airborne systems failures or a crash situation. Items to be considered shall include:</p> <p>a) flammability of cabin interior materials;</p>	<p>CS 27/29 855-863, 1195.</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) fire resistance and, for helicopters of a maximum certificated take-off mass greater than 3 175 kg, the generation of smoke;</p> <p>c) provision of safety features to allow for safe evacuation; and</p> <p>d) fire suppression equipment.</p>								
8.4	<p>8.4 Evacuation</p> <p>The helicopter shall be equipped with sufficient emergency exits to allow for cabin evacuation within an appropriate time period. Items to be considered, appropriate to the size and category of the helicopter, shall include:</p> <p>a) number of seats and seating configuration;</p> <p>b) number, location and size of exits;</p> <p>c) marking of exits and provision of instructions for use;</p> <p>d) likely blockages of exits;</p> <p>e) operation of exits; and</p> <p>f) positioning and weight of evacuation equipment at exits, e.g. slides and rafts.</p>	CS 27.805-807. CS 29.803-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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8.5	8.5 Lighting and marking For helicopters with 10 or more passenger seats, emergency lighting shall be provided and shall have the following characteristics: a) independence from main electrical supply; b) for helicopters for which application for certification was submitted on or after 24 February 2013, automatic activation upon loss of normal power/impact; c) visual indication of emergency exits; and d) illumination both inside and outside the helicopter during evacuation.	CS 29.812	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
9.1	CHAPTER 9. OPERATING ENVIRONMENT AND HUMAN FACTORS 9.1 General	21.B.80Reg. (EU) 748/2012CS 27.51CS 29.51CS 27.141CS	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	There are no formal HF requirements addressing	This is currently a safety focus area

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	<p>The helicopter shall be designed to allow safe operation within the performance limitations of its passengers and those who operate, maintain and service it.</p> <p><i>Note.— The human/machine interface is often the weak link in an operating environment; so, it is necessary to ensure that the helicopter is capable of being controlled at all phases of the flight (including any degradation due to failures) and that neither the crew nor passengers are harmed by the environment in which they have been placed for the duration of the flight.</i></p>	29.141CS 27.771CS 29.771CS 27.775CS 29.775CS 27.779CS 29.779CS 27.831CS 29.831CS 27.610CS 29.610CS 27.807CS 29.807CS 27.1529CS 29.1529CS- 27 Subpart G CS-29 Subpart G							design for maintainability	within the European Aviation Safety Programme and may be the subject of future rulemaking.
9.2.1	9.2 Flight crew 9.2.1 The helicopter shall be designed in such a way as to allow safe and efficient control by the flight crew. The design shall allow for variations in flight crew skill and physiology commensurate with flight crew licensing limits. Account shall be taken of the different expected operating conditions of the helicopter in its environment, including operations degraded by failures.	CS 27/29.771; 51; 141	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.2	9.2.2 The workload imposed on the flight crew by the design of the helicopter shall be reasonable at all stages of flight. Particular consideration shall be given to	CS 27/29.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	critical stages of flight and critical events which may reasonably be expected to occur during the service life of the helicopter, such as engine failure. <i>Note.— Workload can be affected by both cognitive and physiological factors.</i>									
9.3	<p>9.3 Ergonomics</p> <p>During design of the helicopter, account shall be taken of ergonomics factors including:</p> <ul style="list-style-type: none"> a) ease of use and prevention of inadvertent misuse; b) accessibility; c) flight crew working environment; d) cockpit standardization; and e) maintainability. 	<p>CS 27/29.771.</p> <p>CS 27/29.1321.</p> <p>CS 27/29.1523.</p> <p>CS 27/29.1529</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.4	<p>9.4 Operating environmental factors</p> <p>The design of the helicopter shall take into consideration the flight crew operating environment including:</p>	<p>CS 27/29.781;</p> <p>51;</p> <p>141</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>a) effect of aeromedical factors such as noise and vibration; and</p> <p>b) effect of physical forces during normal flight.</p>									
1.1.1	<p>PART V. SMALL AEROPLANES</p> <p>PART VA. AEROPLANES OVER 750 KG BUT NOT EXCEEDING 5 700 KG FOR WHICH APPLICATION FOR CERTIFICATION WAS SUBMITTED ON OR AFTER 13 DECEMBER 2007 BUT BEFORE 7 MARCH 2021</p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all aeroplanes designated in 1.1.2 for which an application for the issue of a Type Certificate was</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.1CS 23.2000</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	submitted to the appropriate national authorities on or after 13 December 2007 but before 7 March 2021.									
1.1.2	<p>1.1.2 Except for those Standards and Recommended Practices which specify a different applicability, the Standards and Recommended Practices of this part shall apply to all aeroplanes having a maximum certificated take-off mass greater than 750 kg but not exceeding 5 700 kg intended for the carriage of passengers or cargo or mail in international air navigation.</p> <p><i>Note 1.— The aeroplanes described in 1.1.2 are known in some States as normal, utility and aerobatic category aeroplanes.</i></p> <p><i>Note 2.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p>	CS 23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.3	<p>1.1.3 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the aeroplanes designated in 1.1.2 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.</p>	CS 23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.4	<p>1.1.4 Unless otherwise stated, the Standards apply to the complete aeroplane including its powerplant, systems and equipment.</p>	CS 23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.2.1	<p>1.2 Operating limitations</p> <p>1.2.1 Limiting conditions shall be established for the aeroplane, its powerplant, systems and equipment (see 7.2). Compliance with the Standards of this part shall be established assuming that the aeroplane is operated within the limitations specified. The limitations shall include a margin of safety to render the likelihood of accidents arising therefrom extremely remote.</p>	CS 23 Subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.2	<p>1.2.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the aeroplane, e.g. mass, centre of gravity location, load distribution, speeds, ambient air temperature and altitude, shall be established within which compliance with all the pertinent Standards in this part is shown.</p> <p><i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each separate operating condition, e.g. take-off, en route, landing.</i></p> <p><i>Note 2.— Maximum operating mass may be limited by the application of noise certification Standards (see Annex 16 — Environmental Protection, Volume I — Aircraft Noise, and Annex 6 — Operation of Aircraft, Part I — International Commercial Air Transport — Aeroplanes and Part II — International General Aviation — Aeroplanes).</i></p>	CS 23.23-33; 45; Subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.3	1.3 Unsafe features and characteristics Under all anticipated operating conditions, the aeroplane shall not possess any feature or characteristic that renders it unsafe.	21.A.21 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4	1.4 Proof of compliance The means by which compliance with the appropriate airworthiness requirements is demonstrated shall ensure that in each case the accuracy achieved will be such as to provide reasonable assurance that the aeroplane, its components and equipment comply with the requirements and are reliable and function correctly under the anticipated operating conditions.	21.A.20, 33, 35 Reg. (EU) 748/2012. CS 23.21, 307	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.1	CHAPTER 2. FLIGHT 2.1 General	21.A.20, 33, 35 Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon an aeroplane or aeroplanes of the type for which a Type Certificate is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.									
2.1.2	2.1.2 Compliance with each Standard shall be established for all applicable combinations of aeroplane mass and centre of gravity position, within the range of loading conditions for which certification is sought.	CS 23.21, 23	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.3	2.1.3 Where necessary, appropriate aeroplane configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the aeroplane's flying qualities.	CS 23.45	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1	2.2 Performance 2.2.1 Sufficient data on the performance of the aeroplane shall be determined and scheduled in the flight manual to provide operators with the necessary information for the purpose of determining the total mass of the aeroplane on the basis of the values, peculiar to the proposed flight, of the relevant operational parameters, in order that the flight may be made with reasonable	CS 23.45.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	assurance that a safe minimum performance for that flight will be achieved.									
2.2.2	<p>2.2.2 Achieving the performance scheduled for the aeroplane shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the flight crew.</p> <p><i>Note.— Guidance material on human performance can be found in the Human Factors Training Manual (Doc 9683).</i></p>	CS 23.45 (f); (h)5	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.3	2.2.3 The scheduled performance of the aeroplane shall be consistent with compliance with 1.2.1 and with the operation in logical combinations of those of the aeroplane's systems and equipment, the operation of which may affect performance.	CS 23.45 (c); (f)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.4	<p>2.2.4 Minimum performance</p> <p>Minimum performance shall be scheduled for aeroplanes with more than one engine that are turbine-powered or have a maximum certificated take-off mass of over 2 721 kg as follows:</p> <p>a) at the maximum masses scheduled (see 2.2.7) for take-off and for landing, as functions of the aerodrome elevation or pressure-altitude either in the standard atmosphere or in specified still air atmospheric conditions; and</p>	CS 23.45 (b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) for seaplanes in specified conditions in smooth water,</p> <p>the aeroplane shall be capable of accomplishing the minimum performances specified in 2.2.5 a) and 2.2.6 a) respectively, not considering obstacles, or runway or water run length.</p> <p><i>Note.— This Standard permits the maximum take-off mass and maximum landing mass to be scheduled in the flight manual against, for example:</i></p> <ul style="list-style-type: none"> — aerodrome elevation, or — pressure-altitude at aerodrome level, or — pressure-altitude and atmospheric temperature at aerodrome level, <p><i>so as to be readily usable when applying the national code on aeroplane performance operating limitations.</i></p>									
2.2.5	<p>2.2.5 Take-off</p> <p>a) For aeroplanes with more than one engine that are turbine-powered or have a maximum certificated take-off mass of over 2 721 kg, after the end of the period during which the take-off power or thrust may be used, the aeroplane shall be capable of continuing to climb, with the critical engine inoperative and the remaining engine(s) operated within their maximum</p>	CS 23.63 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>continuous power or thrust limitations, up to a height that it can maintain and at which it can continue safe flight and landing.</p> <p>b) The minimum performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is scheduled (see 2.2.7), the departure from the scheduled values is not disproportionate.</p>								
2.2.6	<p>2.2.6 Landing</p> <p>a) For aeroplanes for which application for certification was submitted on or after 24 February 2013, aeroplanes with one engine, or a single propeller, or aeroplanes with more than one engine that cannot maintain a positive climb gradient following an engine or propeller failure, the design shall, in the case of engine or propeller failure, enable the aeroplane to be operated to a safe forced landing in favourable conditions.</p> <p>b) For aeroplanes with more than one engine that are turbine-powered or have a maximum certificated take-off mass of over 2 721 kg, starting from the approach configuration and with the critical engine inoperative, the aeroplane shall be capable, in the event of a missed approach, of continuing the flight to a point from which another approach can be made.</p> <p>c) Starting from the landing configuration, the aeroplane shall be capable, in the event of a balked</p>	CS 23.49 (c) and (d). CS 23.67(c)4. CS 23.77	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	landing, of making a climb-out, with all engines operating.									
2.2.7	<p>2.2.7 Scheduling of performance</p> <p>Performance data shall be determined and scheduled in the flight manual in order to provide a safe relationship between the performance of the aeroplane and the aerodromes and routes on which it is capable of being operated. Performance data shall be determined and scheduled for the following stages for the ranges of mass, altitude or pressure-altitude, wind velocity, gradient of the take-off and landing surface for landplanes; water surface conditions, density of water and strength of current for seaplanes; and for any other operational variables for which the aeroplane is to be certificated.</p> <p>a) <i>Take-off.</i> The take-off performance data shall include the distance required to take-off and climb to a selected height above the take-off surface. It shall be determined for each mass, altitude and temperature within the operational limits established for take-off with:</p> <ul style="list-style-type: none"> — take-off power on each engine; — wing flaps in the take-off position(s); and — landing gear extended. <p>b) <i>En route.</i> For aeroplanes with more than one engine, the en-route climb performance shall be the climb (or descent) performance with the aeroplane in</p>	CS 23.45. CS 23.53; 67. CS 23.69. CS 23.75	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>the en-route configuration with the critical engine inoperative. The operating engine(s) shall not exceed maximum continuous power or thrust.</p> <p>c) <i>Landing.</i> The landing distance shall be the horizontal distance traversed by the aeroplane from a point on the approach flight path at a selected height above the landing surface to the point on the landing surface at which the aeroplane comes to a complete stop, or, for a seaplane, comes to a satisfactorily low speed. The selected height above the landing surface and the approach speed shall be appropriately related to operating practices. This distance may be supplemented by such distance margin as may be necessary; if so, the selected height above the landing surface, the approach speed and the distance margin shall be appropriately interrelated and shall make provision for both normal operating practices and reasonable variations therefrom.</p>									
2.3.1	<p>2.3 Flying qualities</p> <p>2.3.1 The aeroplane shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the aeroplane is approved.</p>	CS 23.141;143	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.1	2.3.2 Controllability	CS 23.143 (b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>2.3.2.1 The aeroplane shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power or thrust, changes of aeroplane configurations) without requiring exceptional skill, alertness or strength on the part of the pilot even in the event of failure of any engine. A technique for safely controlling the aeroplane shall be established for all stages of flight and aeroplane configurations for which performance is scheduled.</p> <p><i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i></p>									
2.3.2.2	<p>2.3.2.2 <i>Controllability on the ground (or water).</i> The aeroplane shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.</p>	CS 23.231-239	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.3	<p>2.3.2.3 <i>Controllability during take-off.</i> The aeroplane shall be controllable in the event of sudden failure of the critical engine at any point in the take-off.</p>	CS 23.149	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.3.2.4	2.3.2.4 <i>Take-off safety speed.</i> The take-off safety speeds assumed when the performance of the aeroplane (after leaving the ground or water) during the take-off is determined shall provide an adequate margin above the stall and above the minimum speed at which the aeroplane remains controllable after sudden failure of the critical engine.	CS 23.51	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.3	2.3.3 Trim The aeroplane shall have such trim characteristics as to ensure that the demands made on the pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. This shall apply both in normal operation and in the conditions associated with the failure of one or more engines for which performance characteristics are established.	CS 23.161	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.1	2.4 Stability and control 2.4.1 Stability The aeroplane shall have such stability in relation to its other flight characteristics, performance, structural strength and most probable operating conditions (e.g. aeroplane configurations and speed ranges) as to ensure that demands made on the pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the aeroplane shall not, however, be such that excessive demands are made on the pilot's	CS 23.143; 171 - 181; 672	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	strength or that the safety of the aeroplane is prejudiced by lack of manoeuvrability in emergency conditions. The stability may be achieved by natural or artificial means, or a combination of both. In those cases where artificial stability is necessary to show compliance with the Standards of this part, it shall be shown that any failure or condition that would result in the need for exceptional pilot skill or strength for recovery of aeroplane stability is extremely improbable.									
2.4.2.1	<p>2.4.2 Stalling</p> <p>2.4.2.1 <i>Stall warning.</i> When the aeroplane approaches a stall both in straight and turning flight, a clear and distinctive stall warning shall be apparent to the pilot with the aeroplane in all permissible configurations and powers or thrusts, except those which are not considered to be essential for safe flying. The stall warning and other characteristics of the aeroplane shall be such as to enable the pilot to arrest the development of the stall after the warning begins and, without altering the engine power or thrust, to maintain full control of the aeroplane.</p>	21.B.80Reg. (EU) 748/2012CS-25 CS 25.201 CS 25.203 CS 25.207CS-23 CS 23.201CS 23.203 CS 23.207CS 23.2150	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2.2	<p>2.4.2.2 <i>Behaviour following a stall.</i> In any configuration and at any level of power or thrust in which it is considered that the ability to recover from a stall is essential, the behaviour of the aeroplane following a stall shall not be so extreme as to make difficult a prompt recovery without exceeding the airspeed or strength limitations of the aeroplane.</p>	CS 23.201-203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.4.2.3	2.4.2.3 <i>Stalling speeds.</i> The stalling speeds or minimum steady flight speeds in configurations appropriate for each stage of flight (e.g. take-off, en route, landing) shall be established. One of the values of the power or thrust used in establishing the stalling speeds shall be not more than that necessary to give zero thrust at a speed just above the stall.	CS 23.49	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3.1	2.4.3 Flutter and vibration 2.4.3.1 It shall be demonstrated by suitable tests, analyses or any acceptable combination of tests and analyses that all parts of the aeroplane are free from flutter and excessive vibration in all aeroplane configurations under all speed conditions within the operating limitations of the aeroplane (see 1.2.2). There shall be no vibration or buffeting severe enough to cause structural damage.	CS 23.629; 23.251-253	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3.2	2.4.3.2 There shall be no vibration or buffeting severe enough to interfere with control of the aeroplane or to cause excessive fatigue to the flight crew. <i>Note.— Buffeting as a stall warning is considered desirable and discouragement of this type of buffeting is not intended.</i>	CS 25.143, 251, 253CS 23.143, 251, 253	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.4	2.4.4 Spinning It shall be demonstrated that the aeroplane during normal operation does not exhibit any tendency to inadvertently	CS 23.221; 203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	enter into a spin. If the design is such that spinning is allowed or for aeroplanes with one engine inadvertently possible, it shall be demonstrated that with normal use of the controls and without the use of exceptional piloting skill the aeroplane can be recovered from a spin within appropriate recovery limits.									
3.1	<p>CHAPTER 3. STRUCTURE</p> <p style="text-align: center;">3.1 General</p> <p>The aeroplane structure shall be designed, manufactured and provided with instructions for its maintenance and repair with the objective of avoiding catastrophic failure throughout its operational life.</p>	21.B.80Reg. (EU) 748/2012CS-23 Subpart C CS 23.1529CS 23.2340	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Current CS 25/23 does not mandate the provision of structural repair manuals. Hazardous not specifically addressed in relation to fatigue.	Provision of structural repair manuals is a normal industry practice.
3.2	<p style="text-align: center;">3.2 Mass and mass distribution</p> <p>Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse</p>	21.B.80Reg. (EU) 748/2012CS 23.321CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	.	.

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	manner, within the operating limitations on the basis of which certification is sought.	23.2200								
3.3	3.3 Limit loads Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.6 shall be considered as limit loads.	CS 23.301(a);	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.4	3.4 Strength and deformation In the various loading conditions prescribed in 3.6, no part of the aeroplane structure shall sustain detrimental deformation at any load up to and including the limit load, and the aeroplane structure shall be capable of supporting the ultimate load.	CS 23.305 (b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.5.1	3.5 Airspeeds 3.5.1 Design airspeeds Design airspeeds shall be established for which the aeroplane structure is designed to withstand the corresponding manoeuvring and gust loads. To avoid inadvertent exceedences due to upsets or atmospheric	CS 23.335	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	variations, the design airspeeds shall provide sufficient margin for the establishment of practical operational limiting airspeeds. In addition, the design airspeeds shall be sufficiently greater than the stalling speed of the aeroplane to safeguard against loss of control in turbulent air. Consideration shall be given to a design manoeuvring speed, a design cruising speed, a design dive speed and any other design airspeeds necessary for configurations with high lift or other special devices.									
3.5.2	<p style="text-align: center;">3.5.2 Limiting airspeeds</p> <p>Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the flight manual as part of the operating limitations (see 7.2).</p>	CS 23.1505; 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.6.1	<p style="text-align: center;">3.6 Strength</p> <p>3.6.1 All structural elements shall be designed to withstand the maximum loads expected in service under all anticipated operating conditions without failure, permanent distortion or loss of functionality. In determining these loads, account shall be taken of:</p> <p>a) the expected operational life of the aeroplane;</p> <p>b) the vertical and horizontal gust environment, taking into consideration the expected variations in mission profile and loading configurations;</p>	CS 23; Subpart C. CS 23.571-575. CS 23.333-341. CS 23.333. CS 23.347; 367; 331. CS 23.521-537. CS 23.335. CS 23.629. CS 23.571-575, 609. CS 23.365;	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>c) the manoeuvre spectrum, taking into account variations in mission profile and loading configurations;</p> <p>d) asymmetrical as well as symmetrical loading;</p> <p>e) the ground and water loads, including taxi, landing and take-off loads, and ground/water handling loads;</p> <p>f) the speed range of the aeroplane, taking into account the aeroplane characteristics and operation limitations;</p> <p>g) vibration and buffeting loads;</p> <p>h) corrosion or other degradation, given the maintenance specified, and various operating environments; and</p> <p>i) any other loads, such as flight control loads, cabin pressurization loads, engine loads, or dynamic loads due to changes to the steady state configuration.</p>	361									
3.6.2	3.6.2 The air, inertia and other loads resulting from the specific loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively.	CS 23.301 (b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>							

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3.7	<p>3.7 Survivability</p> <p>The aeroplane shall be designed so as to provide the occupants with the maximum practicable protection in the event of structural failure, or in the event of damage due to ground, water or object impact. Consideration shall be given to at least the following:</p> <p>a) energy absorption by the airframe, occupant seats and restraints; and</p> <p>b) allowing egress in the shortest practicable time.</p>	CS 23.561-562	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.8	<p>3.8 Structural durability</p> <p>The structure of the aeroplane shall conform to damage tolerance, safe-life or failsafe principles and shall be such as to avoid catastrophic failure during the operational life, taking into account, where appropriate:</p> <p>a) the expected environment;</p> <p>b) the expected repeated loads applied in service;</p> <p>c) expected vibrations from aerodynamic interaction or internal sources;</p> <p>d) thermal cycles;</p>	CS 23.571-575	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<ul style="list-style-type: none"> e) accidental and discrete source damage; f) likely corrosion or other degradation; g) specified maintenance; and h) likely structural repairs. 								
3.9	<p>3.9 Special factors</p> <p>For aeroplanes for which application for certification was submitted on or after 24 February 2013, design features (e.g. castings, bearings or fittings), the strength of which is subject to variability in manufacturing processes, deterioration in service or any other cause, shall be accounted for by a suitable factor.</p>	CS 23.619-625	<input checked="" type="checkbox"/>	<input type="checkbox"/>					
4.1.1	<p>CHAPTER 4. DESIGN AND CONSTRUCTION</p> <p>4.1 General</p> <p>4.1.1 Details of design and construction shall be such as to give reasonable assurance that all aeroplane</p>	CS 23 Subpart D	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles. <i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i>									
4.1.2	4.1.2 Substantiation of moving parts The functioning of all moving parts essential to the safe operation of the aeroplane shall be demonstrated in order to ensure that they will function correctly under all operating conditions for such parts.	CS 23.651; 659, 683, 685	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.3	4.1.3 Materials All materials used in parts of the aeroplane essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design.	CS 23.603 + Reg (EU) 1907/2006	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.4	4.1.4 Manufacturing methods	CS 23.605	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	The methods of manufacturing and assembly shall be such as to produce a consistently sound structure which shall be reliable with respect to maintenance of strength in service.									
4.1.5	4.1.5 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the aeroplane will receive.	CS 23.609	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.6	4.1.6 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the aeroplane that require such attention, either periodically or after unusually severe operations.	CS 23.611	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.2	4.2 Systems design features Special consideration shall be given to design features that affect the ability of the flight crew to maintain controlled flight. This shall include at least the following: a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such	CS 23.683; 679; 685. CS 23.671. CS 23.685. CS 23.1309. CS 23.771; 777; 781. CS 23.773-775.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>as to minimize the possibility of jamming, inadvertent operation including prevention of mis-assembly, and unintentional engagement of control surface locking devices.</p> <p>1) Each control and control system shall operate with the ease, smoothness and precision appropriate to its functions.</p> <p>2) Each element of each flight control system shall be designed, or distinctively and permanently marked, to minimize the probability of any incorrect assembly that could result in the malfunction of the system.</p> <p>b) <i>System survivability.</i> Aeroplane systems shall be designed and arranged to maximize the potential for continued safe flight and landing after events resulting in damage to the aeroplane structure or systems.</p> <p>c) <i>Crew environment.</i> The design of the flight crew compartment shall be such as to minimize the possibility of incorrect or restricted operation of the controls by the crew, due to fatigue, confusion or interference. Consideration shall be given at least to the following: layout and identification of controls and instruments, rapid identification of emergency situations, sense of controls, ventilation, heating and noise.</p> <p>d) <i>Pilot vision.</i> The arrangement of the flight crew compartment shall be such as to afford a sufficiently extensive, clear and undistorted field of vision for the safe operation of the aeroplane, and to</p>	<p>CS 23.1309; 1322; 1585. CS 23.851-865. CS 23.855. CS 23.831; 841; 1441; 1455. 851-865</p>							
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	<p>prevent glare and reflections that would interfere with the pilot's vision. The design features of the windshield shall permit, under precipitation conditions of moderate rain, sufficient vision for the normal conduct of flight and for the execution of approaches and landings.</p> <p>e) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the flight crew to deal with, emergencies resulting from foreseeable failures of equipment and systems, the failure of which would endanger the aeroplane. Reasonable provisions shall be made for continuation of essential services following engine or system failures to the extent that such failures are catered for in the performance and operating limitations specified in the Standards in this Annex and in Annex 6, Parts I and II.</p> <p>f) <i>Fire precautions.</i> The design of the aeroplane and the materials used in its manufacture shall be such so as to minimize the risk of in-flight and ground fires, and to minimize the production of smoke and toxic gases in the event of a fire.</p> <p>g) <i>Cargo compartment protection.</i></p> <p>1) Sources of heat within the compartment which are capable of igniting the cargo or baggage shall be shielded or insulated to prevent such ignition; and</p> <p>2) Each cargo and baggage compartment shall be constructed of materials which are at least flame resistant.</p>								
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	h) <i>Incapacitation of occupants.</i> Design precautions shall be taken to protect against possible instances of cabin depressurization and against the presence of smoke or other toxic gases that could incapacitate the occupants of the aeroplane.									
4.3	4.3 Aeroelasticity The aeroplane shall be free from flutter, structural divergence, control reversal, loss of control due to structural deformation and aeroelastic effects, at all speeds within and sufficiently beyond the design envelope to comply with 1.2.1. Account shall be taken of the characteristics of the aeroplane.	CS 23.629	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.4.1	4.4 Occupant accommodation features 4.4.1 Seating and restraints Adequate seating and restraints shall be provided for the occupants, taking account of the likely flight and emergency landing loads to be encountered. Attention shall be paid to minimizing injury to occupants due to contact with surrounding structures during the operation of the aeroplane.	CS 23.785	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.4.2	4.4.2 Cabin environment Ventilation, heating and, where applicable, pressurization systems shall be designed to provide the cabin with an adequate environment during the anticipated flight and ground or water operating conditions. The systems design shall also consider likely emergency conditions.	CS 23.831, 841, 843, 1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.5.1	4.5 Electrical bonding and protection against lightning and static electricity 4.5.1 Electrical bonding and protection against lightning and static electricity shall be such as to: a) protect the aeroplane, its systems, its occupants and those who come in contact with the aeroplane on the ground or water from the dangerous effects of lightning discharge and electrical shock; and b) prevent dangerous accumulation of electrostatic charge.	CS 23.867	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.5.2	4.5.2 The aeroplane shall also be protected against catastrophic effects of lightning. Due account shall be taken of the material used in the construction of the aeroplane.	CS 23.867	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.6.1	4.6 Emergency landing provisions 4.6.1 Provisions shall be made in the design of the aeroplane to protect the occupants, in the event of an emergency landing, from fire and from the direct effects of deceleration forces as well as from injuries arising from the effect of deceleration forces on the aeroplane's interior equipment.	CS 23.561-562	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.2	4.6.2 Facilities shall be provided for the rapid evacuation of the aeroplane in conditions likely to occur following an emergency landing. Such facilities shall be related to the passenger and crew capacity of the aeroplane and shall be shown to be suitable for their intended purpose.	CS 23 803; 815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.7	4.7 Ground handling Design provisions and procedures for safe ground-handling (e.g. towing, jacking) shall be defined. The protection that any limitations and instructions for such operations might provide may be taken into account.	CS 23.507-509	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.1	CHAPTER 5. POWERPLANT	21.A.21 Reg. (EU) 748/2012.	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>5.1 Engines</p> <p>The Standards of Part VI of this Annex shall apply to each engine that is used on the aeroplane as a primary propulsion unit.</p>	CS-E								
5.2	<p>5.2 Propellers</p> <p>The Standards of Part VII of this Annex shall apply to each propeller that is used on the aeroplane.</p>	21.A.21 Reg. (EU) 748/2012, CS-P	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.1	<p>5.3 Powerplant installation</p> <p>5.3.1 Compliance with engine and propeller limitations</p> <p>The powerplant installation shall be so designed that the engines and propellers (if applicable) are capable of functioning reliably in the anticipated operating conditions. In conditions established in the flight manual, the aeroplane shall be capable of being operated without exceeding the limitations established for the engines and propellers in accordance with this chapter and Parts VI and VII.</p>	CS 23.901	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.3.2	5.3.2 Control of engine rotation In those installations where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided for the crew to stop the rotation of the failed engine in flight or to reduce it to a safe level.	CS 23.903 (d) (e)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.3	5.3.3 Turbine engine installation For a turbine engine installation: a) the design shall minimize the hazards to the aeroplane in the event of failure of engine rotating parts, or an engine fire which burns through the engine case; and b) the powerplant installation shall be designed to give reasonable assurance that those engine operating limitations that adversely affect the structural integrity of rotating parts shall not be exceeded in service.	CS 23.903 (b)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.4	5.3.4 Engine restarting Means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.	CS 23.903	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.3.5.1	<p>5.3.5 Arrangement and functioning</p> <p>5.3.5.1 <i>Independence of engines.</i> For aeroplanes for which application for certification was submitted before 24 February 2013, the powerplant shall be arranged and installed so that each engine together with its associated systems is capable of being controlled and operated independently from the others and so that there is at least one arrangement of the powerplant and systems in which any failure, unless the probability of its occurrence is extremely remote, cannot result in a loss of more power than that resulting from complete failure of the critical engine.</p>	CS 23.903 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.2	<p>5.3.5.2 <i>Independence of engines and associated systems.</i> For aeroplanes for which application for certification was submitted on or after 24 February 2013, the engines together with their associated systems shall be arranged and isolated from each other to allow operation, in at least one configuration, so that the failure or malfunction of any engine, or the failure or malfunction (including destruction by fire in the engine compartment) of any system that can affect an engine (other than a fuel tank if only one fuel tank is installed), will not:</p> <p>a) prevent the continued safe operation of the remaining engine(s); or</p> <p>b) require immediate action by any crew member for continued safe operation of the remaining engine(s).</p>	CS 23.903 (c)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.3.5.3	5.3.5.3 <i>Propeller vibration.</i> The propeller vibration stresses shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the aeroplane.	CS 23.907	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.4	5.3.5.4 <i>Cooling.</i> The cooling system shall be capable of maintaining the temperature of powerplant components and fluids within the established limits (see 5.3.1) at ambient air temperatures up to the maximum air temperature appropriate to the intended operation of the aeroplane.	CS 23.1041-1063	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.5	5.3.5.5 <i>Associated systems.</i> The fuel, oil, air induction and other systems associated with the powerplant shall be capable of supplying each engine in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power or thrust, aeroplane attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.	CS 23.Subpart E; 951; 1011; 1091; 1111, 1121	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.6	5.3.5.6 <i>Fire protection.</i> For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.2 f). a) <i>Isolation.</i> Such regions shall be isolated by fireproof material from other regions of the aeroplane where the presence of fire would jeopardize continued	CS 23.1181-1203	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>flight, taking into account the probable points of origin and paths of propagation of fire.</p> <p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be fire resistant. Drainage of each region shall be provided to minimize hazards resulting from the failure of any component containing flammable fluids. Means shall be provided for the crew to shut off the flow of flammable fluids into such regions if a fire occurs. Where sources of flammable fluid exist in such regions, the whole of the related system within the region, including supporting structure, shall be fireproof or shielded from the effects of the fire.</p> <p>c) <i>Fire detection.</i> A sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions of the following aeroplane types: aeroplanes with more than one engine powered by turbine or turbo-charged engines, or aeroplanes where the engine(s) are not readily visible from the cockpit.</p>								
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6.1.1	CHAPTER 6. SYSTEMS AND EQUIPMENT 6.1 General	CS 23 Subpart F	<input checked="" type="checkbox"/>	<input type="checkbox"/>		This reference should reflect the				
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	<p>6.1.1 The aeroplane shall be provided with approved instruments, equipment and systems, including guidance and flight management systems necessary for the safe operation of the aeroplane in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the crew to operate the aeroplane within its operating limitations. Instruments and equipment design shall consider human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Parts I and II, for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>									situation as explained for Large aeroplanes in part III.
6.1.2	<p>6.1.2 The design of the instruments, equipment and systems required by 6.1.1 and their installation shall be such that:</p> <p>a) an inverse relationship exists between the probability of a failure condition and the severity of its effect on the aircraft and its occupants, as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p>	CS 23.1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
				Level of implementation of SARPs						
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	c) electromagnetic interference between them is minimized.									
6.1.3	6.1.3 Means shall be provided to warn the crew of unsafe system operating conditions and to enable them to take corrective action.	CS 23.1322, 1309 (b)(3)	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
6.1.4	6.1.4 Electrical power supply The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations and shall also be such that no single failure or malfunction could impair the ability of the system to supply essential loads for safe operation.	CS 23.1309 (c), 1351	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		
6.1.5	6.1.5 Development assurance of complex electronic hardware and system software For aeroplanes for which application for certification was submitted on or after 24 February 2013, complex electronic hardware and system software shall be developed, verified and validated such as to ensure that the systems in which they are used perform their intended functions at a level of safety that complies with the requirements of this part, notably those of 6.1.2 a) and 6.1.2 b). <i>Note.— Some States accept the use of national or international industry standards for the development</i>	21.B.75Reg. (EU) 748/2012	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Not specifically addressed in CS 25 and CS 23. However, EASA Certification Memo (CM-SWCEH-001) is guidance for the	

