

TYPE-CERTIFICATE DATA SHEET

NO. EASA.A.065

for **H 36**

Type Certificate Holder Diamond Aircraft Industries GmbH

> Nikolaus-August-Otto Str. 5 2700 Wiener Neustadt Austria

 For models:
 H 36
 "DIMONA"

 HK 36
 "SUPER DIMONA"

 HK 36 R
 "SUPER DIMONA"

 HK 36 TS
 HK 36 TC

 HK 36 TTC
 HK 36 TTC-ECO



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SECTION A: <u>H 36 "DIMONA"</u>

A.I <u>General</u>

2.

- 1. Type/ Model/ Variant
 - Type: H 36 "DIMONA" Model: ---Variant: ---Airworthiness Category Utility
- 3. Manufacturer

Utility Hoffmann Flugzeugbau Friesach Gesellschaft mbH 9322 Hirth/Friesach Austria

Hoffmann Aircraft Flugzeugproduktion und Entwicklung GmbH Richard Neutra-Gasse 5 1214 Wien Austria

4. EASA Type Certification Application Date:

See Note 6

Note: State of Design Authority certification application date for grandfathered products

- 5. State of Design Authority Initial: Austria
- 6. State of Design Authority Type Certificate Date

See Note 6

- The EASA Type Certificate replaces the Austrian Type Certificate SF 3/82.
- 7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

A.II EASA Certification Basis

- 1 Reference Date for determining the applicable requirements
- 2. Airworthiness Requirements JAR-22, Change -, issued 15-Mar-1982
- 3. Special Conditions
- 4. Exemptions
- 5. (Reserved) Deviations
- 6. Equivalent Safety Findings
- 7. Environmental Protection
- None None
- None None
- Zivilluftfahrzeug-Lärmzulässigkeitverordnung BGBI. 700/1986

A.III Technical Characteristics and Operational Limitations

Type Design Definition		H36 Top Drawing Set and following approved Design			
2.	Descript	ion	Single engine, two-seated cantilever low wing airplane, GFRP-construction, T-tail, side by side seating configuration, fixed two-legged landing gea		
3.	Equipme	nt	Minimum Equi	pment:	
0.	Equiptine		1 airspeed indi	cator (range up to 300 k	m/h)
			1 altimeter wit	h mbar barometric dial	
			1 magnetic cor	mpass with deviation tab	ole
			1 RPM indicate	or	
			1 running time	meter	
			1 oil pressure g	gauge	
			1 on temperati	ure gauge	
			1 cylinder hear	temperature gauge	
			1 at least 4-poi	int harness for each seat	
			1 voltmeter		
4.	Dimensio	ons	Span	16 m	
			Length	6.85 m	
			Height	-	
-	F		Wing Area	15.2 m	
5	Engine				
	Model		L 2000 EB 1.C OF L 2000 EB 1.AC (see Note 5)		
		Type Certificate	EASA.E.083		
		Limitations	Max take-off ro Max continuou For power-plar	otational speed us rotational speed nts limits refer to AFM	3400 RPM 3000 RPM
6	Propelle	r			
		Model	Hoffmann HO-V62-R/L 160 T or 1 Hoffmann HO-V62-R/L 160 BT Low pitch setting/ Static BPM: 2800+/- 100		- 100
	Type Certificate		EASA.P.065		
		Number of blades	2		
		Diameter	1600 mm		
		Sense of Rotation	CCW		
7.	Fuel cap	acities			
		Tank in the fuselage	Total: 80 liter Usable: 80 liter	rs rs	
8.	Launchir	ng Hooks	N/A		
9.	Weak Lir	nks	N/A		
10.	Load Fac	tors	see AFM		
11.	L1. Air Speeds		Design manoeuvring speed v _A : 176 km/h Maximum rough air speed v _{ra} : 210 km/h		

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		Never exceed s	speed v _{NE} :	275 km/h
12.	Approved Operations Capability	VFR Day		
13.	Launch methods	Self-launch		
14.	Maximum Masses	Take-off		770 kg
		Maximum mas	s of non lifting parts	560 kg
15.	Centre of Gravity Range			
	Forward limit:	270 mm behind	d Datum for all masses	
	Rear limit:	up to 740 kg	385 mm behind Datum	l
		at 770 kg	370 mm behind Datum	l
		varying	g linearly with mass in be	etween
16.	Datum	wing leading e	dge at root rib	
17.	Levelling Means	tangent to win	g lower surface at root r	ib
		(0.6 m beside p	plane of symmetry) horiz	ontal
18.	Control Surface Deflections	See AMM		
19.	Minimum Flight Crew	1 (Pilot)		
20.	Maximum Passenger Seating Capacity	2		
21.	Baggage/ Cargo Compartments	Behind seats	12 kg	
22.	Lifetime limitations	See AMM		

A.IV Operating and Service Instructions

1.	Flight Manual	Airplane Flight Manual, Issue May 1984, BAZ approved, valid for S/Ns. 36.01 – 36.193 and S/N 35.01 – 35.39 inclusive		
		Airplane Flight Manual, Issue 15-Nov-1985, BAZ approved, valid for S/Ns. 36.151 – 36.153 and S/Ns. 36.204 and subsequent		
2.	Maintenance Manual	Maintenance Manual, Issue May 1984, valid for S/Ns 36.01 – 36.193 and S/Ns. 35.01 – 35.39 inclusive		
		Maintenance Manual, Issue 15-Nov-1985, valid for S/Ns. 36.151 – 36.153 and from S/Ns. 36.204 inclusive		

All Master Manuals are issued in German Language only.

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A.V <u>Notes</u>

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. In individual cases, deviations are permitted only in agreement with the type certificate holder.
- 3. The installation and use of a differential braking system in accordance with HOAC/DAI SB 42, latest issue, is permitted.
- 4. Use of unleaded fuel, min. ROZ 96, in accordance with HOAC/DAI SB 56, latest issue, is permitted.
- 5. Engine type designation in accordance with Limbach Technical Bulletin 17.
- 6. Initial Certification carried out by LBA- Germany TC 820 and transferred to Austria TC SF 3/82 before production start.



SECTION B: HK 36 "SUPER DIMONA"

- B.I General
- 1. Type/ Model/ Variant

Model:

Variant:

- 2. Airworthiness Category
- 3. Manufacturer

HK 36 "SUPER DIMONA" ---Utility Hoffmann Aircraft GesmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

H 36 "DIMONA"

HOAC Austria GesmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

Diamond Aircraft Industries GmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

4. EASA Type Certification Application Date: --

Note: State of Design Authority certification application date for grandfathered products

- 5. State of Design Authority Initial: Austria
- 6. State of Design Authority Type Certificate Date
 - 15-May-1990
 - The EASA Type Certificate replaces the Austrian Type Certificate SF 3/82.
- 7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

B.II EASA Certification Basis

- 1. Reference Date for determining the applicable requirements
- 2. Airworthiness Requirements JAR-22, Change 4, issued 07-May-1987
 - Special Conditions
- 4. Exemptions
- 5. (Reserved) Deviations
- 6. Equivalent Safety Findings
- 7. Environmental Protection
- JAR-22, Change 4, issued 07-May-1987 None
- None
 - None None
 - None Zivilluftfahrzeug-Lärmzulässigkeitverordnung BGBI.
 - 700/1986



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3.

B.III Technical Characteristics and Operational Limitations

1.	Type De	sign Definitio	n	HK36 Top Drav	ving Set and following a	pproved
2.	Descript	ion		Design Changes (AM – System) Single engine, two-seated cantilever low wing airplane, GFRP-construction, two main wheels on fixed spring bow and steered tail wheel, T-tail, air		w wing wheels on I, T-tail, air
3.	Equipment		brakes on upper Minimum Equi 1 airspeed indi 1 altimeter wit 1 magnetic cor 1 RPM indicato 1 running time 1 oil pressure g 1 oil temperato 1 cylinder head 1 fuel quantity 1 manifold pre	er wing surface pment: cator (range up to 300 k h mbar barometric dial npass with deviation tak or meter gauge ure gauge d temperature gauge gauge	m/h) ble	
				1 fuel pressure control light		
				1 ammeter	and far analy sout	
4.	Dimensi	ons		1 4-point name Span	16.2 m	
	Differior			Length	7.1 m	
				Height	1.76 m	
F	Engino			Wing Area	15.3 m²	
5.	Engine	Madal		1 2400 EP 1 C a		oto 1)
		Tupo Cortific	ata		DI L 2400 EB 1.AC (SEE N	018 4)
		Limitations	ale	EASA E.U84	atational chood	2200 0014
		LIIIIIIduoiis		Max continuo	s retational speed	2000 RPIVI
		For power p	lants limits rafor	to Elight Manual	is rotational speed	5000 KPIVI
6.	Propelle	roi power-p r			I	
	·	Model		mt-Propeller N	1TV-1-A/L 160-03 Consta	ant Speed
		Type Certific	ate	32.130/53		
		Number of b	lades	2		
		Diameter		1600 mm		
		Sense of Rotation		CCW		
		Settings		Low pitch setti	ng/ Static Rpm: 2950 +	/-100
7.	Fuel cap	acities				
		Standard Tai	nkTotal	55 liters		
			Usable	54 liter		
		Optional	Total	80 liters		
0	Laurah 1		Usable	79 liter		
ð.	Launching Hooks N/A					



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9. 10	Weak Links Load Factors	N/A See AFM		
11.	Air Speeds	Design Manoeuvring Sp	beed V _A :	176 km/h
		Maximum rough air spe	eed V _c :	210 km/h.
		Never exceed speed V _N	E:	261 km/h
12.	Approved Operations Capability	VFR Day		
13.	Launch methods	Self-launch		
14.	Maximum Masses	Take-off		770 kg
		Maximum mass of non	lifting parts	590 kg
15.	Centre of Gravity Range	Forward limit	318 mm behin	d Datum
		Rear limit	430 mm behin	d Datum
16.	Datum	wing leading edge at ro	ot rib	
17.	Levelling Means	wedge 1000 : 52.5 hori	zontal on fusela	age tube
18.	Control Surface Deflections	See AMM		
19.	Minimum Flight Crew	1 (Pilot)		
20.	Maximum Passenger Seating Capacity	2		
21.	Baggage/ Cargo Compartments	Behind Rear Seats	12 kg	
22.	Lifetime limitations	see AMM		

B.IV Operating and Service Instructions

1.	Flight Manual	Airplane Flight Manual, HK 36 "SUPER DIMONA", issued April 1990, BAZ approved, valid for S/Ns. 36.301 and subsequent
2.	Maintenance Manual (incl. Airworthine	Airplane Maintenance Manual, HK 36 "SUPER DIMONA", Doc. No. 3.02.21 or Doc. No. 3.02.04 (German Version) See Note 5
		Service Information's and Service Bulletins

B.V <u>Notes</u>

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations in accordance to the maintenance manual are permitted.
- 3. The installation and use of a differential braking system, in accordance with the HOAC/DAI SB 42, latest issue, is permitted.
- 4. Engine type designation in accordance with Limbach Technical Bulletin 17.
- 5. The HK 36 Series AMM Doc. No. 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E, which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 6. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted



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SECTION C: HK 36 R "SUPER DIMONA"

C.I General

- 1. Type/ Model/ Variant
 - Type:
 - Model:
 - Variant:
- 2. **Airworthiness Category**
- 3. Manufacturer

---Utility Hoffmann Aircraft GesmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

HK 36 R "SUPER DIMONA"

H 36 "DIMONA"

HOAC Austria GesmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

Diamond Aircraft Industries GmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

- EASA Type Certification Application Date: --4.
- 5. State of Design Authority Initial: Austria
- State of Design Authority Type Certificate Date 6. 06-Sep-1990 The EASA Type Certificate replaces the Austrian Type Certificate SF 3/82. 7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

C.II EASA Certification Basis

- 1. Reference Date for determining the applicable requirements
- 2. JAR-22, Change 4, issued 07-May-1987 **Airworthiness Requirements**
- CRI O-3 "Tow Cable Retraction mechanism" 3. **Special Conditions** 4. Exemptions
- 5. (Reserved) Deviations
- None
- None
- 7. **Environmental Protection**

Equivalent Safety Findings

CRI A-9 "Deviations of Serial No. 36307" Zivilluftfahrzeug-Lärmzulässigkeitverordnung BGBI. 738/1993



6.