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	<i>assurance (development, verification and validation) of complex electronic hardware and systems software.</i>								development assurance of CEH and SW and applied in certification project in Special Conditions . This provides guidance to comply with 6.1.2(a) and 6.1.2(b).	
6.2	6.2 Installation Instrument and equipment installations shall comply with the Standards of Chapter 4.	CS 23 Subpart F, 1309	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.3	6.3 Safety and survival equipment Prescribed safety and survival equipment that the crew or passengers are expected to use or operate at the time of an emergency shall be reliable, readily accessible and easily	CS 23.1411	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	identified, and its method of operation shall be plainly marked.									
6.4.1	<p>6.4 Navigation lights and anti-collision lights</p> <p>6.4.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by aeroplanes in flight or operating on the movement area of an aerodrome shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft or personnel on the ground with as much time as possible for interpretation and for subsequent manoeuvre necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry sky, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i></p>	CS 23.1383-1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.4.2	6.4.2 Lights shall be installed in aeroplanes so as to minimize the possibility that they will adversely affect the satisfactory performance of the flight crews' duties.	CS 23.773, 1381, 1383, 1401	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>Note.— In order to avoid the effects mentioned in 6.4.2, it will be necessary in some cases to provide means whereby the pilot can adjust the intensity of the flashing lights.</i>									
6.5	6.5 Electromagnetic interference protection Aeroplane electronic systems, particularly flight-critical and flight-essential systems, shall be protected against electromagnetic interference from both internal and external sources.	AMC 20-136, 158 CS 23.2520	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.6	6.6 Ice protection If certification for flight in icing conditions is requested, the aeroplane shall be shown to be able to operate safely in icing conditions likely to be encountered in all anticipated operating environments.	CS 23.1419	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.1	CHAPTER 7. OPERATING LIMITATIONS AND INFORMATION	CS 23.Subpart G; 1541, 1581	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	7.1 General The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the aeroplane, shall be made available by means of a flight manual, markings and placards, and such other means as may effectively accomplish the purpose.									
7.2.1	7.2 Operating limitations 7.2.1 Limitations which might be exceeded in flight and which are defined quantitatively shall be expressed in suitable units. These limitations shall be corrected if necessary for errors in measurements so that the flight crew can, by reference to the instruments available to them, readily determine when the limitations are reached.	CS 23.1501, 1541, 1583	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.2	7.2.2 Loading limitations The loading limitations shall include all limiting masses, centre of gravity positions, mass distributions and floor loadings (see 1.2.2).	CS 23.1519	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.3	7.2.3 Airspeed limitations The airspeed limitations shall include all speeds (see 3.5.2) that are limiting from the standpoint of structural	CS 23.1505-1513	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	integrity or flying qualities of the aeroplane, or from other considerations. These speeds shall be identified with respect to the appropriate aeroplane configurations and other pertinent factors.									
7.2.4	7.2.4 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the aeroplane (see 5.3.1 and 5.3.5.4).	CS 23.1521, 1522	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.5	7.2.5 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the aeroplane.	CS 23.1583 (m)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.6	7.2.6 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the aeroplane (see 1.2.1).	CS 23.1501, 1522; 1524; 1527	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.7	7.2.7 Flight crew limitations The flight crew limitations shall include the minimum number of flight crew personnel necessary to operate the aeroplane, having regard, among other things, to the	CS 23.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>accessibility to the appropriate crew members of all necessary controls and instruments and to the execution of the established emergency procedures.</p> <p><i>Note.— The circumstances in which the flight crew shall include members in addition to the minimum flight crew are defined in Annex 6, Parts I and II.</i></p>									
7.3.1	<p>7.3 Operating information and procedures</p> <p>7.3.1 Types of eligible operations</p> <p>The particular types of operations for which the aeroplane has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.</p>	CS 23.1525	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.2	<p>7.3.2 Loading information</p> <p>The loading information shall include the empty mass of the aeroplane, together with a definition of the condition of the aeroplane at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the crew and payload, the usable fuel supply and the drainable oil; it includes the mass of all fixed ballast,</i></p>	CS 23.1589	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>unusable fuel supply, undrainable oil, total quantity of engine coolant and total quantity of hydraulic fluid.</i>									
7.3.3	7.3.3 Operating procedures A description shall be given of normal and emergency operating procedures which are peculiar to the particular aeroplane and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.	CS 23.1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.4	7.3.4 Handling information Sufficient information shall be given on any significant or unusual features of the aeroplane characteristics. Those stalling speeds or minimum steady flight speeds required to be established by 2.4.2.3 shall be scheduled.	CS 23.1557, 1581, 1583, 1585	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.4	7.4 Performance information The performance of the aeroplane shall be scheduled in accordance with 2.2. There shall be included information regarding the various aeroplane configurations and powers or thrusts involved and the relevant speeds, together with information that would assist the flight crew in attaining the performance as scheduled.	CS 23.1587	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.5	7.5 Flight manual A flight manual shall be made available. It shall identify clearly the specific aeroplane or series of aeroplanes to which it is related. The flight manual shall include at least the limitations, information and procedures specified in 7.2, 7.3, 7.4 and 7.6.1.	CS 23.1581-1589	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.1	7.6 Markings and placards 7.6.1 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the flight crew during flight.	CS 23.1541-1567	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.2	7.6.2 Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in ground servicing (towing, refuelling, etc.) that could pass unnoticed and that could jeopardize the safety of the aeroplane in subsequent flights.	CS 23.1557 and appendix G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.1	7.7 Continuing airworthiness — maintenance information 7.7.1 General	CS 23.1529 and Appendix G. Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	Information for use in developing procedures for maintaining the aeroplane in an airworthy condition shall be made available. The information shall include that described in 7.7.2, 7.7.3 and 7.7.4.									
7.7.2	7.7.2 Maintenance information Maintenance information shall include a description of the aeroplane and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.	CS 23.1529 and Appendix G. Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.3	7.7.3 Maintenance programme information Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed. <i>Note.— The development of initial maintenance programme information at the time of aircraft type certification is sometimes referred to as the Maintenance Review Board (MRB) process or the process of developing instructions for continued airworthiness.</i>	CS 23.1529 and Appendix G. M.A.302 Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.4	7.7.4 Mandatory maintenance requirements resulting from the type design approval Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of	CS 23.1529 and Appendix G. Reg. (EU) 1321/2014	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>the type design shall be identified as such and included in the maintenance information of 7.7.3.</p> <p><i>Note.— Mandatory requirements identified as part of the type design approval are often referred to as Certification Maintenance Requirements (CMR) and/or airworthiness limitations.</i></p>									
8.1	<p>CHAPTER 8. CRASHWORTHINESS AND CABIN SAFETY</p> <p>8.1 General</p> <p>Crashworthiness shall be taken into account in the design of aeroplanes to improve the probability of occupant survival.</p>	CS 23.561; 562; 783; 865	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.2.1	<p>8.2 Design emergency landing loads</p> <p>8.2.1 For aeroplanes for which application for certification was submitted before 24 February 2013, emergency landing (crash) loads shall be determined for all categories of aeroplanes so that the interiors, furnishings, support structure and safety equipment can be designed to maximize survivability for the occupants. Items to be considered shall include:</p>	CS 23. 561; 562; 787; 807 (b) 4; 967 (c); 1351; 1365	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<ul style="list-style-type: none"> a) dynamic effects; b) restraint criteria for items that could cause a hazard; c) distortion of the fuselage in the areas of emergency exits; d) fuel cell integrity and position; and e) integrity of electrical systems to avoid sources of ignition. 								
8.2.2	<p>8.2.2 For aeroplanes for which application for certification was submitted on or after 24 February 2013, emergency landing (crash) loads shall be determined so that the interiors, furnishings, support structure and safety equipment can be designed to protect the occupants under emergency landing conditions. Items to be considered shall include:</p> <ul style="list-style-type: none"> a) dynamic effects; b) restraint criteria for items that could cause a hazard; c) deformation of the fuselage in the areas of emergency exits; d) fuel cell integrity and position; and e) integrity of electrical systems to avoid sources of ignition. 	CS 23. 561; 562; 787; 807 (b) 4; 967 (c); 1351; 1365	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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8.3	<p>8.3 Cabin fire protection</p> <p>The cabin shall be so designed as to provide fire protection to the occupants in the event of airborne systems failures or a crash situation. Items to be considered shall include:</p> <p>a) flammability of cabin interior materials;</p> <p>b) fire resistance and the generation of smoke and toxic fumes;</p> <p>c) provision of safety features to allow for safe evacuation; and</p> <p>d) fire detection and suppression equipment.</p>	CS 23.853; 831; 803-815; 851; 855, 859	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.4	<p>8.4 Evacuation</p> <p>The aeroplane shall be equipped with sufficient emergency exits to allow for cabin evacuation within an appropriate time period. Items to be considered, appropriate to the size of the aeroplane, shall include:</p> <p>a) number of seats and seating configuration;</p> <p>b) number, location and size of exits;</p>	CS 23.803-815	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<ul style="list-style-type: none"> c) marking of exits and provision of instructions for use; d) likely blockages of exits; e) operation of exits; and f) positioning and weight of evacuation equipment at exits, e.g. rafts. 								
8.5	<p>8.5 Lighting and marking</p> <p>Emergency lighting, if installed, shall have the following characteristics:</p> <ul style="list-style-type: none"> a) independence from main electrical supply; b) automatic activation upon loss of normal power/impact; c) visual indication of emergency exits; d) illumination both inside and outside the aeroplane during evacuation; and e) no additional hazards in the event of fuel spillage, emergency landings and minor crash events. 	CS 23.811-812	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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9.1	<p>CHAPTER 9. OPERATING ENVIRONMENT AND HUMAN FACTORS</p> <p>9.1 General</p> <p>The aeroplane shall be designed to allow safe operation within the performance limitations of its passengers and those who operate, maintain and service it.</p> <p><i>Note.— The human/machine interface is often the weak link in an operating environment;so, it is necessary to ensure that the aeroplane is capable of being controlled at all phases of the flight (including any degradation due to failures) and that neither the crew nor passengers are harmed by the environment in which they have been placed for the duration of the flight.</i></p>	CS 23. 141; 771; 773; 777-781; 831; 841; 1321. 1322; Subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.1	<p>9.2 Flight crew</p> <p>9.2.1 The aeroplane shall be designed in such a way as to allow safe and efficient control by the flight crew. The design shall allow for variations in flight crew skill and physiology commensurate with flight crew licensing limits. Account shall be taken of the different</p>	CS 23.45;141; 149;672;745; 771; 773; 777-781; 831; 841; 1321. 1322; Subpart	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	expected operating conditions of the aeroplane in its environment, including operations degraded by failures.	G								
9.2.2	<p>9.2.2 The workload imposed on the flight crew by the design of the aeroplane shall be reasonable at all stages of flight. Particular consideration shall be given to critical stages of flight and critical events which may reasonably be expected to occur during the service life of the aeroplane, such as a contained engine failure or windshear encounter.</p> <p><i>Note.— Workload can be affected by both cognitive and physiological factors.</i></p>	CS 23.1523	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.3	<p>9.3 Ergonomics</p> <p>During design of the aeroplane, account shall be taken of ergonomic factors including:</p> <p>a) ease of use and prevention of inadvertent misuse;</p> <p>b) accessibility;</p> <p>c) flight crew working environment;</p> <p>d) cockpit standardization; and</p> <p>e) maintainability.</p>	CS 23. 141; 771; 773; 777-781; 831; 841; 1321. 1322; Subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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9.4	<p>9.4 Operating environmental factors</p> <p>The design of the aeroplane shall take into consideration the flight crew operating environment including:</p> <p>a) effect of aeromedical factors such as level of oxygen, temperature, humidity, noise and vibration;</p> <p>b) effect of physical forces during normal flight;</p> <p>c) effect of prolonged operation at high altitude; and</p> <p>d) physical comfort.</p>	CS 23. 141; 143; 771; 773; 777-781; 831; 841; 1321. 1322; Subpart G	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.1	<p>PART VB. AEROPLANES NOT EXCEEDING 5 700 KG FOR WHICH APPLICATION FOR CERTIFICATION IS SUBMITTED ON OR AFTER 7 MARCH 2021</p> <p>CHAPTER 1. GENERAL</p>	21.B.80 to Annex I, 21L.B.43 to Annex Ib to Reg (EU) 748/2012 CS-23 Amendment 5	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all aeroplanes designated in 1.1.2 for which an application for the issue of a Type Certificate is submitted to the appropriate national authorities on or after 7 March 2021.</p>									
1.1.2	<p>1.1.2 Except for those Standards and Recommended Practices which specify a different applicability, the Standards and Recommended Practices of this part shall apply to all aeroplanes having a maximum certificated take-off mass not exceeding 5 700 kg intended for the carriage of passengers or cargo or mail in international air navigation.</p> <p><i>Note 1.— Guidance material concerning the appropriate airworthiness safety levels commensurate with acceptable risk levels is contained in the Airworthiness Manual (Doc 9760).</i></p> <p><i>Note 2.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p>	21.B.80 to Annex I, 21L.B.43 to Annex Ib to Reg (EU) 748/2012 CS-23 Amendment 5	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.3	1.1.3 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed	21.B.80 to Annex I,	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	national code referred to in 1.2.1 of Part II for the aeroplanes designated in 1.1.2 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.	21L.B.43 to Annex Ib to Reg (EU) 748/2012 CS-23 Amendment 5								
1.1.4	1.1.4 Unless otherwise stated, the Standards apply to the complete aeroplane including its powerplant, systems and equipment.	21.B.80 to Annex I, 21L.B.43 to Annex Ib to Reg (EU) 748/2012 CS-23 Amendment 5	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.1	1.2 Operating limitations 1.2.1 Limiting conditions shall be established for the aeroplane, its powerplant, systems and equipment (see 7.2). Compliance with the Standards of this part shall be established assuming that the aeroplane is operated within the limitations specified. The limitations shall include a margin of safety to render the likelihood of accidents arising therefrom extremely remote.	21.B.80 to Annex I, 21L.B.43 to Annex Ib to Reg (EU) 748/2012 CS-23 Amendment 5 CS 23.2170 CS 23.2200 CS 23.23.2445	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.2	1.2.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the aeroplane, e.g. mass, centre of gravity location, load distribution, speeds, ambient air temperature and altitude,	21.B.80 to Annex I, 21L.B.43 to Annex Ib to	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>shall be established within which compliance with all the pertinent Standards in this part is shown.</p> <p><i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each separate operating condition, e.g. take-off, en route, landing.</i></p> <p><i>Note 2.— Maximum operating mass may be limited by the application of Noise Certification Standards (see Annex 16 — Environmental Protection, Volume I — Aircraft Noise, and Annex 6 — Operation of Aircraft, Part I — International Commercial Air Transport — Aeroplanes and Part II — International General Aviation — Aeroplanes).</i></p>	Reg (EU) 748/2012 CS-23 Amendment 5 CS 23.2100 CS 23.2105								
1.3	<p>1.3 Unsafe features and characteristics</p> <p>Under all anticipated operating conditions, the aeroplane shall not possess any feature or characteristic that renders it unsafe.</p>	21L.A.27 Annex Ib to Reg (EU) 748/2012 21.A.21Reg. (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4	<p>1.4 Proof of compliance</p> <p>The means by which compliance with the appropriate airworthiness requirements is demonstrated shall ensure that in each case the accuracy achieved will be such as to provide reasonable assurance that the aeroplane, its components and equipment comply with the requirements</p>	21.A.20 21.A.33 21.A.35Reg. (EU) 748/2012CS 23.2100CS 23.2235	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	and are reliable and function correctly under the anticipated operating conditions. <i>Note.— Guidance material on the proportionality approach in respect of reasonable assurance for compliance with appropriate airworthiness requirements is contained in Doc 9760.</i>	21L.A.25 Annex Ib to Reg (EU) 748/2012								
2.1.1	CHAPTER 2. FLIGHT 2.1 General 2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon an aeroplane or aeroplanes of the type for which a Type Certificate is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.	21.A.20 21.A.3321.A.35 Reg. (EU) 748/2012 21L.A.25 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1.2	2.1.2 Compliance with each Standard shall be established for all applicable combinations of aeroplane mass and centre of gravity position, within the range of loading conditions for which certification is sought.	21.B.80Reg. (EU) 748/2012CS 23.2100	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		21L.B.43 Annex Ib to Reg (EU) 748/2012								
2.1.3	2.1.3 Where necessary, appropriate aeroplane configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the aeroplane's flying qualities.	21.B.80Reg. (EU) 748/2012CS 23.2100CS 23.2105 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.1	2.2 Performance 2.2.1 Sufficient data on the performance of the aeroplane shall be determined and furnished in the flight manual to provide operators with the necessary information for the purpose of determining the maximum total mass of the aeroplane at the time of take-off that would allow the flight to be made with reasonable assurance that a safe minimum performance for that flight will be achieved considering the values of the operational parameters peculiar to the proposed flight.	21.B.80Reg. (EU) 748/2012CS 23.2105CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.2	2.2.2 Achieving the performance scheduled for the aeroplane shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the flight crew.	21.B.80Reg. (EU) 748/2012CS 23.2105	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>Note.— Guidance material on human performance can be found in the Human Factors Training Manual (Doc 9683).</i>	21L.B.43 Annex Ib to Reg (EU) 748/2012								
2.2.3	2.2.3 The scheduled performance of the aeroplane shall be consistent with compliance with 1.2.1 and with the operation in logical combinations of those of the aeroplane's systems and equipment, the operation of which may affect performance.	21.B.80Reg. (EU) 748/2012CS 23.2105 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2.4	2.2.4 Minimum performance Minimum performance shall be scheduled for aeroplanes with more than one engine that are turbine-powered or have a maximum certificated take-off mass of over 2 721 kg as follows: a) at the maximum masses scheduled (see 2.2.7) for take-off and for landing, as functions of the aerodrome elevation or pressure-altitude either in the standard atmosphere or in specified still air atmospheric conditions; and b) for seaplanes in specified conditions in smooth water, the aeroplane shall be capable of accomplishing the minimum performances specified in 2.2.5 a) and 2.2.6 a)	21.B.80Reg. (EU) 748/2012CS 23.2105 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>respectively, not considering obstacles, or runway or water run length.</p> <p><i>Note.— This Standard permits the maximum take-off mass and maximum landing mass to be scheduled in the flight manual against, for example:</i></p> <ul style="list-style-type: none"> — aerodrome elevation, or — pressure-altitude at aerodrome level, or — pressure-altitude and atmospheric temperature at aerodrome level, <p><i>so as to be readily usable when applying the national code on aeroplane performance operating limitations.</i></p>									
2.2.5	<p>2.2.5 Take-off</p> <p>a) For aeroplanes with more than one engine that are turbine-powered or have a maximum certificated take-off mass of over 2 721 kg, after the end of the period during which the take-off power or thrust may be used, the aeroplane shall be capable of continuing to climb, with the critical engine inoperative and the remaining engine(s) operated within their maximum continuous power or thrust limitations, up to a height that it can maintain and at which it can continue safe flight and landing.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2120</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>b) The minimum performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is scheduled (see 2.2.7), the departure from the scheduled values is not disproportionate.</p>								
2.2.6	<p>2.2.6 Landing</p> <p>a) For aeroplanes for which application for certification was submitted on or after 24 February 2013, aeroplanes with one engine, or a single propeller, or aeroplanes with more than one engine that cannot maintain a positive climb gradient following an engine or propeller failure, the design shall, in the case of engine or propeller failure, enable the aeroplane to be operated to a safe forced landing in favourable conditions.</p> <p>b) For aeroplanes with more than one engine that are turbine-powered or have a maximum certificated take-off mass of over 2 721 kg, starting from the approach configuration and with the critical engine inoperative, the aeroplane shall be capable, in the event of a missed approach, of continuing the flight to a point from which another approach can be made.</p> <p>c) Starting from the landing configuration, the aeroplane shall be capable, in the event of a balked landing, of making a climb-out, with all engines operating.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2110CS 23.2125CS 23.2130</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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2.2.7	<p>2.2.7 Scheduling of performance</p> <p>Performance data shall be determined and scheduled in the flight manual in order to provide a safe relationship between the performance of the aeroplane and the aerodromes and routes on which it is capable of being operated. Performance data shall be determined and scheduled for the following stages for the ranges of mass, altitude or pressure-altitude, wind velocity, gradient of the take-off and landing surface for landplanes; water surface conditions, density of water and strength of current for seaplanes; and for any other operational variables for which the aeroplane is to be certificated.</p> <p>a) <i>Take-off.</i> The take-off performance data shall include the distance required to take-off and climb to a selected height above the take-off surface. It shall be determined for each mass, altitude and temperature within the operational limits established for take-off with:</p> <ul style="list-style-type: none"> — take-off power on each engine; — wing flaps in the take-off position(s); and — landing gear extended. <p>b) <i>En route.</i> For aeroplanes with more than one engine, the en-route climb performance shall be the climb (or descent) performance with the aeroplane in the en-route configuration with the critical engine inoperative. The operating engine(s) shall not exceed maximum continuous power or thrust.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2105CS 23.2115CS 23.2125CS 23.2130</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
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	<p>c) <i>Landing.</i> The landing distance shall be the horizontal distance traversed by the aeroplane from a point on the approach flight path at a selected height above the landing surface to the point on the landing surface at which the aeroplane comes to a complete stop, or, for a seaplane, comes to a satisfactorily low speed. The selected height above the landing surface and the approach speed shall be appropriately related to operating practices. This distance may be supplemented by such distance margin as may be necessary; if so, the selected height above the landing surface, the approach speed and the distance margin shall be appropriately interrelated and shall make provision for both normal operating practices and reasonable variations therefrom.</p>									
2.3.1	<p>2.3 Flying qualities</p> <p>2.3.1 The aeroplane shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the aeroplane is approved.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2135</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.1	<p>2.3.2 Controllability</p> <p>2.3.2.1 The aeroplane shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips,</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2135</p> <p>21L.B.43</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	changes of engine power or thrust, changes of aeroplane configurations) without requiring exceptional skill, alertness or strength on the part of the pilot even in the event of failure of any engine. A technique for safely controlling the aeroplane shall be established for all stages of flight and aeroplane configurations for which performance is scheduled. <i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i>	Annex Ib to Reg (EU) 748/2012								
2.3.2.2	2.3.2.2 <i>Controllability on the ground (or water)</i> . The aeroplane shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.	21.B.80Reg. (EU) 748/2012CS 23.2155 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.2.3	2.3.2.3 <i>Controllability during take-off</i> . The aeroplane shall be controllable in the event of sudden failure of the critical engine at any point in the take-off.	21.B.80Reg. (EU) 748/2012CS 23.2135 21L.B.43 Annex Ib to Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		748/2012								
2.3.2.4	2.3.2.4 <i>Take-off safety speed.</i> The take-off safety speeds assumed when the performance of the aeroplane (after leaving the ground or water) during the take-off is determined shall provide an adequate margin above the stall and above the minimum speed at which the aeroplane remains controllable after sudden failure of the critical engine.	21.B.80Reg. (EU) 748/2012CS 23.2115 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3.3	2.3.3 Trim The aeroplane shall have such trim characteristics as to ensure that the demands made on the pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. This shall apply both in normal operation and in the conditions associated with the failure of one or more engines for which performance characteristics are established.	21.B.80Reg. (EU) 748/2012CS 23.2140 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.1	2.4 Stability and control 2.4.1 Stability The aeroplane shall have such stability in relation to its other flight characteristics, performance, structural strength, and most probable operating conditions (e.g. aeroplane configurations and speed ranges) as to ensure that demands made on the pilot's powers of concentration are not excessive when the stage of the flight at which	21.B.80Reg. (EU) 748/2012CS 23.2135CS 23.2145CS 23.2300 21L.B.43 Annex Ib to Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	these demands occur and their duration are taken into account. The stability of the aeroplane shall not, however, be such that excessive demands are made on the pilot's strength or that the safety of the aeroplane is prejudiced by lack of manoeuvrability in emergency conditions. The stability may be achieved by natural or artificial means, or a combination of both. In those cases where artificial stability is necessary to show compliance with the Standards of this part, it shall be shown that any failure or condition that would result in the need for exceptional pilot skill or strength for recovery of aeroplane stability is extremely improbable.	748/2012								
2.4.2.1	<p>2.4.2 Stalling</p> <p>2.4.2.1 <i>Stall warning.</i> When the aeroplane approaches a stall both in straight and turning flight, a clear and distinctive stall warning shall be apparent to the pilot with the aeroplane in all permissible configurations, except those which are not considered to be essential for safe flying. The stall warning and other characteristics of the aeroplane shall be such as to enable the pilot to arrest the development of the stall after the warning begins and, without altering the engine power or thrust, to maintain full control of the aeroplane.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2150</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2.2	<p>2.4.2.2 <i>Behaviour following a stall.</i> In any configuration and at any level of power or thrust in which it is considered that the ability to recover from a stall is essential, the behaviour of the aeroplane following a stall shall not be so extreme as to make difficult a prompt</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2150</p> <p>21L.B.43</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	recovery without exceeding the airspeed or strength limitations of the aeroplane.	Annex Ib to Reg (EU) 748/2012								
2.4.2.3	2.4.2.3 <i>Stalling speeds.</i> The stalling speeds or minimum steady flight speeds in configurations appropriate for each stage of flight (e.g. take-off, en route, landing) shall be established. One of the values of the power or thrust used in establishing the stalling speeds shall be not more than that necessary to give zero thrust at a speed just above the stall.	21.B.80Reg. (EU) 748/2012CS 23.2110 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3.1	2.4.3 Flutter and vibration 2.4.3.1 It shall be demonstrated by suitable tests, analyses or any acceptable combination of tests and analyses that all parts of the aeroplane are free from flutter and excessive vibration in all aeroplane configurations under all speed conditions within the operating limitations of the aeroplane (see 1.2.2). There shall be no vibration or buffeting severe enough to cause structural damage.	21.B.80Reg. (EU) 748/2012CS 23.2245CS 23.2160 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.3.2	2.4.3.2 There shall be no vibration or buffeting severe enough to interfere with control of the aeroplane or to cause excessive fatigue to the flight crew. <i>Note.— Buffeting as a stall warning is considered desirable and discouragement of this type of buffeting is not intended.</i>	21.B.80Reg. (EU) 748/2012CS 23.2135CS 23.2160 21L.B.43 Annex Ib to	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		Reg (EU) 748/2012								
2.4.4	2.4.4 Spinning It shall be demonstrated that the aeroplane during normal operation does not exhibit any tendency to inadvertently enter into a spin. If the design is such that spinning is allowed or for aeroplanes with one engine inadvertently possible, it shall be demonstrated that with normal use of the controls and without the use of exceptional piloting skill the aeroplane can be recovered from a spin within appropriate recovery limits.	21.B.80Reg. (EU) 748/2012CS 23.2150 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.1	CHAPTER 3. STRUCTURE 3.1 General The aeroplane structure shall be designed, manufactured and provided with instructions for its maintenance and repair with the objective of avoiding catastrophic failure throughout its operational life.	21.B.80Reg. (EU) 748/2012CS 23.2625 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.2	3.2 Mass and mass distribution Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the	21.B.80Reg. (EU) 748/2012CS 23.2200	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.	21L.B.43 Annex Ib to Reg (EU) 748/2012								
3.3	3.3 Limit loads Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.6 shall be considered as limit loads.	21.B.80Reg. (EU) 748/2012CS 23.2210 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.4	3.4 Strength and deformation In the various loading conditions prescribed in 3.6, no part of the aeroplane structure shall sustain detrimental deformation at any load up to and including the limit load, and the aeroplane structure shall be capable of supporting the ultimate load.	21.B.80Reg. (EU) 748/2012CS 23.2235 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.5.1	3.5 Airspeeds 3.5.1 Design airspeeds Design airspeeds shall be established for which the aeroplane structure is designed to withstand the	21.B.80Reg. (EU) 748/2012CS 23.2200 21L.B.43 Annex Ib to	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	corresponding manoeuvring and gust loads. To avoid inadvertent exceedences due to upsets or atmospheric variations, the design airspeeds shall provide sufficient margin for the establishment of practical operational limiting airspeeds. In addition, the design airspeeds shall be sufficiently greater than the stalling speed of the aeroplane to safeguard against loss of control in turbulent air. Consideration shall be given to a design manoeuvring speed, a design cruising speed, a design dive speed and any other design airspeeds necessary for configurations with high lift or other special devices.	Reg (EU) 748/2012								
3.5.2	3.5.2 Limiting airspeeds Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the flight manual as part of the operating limitations (see 7.2).	21.B.80Reg. (EU) 748/2012CS 23.2610CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.6.1	3.6 Strength 3.6.1 All structural elements shall be designed to withstand the maximum loads expected in service under all anticipated operating conditions without failure, permanent distortion or loss of functionality. In determining these loads, account shall be taken of:	21.B.80Reg. (EU) 748/2012CS 23.2015CS 23.2200CS 23.2210CS 23.2225CS 23.2240CS 23.2245CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>a) the expected operational life of the aeroplane;</p> <p>b) the vertical and horizontal gust environment, taking into consideration the expected variations in mission profile and loading configurations;</p> <p>c) the manoeuvre spectrum, taking into account variations in mission profile and loading configurations;</p> <p>d) asymmetrical as well as symmetrical loading;</p> <p>e) the ground and water loads, including taxi, landing and take-off loads, and ground/water handling loads;</p> <p>f) the speed range of the aeroplane, taking into account the aeroplane characteristics and operation limitations;</p> <p>g) vibration and buffeting loads;</p> <p>h) corrosion or other degradation, given the maintenance specified, and various operating environments; and</p> <p>i) any other loads, such as flight control loads, cabin pressurization loads, engine loads, or dynamic loads due to changes to the steady state configuration.</p>	<p>23.2255</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>							
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3.6.2	3.6.2 The air, inertia and other loads resulting from the specific loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively.	21.B.80Reg. (EU) 748/2012CS 23.2230 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.7	3.7 Survivability The aeroplane shall be designed so as to provide the occupants with the maximum practicable protection in the event of structural failure, or in the event of damage due to ground, water or object impact. Consideration shall be given to at least the following: a) energy absorption by the airframe, occupant seats and restraints; and b) allowing egress in the shortest practicable time.	21.B.80Reg. (EU) 748/2012CS 23.2270 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.8	3.8 Structural durability The structure of the aeroplane shall conform to damage tolerance, safe-life or failsafe principles and shall be such as to avoid catastrophic failure during the operational life, taking into account, where appropriate:	21.B.80Reg. (EU) 748/2012CS 23.2240 21L.B.43 Annex Ib to	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<ul style="list-style-type: none"> a) the expected environment; b) the expected repeated loads applied in service; c) expected vibrations from aerodynamic interaction or internal sources; d) thermal cycles; e) accidental and discrete source damage; f) likely corrosion or other degradation; g) specified maintenance; and h) likely structural repairs. 	Reg (EU) 748/2012								
3.9	<p>3.9 Special factors</p> <p>For aeroplanes for which application for certification was submitted on or after 24 February 2013, design features (e.g. castings, bearings or fittings), the strength of which is subject to variability in manufacturing processes, deterioration in service, or any other cause, shall be accounted for by a suitable factor.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2265</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.1.1	<p>CHAPTER 4. DESIGN AND CONSTRUCTION</p> <p>4.1 General</p> <p>4.1.1 Details of design and construction shall be such as to give reasonable assurance that all aeroplane parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2250</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.2	<p>4.1.2 Substantiation of moving parts</p> <p>The functioning of all moving parts essential to the safe operation of the aeroplane shall be demonstrated in order to ensure that they will function correctly under all operating conditions for such parts.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2235CS 23.2250CS 23.2300</p> <p>21L.B.43 Annex Ib to Reg (EU)</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		748/2012								
4.1.3	<p>4.1.3 Materials</p> <p>All materials used in parts of the aeroplane essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design.</p>	<p>Reg. (EU) 1907/2006CS 23.2250CS 23.2260</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.4	<p>4.1.4 Manufacturing methods</p> <p>The methods of manufacturing and assembly shall be such as to produce a consistently sound structure which shall be reliable with respect to maintenance of strength in service.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2260</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.1.5	<p>4.1.5 Protection</p> <p>The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the aeroplane will receive.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2255</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.1.6	4.1.6 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the aeroplane that require such attention, either periodically or after unusually severe operations.	21.B.80Reg. (EU) 748/2012CS 23.2255 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.2	4.2 Systems design features Special consideration shall be given to design features that affect the ability of the flight crew to maintain controlled flight. This shall include at least the following: a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such as to minimize the possibility of jamming, inadvertent operation including prevention of mis-assembly, and unintentional engagement of control surface locking devices. 1) Each control and control system shall operate with the ease, smoothness and precision appropriate to its functions. 2) Each element of each flight control system shall be designed, or distinctively and permanently marked, to minimize the probability of any incorrect assembly that could result in the malfunction of the system.	21.B.80Reg. (EU) 748/2012CS 23.2250CS 23.2300CS 23.2320CS 23.2325CS 23.2500CS 23.2510CS 23.2600CS 23.2605CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>		Linked to Amdt109: The aim is to complete the transposition of the recommendation in paragraph(g)(3) before it becomes applicable on or after 1 January 2025.				