C.III Technical Characteristics and Operational Limitations

1.	Type De	sign Definition	HK36 Top Dra	awing Set and following a	pproved
2.	Descript	ion	Design Changes (AM – System) Single engine, two-seated cantilever low wing airplane, GFRP-construction, T-tail, side by side seating configuration, fixed two-legged landing gear air brakes on upper wing surface		
3.	Equipme	ent	Minimum Equ	uipment:	
			1 airspeed ind	, dicator (range up to 300 k	km/h)
			1 altimeter w	ith mbar barometric dial	
			1 magnetic co	ompass with deviation tal	ble
			1 RPM indicat	tor	
			1 running tim	ie meter	
			1 oil pressure	e gauge	
			1 oii tempera	iture gauge ad tomporaturo or coolar	t tomporaturo
				au temperature of coolar • (MÄM 36-450 installed)	it temperature
			1 fuel quantit	v gauge	
			1 manifold pr	essure gauge	
			1 fuel pressur	re control light	
			1 ammeter		
			1 at least 4-p	oint harness for each sea	t
4.	Dimensi	ons	Span	16.2 m	
			Length	7.22 m 1.76 m	
			Ming Area	1.70 m^2	
5.	Engine		Wing Area	15.5 11	
		Model	Rotax 912 A2	or Rotax 912 A3	
		Type Certificate	FASA F 121	01 1101011 912 110	
		limitations	Max take_off	rotational speed	
		Linitations	Max care-on	notational speed	
			For nower-nl:	ants limits refer to Flight	SSUU KPIVI Manual
6.	Propelle	r	i oi powei pi		Wandar
	•	Model	For Rotax Eng	gine 912 A2:	
			1 mt-Propeller MTV-1- Δ/I 170-08 Constant Speed		
			2 Hoffmann H014-170 \$ 123		
			3 mt-Propell	er MT-170R125-2Δ	
			Prope	eller type Certificate:	EASA P.006
			For Rotax Eng	zine 912 A3:	
			1. Hoffmann	, HOV-352F-S1/S170FQ	
			2. mt-Propell	er MTV-21-A-C-F/CF 175-	05
			 دوم N	lote 9	
		Settings see ANNN fo	or the relevant or	oneller combination	
		Number of hlades 2			

Number of blades

**** * ****

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	Diameter	1700 mm or 17	'50 mm			
	Sense of Rotat	ion CW				
7.	Fuel capacities					
	Standard Fuel	Tank	Total	55 liters	S	
			Usable	54 liter		
	Optional		Total	80 liters	S	
			Usable	79 liter		
8.	Launching Hooks		N/A			
9.	Weak Links		N/A			
10.	Load Factors		see AFM			
11.	Air Speeds		Design Manoeu	ivring Sp	eed V _A :	176 km/h
			Maximum roug	h air spe	ed V _{ra} :	210 km/h
			Never exceed s	peed V_N	E:	261 km/h
12.	Approved Operations C	apability	VFR Day, see No	ote4		
13.	Launch methods		Self-launch			
14.	Maximum Masses		Take-off			770 kg
			Maximum mass	s of non	lifting parts	590 kg
15.	Centre of Gravity Range	!	Forward limit		318 mm behind	d Datum
			Rear limit		430 mm behind	d Datum
16.	Datum		wing leading ec	lge at ro	ot rib	
17.	Levelling Means		wedge 1000 : 5	2.5 horiz	zontal on fusela	ge tube
18.	Control Surface Deflect	ons	see AMM			
19.	Minimum Flight Crew		1 (Pilot)			
20.	Maximum Passenger Se	ating Capacity	2			
21.	Baggage/ Cargo Compa	rtments	Behind Rear Se	ats	12 kg	
22.	Lifetime limitations		see AMM			

C.IV Operating and Service Instructions

1. Flight Manual

Airplane Flight Manual, HK 36 R "SUPER DIMONA" issued June 1990, BAZ approved, valid for S/Ns. 36.301 and subsequent, if engine Rotax 912 A2 is installed

Airplane Flight Manual, HK 36 R "SUPER DIMONA", Doc. No. 3.01.04, ACG approved on 22-Jul-1994, latest effective issue, valid for S/Ns. 36.301 and subsequent, if engine Rotax 912 A3 is installed

Flughandbuch für den Motorsegler HK 36 R "SUPER DIMONA", Doc. No. 3.01.03, ACG approved on 03-May-2001, latest effective issue, valid for S/N 36.307, if Rotax 912 A3 and mt-Propeller MTV-21-A-C-F/CF175-05 are installed

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2. Maintenance Manual (incl. Airworthiness Limitations)

Airplane Maintenance Manual, HK 36 "SUPER DIMONA", Doc. No. 3.02.21 or Doc. No. 3.02.04 (German Version), see Note 5

Service Informations and Service Bulletins

C.V Notes

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations, carried out in accordance with the maintenance manual, are permitted.
- 3. The installation and use of a differential braking system, in accordance with HOAC/DAI SB 42, latest issue, is permitted, is permitted.
- 4. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted.
- 5. The HK 36 Series AMM Doc. No 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E, which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 6. The use of the type HK 36 R "Super Dimona" as a towing airplane in accordance with HOAC/DAI SB 40, latest effective issue, is permitted.
- 7. The use of unleaded fuel in accordance with SB No. 36 is permitted.
- 8. The installation and use of the type HK 36 R as a towing airplane with a tow rope retraction unit in accordance with HOAC/DAI SB 61, latest effective issue, is permitted.
- 9. The propeller is only approved for S/N 36.307. The deviations from the basic model are defined in Doc. No. 3.07.01, Chapter R36-003 "Design Deviations". The retrofit in accordance with RÄM 36-003 is permitted.

Propeller type:	mt-Propeller	MTV-21-A-C-F/CF175-05			
Data Sheet No.:	EASA.P.101				
Diameter:	1750 mm ± 0 m	m			
Low Pitch:	12°±0.2°				
Starting Pitch:	14°±1°				
Feathered Pitch:	83°±1°				
Ctrwts. At Low Pitch:28°±1°					
High Pitch:	23°±1°				
Propeller RPM is reduced 1:2.273 to engine RPM					

SECTION D: HK 36 TS

D.I <u>General</u>

nt

	Туре:	H 36 "DIMONA"
	Model:	НК 36 TS
	Variant:	
2.	Airworthiness Category	Utility
3.	Manufacturer	Diamond Aircraft Industries GmbH
		N.A. Otto-Str. 5
		2700 Wiener Neustadt
		Austria
		the state of the s

- 4. EASA Type Certification Application Date
- 5. State of Design Authority Initial: Austria
- State of Design Authority Type Certificate Date 06-Mar-1996
 The EASA Type Certificate replaces the Austrian Type Certificate SF 3/82.
- 7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

D.II EASA Certification Basis

1. Reference Date for determining the applicable requirements

Airworthiness Requirements	JAR-22, Change 4, issued 07-May-1987
Special Conditions	CRI E-1 "Propeller feathering control"
	CRI O-1 "Use as a Tow Plane"
	CRI O-3 "Tow Cable Retraction Mechanism"
Exemptions	None
(Reserved) Deviations	None
Equivalent Safety Findings	CRI D-1 "Middle air brake stop"
	CRI E-2 "Propeller Type Definition"
Environmental Protection	Zivilluftfahrzeug-Lärmzulässigkeitverordnung BGBI. 738/1993
	Airworthiness Requirements Special Conditions Exemptions (Reserved) Deviations Equivalent Safety Findings Environmental Protection



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D.III Technical Characteristics and Operational Limitations

1.	Type De	sign Definition	Drawing List H including Design Design Change	IK 36 TS [gn Chang	Doc. 3.08.01 da ges 1 and follow	ted 01-Jan-1996 ving List of
2.	Descript	ion	Design Changes (AM) HK 36 TS Single engine, two-seated cantilever low wing airplane, GFK/CFK-construction, T-tail, side by side seating configuration, tail wheel, fixed two-legged			ow wing , side by side I two-legged
3	Fauinme	ont	Minimum Fau	inment.	s on upper wing	surface
5.	Equipine		1 airspeed ind	licator (ra	ange up to 300	km/h)
			1 altimeter wi	th mbar l	barometric dial	
			1 magnetic co	mpass w	ith deviation ta	ble
			1 RPM indicate	or (Prop	RPM)	
			1 oil pressure	gauge		
			1 oil temperat	ure gaug	je	
			1 cylinder hea gauge	d tempe (MÄM 3	rature or coolai 6-450 installed	nt temperature)
			1 fuel quantity	y gauge		
			1 manifold pre	essure ga e control	luge light	
			1 ammeter	control	iigiit	
			1 4-point harn	ness for e	ach seat	
4.	Dimensions		Span	16.33 r	n (incl. Winglet)
			Length	7.28 m		
			Height	1.78 m		
_			Wing Area	15.3 m	2	
5.	Engine					
		Model	Rotax 912 A3			
		Type Certificate	EASA.E.121			
		Limitations	Max take-off r	rotationa	l speed	5800 RPM
			Max continuo For power-pla	us rotation Ints limits	onal speed s refer to Flight	5500 RPM Manual
6.	Propelle	r				_
		Model	mt-Propeller N	MTV-21-A	A-C-F/CF 175-05	D
		Type Certificate	EASA.P.101			
		Number of blades	2			
		Diameter	1750 mm			
		Sense of Rotation	CW			
		Settings	Low pitch sett	ing:	12°±0.2°	
			Starting Pitch: Feathered Pitc Ctrwts. At Low	ch: v Pitch:	14°±1° 83°±1° 28°±1°	
			High pitch set	ting:	23°±1°	



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		Gearbox Ratio		1:2.273	
7.	Fuel capacities				
	Standard Fuel Tank	Total	55 liters	5	
		Usable	54 liter		
	Optional	Total	79 liters	5	
		Usable	77 liter		
8.	Launching Hooks	N/A			
9.	Weak Links	N/A			
10.	Load Factors	see AFM			
11.	Air Speeds	Design Manoeuvring Speed V _A :			176 km/h
		Maximum roug	gh air spe	ed V _{ra} :	210 km/h
		Never exceed s	speed V_{N}	E:	261 km/h
12.	Approved Operations Capability	VFR Day, see N	ote4		
13.	Launch methods	Self-launch			
14.	Maximum Masses	Take-off			770 kg
		Maximum mas	s of non	lifting parts	590 kg
15.	Centre of Gravity Range	Forward limit		318 mm behin	d Datum
		Rear limit		430 mm behin	d Datum
16.	Datum	wing leading e	dge at ro	ot rib	
17.	Levelling Means	wedge 1000 : 5	2.5 horiz	zontal on fusela	ge tube
18.	Control Surface Deflections	see AMM			
19.	Minimum Flight Crew	1 (Pilot)			
20.	Maximum Passenger Seating Capacity	2			
21.	Baggage/ Cargo Compartments	Behind Rear Se	ats	12 kg	
22.	Lifetime limitations	see AMM			

D.IV Operating and Service Instructions

1. Flight Manual

Airplane Flight Manual, HK 36 TS, Doc. No. 3.01.06, ACG approved, issued 30-Jan-1996, see Note. 9

2. Maintenance Manual (incl. Airworthiness Limitations)

Airplane Maintenance Manual, HK 36 "SUPER DIMONA", Doc. No. 3.02.21 or Doc. No. 3.02.04 (German Version) See Note 5 Service Informations and Service Bulletins

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D.V Notes

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations in accordance with the maintenance manual are permitted.
- 3. Certification valid for S/Ns. 36.415 36.416 and S/Ns. 36.501 and subsequent, excluding Serial No 36.713, 36.717, 36.719, 36.725 and 36.729.