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	<p>b) <i>System survivability.</i> Aeroplane systems shall be designed and arranged to maximize the potential for continued safe flight and landing after events resulting in damage to the aeroplane structure or systems.</p> <p>c) <i>Crew environment.</i> The design of the flight crew compartment shall be such as to minimize the possibility of incorrect or restricted operation of the controls by the crew, due to fatigue, confusion or interference. Consideration shall be given at least to the following: layout and identification of controls and instruments, rapid identification of emergency situations, sense of controls, ventilation, heating and noise.</p> <p>d) <i>Pilot vision.</i> The arrangement of the flight crew compartment shall be such as to afford a sufficiently extensive, clear and undistorted field of vision for the safe operation of the aeroplane, and to prevent glare and reflections that would interfere with the pilot's vision. The design features of the windshield shall permit, under precipitation conditions of moderate rain, sufficient vision for the normal conduct of flight and for the execution of approaches and landings.</p> <p>e) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the flight crew to deal with, emergencies resulting from foreseeable failures of equipment and systems, the failure of which would endanger the aeroplane. Reasonable provisions shall be made for continuation of essential services following engine or system failures to the extent that such failures are catered for in the</p>								
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	<p>performance and operating limitations specified in the Standards in this Annex and in Annex 6, Parts I and II.</p> <p>f) <i>Fire precautions.</i> The design of the aeroplane and the materials used in its manufacture shall be such so as to minimize the risk of in-flight and ground fires, and to minimize the production of smoke and toxic gases in the event of a fire.</p> <p>g) <i>Cargo compartment protection.</i></p> <p>1) Sources of heat within the compartment which are capable of igniting the cargo or baggage shall be shielded or insulated to prevent such ignition; and</p> <p>2) Each cargo and baggage compartment shall be constructed of materials which are at least flame resistant.</p> <p>3) Recommendation.— <i>As of 7 March 2025, the elements of the aeroplane design associated with cargo compartment fire protection, and a summary of the demonstrated standards that were considered in the process of aeroplane certification should be included in the required aeroplane documentation and made available to the operator.</i></p> <p>h) <i>Incapacitation of occupants.</i> Design precautions shall be taken to protect against possible instances of cabin depressurization and against the</p>								
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	presence of smoke or other toxic gases that could incapacitate the occupants of the aeroplane.									
4.3	4.3 Aeroelasticity The aeroplane shall be free from flutter, structural divergence, control reversal, loss of control due to structural deformation and aeroelastic effects, at all speeds within and sufficiently beyond the design envelope to comply with 1.2.1. Account shall be taken of the characteristics of the aeroplane.	21.B.80Reg. (EU) 748/2012CS 23.2245 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.4.1	4.4 Occupant accommodation features 4.4.1 Seating and restraints Adequate seating and restraints shall be provided for the occupants, taking account of the likely flight and emergency landing loads to be encountered. Attention shall be paid to minimizing injury to occupants due to contact with surrounding structures during the operation of the aeroplane.	21.B.80Reg. (EU) 748/2012CS 23.2265 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.4.2	4.4.2 Cabin environment Ventilation, heating and, where applicable, pressurization systems shall be designed to provide the cabin with an adequate environment during the anticipated flight and	21.B.80Reg. (EU) 748/2012CS 23.2320CS 23.2500	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	ground or water operating conditions. The systems design shall also consider likely emergency conditions.	21L.B.43 Annex Ib to Reg (EU) 748/2012								
4.5.1	<p>4.5 Electrical bonding and protection against lightning and static electricity</p> <p>4.5.1 Electrical bonding, protection against static electricity and lightning protection when appropriate for the type of approved operations shall be such as to:</p> <p>a) protect the aeroplane, its systems, its occupants and those who come in contact with the aeroplane on the ground or water from the dangerous effects of lightning discharge and electrical shock; and</p> <p>b) prevent dangerous accumulation of electrostatic charge.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2335</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.5.2	4.5.2 When appropriate for the type of approved operation, the aeroplane shall also be protected against catastrophic effects of lightning. Due account shall be taken of the material used in the construction of the aeroplane.	<p>21.B.80Reg. (EU) 748/2012CS 23.2335</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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4.6.1	<p>4.6 Emergency landing provisions</p> <p>4.6.1 Provisions shall be made in the design of the aeroplane to protect the occupants, in the event of an emergency landing, from fire and from the direct effects of deceleration forces as well as from injuries arising from the effect of deceleration forces on the aeroplane's interior equipment.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2270</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.6.2	<p>4.6.2 Facilities shall be provided for the rapid evacuation of the aeroplane in conditions likely to occur following an emergency landing. Such facilities shall be related to the passenger and crew capacity of the aeroplane and shall be shown to be suitable for their intended purpose.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2315</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
4.7	<p>4.7 Ground handling</p> <p>Design provisions and procedures for safe ground handling (e.g. towing, jacking) shall be defined. The protection that any limitations and instructions for such operations might provide may be taken into account.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2210</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.1	<p>CHAPTER 5. POWERPLANT</p> <p>5.1 Engines</p> <p>The Standards of Part VI of this Annex shall apply to each engine that is used on the aeroplane as a primary propulsion unit.</p>	<p>21.A.21Reg. (EU) 748/2012CS-E 21L.A.27 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.2	<p>5.2 Propellers</p> <p>The Standards of Part VII of this Annex shall apply to each propeller that is used on the aeroplane.</p>	<p>21.A.21Reg. (EU) 748/2012CS-P 21L.A.27 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.1	<p>5.3 Powerplant installation</p> <p>5.3.1 Compliance with engine and propeller limitations</p> <p>The powerplant installation shall be so designed that the engines and propellers (if applicable) are capable of functioning reliably in the anticipated operating conditions. In conditions established in the flight manual, the aeroplane shall be capable of being operated without exceeding the limitations established for the engines and</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2400 21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	propellers in accordance with this chapter and Parts VI and VII.									
5.3.2	<p>5.3.2 Control of engine rotation</p> <p>In those installations where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided for the crew to stop the rotation of the failed engine in flight or to reduce it to a safe level.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2400</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.3	<p>5.3.3 Turbine engine installation</p> <p>For a turbine engine installation:</p> <p>a) the design shall minimize the hazards to the aeroplane in the event of failure of engine rotating parts, or an engine fire which burns through the engine case; and</p> <p>b) the powerplant installation shall be designed to give reasonable assurance that those engine operating limitations that adversely affect the structural integrity of rotating parts shall not be exceeded in service.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2400</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.4	<p>5.3.4 Engine restarting</p> <p>Means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2400</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		21L.B.43 Annex Ib to Reg (EU) 748/2012								
5.3.5.1	5.3.5 Arrangement and functioning 5.3.5.1 <i>Independence of engines.</i> The powerplant shall be arranged and installed so that each engine together with its associated systems is capable of being controlled and operated independently from the others and so that there is at least one arrangement of the powerplant and systems in which any failure, unless the probability of its occurrence is extremely remote, cannot result in a loss of more power than that resulting from complete failure of the critical engine.	21.B.80Reg. (EU) 748/2012CS 23.2400 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.2	5.3.5.2 <i>Independence of engines and associated systems.</i> The engines together with their associated systems shall be arranged and isolated from each other to allow operation, in at least one configuration, so that the failure or malfunction of any engine, or the failure or malfunction (including destruction by fire in the engine compartment) of any system that can affect an engine (other than a fuel tank if only one fuel tank is installed), will not: a) prevent the continued safe operation of the remaining engine(s); or	21.B.80Reg. (EU) 748/2012CS 23.2400 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	b) require immediate action by any crew member for continued safe operation of the remaining engine(s).									
5.3.5.3	5.3.5.3 <i>Propeller vibration.</i> The propeller vibration stresses shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the aeroplane.	21.B.80Reg. (EU) 748/2012CS 23.2400 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.4	5.3.5.4 <i>Cooling.</i> The cooling system shall be capable of maintaining the temperature of powerplant components and fluids within the established limits (see 5.3.1) at ambient air temperatures up to the maximum air temperature appropriate to the intended operation of the aeroplane.	21.B.80Reg. (EU) 748/2012CS 23.2400 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
5.3.5.5	5.3.5.5 <i>Associated systems.</i> The fuel, oil, air induction and other systems associated with the powerplant shall be capable of supplying each engine in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power or thrust, aeroplane attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.	21.B.80Reg. (EU) 748/2012CS 23.2400CS 23.2435 21L.B.43 Annex Ib to Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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5.3.5.6	<p>5.3.5.6 <i>Fire protection.</i> For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.2 f).</p> <p>a) <i>Isolation.</i> Such regions shall be isolated by fireproof material from other regions of the aeroplane where the presence of fire would jeopardize continued flight, taking into account the probable points of origin and paths of propagation of fire.</p> <p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be fire resistant. Drainage of each region shall be provided to minimize hazards resulting from the failure of any component containing flammable fluids. Means shall be provided for the crew to shut off the flow of flammable fluids into such regions if a fire occurs. Where sources of flammable fluid exist in such regions, the whole of the related system within the region, including supporting structure, shall be fireproof or shielded from the effects of the fire.</p> <p>c) <i>Fire detection.</i> A sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions of the following aeroplane types: aeroplanes with more than one engine powered by turbine or turbo-charged engines, or aeroplanes where the engine(s) are not readily visible from the cockpit.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2440</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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6.1.1	<p>CHAPTER 6. SYSTEMS AND EQUIPMENT</p> <p style="text-align: center;">6.1 General</p> <p>6.1.1 The aeroplane shall be provided with approved instruments, equipment and systems, including guidance and flight management systems necessary for the safe operation of the aeroplane in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the crew to operate the aeroplane within its operating limitations. Instruments and equipment design shall consider human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Parts I and II, for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683).</i></p>	21.B.80Reg. (EU) 748/2012CS-23 Subpart F 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
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6.1.2	<p>6.1.2 The design of the instruments, equipment and systems required by 6.1.1 and their installation shall be such that:</p> <p>a) an inverse relationship exists between the probability of a failure condition and the severity of its effect on the aircraft and its occupants, as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p> <p>c) electromagnetic interference between them is minimized.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2500</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.1.3	<p>6.1.3 Means shall be provided to warn the crew of unsafe system operating conditions and to enable them to take corrective action.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2500CS 23.2605</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.1.4	<p>6.1.4 Electrical power supply</p> <p>The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations and shall also be such that no single failure or malfunction could impair the ability of the system to supply essential loads for safe operation.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2500CS 23.2605</p> <p>21L.B.43</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		Annex Ib to Reg (EU) 748/2012								
6.1.5	<p>6.1.5 Development assurance of complex electronic hardware and system software</p> <p>For aeroplanes for which application for certification was submitted on or after 24 February 2013, complex electronic hardware and system software shall be developed, verified and validated such as to ensure that the systems in which they are used perform their intended functions at a level of safety that complies with the requirements of this part, notably those of 6.1.2 a) and 6.1.2 b).</p> <p><i>Note.— Some States accept the use of national or international industry standards for the development assurance (development, verification and validation) of complex electronic hardware and systems software.</i></p>	<p>21.B.75Reg. (EU) 748/2012</p> <p>21L.B.44 Annex Ib to Reg (EU) 748/2012</p>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	Not specifically addressed in CS-25 and CS-23. However, EASA Certification Memo (CM-SWCEH-001) is guidance for the development assurance of CEH and SW and applied in certification project in Special Conditions. This provides guidance	

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									to comply with 6.1.2 (a) and 6.1.2 (b).	
6.2	6.2 Installation Instrument and equipment installations shall comply with the Standards of Chapter 4.	21.B.80Reg. (EU) 748/2012CS 23.2500 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.3	6.3 Safety and survival equipment Prescribed safety and survival equipment that the crew or passengers are expected to use or operate at the time of an emergency shall be reliable, readily accessible and easily identified, and its method of operation shall be plainly marked.		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input checked="" type="checkbox"/>		Aviation rules for civil aircraft registration 12.02.2008 № 5 (as amended by resolution Ministry of transportation