S/Ns. 36.415 and 36.416 have the following deviations according to HOAC Doc. No. 3.07.101, Chapter 2:

- Fuselage structure
- Landing gear mount
- Horizontal stabilizer structure
- Tank drain
- Electric bonding.
- 4. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted.
- 5. The HK 36 Series AMM Doc. No. 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E, which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 6. The engine Rotax 912 A3 has to be modified in accordance with Rotax SB 912-11, ACG approved on 29-Feb-1996, with Propeller Governor WOODWARD A210790 or Rotax SB 912-24, ACG approved, with Propeller Governor McCauley DCFU290D17B/T1.
- 7. The installation and use of the type HK 36 TS as a towing airplane in accordance with DAI SB 40, latest effective issue, is permitted.
- 8. The installation and use of a differential braking system, in accordance with DAI SB 42, latest effective issue, is permitted.
- 9. For Serial Nos. 36.517 and subsequent Airplane Flight Manual HK 36 TS, Doc. No. 3.01.06, Revision 1 or later, ACG approved, is required.
- 10. The change of the propeller designation from MTV-21-A-C-F/C175-05 to MTV-21-A-C-F/CF175-05 in accordance with DAI SB 52, ACG approved, is permitted.
- 11. The installation of a tow rope retraction unit and use of the type HK 36 TS as a towing airplane in accordance with DAI SB 61, latest effective issue, is permitted.

SECTION E: HK 36 TC

E.I <u>General</u>

1.	Type/ Model/ Variant	
	Туре:	H 36 "DIMONA"
	Model:	НК 36 ТС
	Variant:	
2.	Airworthiness Category	Utility
3.	Manufacturer	Diamond Aircraft Industries GmbH
		N.A. Otto-Str. 5
		2700 Wiener Neustadt
		Austria
4.	EASA Type Certification Application Dat	te
		18-Mar-1996
5.	State of Design Authority	Initial: Austria
6.	State of Design Authority Type Certifica	ate Date
		12-Jul-1996
	The EASA Type Certificate replaces the	Austrian Type Certificate SF 3/82
7.	EASA Type Certification Date	21-Dec-2005 (reissue for EASA)

E.II EASA Certification Basis

1. Reference Date for determining the applicable requirements

2.	Airworthiness Requirements	JAR-22, Change 4, issued 07-May-1987
		Amendment 22/90/1, Amendment 22/91/1
		CRI A-1 HK36TC and HK36TC with Rotax 912S
3.	Special Conditions	CRI E-1 "Propeller feathering control"
		CRI O-1 "Use as a Tow Plane"
		CRI O-3 "Tow Cable Retraction Mechanism"
4.	Exemptions	None
5.	(Reserved) Deviations	None
6.	Equivalent Safety Findings	CRI D-1 "Middle air brake stop"
		CRI E-2 "Propeller Type Definition"
7.	Environmental Protection	Zivilluftfahrzeug-Lärmzulässigkeitverordnung BGBI.
		738/1993

E.III Technical Characteristics and Operational Limitations

1.	Type De	sign Definition	Drawing List H including Desi	IK 36 TC gn Chang	Doc. 3.08.01 dat ges 14 andfollow	ed 12-Jul-1996 ⁄ing List of
2.	Descript	ion	Single engine, two-seated cantilever low wing airplane, GFK/CFK-construction, T-tail, side by side seating configuration, fixed two-legged tri-cycle			w wing side by side I tri-cycle surface
3.	Equipme	ent	Minimum Equ	lipment:	s on upper wing	Surface
			1 airspeed ind	licator (ra	ange up to 300 k	m/h)
			1 altimeter wi	th mbar	barometric dial	
			1 magnetic co	mpass w	vith deviation tab	ble
			1 KPIVI Indicat	or (Prop	RPIVI)	
			1 oil pressure	gauge		
			1 oil temperat	ture gaug	ge	
			1 cylinder hea gauge	d tempe (MÄM 3	rature or coolan 6-450 installed)	t temperature
			1 fuel quantity	y gauge		
			1 manifold pre	essure ga	luge	
			1 ammeter			
			1 4-point harr	ness for e	ach seat	
4.	Dimensions		Span	16.33 ı	m (incl. Winglet)	
			Length	7.28 m	1	
			Height	1.78 m	1	
			Wing Area	15.3 m	2	
5.	Engine					
		Model	Rotax 912 A3	or Rotax	912 S3	
		Type Certificate	EASA.E.121			
		Limitations	Max take-off	rotationa	ll speed	5800 RPM
			Max continuous rotational speed 5500 RPM			5500 RPM
			for Ro	tax 912 S	53 see Note 11	
			For po	ower-plai	nts limits refer to	o Flight Manual
6.	Propelle	r				
		Model	mt-Propeller N	MTV-21-/	A-C-F/CF 175-05	
		Type Certificate	EASA.P.101			
		Number of blades	2			
		Diameter	1750 mm			
		Sense of Rotation	CW			
		Settings	For Rotax 912	A3:		
			Low pitch sett	ing:	12°±0.2°	
			Starting Pitch:	:	14°±1°	

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		Feathered Pitcl Ctrwts. At Low High pitch setti Gearbox Ratio	h: Pitch: ng:	83°±1° 28°±1° 23°±1° 1:2.273	
		For Rotax 912	43:		
		Low pitch setting	ng:	14°±0.2°	
		Starting Pitch: Feathered Pitch Ctrwts. At Low	h: Pitch:	14°±1° 83°±1° 28°±1°	
		Gearbox Ratio	ng:	20 ±1 1:2 4286	
7.	Fuel capacities			11211200	
	Standard Fuel Tank	Total	55 liter	S	
		Usable fuel	54 liter		
	Optional	Total	79 liter	S	
8. 9.	Launching Hooks Weak Links	Usable fuel N/A N/A	77 liter		
10. 11.	Air Speeds	Design Manoeu Maximum roug Never exceed s	uvring Sp gh air spo speed V _N	beed V _A : eed V _{RA} : _{IE} :	176 km/h 210 km/h. 261 km/h
12.	Approved Operations Capability	VFR Day, see N	ote4		
13. 14.	Launch methods Maximum Masses	Self-launch Take-off Maximum mas	s of non	lifting parts	770 kg 610 kg
15.	Centre of Gravity Range	Forward limit Rear limit	0.505	318 mm behind 430 mm behind	d Datum Datum
 16. 17. 18. 19. 20. 21. 22. 	Datum Levelling Means Control Surface Deflections Minimum Flight Crew Maximum Passenger Seating Capacity Baggage/ Cargo Compartments Lifetime limitations	wing leading ed wedge 1000 : 5 see AMM 1 (Pilot) 2 Behind Seats see AMM	dge at Y 2.5 hori	= 0.6 m zontal on fusela, 12 kg	ge tube

E.IV Operating and Service Instructions

1.	Flight Manual	Airplane Flight Manual, HK 36 TC, Doc. No. 3.01.10-E,
		ACG approved, for powerplant Rotax 912 A3, issued
		15-May-1996 (see Note 8)



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Airplane Flight Manual, HK 36 TC, Doc. No. 3.01.12-E, ACG approved, for powerplant Rotax 912 S3, issued 09-Jan-2002 (see Note 11)

2. Maintenance Manual (incl. Airworthiness Limitations)

Airplane Maintenance Manual, HK 36 "SUPER DIMONA", Doc 3.02.21 or Doc. 3.02.04 (German Version) See Note 5

Service Informations and Service Bulletins

E.V Notes

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations in accordance with the maintenance manual are permitted.
- 3. Certification valid for S/N 36.505 andS/N 36.517 and subsequent except S/N 36.713, 36.717, 36.719, 36.725, 36.729 and 36.735.
- 4. S/N 36.505 has the following deviations according to HOAC Doc. No. 3.07.101, Chapter 2:
 - Wing structure
 - Main bulkhead structure
 - Air brake system
- 5. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted.
- 6. The HK 36 Series AMM Doc. No. 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E, which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 7. The engine Rotax 912 A3 has to be modified in accordance with Rotax SB 912-11, ACG approved on 29-Feb-1996, with Propeller Governor WOODWARD A210790 or Rotax SB 912-24, ACG approved, with Propeller Governor McCauley DCFU290D17B/T1.
- 8. The use of the type HK 36 TC as a towing airplane in accordance with DAI SB 40, latest effective issue, is permitted.
- 9. For S/N 36.505, in addition to the Airplane Flight Manual Supplement 4 is valid, ACG approved on 7-Oct-1996
- 10. The change of the propeller designation from MTV-21-A-C-F/C175-05 to MTV-21-A-C-F/CF175-05 in accordance with DAI SB 52, ACG approved, is permitted.
- 11. The installation of a tow rope retraction unit in accordance with DAI SB No. 61 in conjunction with DAI SB 40, use as a tow-plane, is permitted.
- 12. The optional installation of the engine Rotax 912 S3 by the manufacturer in accordance with OÄM 36-200 is permitted for serial no. 36.640 and subsequent. The retrofit installation between Engine 1 and Engine 2 is permitted for all effective serial numbers, in accordance with OSB 36-078.



SECTION F: HK 36 TTS

F.I General

1.	Type/ Model/ Variant		
	Туре:	H 36 "DIMONA"	
	Model:	НК 36 ТТЅ	
	Variant:		
2.	Airworthiness Category Utility		
3.	Manufacturer	Diamond Aircraft Industries GmbH	
		N.A. Otto-Str. 5	
		2700 Wiener Neustadt	
		Austria	
4.	EASA Type Certification Application Date		
		07-May-1996	
5.	State of Design Authority	Initial: Austria	
6.	State of Design Authority Type Ce	ertificate Date	
		20-Dec-1996	
	The EASA Type Certificate replace	es the Austrian Type Certificate SF 3/82	

7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

F.II EASA Certification Basis

1. Reference Date for determining the applicable requirements

		-
2.	Airworthiness Requirements	JAR-22, Change 5
3.	Special Conditions	CRI A-1 HK36TTC and HK36TTS CRI E-1 "Propeller feathering control" CRI O-1 "Use as a Tow Plane"
		CRI O-3 "Tow Cable Retraction Mechanism"
4.	Exemptions	None
5.	(Reserved) Deviations	None
6.	Equivalent Safety Findings	CRI D-1 "Middle air brake stop"
		CRI E-2 "Propeller Type Definition"
		CRI G-1 "Engine Operating Limitation"
7.	Environmental Protection	Zivilluftfahrzeug-Lärmzulässigkeitverordnung BGBl. 738/1993

F.III Technical Characteristics and Operational Limitations

1.	Type De	sign Definition		Drawing List 1996 includir of Design Cha	HK 36 T*' Ig Design Anges (ÄN	* Doc. 3.08.01 dated 20-Dec- Changes 57 and following List A) HK 36 T**
2.	Descript	tion		Single engine airplane, GFK seating config tail wheel, air	, two-sea /CFK-con guration, r brakes c	ted cantilever low wing struction, T-tail, side by side fixed two-legged landing gear, on upper wing surface
3.	Equipm	ent		Minimum Eq	uipment:	
				1 airspeed i	ndicator	(range up to 300 km/h)
				1 altimeter	with mba	ır barometric dial
				1 magnetic	compass	with deviation table
				1 RPM indic	ator (Pro	p RPM)
				1 running ti	me mete	r
				1 oil pressu	re gauge	
				1 oil tempe	rature ga	uge
				1 cylinder h gauge (MÄI	ead temp VI 36-450	erature or coolant temperature installed)
				1 fuel quan	tity gauge	2
				1 manifold	pressure	gauge
				1 fuel press	ure contr	ol light
				1 ammeter		
				1 4-point ha	arness for	each seat
				1 temperat	ure contr	ol light (EGT, airbox)
				1 generator	warning	light
				1 TCU contr	ol light	
				1 TCU warn	ing light	
4.	Dimensi	ions		Span	16.33	m (incl. Winglet)
				Length	7.28 m	1
				Height	1.78 m	1
-	F			Wing Area	15.3 m	1 ²
5.	Engine			~ ~ ~ ~ ~		
		ModelRotax 914 F3	or Rotax	914 F4		
		Type Certificate	EASA.E	.122	(5	
		Limitations		Max take-off	(5 min)	5800 RPM / 38.4 inHg or 39.0 inHg max. 39.9 inHg
				Max continuo	ous	5500 RPM / 34.0 inHg or 34.9 inHg max. 35.4 inHg
C	Data di			see N	lote 11	
ь.	Propelle	er Nastal				
		Model		mt-Propeller	WIV-21-	A-C-F/CF 1/5-05