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										and communications The Republic of Belarus May 29, 2020 No. 29) paragraph 112
6.4.1	<p>6.4 Navigation lights and anti-collision lights</p> <p>6.4.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by aeroplanes in flight or operating on the movement area of an aerodrome shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft or personnel on the ground with as much time as possible for interpretation and for subsequent manoeuvre necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry sky, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in</i></p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2530</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i>									
6.4.2	<p>6.4.2 Lights shall be installed in aeroplanes so as to minimize the possibility that they will adversely affect the satisfactory performance of the flight crews' duties.</p> <p><i>Note.— In order to avoid the effects mentioned in 6.4.2, it will be necessary in some cases to provide means whereby the pilot can adjust the intensity of the flashing lights.</i></p>	21.B.80Reg. (EU) 748/2012CS 23.2530CS 23.2600 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.5	<p>6.5 Electromagnetic interference protection</p> <p>Aeroplane electronic systems, particularly flight-critical and flight-essential systems, shall be protected against electromagnetic interference from both internal and external sources.</p>	21.B.80Reg. (EU) 748/2012AM C 20-136, 158CS 23.2500CS 23.2515CS 23.2520 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
6.6	6.6 Ice protection	21.B.80Reg. (EU) 748/2012CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	If certification for flight in icing conditions is requested, the aeroplane shall be shown to be able to operate safely in icing conditions likely to be encountered in all anticipated operating environments.	23.2540 21L.B.43 Annex Ib to Reg (EU) 748/2012								
7.1	CHAPTER 7. OPERATING LIMITATIONS AND INFORMATION 7.1 General The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the aeroplane, shall be made available by means of a flight manual, markings and placards, and such other means as may effectively accomplish the purpose.	21.B.80Reg. (EU) 748/2012CS 23.2610CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.1	7.2 Operating limitations 7.2.1 Limitations which might be exceeded in flight and which are defined quantitatively shall be expressed in suitable units. These limitations shall be corrected if necessary for errors in measurements so that the flight crew can, by reference to the instruments	21.B.80Reg. (EU) 748/2012CS 23.2170CS 23.2610CS 23.2620	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	available to them, readily determine when the limitations are reached.	21L.B.43 Annex Ib to Reg (EU) 748/2012								
7.2.2	7.2.2 Loading limitations The loading limitations shall include all limiting masses, centre of gravity positions, mass distributions and floor loadings (see 1.2.2).	21.B.80Reg. (EU) 748/2012CS 23.2170CS 23.2610 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.3	7.2.3 Airspeed limitations The airspeed limitations shall include all speeds (see 3.5.2) that are limiting from the standpoint of structural integrity or flying qualities of the aeroplane, or from other considerations. These speeds shall be identified with respect to the appropriate aeroplane configurations and other pertinent factors.	21.B.80Reg. (EU) 748/2012CS 23.2610 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.4	7.2.4 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the aeroplane (see 5.3.1 and 5.3.5.4).	21.B.80Reg. (EU) 748/2012CS 23.2610 21L.B.43 Annex Ib to	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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		Reg (EU) 748/2012								
7.2.5	7.2.5 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the aeroplane.	21.B.80Reg. (EU) 748/2012CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.6	7.2.6 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the aeroplane (see 1.2.1).	21.B.80Reg. (EU) 748/2012CS 23.2170CS 23.2610 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.2.7	7.2.7 Flight crew limitations The flight crew limitations shall include the minimum number of flight crew personnel necessary to operate the aeroplane, having regard, among other things, to the accessibility to the appropriate crew members of all necessary controls and instruments and to the execution of the established emergency procedures.	21.B.80Reg. (EU) 748/2012CS 23.2610 21L.B.43 Annex Ib to Reg (EU)	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>Note.— The circumstances in which the flight crew shall include members in addition to the minimum flight crew are defined in Annex 6, Part I and Part II.</i>	748/2012								
7.3.1	<p>7.3 Operating information and procedures</p> <p>7.3.1 Types of eligible operations</p> <p>The particular types of operations for which the aeroplane has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2610</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.2	<p>7.3.2 Loading information</p> <p>The loading information shall include the empty mass of the aeroplane, together with a definition of the condition of the aeroplane at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the crew and payload, the usable fuel supply and the drainable oil; it includes the mass of all fixed ballast, unusable fuel supply, undrainable oil, total quantity of engine coolant and total quantity of hydraulic fluid.</i></p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2620</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.3.3	7.3.3 Operating procedures A description shall be given of normal and emergency operating procedures which are peculiar to the particular aeroplane and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.	21.B.80Reg. (EU) 748/2012CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.3.4	7.3.4 Handling information Sufficient information shall be given on any significant or unusual features of the aeroplane characteristics. Those stalling speeds or minimum steady flight speeds required to be established by 2.4.2.3 shall be scheduled.	21.B.80Reg. (EU) 748/2012CS 23.2610CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.4	7.4 Performance information The performance of the aeroplane shall be furnished in accordance with 2.2. There shall be included information regarding the various aeroplane configurations and powers or thrusts involved and the relevant speeds, together with information that would assist the flight crew in attaining the performance as furnished.	21.B.80Reg. (EU) 748/2012CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.5	7.5 Flight manual A flight manual shall be made available. It shall identify clearly the specific aeroplane or series of aeroplanes to which it is related. The flight manual shall include at least the limitations, information and procedures specified in 7.2, 7.3, 7.4 and 7.6.1.	21.B.80Reg. (EU) 748/2012CS 23.2620 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.1	7.6 Markings and placards 7.6.1 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the flight crew during flight.	21.B.80Reg. (EU) 748/2012CS 23.2610 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.6.2	7.6.2 Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in ground servicing (towing, refuelling, etc.) that could pass unnoticed and that could jeopardize the safety of the aeroplane in subsequent flights.	21.B.80Reg. (EU) 748/2012CS 23.2610 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.1	7.7 Continuing airworthiness — maintenance information	21.B.80 Reg. (EU) 748/2012CS	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	7.7.1 General Information for use in developing procedures for maintaining the aeroplane in an airworthy condition shall be made available. The information shall include that described in 7.7.2, 7.7.3 and 7.7.4.	23.2625 21L.B.43 Annex Ib to Reg (EU) 748/2012								
7.7.2	7.7.2 Maintenance information Maintenance information shall include a description of the aeroplane and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.	21.B.80 Reg. (EU) 748/2012CS 23.2625 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
7.7.3	7.7.3 Maintenance programme information Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed. <i>Note.— The development of initial maintenance programme information at the time of aircraft type certification is sometimes referred to as the Maintenance Review Board (MRB) process or the process of developing instructions for continued airworthiness.</i>	21.B.80 Reg. (EU) 748/2012CS 23.2625 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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7.7.4	<p>7.7.4 Mandatory maintenance requirements resulting from the type design approval</p> <p>Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 7.7.3.</p> <p><i>Note.— Mandatory requirements identified as part of the type design approval are often referred to as Certification Maintenance Requirements (CMR) and/or airworthiness limitations.</i></p>	<p>21.B.80 Reg. (EU) 748/2012CS 23.2625</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.1	<p>CHAPTER 8. CRASHWORTHINESS AND CABIN SAFETY</p> <p>8.1 General</p> <p>Crashworthiness shall be taken into account in the design of aeroplanes to improve the probability of occupant survival.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2270CS 23.2315CS 23.2320</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.2	<p>8.2 Design emergency landing loads</p> <p>Emergency landing (crash) loads shall be determined so that the interiors, furnishings, support structure and safety equipment can be designed to protect the occupants under</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2270CS 23.2270CS</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>emergency landing conditions. Items to be considered shall include:</p> <p>a) dynamic effects;</p> <p>b) restraint criteria for items that could cause a hazard;</p> <p>c) deformation of the fuselage in the areas of emergency exits;</p> <p>d) fuel cell integrity and position; and</p> <p>e) integrity of electrical systems to avoid sources of ignition.</p>	<p>23.2325CS 23.2330CS 23.2430</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>							
8.3	<p>8.3 Cabin fire protection</p> <p>The cabin shall be so designed as to provide fire protection to the occupants in the event of airborne systems failures or a crash situation. Items to be considered shall include:</p> <p>a) flammability of cabin interior materials;</p> <p>b) fire resistance and the generation of smoke and toxic fumes;</p> <p>c) provision of safety features to allow for safe evacuation; and</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2270CS 23.2270CS 23.2325CS 23.2330CS 23.2430</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>					

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	d) fire detection and suppression equipment.									
8.4	<p>8.4 Evacuation</p> <p>The aeroplane shall be equipped with sufficient emergency exits to allow for cabin evacuation within an appropriate time period. Items to be considered, appropriate to the size of the aeroplane, shall include:</p> <p>a) number of seats and seating configuration;</p> <p>b) number, location and size of exits;</p> <p>c) marking of exits and provision of instructions for use;</p> <p>d) likely blockages of exits;</p> <p>e) operation of exits; and</p> <p>f) positioning and weight of evacuation equipment at exits, e.g. rafts.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2315</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
8.5	<p>8.5 Lighting and marking</p> <p>Emergency lighting, if installed, shall have the following characteristics:</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2315</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<p>a) independence from main electrical supply;</p> <p>b) automatic activation upon loss of normal power/impact;</p> <p>c) visual indication of emergency exits;</p> <p>d) illumination both inside and outside the aeroplane during evacuation; and</p> <p>e) no additional hazards in the event of fuel spillage, emergency landings and minor crash events.</p>	21L.B.43 Annex Ib to Reg (EU) 748/2012								
9.1	<p>CHAPTER 9. OPERATING ENVIRONMENT AND HUMAN FACTORS</p> <p>9.1 General</p> <p>The aeroplane shall be designed to allow safe operation within the performance limitations of its passengers and those who operate, maintain and service it.</p> <p><i>Note.— The human/machine interface is often the weak link in an operating environment; so, it is necessary to ensure that the aeroplane is capable of</i></p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2135CS 23.2600CS 23.2605CS 23.2610</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>being controlled at all phases of the flight (including any degradation due to failures) and that neither the crew nor passengers are harmed by the environment in which they have been placed for the duration of the flight.</i>									
9.2.1	<p align="center">9.2 Flight crew</p> <p>9.2.1 The aeroplane shall be designed in such a way as to allow safe and efficient control by the flight crew. The design shall allow for variations in flight crew skill and physiology commensurate with flight crew licensing limits. Account shall be taken of the different expected operating conditions of the aeroplane in its environment, including operations degraded by failures.</p>	21.B.80Reg. (EU) 748/2012CS 23.2105CS 23.2135CS 23.2300CS 23.2320CS 23.2600CS 23.2605CS 23.2610 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.2.2	<p>9.2.2 The workload imposed on the flight crew by the design of the aeroplane shall be reasonable at all stages of flight. Particular consideration shall be given to critical stages of flight and critical events which may reasonably be expected to occur during the service life of the aeroplane, such as a contained engine failure or windshear encounter.</p> <p><i>Note.— Workload can be affected by both cognitive and physiological factors.</i></p>	21.B.80Reg. (EU) 748/2012CS 23.2600 21L.B.43 Annex Ib to Reg (EU) 748/2012	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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9.3	<p>9.3 Ergonomics</p> <p>During design of the aeroplane, account shall be taken of ergonomic factors including:</p> <p>a) ease of use and prevention of inadvertent misuse;</p> <p>b) accessibility;</p> <p>c) flight crew working environment;</p> <p>d) cockpit standardization; and</p> <p>e) maintainability.</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2135CS 23.2300CS 23.2320CS 23.2600CS 23.2605CS 23.2610</p> <p>21L.B.43 Annex Ib to Reg (EU) 748/2012</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
9.4	<p>9.4 Operating environmental factors</p> <p>The design of the aeroplane shall take into consideration the flight crew operating environment including:</p> <p>a) effect of aeromedical factors such as level of oxygen, temperature, humidity, noise and vibration;</p> <p>b) effect of physical forces during normal flight;</p>	<p>21.B.80Reg. (EU) 748/2012CS 23.2135CS 23.2300CS 23.2320CS 23.2600CS 23.2605CS 23.2610</p> <p>21L.B.43</p>	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes						Significant Difference
				Level of implementation of SARPs						
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	<p>c) effect of prolonged operation at high altitude; and</p> <p>d) physical comfort.</p>	Annex Ib to Reg (EU) 748/2012								
1.1.1	<p>PART VI. ENGINES</p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 Except as noted below, the Standards of this part are applicable to engines of all types, used as primary propulsion units, as required in Parts IIIB, IVB and V. The Standards of this part are applicable to an engine type at the time of submission of an application to the appropriate national authority for a type approval.</p> <p><i>Note.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by</i></p>	Part 21 Annex I and Part 21 Light Annex Ib to Reg (EU) 748/2012; CS-E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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	<i>requirements established, adopted or accepted by Contracting States.</i> v									
1.1.2	1.1.2 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code for the engines designated in 1.1.1 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.	CS E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.1	1.2 Engine installation and interfaces 1.2.1 All necessary information for the safe and correct interfaces between the engine and the aircraft shall be made available.	CS E.20	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2.2	1.2.2 The installation instructions shall specify those assumptions concerning the conditions that may be imposed on the engine when it is eventually installed in an aircraft.	CS E.30	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.1	1.3 Declared ratings, conditions and limitations 1.3.1 The thrust or power ratings and the conditions of the atmosphere upon which they are based and all operating conditions and limitations which are intended to govern the operation of the engine shall be declared. v	CS E.40	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.3.2	1.3.2 Within the stated limits of 1.3.1, the engine shall produce the thrust or power demanded of it at all required flight conditions, taking into account environmental effects and conditions.	CS E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4.1	1.4 Continuing airworthiness – maintenance information 1.4.1 General Information for use in developing procedures for maintaining the engine in an airworthy condition shall be made available. The information shall include that described in 1.4.2, 1.4.3 and 1.4.4.	CS E.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4.2	1.4.2 Maintenance information Maintenance information shall include a description of the engine and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.	CS E.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.4.3	1.4.3 Maintenance programme information Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed.	CS E.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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1.4.4	1.4.4 Mandatory maintenance requirements resulting from the type design approval Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 1.4.3.	CS E.25	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1	CHAPTER 2. DESIGN AND CONSTRUCTION 2.1 Functioning The engine shall be designed and constructed so as to function reliably within its operating limitations under its anticipated operating conditions when installed in accordance with Parts IIIB, IVB and V of this Annex and, if applicable, fitted with a propeller approved for the installation.	CS E.100(c). CS E Subpart B and D	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.2	2.2 Failure analysis For turbine engines, a safety assessment of the engine shall be conducted to ensure that it functions safely throughout the full range of operating conditions. A	CS E 210; 510	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	summary shall be made of all foreseeable failures and combinations of failures that result in hazardous engine effects. If the primary failure of single elements (for example, disks) is likely to result in hazardous engine effects, reliance shall be placed on meeting prescribed integrity requirements.									
2.3	2.3 Materials and manufacturing methods The selection of materials and the manufacturing methods and processes shall account for the operational environment of the engine expected in service. The materials and manufacturing methods and processes used in the construction of the engine shall result in known and reproducible structural behaviour.	CS E.70	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4	2.4 Integrity The integrity of the engine shall be demonstrated throughout its operating envelope and be maintained for its operational life. The effects of cyclic loading, environmental and operational degradation and likely subsequent part failures shall not reduce the integrity of the engine below acceptable levels. All necessary instructions for ensuring continued airworthiness in this regard shall be promulgated.	CS E.210; 510; 520	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3	<p>CHAPTER 3. TESTS</p> <p>The engine type shall complete satisfactorily such tests as are necessary to verify the validity of the declared ratings, conditions and limitations and to ensure that it will operate satisfactorily and reliably. The tests shall include at least the following:</p> <p>a) <i>Power calibration.</i> Tests shall be conducted to establish the power or thrust characteristics of the engine when new and also after the tests in b) and c). There shall be no excessive decrease in power at the conclusion of all the tests specified.</p> <p>b) <i>Operation.</i> Tests shall be conducted to ensure that starting, idling, acceleration, vibration, over-speeding and other characteristics are satisfactory and to demonstrate adequate margins of freedom from detonation, surge, flutter or other detrimental conditions as may be appropriate to the particular type engine.</p> <p>c) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, thrust, speeds, temperatures and other operating conditions as are necessary to demonstrate reliability and durability of the engine. They shall also include operation under conditions in excess of the declared limits to the extent that such limitations might be exceeded in actual service.</p>	CS E Subpart C and Subpart E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Sigini- ficant Differ- ence			
				Level of implementation of SARPs						
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	<p>d) <i>Operating Environment.</i> Tests shall be conducted to ensure that the engine characteristics are satisfactory with regard to the operating environment.</p> <p><i>Note.— Operating environment may include encounter with birds, rain and hail, electromagnetic interference and lightning.</i></p>									
1.1.1	<p>PART VII. PROPELLERS</p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable to all propellers, as required in Parts IIIB and V. The Standards of this part are applicable to a propeller at the time of submission of an application to the appropriate national authority for a type approval.</p> <p><i>Note.— The following Standards do not include quantitative specifications comparable to those found in</i></p>	Part 21 Annex I and Part 21 Light Annex Ib to Reg (EU) 748/2012; CS-E	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	<i>national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i>									
1.1.2	1.1.2 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code for the propellers designated in 1.1.1 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.	CS P	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.2	1.2 Declared ratings, conditions and limitations The power ratings and all operating conditions and limitations which are intended to govern the operation of the propeller shall be declared.	CS P.50	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.1	1.3 Continuing airworthiness — maintenance information 1.3.1 General Information for use in developing procedures for maintaining the propeller in an airworthy condition shall be made available. The information shall include that described in 1.3.2, 1.3.3 and 1.3.4.	CS P.40	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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				Level of implementation of SARPs						
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1.3.2	1.3.2 Maintenance information Maintenance information shall include a description of the propeller and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on defect diagnosis.	CS P.40	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.3	1.3.3 Maintenance programme information Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed.	CS P.40	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.3.4	1.3.4 Mandatory maintenance requirements resulting from the type design approval Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 1.3.3.	CS P.40	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.1	CHAPTER 2. DESIGN AND CONSTRUCTION 2.1 Functioning	21.B.80Reg. (EU) 748/2012CS P Subpart BCS 23.2400	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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	The propeller assembly shall be designed and constructed so as to function reliably within its operating limitations under its anticipated operating conditions when installed in accordance with Parts IIIB and V of this Annex and shown to be not hazardous.	21L.B.43 Annex Ib Reg (EU) 748/2012								
2.2	2.2 Failure analysis A safety assessment of the propeller shall be conducted to ensure that it functions safely throughout the full range of operating conditions. A summary shall be made of those failures which could result in hazardous propeller effects. If the primary failure of single elements (for example, blades) is likely to result in hazardous propeller effects, reliance shall be placed on meeting prescribed integrity requirements.	CS P.150	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.3	2.3 Materials and manufacturing methods The selection of materials and the manufacturing methods and processes shall account for the operational environment of the propeller expected in service. The materials and manufacturing methods and processes used in the construction of the propeller shall result in known and reproducible structural behaviour.	CS P.170	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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2.4.1	<p>2.4 Pitch control and indication</p> <p>2.4.1 No loss of normal propeller pitch control shall cause a hazardous overspeeding under anticipated operating conditions.</p>	CS P.210; 410	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
2.4.2	<p>2.4.2 No single failure or malfunction in the propeller control system during normal or emergency operation shall result in unintended travel of the propeller blades to a position below the in-flight low-pitch position. Failure of structural elements need not be considered if the occurrence of such a failure is shown to be extremely remote.</p>	CS P .210	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
3.1	<p>CHAPTER 3. TESTS AND INSPECTIONS</p> <p>3.1 Blade retention test</p> <p>Propeller assemblies with detachable blades shall be subjected to a centrifugal load with sufficient margin to ensure that the hub and blade retention system will operate satisfactorily and reliably under the expected loads in service under all anticipated operating conditions.</p>	CS P.350; 370	<input checked="" type="checkbox"/>	<input type="checkbox"/>						

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3.2	<p>3.2 Operational and endurance tests</p> <p>The propeller shall satisfactorily complete such tests as are necessary to ensure that it will operate satisfactorily and reliably within the declared ratings, conditions and limitations. The tests shall include at least the following:</p> <p>a) <i>Function.</i> Tests shall be conducted to demonstrate proper and reliable functioning of the pitch control system.</p> <p>b) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, speeds and other operating conditions as are necessary to demonstrate reliability and durability of the propeller.</p> <p>c) <i>Operating environment.</i> Except for fixed pitch wood propellers, it shall be demonstrated by tests or analysis based on tests or experience on similar designs, that the propeller is capable of withstanding the likely impact of a bird or a lightning strike without causing a hazardous propeller effect.</p>	CS P Subpart C and D, P.400, P390, P360, P380	<input checked="" type="checkbox"/>	<input type="checkbox"/>						
1.1.1	PART VIII. REMOTELY PILOTED AEROPLANES <i>Applicable as of 26 November 2026.</i>		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>		

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	<p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all remotely piloted aeroplanes for which an application for the issue of a Type Certificate is submitted to the appropriate national authorities on or after 26 November 2026.</p> <p><i>Note 1.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p> <p><i>Note 2. — The provisions in this part support remotely piloted aeroplane operations SARPs in Annex 6.</i></p>									
1.1.2	1.1.2 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the remotely piloted aeroplanes designated in 1.1.1 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.		<input type="checkbox"/>							

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1.1.3	1.1.3 Unless otherwise stated, the Standards apply to the complete remotely piloted aeroplane including its powerplant, systems and equipment.		<input type="checkbox"/>							
1.2.1	1.2 Operating limitations 1.2.1 Limiting conditions shall be established for the remotely piloted aeroplane, its powerplant, systems and equipment (see 7.2). Compliance with the Standards of this part shall be established assuming that the remotely piloted aeroplane is operated within the limitations specified. The limitations shall include a margin of safety to render the likelihood of accidents arising therefrom extremely remote.		<input type="checkbox"/>							
1.2.2	1.2.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the remotely piloted aeroplane, e.g. mass, centre of gravity location, load distribution, speeds, ambient air temperature, altitude and C2 Link performance, shall be established within which compliance with all the pertinent Standards in this part is shown. <i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each separate operating condition, e.g. take-off, en route, landing.</i> <i>Note 2.— Maximum operating mass may be limited by the application of noise certification Standards</i>		<input type="checkbox"/>							

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	<i>(see Annex 16 — Environmental Protection, Volume I — Aircraft Noise, and Annex 6 — Operation of Aircraft.</i>									
1.3	1.3 Unsafe features and characteristics Under all anticipated operating conditions, the remotely piloted aeroplane shall not possess any feature or characteristic that renders it unsafe.		<input type="checkbox"/>							
1.4	1.4 Proof of compliance The means by which compliance with the appropriate airworthiness requirements is demonstrated shall ensure that in each case the accuracy achieved will be such as to provide reasonable assurance that the remotely piloted aeroplane, its components and equipment comply with the requirements and are reliable and function correctly under the anticipated operating conditions.		<input type="checkbox"/>							
2.1.1	CHAPTER 2. FLIGHT		<input type="checkbox"/>							

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	<p>2.1 General</p> <p>2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon a remotely piloted aeroplane or remotely piloted aeroplanes of the type for which a Type Certificate is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.</p>								
2.1.2	<p>2.1.2 Compliance with each Standard shall be established for all applicable combinations of remotely piloted aeroplane mass and centre of gravity position, within the range of loading conditions for which certification is sought.</p>		<input type="checkbox"/>						
2.1.3	<p>2.1.3 Where necessary, appropriate remotely piloted aeroplane configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the remotely piloted aeroplane's flying qualities.</p>		<input type="checkbox"/>						
2.2.1	<p>2.2 Performance</p> <p>2.2.1 Sufficient data on the performance of the remotely piloted aeroplane shall be determined and scheduled in the remotely piloted aeroplane flight manual to provide operators with the necessary information for</p>		<input type="checkbox"/>						

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	the purpose of determining the total mass of the remotely piloted aeroplane on the basis of the values, peculiar to the proposed flight, of the relevant operational parameters, in order that the flight may be made with reasonable assurance that a safe minimum performance for that flight will be achieved.									
2.2.2	<p>2.2.2 Achieving the performance scheduled for the remotely piloted aeroplane shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the remote flight crew.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							
2.2.3	2.2.3 The scheduled performance of the remotely piloted aeroplane shall be consistent with compliance with 1.2.1 and with the operation in logical combinations of those of the remotely piloted aeroplane's systems and equipment, the operation of which may affect performance.		<input type="checkbox"/>							
2.2.4	<p>2.2.4 Minimum performance</p> <p>Minimum performance shall be scheduled for remotely piloted aeroplanes with more than one engine:</p> <p>a) at the maximum masses scheduled (see 2.2.7) for take-off and for landing, as functions of the</p>		<input type="checkbox"/>							

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	<p>aerodrome elevation or pressure-altitude either in the standard atmosphere or in specified still air atmospheric conditions; and</p> <p>b) for seaplanes in specified conditions in smooth water, the remotely piloted aeroplane shall be capable of accomplishing the minimum performances specified in 2.2.5 a) and 2.2.6 a) respectively, not considering obstacles, or runway or water run length.</p> <p><i>Note.— This Standard permits the maximum take-off mass and maximum landing mass to be scheduled in the remotely piloted aeroplane flight manual against, for example:</i></p> <ul style="list-style-type: none"> — aerodrome elevation, or — pressure-altitude at aerodrome level, or — pressure-altitude and atmospheric temperature at aerodrome level, <p>so as to be readily usable when applying the national code on remotely piloted aeroplane performance operating limitations.</p>									
2.2.5	2.2.5 Take-off		<input type="checkbox"/>							
	a) For remotely piloted aeroplanes with more than one engine after the end of the period during which the take-off power or thrust may be used, the									

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	<p>remotely piloted aeroplane shall be capable of continuing to climb, with the critical engine inoperative and the remaining engine(s) operated within their maximum continuous power or thrust limitations, up to a height that it can maintain and at which it can continue safe flight and landing.</p> <p>b) The minimum performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data is scheduled (see 2.2.7), the departure from the scheduled values is not disproportionate.</p> <p><i>Note.— For remotely piloted aeroplanes that are assisted during take-off, related SARPs can be found in 11.3 of this part.</i></p>								
2.2.6	<p>2.2.6 Landing</p> <p>a) For remotely piloted aeroplanes with one engine, or a single propeller, or remotely piloted aeroplanes with more than one engine that cannot maintain a positive climb gradient following an engine or propeller failure, the design shall, in the case of engine or propeller failure, enable the remotely piloted aeroplane to be operated to a safe forced landing in favourable conditions, or to initiate the emergency recovery capability, as specified in Chapter 11 of this part.</p> <p>b) For remotely piloted aeroplanes with more than one engine starting from the approach</p>		<input type="checkbox"/>						

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	<p>configuration and with the critical engine inoperative, the remotely piloted aeroplane shall be capable, in the event of a missed approach, of continuing the flight to a point from which another approach can be made.</p> <p>c) Starting from the landing configuration, the remotely piloted aeroplane shall be capable, in the event of a balked landing, of making a climb-out, with all engines operating.</p> <p><i>Note.— For remotely piloted aeroplanes that are assisted during landing recovery, related SARPs can be found in 11.4 of this part.</i></p>								
2.2.7.1	<p>2.2.7 Scheduling of performance</p> <p>2.2.7.1 Performance data shall be determined and scheduled in the remotely piloted aeroplane flight manual in order to provide a safe relationship between the performance of the remotely piloted aeroplane and the aerodromes and routes on which it is capable of being operated. Performance data shall be determined and scheduled for the following stages for the ranges of mass, altitude or pressure-altitude, wind velocity, gradient of the take-off and landing surface for landplanes; water surface conditions, density of water and strength of current for seaplanes; and for any other operational variables for which the remotely piloted aeroplane is to be certificated.</p>		<input type="checkbox"/>						

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	<p>a) <i>Take-off.</i> The take-off performance data shall include the distance required to take-off and climb to a selected height above the take-off surface. It shall be determined for each mass, altitude and temperature within the operational limits established for take-off with:</p> <ul style="list-style-type: none"> — take-off power on each engine; — wing flaps in the take-off position(s); and — landing gear extended. <p>b) <i>En route.</i> For remotely piloted aeroplanes with more than one engine, the en-route climb performance shall be the climb (or descent) performance with the remotely piloted aeroplane in the en-route configuration with the critical engine inoperative. The operating engine(s) shall not exceed maximum continuous power or thrust.</p> <p>c) <i>Landing.</i> The landing distance shall be the horizontal distance traversed by the remotely piloted aeroplane from a point on the approach flight path at a selected height above the landing surface to the point on the landing surface at which the remotely piloted aeroplane comes to a complete stop, or, for a seaplane, comes to a satisfactorily low speed. The selected height above the landing surface and the approach speed shall be appropriately related to operating practices. This distance may be supplemented by such distance margin as may be necessary; if so, the selected height above the landing surface, the approach speed and the distance margin shall</p>								
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	<p>be appropriately interrelated and shall make provision for both normal operating practices and reasonable variations therefrom.</p> <p><i>Note.— If the landing distance includes the distance margin specified in this Standard, it is not necessary to allow for the expected variations in the approach and landing techniques in applying 5.2.11 of Annex 6 — Operation of Aircraft, Part I — International Commercial Air Transport — Aeroplanes.</i></p>									
2.2.7.2	<p>2.2.7.2 Recommendation.— For remotely piloted aeroplanes that are assisted during take-off or landing recovery, the effects of launch and recovery methods on the scheduling of performance should be considered.</p> <p><i>Note.— Related SARPs can be found in 11.3 and 11.4 of this part.</i></p>		<input type="checkbox"/>							
2.3.1	<p>2.3 Flying qualities</p> <p>2.3.1 The remotely piloted aeroplane shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the remotely piloted aeroplane is approved.</p>		<input type="checkbox"/>							
2.3.2.1	2.3.2 Controllability		<input type="checkbox"/>							