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	Type Certificate	EASA.P.101			
	Number of blades	2			
	Diameter	1750 mm			
	Sense of Rotation	CW			
	Settings	Low pitch setti	ng:	16.5°+0.2°	
		High nitch sett	ing:	28°+0 2°	
				28 10.2	
		Starting locks s	etting:	19°±1°	
		Feathered Pitc	h:	83°±1°	
_	F 1	Ctrwts. At Low	Pitch:	32.5°±1°	
7.	Fuel capacities				
	Standard Fuel Tank	Total	55 litei	rs	
		Usable	54 litei	r	
	Optional	Total	79 litei	rs	
		Usable	77 litei	r	
8.	Launching Hooks	N/A			
9.	Weak Links	N/A			
10.	Load Factors	see AFM			
11.	Air Speeds	Design Manoe	uvring S	peed V _A :	176 km/h
		Maximum roug	gh air sp	eed V _{RA} :	210 km/h
		Never exceed s	speed V _r	NE:	261 km/h
		Air Brake in Mi	ddle Sto	op V _{abf} :	150 km/h
12.	Approved Operations Capability	VFR Day, see N	lote4	-	
13.	Launch methods	Self-launch			
14.	Maximum Masses	Take-off			770 kg
		Maximum mas	s of non	lifting parts	590 kg
		For S/N	N 36.511	and 36.517 and	l subsequent
					610 kg
15.	Centre of Gravity Range	Forward limit		318 mm behin	d Datum
		Rear limit		430 mm behin	d Datum
16.	Datum	wing leading e	dge at Y	= 0.6 m	
17.	Levelling Means	wedge 1000 : 5	52.5 hor	izontal on fusela	ge tube
18.	Control Surface Deflections	see AMM			
19.	Minimum Flight Crew	1 (Pilot)			
20.	Maximum Passenger Seating Capacity	2			
21.	Baggage/ Cargo Compartments	Behind Seats	12 kg		
22.	Lifetime limitations	see AMM			



F.IV Operating and Service Instructions

1.	Flight Manual	Airplane Flight Manual, HK 36 TTS,
		Doc. No. 3.01.15-E, ACG approved,
		issued 03-Mar-2017
2.	Maintenance Manual (incl. A	Nirworthiness Limitations)
		Airplane Maintenance Manual, HK 36 "SUPER
		DIMONA", Doc. No. 3.02.21 or Doc. No. 3.02.04
		(German Version), see Note 5
		Service Informations and Service Bulletins

F.V Notes

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations in accordance with the maintenance manual are permitted.
- 3. Certification valid for S/Ns 36.393 and 36.511 and subsequent (see Note 10), excluding S/Ns 36.713, 36.717, 36.719, 36.725 and 36.729.
- 4. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted.
- 5. The HK 36 Series AMM Doc. No. 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E, which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 6. The engine Rotax 914 F has to be modified in accordance with Rotax SB 914-01, ACG approved, with Propeller Governor WOODWARD A210790 or Rotax SB 914-09, ACG approved, with Propeller Governor McCauley DCFU290D17B/T2.
- 7. The use of the type HK 36 TTS as a towing airplane in accordance with DAI SB 40, latest effective issue, is permitted.
- 8. The installation and use of a differential braking system, in accordance with DAI SB 42, latest effective issue, is permitted.
- 9. The installation of a tow rope retraction unit in accordance with DAI SB 61 in conjunction with DAI SB 40, use as a tow-plane, is permitted.
- 10. S/N 36.393 has deviations, in accordance with DAI Doc. No. 3.07.201, Chapter 2. In addition, Supplement No. 4 to the Airplane Flight Manual, ACG approved, must be followed.
- 11. Use of different engine TCU-versions, in accordance with the DAI SB 66, is permitted.

SECTION G: HK 36 TTC

G.I General

1. Type/ Model/ Variant

	Туре:	H 36 "DIMONA"
	Model:	НК 36 ТТС
	Variant:	-
2.	Airworthiness Category	Utility
3.	Manufacturer	Diamond Aircraft Industries GmbH
		N.A. Otto-Str. 5
		2700 Wiener Neustadt
		Austria
4.	EASA Type Certification Application	Date 07-May-1996
5.	State of Design Authority	Initial: Austria
c	Ctate of Design Authority Type Carti	Casta Data

6. State of Design Authority Type Certificate Date

20-Dec-1996

The EASA Type Certificate replaces the Austrian Type Certificate SF 3/82.

7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

G.II EASA Certification Basis

1. Reference Date for determining the applicable requirements

2.	Airworthiness Requirements	JAR-22, Change 5
		CRI A-1 HK36TTC and HK36TTS
3.	Special Conditions	CRI E-1 "Propeller feathering control"
		CRI O-1 "Use as a Tow Plane"
		CRI O-2 "Tow Cable Retraction Mechanism"
4.	Exemptions	None
5.	(Reserved) Deviations	None
6.	Equivalent Safety Findings	CRI D-1 "Middle air brake stop"
		CRI E-2 "Propeller Type Definition"
		CRI G-1 "Engine Operating Limitation"
7.	Environmental Protection	Zivilluftfahrzeug-Lärmzulässigkeitverordnung
		BGBI. 738/1993