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	<p>2.3.2.1 The remotely piloted aeroplane shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power or thrust, changes of remotely piloted aeroplane configurations) without requiring exceptional skill or alertness on the part of the remote pilot even in the event of failure of any engine. A means or technique for safely controlling the remotely piloted aeroplane shall be established for all stages of flight and remotely piloted aeroplane configurations for which performance is scheduled.</p> <p><i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i></p>									
2.3.2.2	<p>2.3.2.2 <i>Controllability on the ground (or water).</i> The remotely piloted aeroplane shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.</p>		<input type="checkbox"/>							
2.3.2.3	<p>2.3.2.3 <i>Controllability during take-off.</i> The remotely piloted aeroplane shall be controllable in the event of sudden failure of the critical engine at any point in the take-off.</p>		<input type="checkbox"/>							

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2.3.2.4	2.3.2.4 <i>Take-off safety speed.</i> The take-off safety speeds assumed when the performance of the remotely piloted aeroplane (after leaving the ground or water) during the take-off is determined shall provide an adequate margin above the stall and above the minimum speed at which the remotely piloted aeroplane remains controllable after sudden failure of the critical engine.		<input type="checkbox"/>							
2.3.3	2.3.3 Trim The remotely piloted aeroplane shall have such trim characteristics as to ensure that the demands made on the remote pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. This shall apply both in normal operation and in the conditions associated with the failure of one or more engines for which performance characteristics are established.		<input type="checkbox"/>							
2.4.1	2.4 Stability and control 2.4.1 Stability The remotely piloted aeroplane shall have such stability in relation to its other flight characteristics, performance, structural strength, and most probable operating		<input type="checkbox"/>							

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	conditions (e.g. remotely piloted aeroplane configurations and speed ranges) as to ensure that demands made on the remote pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the remotely piloted aeroplane shall not, however, be such that excessive demands are made on the remote pilot's skill or that the safety of the remotely piloted aeroplane is prejudiced by lack of manoeuvrability in emergency conditions. The stability may be achieved by natural or artificial means, or a combination of both. In those cases where artificial stability is necessary to show compliance with the Standards of this part, it shall be shown that any failure or condition that would result in the need for exceptional remote pilot skill for recovery of remotely piloted aeroplane stability is extremely improbable.								
2.4.2.1	2.4.2 Stalling 2.4.2.1 <i>Stall warning.</i> When the remotely piloted aeroplane approaches a stall both in straight and turning flight, a clear and distinctive stall warning shall be apparent to the remote pilot with the remotely piloted aeroplane in all permissible configurations and powers or thrusts, except those which are not considered to be essential for safe flying. The stall warning and other characteristics of the remotely piloted aeroplane shall be such as to enable the remote pilot to arrest the development of the stall after the warning begins and, without altering the engine power or thrust, to maintain full control of the remotely piloted aeroplane.		<input type="checkbox"/>						

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2.4.2.2	2.4.2.2 <i>Behaviour following a stall.</i> In any configuration and at any level of power or thrust in which it is considered that the ability to recover from a stall is essential, the behaviour of the remotely piloted aeroplane following a stall shall not be so extreme as to make difficult a prompt recovery without exceeding the airspeed or strength limitations of the remotely piloted aeroplane.		<input type="checkbox"/>							
2.4.2.3	2.4.2.3 <i>Stalling speeds.</i> The stalling speeds or minimum steady flight speeds in configurations appropriate for each stage of flight (e.g. take-off, en route, landing) shall be established. One of the values of the power or thrust used in establishing the stalling speeds shall be not more than that necessary to give zero thrust at a speed just above the stall.		<input type="checkbox"/>							
2.4.3.1	2.4.3 Flutter and vibration 2.4.3.1 It shall be demonstrated by suitable tests, analyses or any acceptable combination of tests and analyses that all parts of the remotely piloted aeroplane are free from flutter and excessive vibration in all remotely piloted aeroplane configurations under all speed conditions within the operating limitations of the remotely piloted aeroplane (see 1.2.2). There shall be no vibration or buffeting severe enough to cause structural damage.		<input type="checkbox"/>							
2.4.3.2	2.4.3.2 There shall be no vibration or buffeting severe enough to interfere with normal functioning of on-		<input type="checkbox"/>							

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	board equipment used for control of the remotely piloted aeroplane.. <i>Note.— Buffeting as a stall warning is considered desirable and discouragement of this type of buffeting is not intended.</i>									
2.4.4	2.4.4 Spinning It shall be demonstrated that the remotely piloted aeroplane during normal operation does not exhibit any tendency to inadvertently enter into a spin. If the design is such that spinning is allowed or for remotely piloted aeroplanes with one engine inadvertently possible, it shall be demonstrated that without the use of exceptional piloting skill the remotely piloted aeroplane can be recovered from a spin within appropriate recovery limits.		<input type="checkbox"/>							
3.1	CHAPTER 3. STRUCTURE 3.1 General The remotely piloted aeroplane structure shall be designed, manufactured and provided with instructions for its maintenance and repair with the objective of		<input type="checkbox"/>							

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	avoiding catastrophic failure throughout its operational life.									
3.2	3.2 Mass and mass distribution Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.		<input type="checkbox"/>							
3.3	3.3 Limit loads Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.6 shall be considered as limit loads.		<input type="checkbox"/>							
3.4	3.4 Strength and deformation In the various loading conditions prescribed in 3.6, no part of the remotely piloted aeroplane structure shall sustain detrimental deformation at any load up to and including the limit load, and the remotely piloted aeroplane structure shall be capable of supporting the ultimate load.		<input type="checkbox"/>							

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3.5.1	<p>3.5 Airspeeds</p> <p>3.5.1 Design airspeeds</p> <p>Design airspeeds shall be established for which the remotely piloted aeroplane structure is designed to withstand the corresponding manoeuvring and gust loads. To avoid inadvertent exceedances due to upsets or atmospheric variations, the design airspeeds shall provide sufficient margin for the establishment of practical operational limiting airspeeds. In addition, the design airspeeds shall be sufficiently greater than the stalling speed of the remotely piloted aeroplane to safeguard against loss of control in turbulent air. Consideration shall be given to a design manoeuvring speed, a design cruising speed, a design dive speed, and any other design airspeeds necessary for configurations with high lift or other special devices.</p>		<input type="checkbox"/>							
3.5.2	<p>3.5.2 Limiting airspeeds</p> <p>Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the remotely piloted aeroplane flight manual as part of the operating limitations (see 7.2).</p>		<input type="checkbox"/>							
3.6.1	3.6 Strength		<input type="checkbox"/>							

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	<p>3.6.1 All structural elements shall be designed to withstand the maximum loads expected in service under all anticipated operating conditions without failure, permanent distortion or loss of functionality. In determining these loads, account shall be taken of:</p> <p>a) the expected operational life of the remotely piloted aeroplane;</p> <p>b) the vertical and horizontal gust environment, taking into consideration the expected variations in mission profile and loading configurations;</p> <p>c) the manoeuvre spectrum, taking into account variations in mission profiles, and loading configurations;</p> <p>d) asymmetrical as well as symmetrical loading;</p> <p>e) the ground and water loads, including taxi, landing and take-off loads, launch and recovery loads as required in Chapter 11, and ground/water handling loads;</p> <p>f) the speed range of the remotely piloted aeroplane, taking into account the remotely piloted aeroplane characteristics and operation limitations;</p> <p>g) vibration and buffeting loads;</p>								
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	<p>h) corrosion or other degradation, given the maintenance specified, and various operating environments; and</p> <p>i) any other loads, such as flight control loads, pressurization loads, engine loads, or dynamic loads due to changes to the steady state configuration.</p>									
3.6.2	3.6.2 The air, inertia and other loads resulting from the specific loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively.		<input type="checkbox"/>							
3.7	<p>3.7 Structural durability</p> <p>The structure of the remotely piloted aeroplane shall conform to damage tolerance, safe-life or failsafe principles and shall be such as to avoid catastrophic failure during the operational life, taking into account, where appropriate:</p> <p>a) the expected environment; and</p> <p>b) the expected repeated loads applied in service.</p>		<input type="checkbox"/>							
3.8	3.8 Special factors		<input type="checkbox"/>							

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	For remotely piloted aeroplanes, design features (e.g. castings, bearings or fittings), the strength of which is subject to variability in manufacturing processes, deterioration in service, or any other cause, shall be accounted for by a suitable factor.									
4.1.1	<p>CHAPTER 4. DESIGN AND CONSTRUCTION</p> <p>4.1 General</p> <p>4.1.1 Details of design and construction shall be such as to give reasonable assurance that all remotely piloted aeroplane parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							
4.1.2	4.1.2 Substantiation of moving parts		<input type="checkbox"/>							

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	The functioning of all moving parts essential to the safe operation of the remotely piloted aeroplane shall be demonstrated in order to ensure that they will function correctly under all operating conditions for such parts.									
4.1.3	4.1.3 Materials All materials used in parts of the remotely piloted aeroplane essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design.		<input type="checkbox"/>							
4.1.4	4.1.4 Manufacturing methods The methods of manufacturing and assembly shall be such as to produce a consistently sound structure which shall be reliable with respect to maintenance of strength in service.		<input type="checkbox"/>							
4.1.5	4.1.5 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed, taking into account the maintenance the remotely piloted aeroplane will receive.		<input type="checkbox"/>							

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4.1.6	4.1.6 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the remotely piloted aeroplane that require such attention, either periodically or after unusually severe operations.		<input type="checkbox"/>							
4.2	4.2 Systems design features Special consideration shall be given to design features that affect the ability of the remote flight crew member to maintain controlled flight. This shall include at least the following: a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such as to minimize the possibility of jamming, inadvertent operation including prevention of mis-assembly, and unintentional engagement of control surface locking devices. 1) each control and control system shall operate with the ease, smoothness and precision appropriate to its functions; 2) each element of each flight control system shall be designed, or distinctively and permanently marked, to minimize the probability of any		<input type="checkbox"/>							

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	<p>incorrect assembly that could result in the malfunction of the system; and</p> <p>3) the influence of any systems design features and their failure conditions that affect structural performance must be taken into account when showing compliance with the requirements in Chapters 3 and 4;</p> <p>b) <i>System survivability.</i> Remotely piloted aeroplane systems shall be designed and arranged to maximize the potential for continued safe flight and landing after events resulting in damage to the remotely piloted aeroplane structure or systems.</p> <p>c) <i>Pilot vision.</i> The arrangement of design features for remote pilot vision, if implemented on the remotely piloted aeroplane, shall permit, under normal and precipitation conditions of moderate rain, sufficient vision for the normal conduct of flight and for the execution of approaches and landings as assumed in the design to support safe operation of the remotely piloted aeroplane.</p> <p>d) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the remote flight crew to deal with, emergencies resulting from foreseeable failures of equipment, systems, the C2 Link, and the remote pilot station, the failure of which would endanger the remotely piloted aeroplane. Reasonable provisions shall be made for continuation of essential services following engine or systems' failures to the extent that such failures are catered for in the</p>								
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	<p>performance and operating limitations specified in the Standards in this Annex and in Annex 6.</p> <p>e) <i>Fire precautions.</i> The design of the remotely piloted aeroplane and the materials used in its manufacture shall be such so as to minimize the risk of in-flight and ground fires, to minimize the production of smoke and toxic gases in the event of a fire.</p> <p>f) <i>Cargo compartment protection.</i></p> <p>1) sources of heat which are capable of igniting the cargo shall be shielded or insulated to prevent such ignition; and</p> <p>2) each cargo compartment shall be constructed of materials which are at least flame resistant.</p>									
4.3	<p>4.3 Aeroelasticity</p> <p>The remotely piloted aeroplane shall be free from flutter, structural divergence, and loss of control due to structural deformation and aeroelastic effects, at all speeds within and sufficiently beyond the design envelope to comply with 1.2.1. Account shall be taken of the characteristics of the remotely piloted aeroplane, the lack of the proprioceptive sensory information (e.g. vibration and acceleration), and variations in remote pilot skill and workload.</p>		<input type="checkbox"/>							
4.4.1	4.4 Electrical bonding and protection against		<input type="checkbox"/>							

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	<p align="center">lightning and static electricity</p> <p>4.4.1 Electrical bonding and protection against lightning and static electricity shall be such as to:</p> <p>a) protect the remotely piloted aeroplane, its systems and persons who come in contact with the remotely piloted aeroplane on the ground or water from the dangerous effects of lightning discharge and electrical shock; and</p> <p>b) prevent dangerous accumulation of electrostatic charge.</p>									
4.4.2	4.4.2 The remotely piloted aeroplane shall also be protected against catastrophic effects of lightning. Due account shall be taken of the material used in the construction of the remotely piloted aeroplane.		<input type="checkbox"/>							
4.5	<p>4.5 Ground handling</p> <p>Design provisions and procedures for safe ground handling (e.g. towing, jacking) shall be defined. The protection that any limitations and instructions for such operations might provide may be taken into account.</p>		<input type="checkbox"/>							
5.1	CHAPTER 5. POWERPLANT		<input type="checkbox"/>							

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	<p>5.1 Engines</p> <p>The Standards of Part VI of this Annex shall apply to each engine that is used on the remotely piloted aeroplane as a primary propulsion unit.</p>								
5.2	<p>5.2 Propellers</p> <p>The Standards of Part VII of this Annex shall apply to each propeller that is used on the remotely piloted aeroplane.</p>		<input type="checkbox"/>						
5.3.1	<p>5.3 Powerplant installation</p> <p>5.3.1 Compliance with engine and propeller limitations</p> <p>The powerplant installation shall be so designed that the engines and propellers (if applicable) are capable of functioning reliably in the anticipated operating conditions. In conditions established in the remotely piloted aeroplane flight manual, the remotely piloted aeroplane shall be capable of being operated without exceeding the limitations established for the engines and propellers in accordance with this chapter and Parts VI and VII.</p>		<input type="checkbox"/>						

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5.3.2	5.3.2 Control of engine rotation In those installations where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided to stop the rotation of the failed engine in flight or to reduce it to a safe level.		<input type="checkbox"/>							
5.3.3	5.3.3 Turbine engine installation For a turbine engine installation: a) the design shall minimize the hazards to the remotely piloted aeroplane in the event of failure of engine rotating parts, or an engine fire which burns through the engine case; and b) the powerplant installation shall be designed to give reasonable assurance that those engine operating limitations that adversely affect the structural integrity of rotating parts shall not be exceeded in service.		<input type="checkbox"/>							
5.3.4	5.3.4 Engine restarting Means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude, unless the remotely piloted aeroplane can be safely controlled without engine restart by means of an approved		<input type="checkbox"/>							

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	emergency recovery capability, as specified in Chapter 11 of this part.									
5.3.5.1	<p>5.3.5 Arrangement and functioning</p> <p>5.3.5.1 <i>Independence of engines and associated systems.</i> For remotely piloted aeroplanes, the engines together with their associated systems shall be arranged and isolated from each other to allow operation, in at least one configuration, so that the failure or malfunction of any engine, or the failure or malfunction (including destruction by fire in the engine compartment) of any system that can affect an engine (other than a fuel tank if only one fuel tank is installed), will not:</p> <p>a) prevent the continued safe operation of the remaining engine(s); or</p> <p>b) require immediate action for continued safe operation.</p>		<input type="checkbox"/>							
5.3.5.2	<p>5.3.5.2 <i>Propeller vibration.</i> The propeller vibration stresses shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the remotely piloted aeroplane.</p>		<input type="checkbox"/>							
5.3.5.3	<p>5.3.5.3 <i>Cooling.</i> The cooling system shall be capable of maintaining the temperature of powerplant components and fluids within the established limits (see 5.3.1) at ambient air temperatures up to the maximum air temperature appropriate to the intended operation of the remotely piloted aeroplane.</p>		<input type="checkbox"/>							

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5.3.5.4	5.3.5.4 <i>Associated systems.</i> The fuel, oil, air induction, and other systems associated with the powerplant shall be capable of supplying each engine in accordance with its established requirements, under all conditions affecting the functioning of the systems (e.g. engine power or thrust, remotely piloted aeroplane attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.		<input type="checkbox"/>							
5.3.5.5	5.3.5.5 <i>Fire protection.</i> For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.2 f). a) <i>Isolation.</i> Such regions shall be isolated by fireproof material from other regions of the remotely piloted aeroplane where the presence of fire would jeopardize continued flight, taking into account the probable points of origin and paths of propagation of fire. b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be fire resistant. Drainage of each region shall be provided to minimize hazards resulting from the failure of any component containing flammable fluids. Means shall be provided for the remote flight crew to shut off the flow of flammable fluids into such regions if a fire occurs. Where sources of flammable fluid exist in such regions, the whole of the related system within the region, including		<input type="checkbox"/>							

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	<p>supporting structure, shall be fire proof or shielded from the effects of the fire.</p> <p>c) <i>Fire detection.</i> A sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions.</p>								
6.1.1	<p>CHAPTER 6. SYSTEMS AND EQUIPMENT</p> <p>6.1 General</p> <p>6.1.1 The remotely piloted aeroplane shall be provided with approved equipment and systems, including guidance and flight management systems necessary for the safe operation of the remotely piloted aeroplane in the anticipated operating conditions. These shall include the equipment necessary to enable the remote flight crew to operate the remotely piloted aeroplane within its operating limitations. Equipment design shall consider human factors principles.</p> <p><i>Note 1.— Equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Parts I and II, for particular circumstances or on particular kinds of</i></p>		<input type="checkbox"/>						

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	<p><i>routes. Systems are addressed in Part X — Remote Pilot Station of this Annex.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>									
6.1.2	<p>6.1.2 The design of the equipment and systems required by 6.1.1 and their installation shall be such that:</p> <p>a) an inverse relationship exists between the probability of a failure condition and the severity of its effect, as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p> <p>c) electromagnetic interference between them is minimized.</p> <p><i>Note.— The system safety assessment process includes integration of the RPS and the specification of the C2 Link. See also 10.3.3 of this part.</i></p>		<input type="checkbox"/>							
6.1.3	<p>6.1.3 Means shall be provided to warn the remote crew of unsafe system operating conditions on both the remote pilot station and the remotely piloted aeroplane and to enable corrective action to be taken automatically or by the remote crew.</p>		<input type="checkbox"/>							

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6.1.4	6.1.4 Electrical power supply The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations and shall also be such that no single failure or malfunction could impair the ability of the system to supply essential loads for safe operation.		<input type="checkbox"/>							
6.1.5	6.1.5 Development assurance of complex electronic hardware and system software Complex electronic hardware and system software shall be developed, verified and validated such as to ensure that the systems in which they are used perform their intended functions at a level of safety that complies with the requirements of this part, notably those of 6.1.2 a) and 6.1.2 b). <i>Note.— Some States accept the use of national or international industry standards for the development assurance (development, verification and validation) of complex electronic hardware and systems software.</i>		<input type="checkbox"/>							
6.2	6.2 Installation Instrument and equipment installations shall comply with the Standards of Chapter 4.		<input type="checkbox"/>							
6.3.1	6.3 Navigation lights and anti-collision lights		<input type="checkbox"/>							

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	<p>6.3.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by remotely piloted aeroplanes in flight or operating on the movement area of an aerodrome shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft, or remote pilot of another remotely piloted aircraft, or personnel on the ground with as much time as possible for interpretation and the subsequent manoeuvre necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry night, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i></p>								
6.3.2	<p>6.3.2 Lights shall be installed in remotely piloted aeroplanes so as to minimize the possibility that they will adversely affect the satisfactory performance of any required sensors.</p> <p><i>Note.— In order to avoid the effects mentioned in 6.3.2, it will be necessary in some cases to provide means whereby the remote pilot can adjust the intensity of the flashing lights.</i></p>		<input type="checkbox"/>						

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6.4	6.4 Electromagnetic interference protection Remotely piloted aeroplane electronic systems, particularly flight-critical and flight-essential systems, shall be protected as appropriate against electromagnetic interference from internal and external sources.		<input type="checkbox"/>							
6.5	6.5 Ice protection If certification for flight in icing conditions is requested, the remotely piloted aeroplane shall be shown to be able to operate safely in icing conditions likely to be encountered in all anticipated operating environments.		<input type="checkbox"/>							
7.1	CHAPTER 7. OPERATING LIMITATIONS AND INFORMATION 7.1 General The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the remotely piloted aeroplane, shall be made available by means of a remotely piloted aeroplane flight manual,		<input type="checkbox"/>							

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	markings and placards, and such other means as may effectively accomplish the purpose.									
7.2.1	<p>7.2 Operating limitations</p> <p>7.2.1 Limitations which might be exceeded in flight and which are defined quantitatively shall be expressed in suitable units. These limitations shall be corrected if necessary for errors in measurements so that the remote flight crew can, by reference to the instruments available to them, readily determine when the limitations are reached.</p>		<input type="checkbox"/>							
7.2.2	<p>7.2.2 Loading limitations</p> <p>The loading limitations shall include all limiting masses, centres of gravity positions, mass distributions and floor loadings (see 1.2.2).</p>		<input type="checkbox"/>							
7.2.3	<p>7.2.3 Airspeed limitations</p> <p>The airspeed limitations shall include all speeds (see 3.5.2) that are limiting from the standpoint of structural integrity or flying qualities of the remotely piloted aeroplane, or from other considerations. These speeds shall be identified with respect to the appropriate remotely piloted aeroplane configurations and other pertinent factors.</p>		<input type="checkbox"/>							

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7.2.4	7.2.4 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the remotely piloted aeroplane (see 5.3.1 and 5.3.5.3).		<input type="checkbox"/>							
7.2.5	7.2.5 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the remotely piloted aeroplane.		<input type="checkbox"/>							
7.2.6	7.2.6 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the remotely piloted aeroplane (see 1.2.1).		<input type="checkbox"/>							
7.2.7	7.2.7 Remote flight crew limitations The remote flight crew limitations shall include the minimum number of remote flight crew personnel necessary to operate the remotely piloted aeroplane.		<input type="checkbox"/>							

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	<i>Note.— The circumstances in which the remote flight crew shall include members in addition to the minimum remote flight crew are defined in Annex 6 — Operation of Aircraft.</i>									
7.3.1	7.3 Operating information and procedures 7.3.1 Types of eligible operations The particular types of operations for which the remotely piloted aeroplane has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.		<input type="checkbox"/>							
7.3.2	7.3.2 Loading information The loading information shall include the empty mass of the remotely piloted aeroplane, together with a definition of the condition of the remotely piloted aeroplane at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related. <i>Note.— Usually the empty mass excludes the mass of the payload, the usable fuel supply and the drainable oil; it includes the mass of all fixed ballast, unusable fuel supply, undrainable oil, total quantity of engine coolant and total quantity of hydraulic fluid.</i>		<input type="checkbox"/>							

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7.3.3	7.3.3 Operating procedures A description shall be given of normal and emergency operating procedures which are peculiar to the particular remotely piloted aeroplane and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.		<input type="checkbox"/>							
7.3.4	7.3.4 Handling information Sufficient information shall be given on any significant or unusual features of the remotely piloted aeroplane characteristics. Those stalling speeds or minimum steady flight speeds required to be established by 2.4.2.3 shall be scheduled.		<input type="checkbox"/>							
7.4	7.4 Performance information The performance of the remotely piloted aeroplane shall be scheduled in accordance with 2.2. There shall be included information regarding the various remotely piloted aeroplane configurations and powers or thrusts involved and the relevant speeds, together with information that would assist the remote flight crew in attaining the performance as scheduled.		<input type="checkbox"/>							

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7.5	<p>7.5 Remotely piloted aeroplane flight manual</p> <p>A remotely piloted aeroplane flight manual shall be made available. It shall identify clearly the specific remotely piloted aeroplane or series of remotely piloted aeroplanes to which it is related. The remotely piloted aeroplane flight manual shall include at least the limitations, information and procedures specified in 7.2, 7.3, 7.4 of this part and Part X of this Annex.</p>		<input type="checkbox"/>							
7.6	<p>7.6 Markings and placards</p> <p>Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in taxi, launch, recovery, and ground servicing (towing, refuelling, etc.) that could pass unnoticed and that could jeopardize the safety of the remotely piloted aeroplane in subsequent flights and safety of the ground crew.</p>		<input type="checkbox"/>							
7.7.1	<p>7.7 Continuing airworthiness — maintenance information</p> <p>7.7.1 General</p> <p>Information for use in developing procedures for maintaining the remotely piloted aeroplane in an airworthy condition shall be made available. The</p>		<input type="checkbox"/>							

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	information shall include that described in 7.7.2, 7.7.3 and 7.7.4.									
7.7.2	<p>7.7.2 Maintenance information</p> <p>Maintenance information shall include a description of the remotely piloted aeroplane and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on transportation, storage and assembly, and defect diagnosis.</p>		<input type="checkbox"/>							
7.7.3	<p>7.7.3 Maintenance programme information</p> <p>Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed.</p> <p><i>Note.— The development of initial maintenance programme information at the time of remotely piloted aeroplane type certification can take advantage of the Maintenance Review Board (MRB) process or the process of developing instructions for continuing airworthiness.</i></p>		<input type="checkbox"/>							
7.7.4	<p>7.7.4 Mandatory maintenance requirements resulting from the type design approval</p> <p>Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of</p>		<input type="checkbox"/>							

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	<p>the type design shall be identified as such and included in the maintenance information of 7.7.3.</p> <p><i>Note.— Mandatory requirements identified as part of the type design approval are often referred to as Certification Maintenance Requirements (CMR) and/or airworthiness limitations.</i></p>								
7.8	<p>7.8 C2 Link information</p> <p>Sufficient information shall be given on any relevant C2 Link relating to configuration, operation, performance, emergency procedures, and operating limitations.</p>		<input type="checkbox"/>						
9.1	<p>CHAPTER 9. OPERATING ENVIRONMENT AND HUMAN FACTORS</p> <p>9.1 General</p> <p>The remotely piloted aeroplane shall be designed to allow safe operation within the performance limitations of those who operate, maintain and service it.</p> <p><i>Note.— The human/machine interface is often the weak link in an operating environment and so it is necessary to ensure that the remotely piloted aeroplane is</i></p>		<input type="checkbox"/>						

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	<i>capable of being controlled at all phases of the flight (including any degradation due to failures).</i>									
9.2.1	<p>9.2 Remote flight crew</p> <p>9.2.1 The remotely piloted aeroplane shall be designed in such a way as to allow safe and efficient control by the remote flight crew. The design shall allow for variations in remote flight crew skill and physiology commensurate with remote flight crew licensing limits. Account shall be taken of the different expected operating conditions of the remotely piloted aeroplane in its environment, including operations degraded by failures.</p>		<input type="checkbox"/>							
9.2.2	<p>9.2.2 The workload imposed on the remote flight crew by the design of the remotely piloted aeroplane shall be reasonable at all stages of flight. Particular consideration shall be given to critical stages of flight and critical events which may reasonably be expected to occur during the service life of the remotely piloted aeroplane, such as a contained engine failure or windshear encounter.</p> <p><i>Note.— Workload can be affected by both cognitive and physiological factors.</i></p>		<input type="checkbox"/>							
9.3	9.3 Ergonomics		<input type="checkbox"/>							

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	<p>During design of the remotely piloted aeroplane, account shall be taken of ergonomic factors, where applicable, including:</p> <ul style="list-style-type: none"> a) ease of use and prevention of inadvertent misuse; b) accessibility; c) maintainability; and d) transportation storage and assembly/disassembly. 								
10.1.1	<p>CHAPTER 10. REMOTE PILOT STATION INTEGRATION</p> <p>10.1 General</p> <p>10.1.1 The Standards of Part X of this Annex shall apply to each remote pilot station that is used to pilot the remotely piloted aeroplane.</p>		<input type="checkbox"/>						
10.1.2	<p>10.1.2 The remote pilot station shall be compatible with the type of remotely piloted aeroplane and appropriate to the intended operation.</p>		<input type="checkbox"/>						

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10.2.1	10.2 Integration 10.2.1 <i>Compliance with remote pilot station limitations.</i> The remotely piloted aeroplane shall be so designed that the remote pilot station is capable of performing satisfactorily and reliably its intended functions in the anticipated operating conditions when connected to the remotely piloted aeroplane. In conditions established in the flight manual, the remotely piloted aeroplane shall be capable of being operated within the limitations established for the remote pilot station in accordance with this chapter and with Part X.		<input type="checkbox"/>							
10.2.2	10.2.2 <i>Integration tests.</i> The remotely piloted aeroplane shall complete satisfactorily tests with all approved types of remote pilot stations, as are necessary to verify the validity of the declared conditions and limitations and to ensure that remote pilot stations will operate satisfactorily and reliably using any specified C2 Link and supporting C2 Link communication service providers, as specified under the anticipated operating conditions.		<input type="checkbox"/>							
10.3.1	10.3 Controls and information 10.3.1 The remote pilot station shall be integrated in such a way as to allow timely control as required for safe and efficient control of the remotely		<input type="checkbox"/>							

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	<p>piloted aeroplane by the remote flight crew. This shall include at least the following:</p> <p>a) processing the data provided by the remotely piloted aeroplane regarding:</p> <ul style="list-style-type: none"> — attitude, altitude, position, heading, speed, vertical speed, turning information; — powerplant and propeller speed; — detect and avoid; — weather conditions; — C2 Link state and performance according to the SARPs defined in the applicable sections of Annex 10 for remotely piloted aircraft systems; and — status of automated systems, including the current lost C2 Link state; <p>b) controlling the remotely piloted aeroplane in the anticipated operating condition;</p> <p>c) controlling the powerplant according to Chapter 5 of this part;</p> <p>d) information on predicted QoSD in the geographical area of the flight based on the QoSr and C2 Link specification; and</p>								
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	e) status of automated systems, including flight controls exceedance or malfunctions.									
10.3.2	<p>10.3.2 All required information shall be provided through the remote pilot station for the remote flight crew to safely and efficiently operate the remotely piloted aeroplane (e.g. set or monitor flight parameters for the flight, navigation, and powerplant) using any specified C2 Link and supporting C2 Link communication service providers in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the remote flight crew to operate the remotely piloted aeroplane within its anticipated operating limitations. Instruments and equipment design shall consider human factors principles.</p> <p><i>Note 1.— Instruments and equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							
10.3.3	<p>10.3.3 The design of the instruments, equipment and systems required by 10.3.2 and their installation shall be such that:</p> <p>a) an inverse relationship exists between the probability of a failure condition and the severity of</p>		<input type="checkbox"/>							

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	its effect, as determined by a system safety assessment process; b) they perform their intended function under all anticipated operating conditions; and c) electromagnetic interference between them is minimized.									
10.3.4	10.3.4 Means shall be provided to warn the remote flight crew of unsafe system operating conditions and to enable them to take corrective action.		<input type="checkbox"/>							
10.3.5	10.3.5 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the remote flight crew during flight.		<input type="checkbox"/>							
10.4.1	10.4 C2 Link 10.4.1 The remotely piloted aeroplane and remote pilot station system architecture shall be compatible with any specified C2 Link and supporting C2 Link communication service providers as specified, to enable the remotely piloted aeroplane to be operated safely under the anticipated operating conditions.		<input type="checkbox"/>							

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10.4.2	10.4.2 Means shall be provided to monitor the C2 Link performance and the C2 Link state according to metrics defined in the applicable parts of Annex 10, reacting according to the transaction completion criteria defined in Annex 6.		<input type="checkbox"/>							
10.5.1	10.5 Flight manual 10.5.1 The remotely piloted aeroplane flight manual shall address all combinations of remote pilot station models listed in the approved type design of the remotely piloted aeroplane. There may be substantial variations between different remote pilot stations used with the same remotely piloted aeroplane.		<input type="checkbox"/>							
10.5.2	10.5.2 In developing the remotely piloted aeroplane flight manual, specific consideration shall be given to human performance aspects, including transfer of control within and between remote pilot stations if envisaged by operational requirements, remote pilot handovers, switchovers of C2 Links or networks constituting the C2 Link, appropriate contingency planning procedures, remote crew communications, e.g. remote pilot to remote pilot, remote pilot to remotely piloted aeroplane observer or other support personnel, and remote pilot to ATC.		<input type="checkbox"/>							
10.5.3	10.5.3 The remotely piloted aeroplane flight manual shall contain all necessary information for operation of the RPAS.		<input type="checkbox"/>							

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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
				Level of implementation of SARPs						
				A) More Exacting or Exceeds	B) Different in character or Other means of compliance	C) Less protective or partially implemented or not implemented				

10.5.4	<p>10.5.4 Recommendation.— <i>In addition to those specified in 7.5, the following procedures should be included, inter alia:</i></p> <p><i>a) remotely piloted aeroplane handover procedures from one RPS to another;</i></p> <p><i>b) C2 Link specifications and procedures for switchover of remotely piloted aeroplane command and control from one C2 Link to another and to respond to temporary interruption or loss of the C2 Link;</i></p> <p><i>c) flight termination procedures, if applicable;</i></p> <p><i>d) security procedures unique to remotely piloted aircraft system (e.g. remote pilot station security, C2 Link, etc.); and</i></p> <p><i>e) detect and avoid.</i></p>		<input type="checkbox"/>							
11.1	CHAPTER 11. REMOTELY PILOTED AEROPLANE UNIQUE CONSIDERATIONS		<input type="checkbox"/>							

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				Level of implementation of SARPs						
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	11.1 General The Standards of this Chapter shall apply to additional aspects of remotely piloted aeroplane features that are not common to manned aviation.									
11.2	11.2 Transportation, storage and assembly Where the remotely piloted aeroplane is designed to be transportable while non-operational, it shall be shown that environmental factors and other foreseeable conditions likely to be encountered during transportation or storage do not adversely affect any requirement of this part. Limitations, information and markings for the safe transport and assembly of the remotely piloted aeroplane shall be developed and made available as defined under Chapter 7 of this part. <i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i>		<input type="checkbox"/>							
11.3.1	11.3 Launch methods 11.3.1 Where the remotely piloted aeroplane is designed to be assisted during launch, the effects of the launch method shall be taken into account in calculating launch loads as required in Chapter 3, and in establishing		<input type="checkbox"/>							

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	operational limitations, markings, and placards as required in Chapter 7.									
11.3.2	11.3.2 <i>Take-off performance.</i> Where the remotely piloted aeroplane is designed to be assisted during launch, the remotely piloted aeroplane shall achieve sufficient energy and controllability at the end of the launch phase to ensure safe and controllable flight under all anticipated operating conditions.		<input type="checkbox"/>							
11.4.1	11.4 Recovery methods 11.4.1 Where the remotely piloted aeroplane is designed to be assisted during normal landing recovery, the effects of the recovery method shall be taken into account in calculating recovery loads as required in Chapter 3, and in establishing operational limitations, markings, and placards as required in Chapter 7 of this part.		<input type="checkbox"/>							
11.4.2	11.4.2 <i>Recovery performance.</i> Where the remotely piloted aeroplane is designed to be assisted during normal landing recovery, the remotely piloted aeroplane flight performance and control characteristics shall be adequate for the intended landing procedures under all anticipated operating conditions.		<input type="checkbox"/>							
11.5	11.5 Emergency recovery		<input type="checkbox"/>							

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	<p>For remotely piloted aeroplanes that have an emergency recovery capability or a flight termination system initiated through remote pilot command or through automatic means with the intent to reduce the risk of fatal injuries to people on the ground in case of emergency landing:</p> <p>a) any systems on board the remotely piloted aeroplane that are critical to an emergency recovery capability to reach a safe area shall perform their intended functions in the entire flight envelope under the remotely piloted aeroplane anticipated operating conditions;</p> <p>b) any systems on board the remotely piloted aeroplane that are critical to a flight termination system, procedure or function that aims to immediately end normal flight shall be shown to perform their intended functions for the entire flight envelope under the remotely piloted aeroplane anticipated operating conditions; and</p> <p>c) operating limitations, procedures, instructions and any additional information necessary for the safe operation of the remotely piloted aeroplane shall be established and provided in the remotely piloted aeroplane flight manual as required in Chapter 7 of this part.</p> <p><i>Note 1.— Emergency recovery capability consists of functions that could be implemented through remote pilot crew command or through an automatic pre-programmed course of action, and are intended to</i></p>									
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	<p><i>navigate the remotely piloted aeroplane to a pre-selected emergency site and then to make an emergency landing.</i></p> <p><i>Note 2.— A flight termination system (e.g. a whole remotely piloted aeroplane recovery parachute) aims to immediately end the flight and to reduce the kinetic energy at impact, but does not necessarily ensure the impact point location.</i></p> <p><i>Note 3.— When considering the protection of people on the ground, in case of emergency landing, items to be considered include:</i></p> <p><i>a) restraint criteria for items that could cause a hazard to people on the ground;</i></p> <p><i>b) fuel cell integrity and position; and</i></p> <p><i>c) integrity of electrical systems to avoid sources of ignition.</i></p>									
11.6	<p>11.6 Automatic taxi, take-off and landing</p> <p>Any systems installed on the remotely piloted aeroplane that are required for automatic taxi, take-off or landing shall ensure that loss, degradation, or interruption of navigational information or C2 Link does not adversely affect safety during taxi, take-off or landing.</p>		<input type="checkbox"/>							
11.7	11.7 C2 Link		<input type="checkbox"/>							

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	<p>The C2 Link, as integrated in the remotely piloted aircraft system, shall perform its intended function under all anticipated operating conditions. Considerations regarding the C2 Link shall include:</p> <p>a) a means to maintain C2 Link through foreseeable operating conditions;</p> <p>b) a means to regain C2 Link in the event that it is temporarily interrupted;</p> <p>c) a means to ensure continued safe flight and landing in the event that the RPAS enters a lost C2 Link state;</p> <p>d) incorporation of C2 Link performance and operational limitations as required in Chapter 7 of this part; and</p> <p>e) a means to monitor the performance and status of the C2 Link.</p>									
11.8	<p>11.8 Detect and avoid, and other equipment</p> <p>Any equipment specifically required for remotely piloted aeroplane operation, such as the detect and avoid system, shall comply with the Standards of Chapter 6.</p>		<input type="checkbox"/>							
11.9	11.9 Mission equipment		<input type="checkbox"/>							

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	The installation of the mission equipment on the remotely piloted aeroplane shall be taken into consideration when showing compliance with the requirements of this part, in order to show that it does not affect the safe flight of the remotely piloted aeroplane.									
11.10.1	11.10 Security 11.10.1 The remotely piloted aeroplane design shall ensure security protection of the remotely piloted aeroplane system from unauthorized physical and electronic access by sources external to the remotely piloted aeroplane, including during maintenance activity.		<input type="checkbox"/>							
11.10.2	11.10.2 Recommendation. — <i>Security threats should be identified and assessed, and risk mitigation strategies should be implemented to protect the remotely piloted aeroplane from adverse impacts on safety, functionality, and continuing airworthiness.</i>		<input type="checkbox"/>							
1.1.1	PART IX. REMOTELY PILOTED HELICOPTERS (RPH) <i>Applicable as of 26 November 2026.</i>		<input type="checkbox"/>							

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	<p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 The Standards of this part are applicable in respect of all remotely piloted helicopters for which an application for the issue of a Type Certificate is submitted to the appropriate national authorities on or after 26 November 2026.</p> <p><i>Note.1 — The provisions in this part support remotely piloted helicopter operation SARPs in Annex 6.</i></p> <p><i>Note.2 — The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p>									
1.1.2	1.1.2 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code referred to in 1.2.1 of Part II for the remotely piloted helicopters designated in 1.1.1 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.		<input type="checkbox"/>							

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1.1.3	1.1.3 Unless otherwise stated, the Standards apply to the complete remotely piloted helicopter including its powerplant, rotors, systems and equipment.		<input type="checkbox"/>							
1.2.1	1.2 Operating limitations 1.2.1 Limiting conditions shall be established for the remotely piloted helicopter, its powerplant, rotors, systems and equipment (see 7.2). Compliance with the Standards of this part shall be established assuming that the remotely piloted helicopter is operated within the limitations specified. The safety implications of exceeding these operating limits shall be considered.		<input type="checkbox"/>							
1.2.2	1.2.2 Limiting ranges of any parameter whose variation may compromise the safe operation of the remotely piloted helicopter, e.g. mass, centre of gravity location, load distribution, speeds, ambient air temperature, altitude and C2 Link performance, shall be established within which compliance with all the pertinent Standards of this part is shown. <i>Note 1.— The maximum operating mass and centre of gravity limits may vary, for example, with each altitude and with each practicably separate operating condition, e.g. take-off, en route, landing.</i> <i>Note 2.— Maximum operating mass may be limited by the application of noise certification Standards (see Annex 16 — Environmental Protection, Volume I — Aircraft Noise and Annex 6 — Operation of Aircraft.</i>		<input type="checkbox"/>							

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1.3	1.3 Unsafe features and characteristics Under all anticipated operating conditions, the remotely piloted helicopter shall not possess any feature or characteristic that renders it unsafe.		<input type="checkbox"/>							
1.4	1.4 Proof of compliance The means by which compliance with the appropriate airworthiness requirements is demonstrated shall ensure that in each case the accuracy achieved will be such as to provide reasonable assurance that the remotely piloted helicopter, its components and equipment comply with the requirements and are reliable and function correctly under the anticipated operating conditions.		<input type="checkbox"/>							
2.1.1	CHAPTER 2. FLIGHT 2.1 General		<input type="checkbox"/>							

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	2.1.1 Compliance with the Standards prescribed in this chapter shall be established by flight or other tests conducted upon a remotely piloted helicopter or remotely piloted helicopters of the type for which a Type Certificate is sought, or by calculations (or other methods) based on such tests, provided that the results obtained by calculations (or other methods) are equal in accuracy to, or conservatively represent, the results of direct testing.									
2.1.2	2.1.2 Compliance with each Standard shall be established for all applicable combinations of remotely piloted helicopter mass and centre of gravity position, within the range of loading conditions for which certification is sought.		<input type="checkbox"/>							
2.1.3	2.1.3 Where necessary, appropriate remotely piloted helicopter configurations shall be established for the determination of performance in the various stages of flight and for the investigation of the remotely piloted helicopter's flying qualities.		<input type="checkbox"/>							
2.2.1	2.2 Performance 2.2.1 Sufficient data on the performance of the remotely piloted helicopter shall be determined and scheduled in the remotely piloted helicopter flight manual to provide operators with the necessary information for the purpose of determining the total mass of the remotely piloted helicopter on the basis of the values, peculiar to		<input type="checkbox"/>							

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	the proposed flight, of the relevant operational parameters, in order that the flight may be made with reasonable assurance that a safe minimum performance for that flight will be achieved.									
2.2.2	<p>2.2.2 Achieving the performance scheduled for the remotely piloted helicopter shall take into consideration human performance and in particular shall not require exceptional skill or alertness on the part of the remote flight crew.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							
2.2.3	2.2.3 The scheduled performance of the remotely piloted helicopter shall be consistent with compliance with 1.2.1 and with the operation in logical combinations of those of the remotely piloted helicopter's systems and equipment, the operation of which may affect performance.		<input type="checkbox"/>							
2.2.4	<p>2.2.4 Minimum performance</p> <p>At the maximum masses scheduled (see 2.2.7) for take-off and for landing as functions of the take-off and landing site pressure-altitude and temperature in still air conditions, and, for water operations, in specified conditions of smooth water, the remotely piloted helicopter shall be capable of accomplishing the minimum performances specified in 2.2.5 and 2.2.6,</p>		<input type="checkbox"/>							

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	respectively, not considering obstacles or final approach and take-off area length.									
2.2.5	<p>2.2.5 Take-off</p> <p>a) the performance at all stages of take-off and climb shall be sufficient to ensure that under conditions of operation departing slightly from the idealized conditions for which data are scheduled (see 2.2.7), the departure from the scheduled values is not disproportionate.</p> <p>b) for Category A remotely piloted helicopters, in the event of critical engine failure at or after the take-off decision point, the remotely piloted helicopter shall be capable of continuing safe flight, the remaining engine(s) being operated within the approved limitations.</p> <p><i>Note.— For remotely piloted helicopters that are assisted during take-off, related SARPs can be found in 11.3 of this part.</i></p>		<input type="checkbox"/>							
2.2.6	<p>2.2.6 Landing</p> <p>a) It shall be possible to make a safe landing on a prepared landing surface after complete power failure occurring during normal cruise, or it shall be possible to initiate the remotely piloted helicopter</p>		<input type="checkbox"/>							

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	<p>emergency recovery capability, as specified in Chapter 11 of this part;</p> <p>b) for Category A remotely piloted helicopters, starting from the landing configuration in the event of critical engine failure at or before the landing decision point, the remotely piloted helicopter shall be capable of continuing safe flight, the remaining engine(s) being operated within the approved limitations.</p> <p><i>Note.— For remotely piloted helicopters that are assisted during landing recovery, related SARPs can be found in 11.4 of this part.</i></p>								
2.2.7.1	<p>2.2.7 Scheduling of performance</p> <p>2.2.7.1 Performance data shall be determined and scheduled in the remotely piloted helicopter flight manual as follows for the ranges of mass, altitude, temperature and other operational variables for which the remotely piloted helicopter is to be certificated, and additionally for amphibians, water surface conditions and strength of current shall be included.</p> <p>a) <i>Hover performance.</i> The hover performance shall be determined for both in-ground effect and out-of-ground effect with all engines operating.</p> <p>b) <i>Climb.</i> The steady rate of climb with the engine(s) operating at or within approved limits shall be established.</p>		<input type="checkbox"/>						

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	<p>c) <i>Height-velocity envelope.</i> If there are any combinations of height and forward velocity (including hover) under which a safe landing cannot be made after failure of the critical engine and with the remaining engine(s) (if applicable) operating within approved limits, a height-velocity envelope shall be established.</p> <p>d) <i>Take-off distance — all engines operating.</i> Where required by the operating rules, the take-off distance — all engines operating shall be the horizontal distance required from the start of the take-off to the point where a selected speed up to the best rate of climb speed (Vy) and selected height above the take-off surface are achieved, all engines operating at approved take-off power required.</p> <p>In addition, for Category A remotely piloted helicopters:</p> <p>e) <i>Minimum performance.</i> The minimum climb performance shall be established for both take-off and landing.</p> <p>f) <i>Take-off decision point.</i> The take-off decision point shall be the point in the take-off phase used in determining take-off performance and from which either a rejected take-off may be made or a take-off safely continued, with the critical engine inoperative.</p> <p>g) <i>Take-off distance required.</i> The take-off distance required shall be the horizontal distance required from the start of the take-off to the point at which the</p>									
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	<p>take-off safety speed (VTOSS), a selected height above the take-off surface, and a positive climb gradient are achieved, following failure of the critical engine at take-off decision point, the remaining engine(s) operating within approved operating limits. If procedures involve rearward flight, the back-up distance shall be included.</p> <p>h) <i>Rejected take-off distance required.</i> The rejected take-off distance required shall be the horizontal distance required from the start of the take-off to the point where the remotely piloted helicopter comes to a complete stop following engine failure and rejection of the take-off at the take-off decision point.</p> <p>i) <i>Take-off path — climb gradients.</i> The take-off path — climb gradient shall be the steady gradient(s) of climb for the appropriate configuration(s) with the critical engine inoperative from the end of the take-off distance required to a defined point above the take-off surface.</p> <p>j) <i>Engine inoperative climb.</i> The engine inoperative climb shall be the steady rate of climb/descent with the critical engine inoperative and the operating engine(s) not exceeding the power for which they are certificated.</p> <p>k) <i>Landing decision point.</i> The landing decision point shall be the latest point in the approach phase from which either a landing may be made or a rejected landing (go-around) safely initiated, with the critical engine inoperative.</p>								
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	l) <i>Landing distance required.</i> The landing distance required shall be the horizontal distance required to land and come to a complete stop from a point on the approach flight path at a selected height above the landing surface with the critical engine inoperative.									
2.2.7.2	<p>2.2.7.2 Recommendation.— <i>For remotely piloted helicopters that are assisted during landing recovery, the effects of recovery methods on the scheduling of performance should be considered.</i></p> <p><i>Note.—Related SARPs can be found in 11.3 and 11.4 of this part.</i></p>		<input type="checkbox"/>							
2.3.1	<p>2.3 Flying qualities</p> <p>2.3.1 The remotely piloted helicopter shall comply with the Standards of 2.3 at all altitudes up to the maximum anticipated altitude relevant to the particular requirement in all temperature conditions relevant to the altitude in question and for which the remotely piloted helicopter is approved.</p>		<input type="checkbox"/>							
2.3.2.1	<p>2.3.2 Controllability</p> <p>2.3.2.1 The remotely piloted helicopter shall be controllable and manoeuvrable under all anticipated operating conditions, and it shall be possible to make smooth transitions from one flight condition to another (e.g. turns, sideslips, changes of engine power, changes of</p>		<input type="checkbox"/>							

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	remotely piloted helicopter configuration) without requiring exceptional skill or alertness on the part of the remote pilot even in the event of failure of any engine. A means or technique for safely controlling the remotely piloted helicopter shall be established for all stages of flight and remotely piloted helicopter configurations for which performance is scheduled. <i>Note.— This Standard is intended, among other things, to relate to operation in conditions of no appreciable atmospheric turbulence and also to ensure that there is no undue deterioration of the flying qualities in turbulent air.</i>									
2.3.2.2	2.3.2.2 <i>Controllability on the ground (or water).</i> The remotely piloted helicopter shall be controllable on the ground (or on the water) during taxiing, take-off and landing under the anticipated operating conditions.		<input type="checkbox"/>							
2.3.2.3	2.3.2.3 <i>Controllability during take-off.</i> The remotely piloted helicopter shall be controllable in the event of sudden failure of the critical engine at any point in the take-off, when the remotely piloted helicopter is handled in the manner associated with the scheduling of the take-off data.		<input type="checkbox"/>							
2.3.3	2.3.3 Trim		<input type="checkbox"/>							

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	The remotely piloted helicopter shall have such trim and handling capabilities as to ensure that the demands made on the remote pilot's attention and ability to maintain a desired flight condition are not excessive when account is taken of the stage of flight at which these demands occur and their duration. In the event of a malfunction of the systems associated with the flight controls, there shall not be any significant deterioration of the handling characteristics.									
2.4.1	2.4 Stability and control 2.4.1 Stability The remotely piloted helicopter shall have such stability in relation to its other flight characteristics, performance, structural strength, and most probable operating conditions (e.g. remotely piloted helicopter configurations and speed ranges) as to ensure that demands made on the remote pilot's powers of concentration are not excessive when the stage of the flight at which these demands occur and their duration are taken into account. The stability of the remotely piloted helicopter shall not, however, be such that excessive demands are made on the remote pilot's skill or that the safety of the remotely piloted helicopter is prejudiced by lack of manoeuvrability in emergency conditions.		<input type="checkbox"/>							
2.4.2.1	2.4.2 Autorotation		<input type="checkbox"/>							

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	2.4.2.1 <i>Rotor speed control.</i> The autorotation characteristics of the remotely piloted helicopter shall be such as to enable control of the rotor speed to within prescribed limits and to maintain full control of the remotely piloted helicopter.									
2.4.2.2	2.4.2.2 <i>Behaviour following a power loss.</i> The behaviour of the remotely piloted helicopter following a power loss shall not be so extreme as to make difficult a prompt recovery of rotor speed without exceeding the airspeed or strength limitations of the remotely piloted helicopter.		<input type="checkbox"/>							
2.4.2.3	2.4.2.3 <i>Autorotation airspeeds.</i> For Category A remotely piloted helicopters, airspeeds for autorotative landings shall be established. For other remotely piloted helicopters, the autorotation airspeeds recommended for maximum range and minimum rate of descent shall be established.		<input type="checkbox"/>							
2.4.3	2.4.3 <i>Vibration</i> There shall be no vibration or buffeting severe enough to interfere with the control of the remotely piloted helicopter.		<input type="checkbox"/>							
2.4.4	2.4.4 <i>Ground resonance</i>		<input type="checkbox"/>							

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	The remotely piloted helicopter shall have no dangerous tendency to oscillate on the ground with the rotor turning.									
3.1	<p>CHAPTER 3. STRUCTURE</p> <p>3.1 General</p> <p>The remotely piloted helicopter structure shall be designed, manufactured and provided with instructions for its maintenance and repair with the objective of avoiding hazardous and catastrophic failure throughout its operational life.</p> <p><i>Note.— Structure includes the airframe, undercarriage, control system, blades and rotorhead, rotor pylon and auxiliary lifting surfaces.</i></p>		<input type="checkbox"/>							
3.2	<p>3.2 Mass and mass distribution</p> <p>Unless otherwise stated, all structural Standards shall be complied with when the mass is varied over the applicable range and is distributed in the most adverse manner, within the operating limitations on the basis of which certification is sought.</p>		<input type="checkbox"/>							

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3.3	<p>3.3 Limit loads</p> <p>Except as might be otherwise qualified, the external loads and the corresponding inertia loads, or resisting loads obtained for the various loading conditions prescribed in 3.7, 3.8 and 3.9 shall be considered as limit loads.</p>		<input type="checkbox"/>							
3.4	<p>3.4 Strength and deformation</p> <p>In the various loading conditions prescribed in 3.7, 3.8 and 3.9, no part of the remotely piloted helicopter structure shall sustain detrimental deformation at any load up to and including the limit load, and the remotely piloted helicopter structure shall be capable of supporting the ultimate load.</p>		<input type="checkbox"/>							
3.5.1	<p>3.5 Airspeeds</p> <p>3.5.1 Design airspeeds</p> <p>Design airspeeds shall be established for which the remotely piloted helicopter structure is designed to withstand the corresponding manoeuvring and gust loads in accordance with 3.7.</p>		<input type="checkbox"/>							

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3.5.2	3.5.2 Limiting airspeeds Limiting airspeeds, based on the corresponding design airspeeds with safety margins, where appropriate, in accordance with 1.2.1, shall be included in the remotely piloted helicopter flight manual as part of the operating limitations (see 7.2.3). When airspeed limitations are a function of mass, mass distribution, altitude, rotor speed, power or other factors, airspeed limitations based on the critical combination of these factors shall be established.		<input type="checkbox"/>							
3.6	3.6 Main rotor(s) rotational speed limits A range of main rotor(s) speeds shall be established that: a) with power on, provides adequate margin to accommodate the variations in rotor speed occurring in any appropriate manoeuvre, and is consistent with the kind of governor or synchronizer used; and b) with power off, allows each appropriate autorotative manoeuvre to be performed throughout the ranges of airspeed and mass for which certification is requested.		<input type="checkbox"/>							
3.7.1	3.7 Loads 3.7.1 The loading conditions of 3.7, 3.8 and 3.9 shall consider the range of mass and mass distributions prescribed in 3.2, the main rotor rpm ranges established in 3.6, and airspeeds established in accordance with 3.5.1. Asymmetrical as well as		<input type="checkbox"/>							

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	symmetrical loading shall be taken into account. The air, inertia and other loads resulting from the specific loading conditions shall be distributed so as to approximate actual conditions closely or to represent them conservatively in consideration of all anticipated operating conditions.									
3.7.2	3.7.2 Manoeuvring loads Manoeuvring loads shall be computed on the basis of manoeuvring load factors appropriate to the manoeuvres permitted by the operating limitations. They shall not be less than values that experience indicates will be adequate for the anticipated operating conditions.		<input type="checkbox"/>							
3.7.3	3.7.3 Gust loads Gust loads shall be computed for vertical and horizontal gust velocities that statistics or other evidence indicate will be adequate for the anticipated operating conditions.		<input type="checkbox"/>							
3.8.1	3.8 Ground and water loads 3.8.1 The structure shall be able to withstand all the loads due to the reactions of the ground or water surface, as applicable, that arise during start-up, ground and water taxiing, lift-off, touchdown and rotor braking.		<input type="checkbox"/>							

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3.8.2	3.8.2 Landing conditions The landing conditions at the maximum certificated take-off mass and at the maximum certificated landing mass shall include such symmetrical and asymmetrical attitudes of the remotely piloted helicopter at ground or water contact, such velocities of descent, and such other factors affecting the loads imposed upon the structure as might be present in the anticipated operating conditions.		<input type="checkbox"/>							
3.9	3.9 Miscellaneous loads In addition to or in conjunction with, the manoeuvring and gust loads and with the ground and water loads, consideration shall be given to all other loads (flight control loads, pilot forces, engine torque, loads due to changes of configuration, external loads, etc.) that are likely to occur in the anticipated operating conditions.		<input type="checkbox"/>							
3.10	3.10 Fatigue strength The strength and fabrication technique of the remotely piloted helicopter structure shall be such as to avoid catastrophic fatigue failure under repeated loads and vibratory loads in the anticipated operating conditions. Environmental degradation, accidental damage and other likely failures shall be considered.		<input type="checkbox"/>							

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3.11	<p>3.11 Special factors</p> <p>Design features (e.g. castings, bearings or fittings), the strength of which are subject to variability in manufacturing processes, deterioration in service or any other cause, shall be accounted for by a suitable factor.</p>		<input type="checkbox"/>							
4.1.1	<p>CHAPTER 4. DESIGN AND CONSTRUCTION</p> <p>4.1 General</p> <p>4.1.1 Details of design and construction shall be such as to give reasonable assurance that all remotely piloted helicopter parts will function effectively and reliably in the anticipated operating conditions. They shall be based upon practices that experience has proven to be satisfactory or that are substantiated by special tests or by other appropriate investigations or both. They shall also consider human factors principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							

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4.1.2	4.1.2 Substantiation of moving parts The functioning of all moving parts essential to the safe operation of the remotely piloted helicopter shall be demonstrated in order to ensure that they will function correctly under all operating conditions for such parts.		<input type="checkbox"/>							
4.1.3	4.1.3 Materials All materials used in parts of the remotely piloted helicopter essential for its safe operation shall conform to approved specifications. The approved specifications shall be such that materials accepted as complying with the specifications will have the essential properties assumed in the design.		<input type="checkbox"/>							
4.1.4	4.1.4 Manufacturing methods The methods of manufacturing and assembly shall be such as to produce consistently sound structure which shall be reliable with respect to maintenance of strength in service.		<input type="checkbox"/>							
4.1.5	4.1.5 Protection The structure shall be protected against deterioration or loss of strength in service due to weathering, corrosion, abrasion or other causes, which could pass unnoticed,		<input type="checkbox"/>							

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	taking into account the maintenance the remotely piloted helicopter will receive.									
4.1.6	4.1.6 Inspection provisions Adequate provision shall be made to permit any necessary examination, replacement or reconditioning of parts of the remotely piloted helicopter that require such attention, either periodically or after unusually severe operations.		<input type="checkbox"/>							
4.1.7	4.1.7 Critical parts All critical parts used in the remotely piloted helicopter shall be identified and procedures shall be established to ensure that the required level of integrity for critical parts is controlled during design, manufacture and throughout the service life of those parts.		<input type="checkbox"/>							
4.2	4.2 Systems design features Special consideration shall be given to design features that affect the ability of the remote flight crew member to maintain controlled flight. This shall include at least the following: a) <i>Controls and control systems.</i> The design of the controls and control systems shall be such		<input type="checkbox"/>							

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	<p>as to minimize the possibility of jamming, inadvertent operation and unintentional engagement of control locking devices.</p> <p>1) each control and control system shall operate with the ease, smoothness and precision appropriate to its function;</p> <p>2) each element of each flight control system shall be designed, or distinctively and permanently marked, to minimize the probability of any incorrect assembly that could result in the malfunction of the system; and</p> <p>3) within the range of adjustment available to the remote pilot, the control system shall not produce hazardous loads on the remotely piloted helicopter or create hazardous deviations in the flight path, under any flight condition appropriate to its use, either during normal operation or in the event of a malfunction, assuming that corrective action begins within a reasonable period of time. Where multiple control systems are installed, subsequent malfunction conditions shall be considered in sequence unless their occurrence is shown to be extremely improbable.</p> <p>b) <i>Pilot vision.</i> The arrangement of design features for remote pilot vision, if implemented on the remotely piloted helicopter, shall permit, under normal and precipitation conditions of moderate rain, sufficient vision for the normal conduct of flight and for the execution of approaches and landings as assumed in the</p>								
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	<p>design to support safe operation of the remotely piloted helicopter.</p> <p>c) <i>Provision for emergencies.</i> Means shall be provided which shall either automatically prevent, or enable the remote flight crew to deal with emergencies resulting from foreseeable failures of equipment, systems, the C2 Link, and the remote pilot station, the failure of which would endanger the remotely piloted helicopter. Reasonable provisions shall be made for continuation of essential services following engine or system failures to the extent that such failures are catered for in the performance and operating limitations specified in the Standards in this Annex and in Annex 6.</p> <p>d) <i>Fire precautions.</i> The remotely piloted helicopter shall have adequate fire protection.</p>									
4.3	<p>4.3 Flutter</p> <p>Each aerodynamic surface of the remotely piloted helicopter shall be free from flutter under each appropriate speed and power condition.</p>		<input type="checkbox"/>							
4.4.1	<p>4.4 Electrical bonding and protection against lightning and static electricity</p>		<input type="checkbox"/>							

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	<p>4.4.1 Electrical bonding and protection against lightning and static electricity shall be such as to:</p> <p>a) protect the remotely piloted helicopter, its systems and persons who come in contact with the remotely piloted helicopter on the ground or water from the dangerous effects of lightning discharge and electrical shock; and</p> <p>b) prevent dangerous accumulation of electrostatic charge.</p>									
4.4.2	4.4.2 The remotely piloted helicopter shall also be protected against catastrophic effects of lightning. Due account shall be taken of the material used in the construction of the remotely piloted helicopter.		<input type="checkbox"/>							
4.5	<p>4.5 Ground handling</p> <p>Adequate provisions shall be made in the design to minimize the risk that normal ground-handling operations (e.g. towing, jacking) may cause damage, which could pass unnoticed, to the parts of the remotely piloted helicopter essential for its safe operation. The protection that any limitations and instructions for such operations might provide may be taken into account.</p>		<input type="checkbox"/>							

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5.1	CHAPTER 5. ROTORS AND POWERPLANT 5.1 Engines The Standards of Part VI of this Annex shall apply to each engine that is used on the remotely piloted helicopter as a primary propulsion unit(s).		<input type="checkbox"/>							
5.2.1	5.2 Rotors and powerplant installation 5.2.1 General The powerplant installation and rotors shall comply with the Standards of Chapter 4 and with the Standards of 5.2.		<input type="checkbox"/>							
5.2.2	5.2.2 Design, construction and functioning a) The rotors and rotor drive systems assembly complete with accessories shall be designed and constructed so as to function reliably within their operating limitations under the anticipated operating conditions when properly fitted to the engine and installed in the remotely piloted helicopter in accordance with this chapter.		<input type="checkbox"/>							

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	<p>b) For remotely piloted helicopters which are certificated to Category A Standard, an assessment shall be conducted for the rotors and rotor drive systems to ensure that they function safely throughout the full range of operating conditions. Where this assessment identifies a failure which could prevent continued safe flight or landing of the remotely piloted helicopter, means shall be prescribed to minimize the likelihood of that failure.</p>								
5.2.3	<p>5.2.3 Declared ratings, conditions and limitations</p> <p>The power ratings and all operating conditions and limitations which are intended to govern the operation of the rotors and rotor drive systems shall be declared.</p> <p>a) <i>Maximum and minimum rotor rotational speed limitations.</i> Maximum and minimum speeds for the rotors in both power-on and power-off conditions shall be established. Any operating conditions (e.g. airspeed) that affect such maxima or minima shall be declared.</p> <p>b) <i>Rotor underspeed warnings for single engine remotely piloted helicopters, and for multi-engine remotely piloted helicopters not having an approved device for automatically increasing power when an engine fails.</i> When the remotely piloted helicopter approaches a rotor rotational speed limit, with or without engines inoperative, clear and distinctive warnings shall be apparent to the remote pilot. The warnings or initial characteristics of the condition shall be such as to enable</p>		<input type="checkbox"/>						

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	the remote pilot or system to arrest the development of the condition after the warning begins and to recover the rotor rotational speed to within prescribed normal limits and to maintain full control of the remotely piloted helicopter.									
5.2.4	<p>5.2.4 Tests</p> <p>Rotors and rotor drive systems shall complete satisfactorily such tests as are necessary to ensure that they will operate satisfactorily and reliably within the declared ratings, conditions and limitations. The tests shall include at least the following:</p> <p>a) <i>Operation.</i> Tests shall be conducted to ensure that strength and vibration characteristics are satisfactory and to demonstrate proper and reliable functioning of pitch changing and control mechanisms and free wheel mechanisms.</p> <p>b) <i>Endurance.</i> Tests of sufficient duration shall be conducted at such powers, engine and rotor speeds, and other operating conditions as are necessary to demonstrate reliability and durability of the rotors and rotor drive systems.</p>		<input type="checkbox"/>							
5.2.5	5.2.5 Compliance with engine, rotor and rotor drive system limitations		<input type="checkbox"/>							

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	The powerplant installation shall be so designed that the engines, rotors and rotor drive systems are capable of functioning reliably in the anticipated operating conditions. In conditions established in the remotely piloted helicopter flight manual, the remotely piloted helicopter shall be capable of being operated without exceeding the limitations established for the engines, rotors and rotor drive systems in accordance with this chapter and Part VI.									
5.2.6	5.2.6 Control of engine rotation For remotely piloted helicopters which are certificated to Category A Standard, where continued rotation of a failed engine would increase the hazard of fire or of a serious structural failure, means shall be provided to stop the rotation of the failed engine in flight or to reduce it to a safe level.		<input type="checkbox"/>							
5.2.7	5.2.7 Engine restarting For remotely piloted helicopters which are certificated to Category A Standard, a means shall be provided for restarting an engine in flight at altitudes up to a declared maximum altitude.		<input type="checkbox"/>							
5.2.8.1	5.2.8 Arrangement and functioning 5.2.8.1 <i>Independence of engines and associated systems.</i> For Category A remotely piloted		<input type="checkbox"/>							

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	helicopters, the engines together with their associated systems shall be arranged and isolated from each other to allow operation, in at least one configuration, so that the failure or malfunction of any engine, or the failure of any system that can affect any engine, will not: a) prevent the continued safe operation of the remaining engine(s); or b) require immediate action, other than normal remote pilot action to maintain safe operation.									
5.2.8.2	5.2.8.2 <i>Rotors and rotor drive systems vibration.</i> The vibration stresses for the rotors and rotor drive systems shall be determined and shall not exceed values that have been found safe for operation within the operating limitations established for the remotely piloted helicopter.		<input type="checkbox"/>							
5.2.8.3	5.2.8.3 <i>Cooling.</i> The cooling system shall be capable of maintaining the temperature of powerplant components and fluids within the established limits (see 5.2.5) at all ambient temperatures approved for operation of the remotely piloted helicopter. The maximum and minimum ambient air temperatures for which the powerplant has been established as being suitable shall be scheduled in the remotely piloted helicopter flight manual.		<input type="checkbox"/>							
5.2.8.4	5.2.8.4 <i>Associated systems.</i> The fuel, oil, air induction, and other systems associated with the powerplant and the rotor(s), shall be capable of supplying the appropriate unit in accordance with its established		<input type="checkbox"/>							

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	requirements, under all conditions affecting the functioning of the systems (e.g. engine power setting, remotely piloted helicopter attitudes and accelerations, atmospheric conditions, fluid temperatures) within the anticipated operating conditions.									
5.2.8.5	<p>5.2.8.5 <i>Fire protection.</i> For regions of the powerplant where the potential fire hazards are particularly serious because of the proximity of ignition sources to combustible materials, the following shall apply in addition to the general Standard of 4.2 d).</p> <p>a) <i>Isolation.</i> Such regions shall be isolated by fire resistant material from other regions of the remotely piloted helicopter where the presence of fire would jeopardize continued flight and landing (Category A remotely piloted helicopters) or would jeopardize safe landing (other remotely piloted helicopters), taking into account the probable points of origin and paths of propagation of fire.</p> <p>b) <i>Flammable fluids.</i> Flammable fluid system components located in such regions shall be fire resistant. Drainage of each region shall be provided to minimize hazards resulting from the failure of any component containing flammable fluids. Means shall be provided to shut off the flow of flammable fluids into such regions if a fire occurs. Where sources of flammable fluid exist in such regions, the whole of the related system within the region, including supporting structure, shall be fireproof or shielded from the effects of fires.</p>		<input type="checkbox"/>							

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	<p>c) <i>Fire detection.</i> For engine installations, a sufficient number of fire detectors shall be provided and located to ensure rapid detection of any fire that might occur in such regions.</p>								
6.1.1	<p>CHAPTER 6. SYSTEMS AND EQUIPMENT</p> <p>6.1 General</p> <p>6.1.1 The remotely piloted helicopter shall be provided with approved equipment and systems, including guidance and flight management systems necessary for the safe operation of the remotely piloted helicopter in the anticipated operating conditions. These shall include the equipment necessary to enable the remote crew to operate the remotely piloted helicopter within its operating limitations. Equipment design shall consider human factors principles.</p> <p><i>Note 1.— Equipment additional to the minimum necessary for the issuance of a Certificate of Airworthiness are prescribed in Annex 6, Part III, for particular circumstances or on particular kinds of routes. Systems are addressed in Part X — Remote Pilot Station (RPS) of this Annex.</i></p>		<input type="checkbox"/>						

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	<i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i>									
6.1.2	<p>6.1.2 The design of the equipment and systems required by 6.1.1 and their installation shall be such that:</p> <p>a) for a Category A remotely piloted helicopter, an inverse relationship exists between the probability of a failure condition and the severity of its effect, as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p> <p>c) electromagnetic interference between them is minimized.</p> <p><i>Note.— The system safety assessment process includes integration of the remote pilot station and the specification of the C2 Link. See also 10.3.3 of this part.</i></p>		<input type="checkbox"/>							
6.1.3	6.1.3 Means shall be provided to warn the remote crew of unsafe system operating conditions on both the remote pilot station and the remotely piloted helicopter and to enable corrective action to be taken automatically or by the remote crew.		<input type="checkbox"/>							
6.1.4	6.1.4 Electrical power supply		<input type="checkbox"/>							

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	The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations and shall also be such that no single failure or malfunction could impair the ability of the system to supply essential loads for safe operation.									
6.1.5	6.1.5 Development assurance of complex electronic hardware and system software Complex electronic hardware and system software shall be developed, verified and validated such as to ensure that the systems in which they are used perform their intended functions at a level of safety that complies with the requirements of this part, notably those of 6.1.2 a) and 6.1.2 b). <i>Note.— Some States accept the use of national or international industry standards for the development assurance (development, verification and validation) of complex electronic hardware and systems software.</i>		<input type="checkbox"/>							
6.2	6.2 Installation Instrument and equipment installations shall comply with the Standards of Chapter 4.		<input type="checkbox"/>							
6.3.1	6.3 Navigation lights and anti-collision lights 6.3.1 The lights required by Annex 2 — <i>Rules of the Air</i> to be displayed by remotely piloted helicopters in flight or operating on the movement area of an		<input type="checkbox"/>							

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	<p>aerodrome shall have intensities, colours, fields of coverage and other characteristics such that they furnish the pilot of another aircraft, the remote pilot of another remotely piloted aircraft, or personnel on the ground with as much time as possible for interpretation and the subsequent manoeuvre necessary to avoid a collision. In the design of such lights, due account shall be taken of the conditions under which they may reasonably be expected to perform these functions.</p> <p><i>Note.— It is likely that lights will be viewed against a variety of backgrounds, such as typical city lighting, clear starry night, moonlit water and daytime conditions of low background luminance. Furthermore, collision risk situations are most likely to arise in terminal control areas in which aircraft are manoeuvring in the intermediate and lower flight levels at closing speeds that are unlikely to exceed 900 km/h (500 kt).</i></p>									
6.3.2	<p>6.3.2 Lights shall be installed in remotely piloted helicopters so as to minimize the possibility that they will adversely affect the satisfactory performance of any required sensors.</p> <p><i>Note.— In order to avoid the effects mentioned in 6.3.2, it will be necessary in some cases to provide means whereby the remote pilot can adjust the intensity of the flashing lights.</i></p>		<input type="checkbox"/>							
6.4	6.4 Electromagnetic interference protection		<input type="checkbox"/>							

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	Remotely piloted helicopter electronic systems, particularly flight-critical and flight-essential systems shall be protected as appropriate against electromagnetic interference from internal and external sources.									
6.5	6.5 Ice protection If certification for flight in icing conditions is requested, the remotely piloted helicopter shall be shown to be able to operate safely in icing conditions likely to be encountered in all anticipated operating environments.		<input type="checkbox"/>							
7.1	CHAPTER 7. OPERATING LIMITATIONS AND INFORMATION 7.1 General The operating limitations within which compliance with the Standards of this Annex is determined, together with any other information necessary to the safe operation of the remotely piloted helicopter, shall be made available by means of a remotely piloted helicopter flight manual, markings and placards, and such other means as may effectively accomplish the purpose.		<input type="checkbox"/>							

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7.2.1	<p>7.2 Operating limitations</p> <p>7.2.1 Limitations which might be exceeded in flight and which are defined quantitatively shall be expressed in suitable units. These limitations shall be corrected if necessary for errors in measurements so that the remote flight crew can, by reference to the instruments available to them, readily determine when the limitations are reached.</p>		<input type="checkbox"/>							
7.2.2	<p>7.2.2 Loading limitations</p> <p>The loading limitations shall include all limiting masses, centres of gravity positions, mass distributions and floor loadings (see 1.2.2).</p>		<input type="checkbox"/>							
7.2.3	<p>7.2.3 Airspeed limitations</p> <p>The airspeed limitations shall include all speeds (see 3.5.2) that are limiting from the standpoint of structural integrity or flying qualities of the remotely piloted helicopter, or from other considerations. These speeds shall be identified with respect to the appropriate remotely piloted helicopter configurations and other pertinent factors.</p>		<input type="checkbox"/>							

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7.2.4	7.2.4 Powerplant limitations The powerplant limitations shall include all those established for the various powerplant components as installed in the remotely piloted helicopter (see 5.2.5 and 5.2.8.3).		<input type="checkbox"/>							
7.2.5	7.2.5 Rotor limitations Limitations on rotor speeds shall include maximum and minimum rotor speeds for power-off (autorotation) and power-on conditions.		<input type="checkbox"/>							
7.2.6	7.2.6 Limitations on equipment and systems The limitations on equipment and systems shall include all those established for the various equipment and systems as installed in the remotely piloted helicopter.		<input type="checkbox"/>							
7.2.7	7.2.7 Miscellaneous limitations Miscellaneous limitations shall include any necessary limitations with respect to conditions found to be prejudicial to the safety of the remotely piloted helicopter (see 1.2.1).		<input type="checkbox"/>							
7.2.8	7.2.8 Remote flight crew limitations		<input type="checkbox"/>							

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	<p>The remote flight crew limitations shall include the minimum number of remote flight crew personnel necessary to operate the remotely piloted helicopter.</p> <p><i>Note.— The circumstances in which the remote flight crew shall include members in addition to the minimum remote flight crew are defined in Annex 6 — Operation of Aircraft.</i></p>									
7.3.1	<p>7.3 Operating information and procedures</p> <p>7.3.1 Types of eligible operations</p> <p>The particular types of operations for which the remotely piloted helicopter has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.</p>		<input type="checkbox"/>							
7.3.2	<p>7.3.2 Loading information</p> <p>The loading information shall include the empty mass of the remotely piloted helicopter, together with a definition of the condition of the remotely piloted helicopter at the time of weighing, the corresponding centre of gravity position, and the reference points and datum lines to which the centre of gravity limits are related.</p> <p><i>Note.— Usually the empty mass excludes the mass of the payload, and the usable fuel supply; it</i></p>		<input type="checkbox"/>							

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	<i>includes the mass of all fixed ballast, unusable fuel supply and total quantity of oil, engine coolant and hydraulic fluid.</i>									
7.3.3	<p>7.3.3 Operating procedures</p> <p>A description shall be given of normal and emergency operating procedures which are peculiar to the particular remotely piloted helicopter and necessary for its safe operation. These shall include procedures to be followed in the event of failure of one or more engines.</p>		<input type="checkbox"/>							
7.3.4	<p>7.3.4 Handling information</p> <p>Sufficient information shall be given on any significant or unusual features of the remotely piloted helicopter characteristics.</p>		<input type="checkbox"/>							
7.4	<p>7.4 Performance information</p> <p>The performance of the remotely piloted helicopter shall be scheduled in accordance with 2.2. There shall be included information regarding the various remotely piloted helicopter configurations and powers involved and the relevant speeds, together with information which will assist the remote flight crew in attaining the performance as scheduled.</p>		<input type="checkbox"/>							
7.5	7.5 Remotely piloted helicopter flight manual		<input type="checkbox"/>							

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	A remotely piloted helicopter flight manual shall be made available. It shall identify clearly the specific remotely piloted helicopter or series of remotely piloted helicopters to which it is related. The remotely piloted helicopter flight manual shall include at least the limitations, information and procedures specified in 7.2, 7.3 and 7.4 of this part and Part X of this Annex.									
7.6	7.6 Markings and placards Markings and placards or instructions shall be provided to give any information that is essential to the ground crew in order to preclude the possibility of mistakes in taxi, take-off, landing, shutdown, and ground servicing (towing, refuelling, etc.), that could pass unnoticed and that could jeopardize the safety of the remotely piloted helicopter in subsequent flights and the safety of the ground crew.		<input type="checkbox"/>							
7.7.1	7.7 Continuing airworthiness — maintenance information 7.7.1 General Information for use in developing procedures for maintaining the remotely piloted helicopter in an		<input type="checkbox"/>							

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	airworthy condition shall be made available. The information shall include that described in 7.7.2, 7.7.3 and 7.7.4.									
7.7.2	7.7.2 Maintenance information Maintenance information shall include a description of the remotely piloted helicopter and recommended methods for the accomplishment of maintenance tasks. Such information shall include guidance on transportation, storage and assembly, and defect diagnosis.		<input type="checkbox"/>							
7.7.3	7.7.3 Maintenance programme information Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed. <i>Note.— The development of initial maintenance programme information at the time of remotely piloted helicopter type certification can take advantage of the Maintenance Review Board (MRB) process or the process of developing instructions for continuing airworthiness.</i>		<input type="checkbox"/>							
7.7.4	7.7.4 Mandatory maintenance requirements resulting from the type design approval		<input type="checkbox"/>							

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	<p>Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 7.7.3.</p> <p><i>Note.— Mandatory requirements identified as part of the type design approval are often referred to as Certification Maintenance Requirements (CMR) and/or airworthiness limitations.</i></p>								
7.8	<p>7.8 C2 Link information</p> <p>Sufficient information shall be given on any relevant C2 Link relating to configuration, operation, performance, emergency procedures, and operating limitations.</p>		<input type="checkbox"/>						
9.1	<p>CHAPTER 9. OPERATING ENVIRONMENT AND HUMAN FACTORS</p> <p>9.1 General</p> <p>The remotely piloted helicopter shall be designed to allow safe operation within the performance limitations of those who operate, maintain and service the aircraft.</p> <p><i>Note.— The human/machine interface is often the weak link in an operating environment and so it is</i></p>		<input type="checkbox"/>						

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	<i>necessary to ensure that the remotely piloted helicopter is capable of being controlled at all phases of the flight (including any degradation due to failures).</i>									
9.2.1	<p>9.2 Remote flight crew</p> <p>9.2.1 The remotely piloted helicopter shall be designed in such a way as to allow safe and efficient control by the remote flight crew. The design shall allow for variations in remote flight crew skill and physiology commensurate with remote flight crew licensing limits. Account shall be taken of the different expected operating conditions of the remotely piloted helicopter in its environment, including operations degraded by failures.</p>		<input type="checkbox"/>							
9.2.2	<p>9.2.2 The workload imposed on the remote flight crew by the design of the remotely piloted helicopter shall be reasonable at all stages of flight. Particular consideration shall be given to critical stages of flight and critical events which may reasonably be expected to occur during the service life of the remotely piloted helicopter, such as engine failure.</p> <p><i>Note.— Workload can be affected by both cognitive and physiological factors.</i></p>		<input type="checkbox"/>							
9.3	9.3 Ergonomics		<input type="checkbox"/>							

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	<p>During design of the remotely piloted helicopter, account shall be taken of ergonomic factors, where applicable, including:</p> <ul style="list-style-type: none"> a) ease of use and prevention of inadvertent misuse; b) accessibility; c) maintainability; and d) transportation, storage and assembly/disassembly. 								
10.1.1	<p>CHAPTER 10. REMOTE PILOT STATION INTEGRATION</p> <p>10.1 General</p> <p>10.1.1 The Standards of Part X of this Annex shall apply to each remote pilot station that is used to pilot the remotely piloted helicopter.</p>		<input type="checkbox"/>						
10.1.2	<p>10.1.2 The remote pilot station shall be compatible with the type of the remotely piloted helicopter and appropriate to the intended operation.</p>		<input type="checkbox"/>						

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10.2.1	<p>10.2 Integration</p> <p>10.2.1 <i>Compliance with remote pilot station limitations.</i> The remotely piloted helicopter shall be so designed that the remote pilot station is capable of performing satisfactorily and reliably its intended functions in the anticipated operating conditions when connected to the remotely piloted helicopter. In conditions established in the flight manual, the remotely piloted helicopter shall be capable of being operated within the limitations established for the remote pilot station in accordance with this chapter and with Part X.</p>		<input type="checkbox"/>							
10.2.2	<p>10.2.2 <i>Integration tests.</i> The remotely piloted helicopter shall complete satisfactorily tests with all approved types of remote pilot stations, as are necessary to verify the validity of the declared conditions and limitations and to ensure that the remote pilot station will operate satisfactorily and reliably using any specified C2 Link and supporting C2 Link communication service providers, as specified under the anticipated operating conditions.</p>		<input type="checkbox"/>							
10.3.1	<p>10.3 Controls and information</p> <p>10.3.1 The remote pilot station shall be integrated in such a way as to allow timely control as required for safe and efficient control of the remotely</p>		<input type="checkbox"/>							

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	<p>piloted helicopter by the remote flight crew. This shall include at least the following:</p> <p>a) processing the data provided by the remotely piloted helicopter regarding:</p> <ul style="list-style-type: none"> — attitude, altitude, position, heading, speed, vertical speed, turning information; — powerplant; — detect and avoid; — weather conditions; — rotor speed; — C2 Link state and performance according to the SARPs defined in the applicable sections of Annex 10 for remotely piloted aircraft systems; and — status of automated systems, including the current lost C2 Link state; <p>b) controlling the remotely piloted helicopter in the anticipated operating condition;</p> <p>c) controlling the powerplant according to Chapter 5 of this part;</p> <p>d) information on predicted QoSD in the geographical area of the flight based on the QoSr and C2 Link specification; and</p>								
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	e) status of automated systems, including flight controls exceedance or malfunctions.									
10.3.2	<p>10.3.2 All required information shall be provided through the remote pilot station for the remote flight crew to safely and efficiently operate the remotely piloted helicopter (e.g. set or monitor flight parameters for the flight, navigation, and powerplant) using any specified C2 Link and supporting C2 Link communication service providers in the anticipated operating conditions. These shall include the instruments and equipment necessary to enable the remote crew to operate the remotely piloted helicopter within its anticipated operating limitations. Instrument and equipment design shall consider human factors principles.</p> <p><i>Note 1.— Instruments and equipment, additional to the minimum necessary for the issuance of a Certificate of Airworthiness, are prescribed in Annex 6 for particular circumstances or on particular kinds of routes.</i></p> <p><i>Note 2.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							
10.3.3	10.3.3 The design of the instruments, equipment and systems required by 10.3.2 and their installation shall be such that:		<input type="checkbox"/>							

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	<p>a) an inverse relationship exists between the probability of a failure condition and the severity of its effect as determined by a system safety assessment process;</p> <p>b) they perform their intended function under all anticipated operating conditions; and</p> <p>c) electromagnetic interference between them is minimized.</p>									
10.3.4	10.3.4 Means shall be provided to warn the remote flight crew of unsafe system operating conditions and to enable the crew to take corrective action.		<input type="checkbox"/>							
10.3.5	10.3.5 Markings and placards on instruments, equipment, controls, etc., shall include such limitations or information as necessary for the direct attention of the remote flight crew during flight.		<input type="checkbox"/>							
10.4.1	10.4 C2 Link 10.4.1 The remotely piloted helicopter and remote pilot station system architecture shall be compatible with any specified C2 Link and supporting C2 Link communication service providers as specified, to enable the remotely piloted helicopter to be operated safely under the anticipated operating conditions.		<input type="checkbox"/>							

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10.4.2	10.4.2 Means shall be provided to monitor the C2 Link performance and the C2 Link state according to metrics defined in the applicable parts of Annex 10, reacting according to the transaction completion criteria defined in Annex 6.		<input type="checkbox"/>							
10.5.1	10.5 Flight manual 10.5.1 The remotely piloted helicopter flight manual shall address all combinations of remote pilot station models listed in the approved type design of the remotely piloted helicopter. There may be substantial variations between different remote pilot stations used with the same remotely piloted helicopter.		<input type="checkbox"/>							
10.5.2	10.5.2 In developing the remotely piloted helicopter flight manual, specific consideration shall be given to human performance aspects, including transfer of control within and between remote pilot station if envisaged by operational requirements, remote pilot handovers, control link switchovers, appropriate contingency planning procedures, remote crew communications, e.g. remote pilot to remote pilot, remote pilot to remotely piloted helicopter observer or other support personnel, and remote pilot to ATC.		<input type="checkbox"/>							

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10.5.3	10.5.3 The remotely piloted helicopter flight manual shall contain all necessary information for operation of the RPAS.		<input type="checkbox"/>							
10.5.4	<p>10.5.4 Recommendation.— <i>In addition to those specified in 7.5, the following procedures should be included, inter alia:</i></p> <p>a) <i>remotely piloted helicopter handover procedures from one remote pilot station to another;</i></p> <p>b) <i>C2 Link specifications and procedures for switchover of remotely piloted helicopter command and control from one C2 Link to another and to respond to temporary interruption or loss of the C2 Link;</i></p> <p>c) <i>flight termination procedures, if applicable;</i></p> <p>d) <i>security procedures unique to remotely piloted aircraft systems (e.g. remote pilot station security, C2 Link, etc.); and</i></p> <p>e) <i>detect and avoid.</i></p>		<input type="checkbox"/>							
11.1	CHAPTER 11. REMOTELY PILOTED HELICOPTER UNIQUE CONSIDERATIONS		<input type="checkbox"/>							

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	<p>11.1 General.</p> <p>The Standards of this Chapter shall apply to additional aspects of remotely piloted helicopter features that are not common to manned aviation.</p>									
11.2	<p>11.2 Transportation, storage and assembly</p> <p>Where the remotely piloted helicopter is designed to be transportable while non-operational, it shall be shown that environmental factors and other foreseeable conditions likely to be encountered during transportation or storage do not adversely affect any requirement of this part. Limitations, information and markings for the safe transport and assembly of the remotely piloted helicopter shall be developed and made available as defined under Chapter 7 of this part.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (Doc 9683) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (Doc 10019).</i></p>		<input type="checkbox"/>							
11.3.1	11.3 Launch methods		<input type="checkbox"/>							

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	11.3.1 Where the remotely piloted helicopter is designed to be assisted during launch, the effects of the launch method shall be taken into account in calculating launch loads as required in Chapter 3, and in establishing operational limitations, markings, and placards as required in Chapter 7.									
11.3.2	11.3.2 Take-off performance. Where the remotely piloted helicopter is designed to be assisted during launch, the remotely piloted helicopter shall achieve sufficient energy and controllability at the end of the launch phase to ensure safe and controllable flight under all anticipated operating conditions.		<input type="checkbox"/>							
11.4.1	11.4 Recovery methods 11.4.1 Where the remotely piloted helicopter is designed to be assisted during normal landing recovery, the effects of the recovery method shall be taken into account in calculating recovery loads as required in Chapter 3, and in establishing operational limitations, markings, and placards as required in Chapter 7.		<input type="checkbox"/>							
11.4.2	11.4.2 Recovery performance. Where the remotely piloted helicopter is designed to be assisted during normal landing recovery, the remotely piloted helicopter flight performance and control characteristics shall be adequate for the intended landing procedures under all anticipated operating conditions.		<input type="checkbox"/>							

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11.5	<p>11.5 Emergency recovery</p> <p>For remotely piloted helicopters that have an emergency recovery capability or a flight termination system initiated through remote pilot command or through automatic means with the intent to reduce the risk of fatal injuries to people on the ground in case of emergency landing:</p> <p>a) any systems on board the remotely piloted helicopter that are critical to an emergency recovery capability to reach a safe area shall perform their intended functions in the entire flight envelope under the remotely piloted helicopter anticipated operating conditions;</p> <p>b) any systems on board the remotely piloted helicopter that are critical to a flight termination system, procedure or function that aims to immediately end normal flight shall be shown to perform their intended functions in the entire flight envelope under the remotely piloted helicopter anticipated operating conditions; and</p> <p>c) operating limitations, procedures, instructions and any additional information necessary for the safe operation of the remotely piloted helicopter shall be established and provided in the remotely piloted helicopter flight manual as required in Chapter 7.</p>		<input type="checkbox"/>							
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	<p><i>Note 1.— A flight termination system (e.g. a whole remotely piloted helicopter recovery parachute) aims to immediately end the flight and to reduce the kinetic energy at impact, but does not necessarily ensure the impact point location.</i></p> <p><i>Note 2.— Emergency recovery capability consists of functions that could be implemented through remote pilot crew command or through an automatic pre-programmed course of action, and are intended to navigate the remotely piloted helicopter to a pre-selected emergency site and then to make an emergency landing.</i></p> <p><i>Note 3.— When considering the protection of people on the ground, in case of emergency landing, items to be considered include:</i></p> <p><i>a) restraint criteria for items that could cause a hazard to people on the ground;</i></p> <p><i>b) fuel cell integrity and position; and</i></p> <p><i>c) integrity of electrical systems to avoid sources of ignition.</i></p>								
11.6	<p>11.6 Automatic taxi, take-off and landing</p> <p>Any system installed on the remotely piloted helicopter that is required for automatic taxi, take-off or landing shall ensure that loss, degradation, interruption of</p>		<input type="checkbox"/>						

Annex Reference & SARP Identifier	European Union Aviation Safety Agency-Annex 8 Amendment 109	State Reference	Difference					Not Applicable	Details of Difference	Remarks
	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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	navigational information or C2 Link does not adversely affect safety during taxi, take-off or landing.										
11.7	<p>11.7 C2 Link</p> <p>The C2 Link, as integrated in the remotely piloted aircraft system, shall perform its intended function under all anticipated operating conditions. Considerations regarding the C2 Link shall include at least:</p> <p>a) a means to maintain C2 Link through foreseeable operating conditions;</p> <p>b) a means to regain C2 Link in the event that it is temporarily interrupted;</p> <p>c) a means to ensure continued safe flight and landing in the event that the RPAS enters a lost C2 Link state;</p> <p>d) incorporation of C2 Link performance and operational limitations as required in Chapter 7 of this part; and</p> <p>e) a means to monitor the performance and status of the C2 Link.</p>		<input type="checkbox"/>								
11.8	11.8 Detect and avoid, and other equipment		<input type="checkbox"/>								

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	Any equipment required for remotely piloted helicopter operation, such as the detect and avoid system, shall comply with the Standards of Chapter 6 of this part.									
11.9	11.9 Mission equipment The installation of the mission equipment on the remotely piloted helicopter shall be taken into consideration when showing compliance with the requirements of this part, in order to show that it does not affect the safe flight of the remotely piloted helicopter.		<input type="checkbox"/>							
11.10.1	11.10 Security 11.10.1 The remotely piloted helicopter design shall ensure system security protection of the remotely piloted helicopter system from unauthorized physical and electronic access by sources external to the remotely piloted helicopter, including during maintenance activity.		<input type="checkbox"/>							
11.10.2	11.10.2 Recommendation. — <i>Security threats should be identified and assessed, and risk mitigation strategies should be implemented to protect the remotely piloted helicopter from adverse impacts on safety, functionality, and continuing airworthiness.</i>		<input type="checkbox"/>							

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				Level of implementation of SARPs						
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1.1.1	<p>PART X. REMOTE PILOT STATION (RPS)</p> <p><i>Applicable as of 26 November 2026.</i></p> <p>CHAPTER 1. GENERAL</p> <p>1.1 Applicability</p> <p>1.1.1 Except as noted below, the Standards of this part are applicable to remote pilot stations of all types as required in Parts VIII and IX. The Standards of this part are applicable to a remote pilot station type at the time of submission of an application to the appropriate national authority for a type approval.</p> <p><i>Note 1.— The following Standards do not include quantitative specifications comparable to those found in national airworthiness codes. In accordance with 1.2.1 of Part II, these Standards are to be supplemented by requirements established, adopted or accepted by Contracting States.</i></p> <p><i>Note.2 — The provisions in this part support RPAS operation SARPs in Annex 6.</i></p>		<input type="checkbox"/>							
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	AIRWORTHINESS OF AIRCRAFT - Annex Standard or Recommended Practice		No	Yes			Significant Difference			
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1.1.2	1.1.2 The level of airworthiness defined by the appropriate parts of the comprehensive and detailed national code for the remote pilot stations designated in 1.1.1 shall be at least substantially equivalent to the overall level intended by the broad Standards of this part.		<input type="checkbox"/>							
1.2	1.2 RPS interfaces and integration All necessary information for the safe and correct interfaces between the remote pilot station and the remotely pilot aircraft shall be made available, including those limitations concerning the C2 Link and information necessary for intended function of any C2 Link as specified in the type design.		<input type="checkbox"/>							
1.3.1	1.3 Continuing airworthiness — maintenance information 1.3.1 <i>General.</i> Information for use in developing procedures for maintaining the remote pilot station in an airworthy condition shall be made available. The information shall include that described in 1.3.2, 1.3.3 and 1.3.4.		<input type="checkbox"/>							
1.3.2	1.3.2 <i>Maintenance information.</i> Maintenance information shall include a description of the remote pilot station and recommended methods for the accomplishment of maintenance tasks. Such information		<input type="checkbox"/>							

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	shall include guidance on defect diagnosis. Such information shall clearly distinguish between: a) defect diagnosis and rectification tasks that may be performed while the remote pilot station is operational if necessary for the safe conclusion of the flight; and b) maintenance tasks that must not be performed when the remote pilot station is operational.									
1.3.3	1.3.3 <i>Maintenance programme information.</i> Maintenance programme information shall include the maintenance tasks and the recommended intervals at which these tasks are to be performed.		<input type="checkbox"/>							
1.3.4	1.3.4 <i>Mandatory maintenance requirements resulting from the type design approval.</i> Mandatory maintenance requirements that have been specified by the State of Design as part of the approval of the type design shall be identified as such and included in the maintenance information of 1.3.3.		<input type="checkbox"/>							
2.1	CHAPTER 2. DESIGN AND CONSTRUCTION 2.1 Fire, smoke and toxic gas protection		<input type="checkbox"/>							

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	Means shall be provided to minimize the risk of fires, the production of smoke and toxic gases in the event of a fire.									
2.2	2.2 Functioning The remote pilot station shall be designed and constructed so as to function reliably within its operating limitations under its anticipated operating conditions when integrated within a remote piloted aircraft system using any C2 Link and supporting communications services, as specified under the anticipated operating conditions in the type design.		<input type="checkbox"/>							
2.3	2.3 Failure analysis A safety assessment of the remote pilot station shall be conducted to ensure that it functions safely throughout the full range of operating conditions. A failure analysis shall be conducted to identify the potential RPS failure conditions, their effect at the RPS level and their probability of occurrence that will allow performing the overall system safety assessment at the RPA level, as required in Chapter 6 of Part VIII or Part IX.		<input type="checkbox"/>							
2.4	2.4 Materials and manufacturing methods The selection of materials and the manufacturing methods and processes shall account for the operational		<input type="checkbox"/>							

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	environment of the remote pilot station expected in service.									
2.5	<p>2.5 Electrical bonding and protection against lightning and static electricity</p> <p>Electrical bonding and protection against lightning and static electricity shall be such as to:</p> <p>a) protect the remote pilot station, its systems, its occupants and those who come in contact with the remote pilot station from the dangerous effects of lightning discharge and electrical shock; and</p> <p>b) prevent dangerous accumulation of electrostatic charge.</p>		<input type="checkbox"/>							
2.6	<p>2.6 Handling of the remote pilot station</p> <p>Design provisions and procedures for safe handling of the remote pilot station shall be defined.</p>		<input type="checkbox"/>							
3.1.1	CHAPTER 3. SYSTEM AND EQUIPMENT		<input type="checkbox"/>							

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	3.1 General 3.1.1 Systems and equipment installed on the RPS shall be so designed and installed as to ensure compliance with all Standards set in this part, as well as those applicable to the RPA controlled by the RPS.									
3.1.2	3.1.2 The RPS shall be able to display to the remote flight crew all information necessary to safely operate the RPA.		<input type="checkbox"/>							
3.1.3	3.1.3 The RPS shall provide means to warn the remote flight crew of unsafe conditions related to its own systems or received from the RPA controlled by the RPS and to enable them to take corrective action.		<input type="checkbox"/>							
3.2	3.2 Electrical power supply The design of the electrical power supply system shall be such as to enable it to supply power loads during normal operations of the remote pilot station.		<input type="checkbox"/>							
3.3	3.3 Electromagnetic interference protection		<input type="checkbox"/>							

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	Electronic systems pertaining to the remote pilot station, particularly those the malfunction of which may adversely affect the safe operation of the remotely piloted aircraft, shall be protected against electromagnetic interference from both internal and external sources.									
3.4	<p>3.4 Development assurance of complex electronic hardware and system software</p> <p>Complex electronic hardware and system software shall be developed, verified and validated to ensure that the systems in which they are used on the remote pilot station perform their intended functions at a level of safety commensurate with the failure condition classification for the remotely piloted aircraft in which RPS certification is sought.</p> <p><i>Note.— Some States accept the use of national or international industry standards for the development assurance (development, verification and validation) of complex electronic hardware and systems software.</i></p>		<input type="checkbox"/>							
4.1	CHAPTER 4. REMOTE FLIGHT CREW COMPARTMENT SAFETY		<input type="checkbox"/>							

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	4.1 Fire protection Means shall be provided for fire protection to the remote flight crew.								
4.2	4.2 Evacuation Means shall be provided to allow adequate evacuation in case of emergency.		<input type="checkbox"/>						
5.1	CHAPTER 5. OPERATING ENVIRONMENT AND HUMAN FACTORS 5.1 General The remote pilot station shall be designed to allow safe operation within the performance limitations of those who operate, maintain and service the remote pilot station. <i>Note.— The human/machine interface is often the weak link in an operating environment and so it is necessary to ensure that the remotely piloted aircraft is</i>		<input type="checkbox"/>						

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	<i>capable of being controlled at all phases of the flight (including any degradation due to failures and/or remote pilot station located on mobile/moving platforms where inputs from equipment-based sources could conflict with sensory sources) and that the remote crew is not harmed by the environment in which they have been placed for the duration of the remotely piloted aircraft operation.</i>									
5.2.1	<p>5.2 Remote flight crew</p> <p>5.2.1 The remote pilot station shall be designed in such a way as to allow safe and efficient control of the remotely piloted aircraft by the remote flight crew. The design shall allow for variations in remote flight crew skill and physiology commensurate with remote flight crew licensing limits. Account shall be taken of the different expected operating conditions of the remotely piloted aircraft, including operations degraded by failures.</p>		<input type="checkbox"/>							
5.2.2	<p>5.2.2 Recommendation.— <i>The human performance implications of the lack of sensory information directly from the remotely piloted aircraft (e.g. vibration, g-load, fumes, flames) resulting from the pilot being remote to the aircraft should be considered and, where necessary, such information should be accordingly addressed.</i></p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training</i></p>		<input type="checkbox"/>							

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	Manual (<i>Doc 9683</i>) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (<i>Doc 10019</i>).									
5.3	<p>5.3 Ergonomics</p> <p>During design of the remote pilot station, account shall be taken of ergonomic factors including:</p> <p>a) ease of use and prevention of inadvertent misuse;</p> <p>b) accessibility;</p> <p>c) remote pilot station design philosophy;</p> <p>and</p> <p>d) maintainability.</p>		<input type="checkbox"/>							
5.4.1	<p>5.4 Operating environmental factors</p> <p>5.4.1 The operating environment of the remote pilot station shall be designed in accordance with human performance principles.</p> <p><i>Note.— Guidance material on human factors principles can be found in the Human Factors Training Manual (<i>Doc 9683</i>) and the Manual on Remotely Piloted Aircraft Systems (RPAS) (<i>Doc 10019</i>).</i></p>		<input type="checkbox"/>							

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5.4.2	5.4.2 Adequate seating shall be provided for the remote flight crew. Attention shall be paid to minimize injury to the remote flight crew due to contact with surrounding structures during the operation of the remotely piloted aircraft.		<input type="checkbox"/>							
6.1	CHAPTER 6. OPERATING LIMITATIONS AND INFORMATION 6.1 General All operating conditions and limitations which are intended to govern the operation of the remote pilot station shall be declared.		<input type="checkbox"/>							
6.2.1	6.2 Operating information and procedures 6.2.1 <i>Types of eligible operations.</i> The particular types of operations for which the remote pilot station has been shown to be eligible by virtue of compliance with the appropriate airworthiness requirements shall be listed.		<input type="checkbox"/>							

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6.2.2	6.2.2 <i>Operating procedures.</i> A description shall be given of normal and emergency operating procedures which are peculiar to the particular remote pilot station and necessary for its safe operation.		<input type="checkbox"/>							
6.3	6.3 RPS operating manual An RPS operating manual shall be provided: a) identifying clearly the specific remote pilot station or series of remote pilot stations to which it is related; b) identifying the specific remotely piloted aircraft/remotely piloted aircraft system or series of remotely piloted aircraft/remotely piloted aircraft systems to which it is related; and c) including at least the limitations, information and procedures specified in 6.1 and 6.2.		<input type="checkbox"/>							
7.1	CHAPTER 7. SECURITY 7.1 Remote pilot station access control		<input type="checkbox"/>							

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	Means shall be provided to adequately prevent unauthorized access to the RPS.									
7.2.1	7.2 Systems security 7.2.1 The remote pilot station design shall ensure security protection of the remotely piloted aircraft system from unauthorized physical and electronic access by sources external to the remote pilot station, including during maintenance activity.		<input type="checkbox"/>							
7.2.2	7.2.2 Recommendation. — <i>Security threats should be identified and assessed, and risk mitigation strategies should be implemented to protect the remote pilot station from adverse impacts on safety, functionality, and continuing airworthiness.</i>		<input type="checkbox"/>							
1.1	APPENDIX. APPROVED MAINTENANCE ORGANIZATION (AMO) CERTIFICATE 1. Purpose and scope		<input type="checkbox"/>							

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	1.1 Recommendation. — <i>The AMO certificate should contain the minimum information required in paragraph 2.</i>									
1.2	1.2 Recommendation. — <i>The AMO certificate should define the scope of approval for which a maintenance organization is authorized.</i> <i>Note.— Detailed guidance material and examples for the completion of the AMO template in paragraph 2 is contained in the Airworthiness Manual (Doc 9760).</i>		<input type="checkbox"/>							
2	<p>2. AMO template</p> <p>APPROVED MAINTENANCE ORGANIZATION CERTIFICATE</p> <hr/> <p>Issuing authority:¹</p> <p>Approval reference number:² Organization name:³ Expiration date (if applicable):⁴</p> <p>Registered address:</p>		<input type="checkbox"/>							

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	Telephone: E-mail:																						
	Class(es) and rating(s) authorized <table style="width: 100%; border-collapse: collapse;"> <thead> <tr> <th style="width: 33%;">Class⁵</th> <th style="width: 33%;">Rating⁶</th> <th style="width: 33%;">Limitations⁷</th> </tr> </thead> <tbody> <tr> <td>Aircraft maintenance</td> <td></td> <td></td> </tr> <tr> <td>Engine maintenance</td> <td></td> <td></td> </tr> <tr> <td>Component maintenance</td> <td></td> <td></td> </tr> <tr> <td>Specialized maintenance</td> <td></td> <td></td> </tr> </tbody> </table>	Class ⁵	Rating ⁶	Limitations ⁷	Aircraft maintenance			Engine maintenance			Component maintenance			Specialized maintenance									
Class ⁵	Rating ⁶	Limitations ⁷																					
Aircraft maintenance																							
Engine maintenance																							
Component maintenance																							
Specialized maintenance																							
	<p style="text-align: center;">Terms of Approval</p> <p>This certificate certifies that⁸ _____ is authorized to engage in activities specified in the Terms of Approval annexed hereto, subject to the compliance with the⁹ _____ and the latest maintenance organization's procedures manual (MOPM).</p> <p>Locations of maintenance facilities: As per¹⁰ _____ of the latest MOPM.</p> <p>This certificate shall remain valid during the period of validity specified above unless it is surrendered, superseded, suspended or revoked.</p>																						

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<p>Name:¹¹ _____ Date of original issue:¹² _____</p> <p>Title:¹³ _____ Date of current issue:¹⁵ _____</p> <p>Signature:¹⁴ _____</p> <p>Notes:</p> <p>A. Name of the authority issuing the approval.</p> <p>B. Unique approval reference number as issued by the State of Registry.</p> <p>C. Registered address, telephone and email.</p> <p>D. Expiry date (dd-mm-yyyy) if applicable, if not applicable, insert N/A.</p> <p>E. Scope of approval using the classes as follows: aircraft, engine, component or specialized maintenance.</p> <p>F. Scope of approval using the ratings as follows:</p> <p>a) aircraft maintenance — large aeroplane, small aeroplane, helicopter, other kind of aircraft (such as glider, balloon, airship, light sport aircraft);</p>								
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	<p>b) engine maintenance — categories of engine (such as reciprocating, turbine and electric);</p> <p>c) components maintenance — standard numbering system (SNS) code derived from ASD/ATA S1000D specification for identifying the aircraft system applicable to the rating (<i>Airworthiness Manual</i> (Doc 9760, Chapter 10, Attachment F refers); and</p> <p>d) specialized maintenance — class of approval necessary for the specialized maintenance using the following ratings: composite material maintenance, surface treatment such as peening, plating, painting, non-destructive testing, welding, other unique processes accepted/approved by the State (Doc 9760, Chapter 10, Attachment F refers).</p> <p>G. Limitation in the scope of approval if required for aircraft, components or specialized maintenance. If the limitations are described in the approved maintenance organization’s procedures manual a reference to the manual should be included in the AMO certificate.</p> <p>H. Name of organization authorized to perform maintenance. In the case where a State does not annex terms of approval to the AMO certificate, the State should amend this item as follows:</p> <p>“This certificate certifies that⁸ _____ is authorized to engage in activities listed in this certificate, subject to compliance with</p>									
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	<p>the _____ and the latest maintenance organization's procedures manual."</p> <p>I. Reference to relevant State regulations.</p> <p>J. Reference to the appropriate section/chapter and paragraph of the maintenance organization's procedures manual in which the approved locations of the organization's facilities are listed; for example, Section/Chapter 1, paragraph 1.1.</p> <p>K. Name of the authority representative signing the AMO certificate.</p> <p>L. Date of original issue (if different from the date of current issue), if not, use N/A.</p> <p>M. Title of the authority representative signing the AMO certificate.</p> <p>N. Signature of the authority representative. In addition, an official stamp may be applied on the AMO certificate.</p> <p>O. Issuance date of the AMO certificate (dd-mm-yyyy).</p>								
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