G.III Technical Characteristics and Operational Limitations

 2. Description Single engine, two-seated cantilever low wing airplane, GFRP-construction, T-tail, side by side seating configuration, fixed two-legged tri cycle landing gear, air brakes on upper wing surface 3. Equipment Minimum Equipment: 1 airspeed indicator (range up to 300 km/h) 1 altimeter with mbar barometric dial 1 magnetic compass with deviation table 1 RPM indicator 1 running time meter 1 oil pressure gauge 1 oil temperature gauge 	rature		
3. Equipment 3. Equipment 3. Equipment 1 airspeed indicator (range up to 300 km/h) 1 altimeter with mbar barometric dial 1 magnetic compass with deviation table 1 RPM indicator 1 running time meter 1 oil pressure gauge 1 oil temperature gauge	rature		
 Equipment Minimum Equipment: 1 airspeed indicator (range up to 300 km/h) 1 altimeter with mbar barometric dial 1 magnetic compass with deviation table 1 RPM indicator 1 running time meter 1 oil pressure gauge 1 oil temperature gauge 1 culinder head temperature or coelect temper 	rature		
1 altimeter with mbar barometric dial 1 magnetic compass with deviation table 1 RPM indicator 1 running time meter 1 oil pressure gauge 1 oil temperature gauge 1 culinder head temperature or coolect temper	rature		
1 magnetic compass with deviation table 1 RPM indicator 1 running time meter 1 oil pressure gauge 1 oil temperature gauge 1 cylinder head temperature or coelect temper	rature		
1 RPM indicator 1 running time meter 1 oil pressure gauge 1 oil temperature gauge 1 cylinder head temperature or coelent temper	rature		
1 running time meter 1 oil pressure gauge 1 oil temperature gauge	rature		
1 oil pressure gauge 1 oil temperature gauge 1 culinder bood temperature er sociant temper	rature		
1 oil temperature gauge	rature		
1 culindar haad tamparatura ar coalant tampar	rature		
gauge (MÄM 36-450 installed)			
1 fuel quantity gauge			
1 manifold pressure gauge			
1 fuel pressure control light			
1 4-point barness for each seat			
1 temperature control light (EGT, airbox)	1 temperature control light (FGT airbox)		
1 generator warning light	1 generator warning light		
1 TCU control light	1 TCU control light		
1 TCU warning light			
4. Dimensions Span 16.33 m (incl. Winglet)			
Length 7.28 m			
Height 1.78 m			
Wing Area 15.3 m ²			
5. Engine			
Model Rotax 914 F3 or Rotax 914 F4			
Type Certificate EASA.E.122			
Limitations			
Max take-off Power (5 min) 5800 RPM / 38.4 inHg or 39.0 inHg max. 39.9 in	ıHg		
Maximum Continuous Power 5500 RPM / 34.0 inHg or 34.9 inHg max. 35.4 in see Note 9	ıHg		
6. Propeller			
Modelmt-Propeller MTV-21-A-C-F/CF 175-05			
Type Certificate EASA.P.101			
Number of blades 2			
Diameter 1750 mm			
Sense of Rotation CW			
Settings Low pitch setting: 16.5°±0.2°			



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		High pitch sett Starting locks s Feathered Pitc Ctrwts. At Low Gearbox Batio	ing: setting: h: Pitch: 1: 2 428	28°±1° 19°±1° 83°±1° 32.5°±1°	
7.	Fuel capacities				
	Standard Fuel Tank	Total	55 liter	s	
		Usable	54 liter		
	Optional	Total	79 liter	S	
	·	Usable	77 liter		
8.	Launching Hooks	N/A			
9.	Weak Links	, N/A			
10.	Load Factors	see AFM			
11.	Air Speeds	Design Manoeuvring Speed V _A : 176 km			176 km/h
		Maximum rough air speed V _{RA} :		210 km/h.	
		Never exceed s	speed V_N	E:	261 km/h
		Air Brake in Mi	ddle Sto	p V _{abf} :	150 km/h
12.	Approved Operations Capability	VFR Day, see N	lote4		
		Cloud flying not permitted			
		Aerobatic man	oeuvres	not permitted	
13.	Launch methods	Self-launch			
14.	Maximum Masses	Take-off			770 kg
		Maximum mas	s of non	lifting parts	610 kg
15.	Centre of Gravity Range	Forward limit		318 mm behind	d Datum
		Rear limit		430 mm behind	d Datum
16.	Datum	wing leading e	dge at Y	= 0.6 m	
17.	Levelling Means	wedge 1000 : 5	52 horizo	ontal on fuselage	e tube
18.	Control Surface Deflections	see AMM			
19.	Minimum Flight Crew	1 (Pilot)			
20.	Maximum Passenger Seating Capacity	2			
21.	Baggage/ Cargo Compartments	Behind Seats		12 kg	
22.	Lifetime limitations	see AMM			

G.IV Operating and Service Instructions

1.	Flight Manual	Airplane Flight Manual, HK 36 TTC, Doc. No. 3.01.20,
		ACG approved, issued 30-July-1996
2.	Maintenance Manual (incl. Airworthine	ess Limitations)
		Airplane Maintenance Manual, HK 36 "SUPER
		DIMONA", Doc.No. 3.02.21 or Doc. 3.02.04 (German
		Version), see Note 5
		Service Informations and Service Bulletins

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G.V Notes

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations, carried out in accordance with the Maintenance Manual, are permitted.
- 3. Certification is valid for S/Ns 36.518 and subsequent except S/Ns 36.713, 36.717, 36.719, 36.725, 36.729 and 36.735.
- 4. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted.
- 5. The HK 36 Series AMM Doc. No. 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 6. The engine Rotax 914 F has to be modified in accordance with Rotax SB 914-01, ACG approved, with Propeller Governor WOODWARD A210790, or Rotax SB 912-24, ACG approved, with Propeller Governor McCauley DCFU290D17B/T2.
- 7. The installation and use of the type HK 36 TTC as a towing airplane in accordance with DAI SB 40, latest effective issue, is permitted.
- 8. The installation of a tow-rope retraction device, in accordance with DAI SB 61, in conjunction with DAI SB 40; use as a tow-plane, is permitted.
- 9. Use of different engine TCU-versions, in accordance with the DAI SB 66, is permitted.



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SECTION H: HK 36 TTC-ECO

- H.I General
- 1. Type/ Model/ Variant

	Туре:	H 36 "DIMONA"
	Model:	HK 36 TTC-ECO
	Variant:	-
2.	Airworthiness Category	Utility
3.	Manufacturer	Diamond Aircraft Industries GmbH
		N.A. Otto-Str. 5
		2700 Wiener Neustadt
		Austria
4.	EASA Type Certification Application Da	ite 26-Mar-1997
5.	State of Design Authority	Initial: Austria
6.	State of Design Authority Type Certific	ate Date

10-Jun-1998

- The EASA Type Certificate replaces the Austrian Type Certificate SF 3/82
- 7. EASA Type Certification Date 21-Dec-2005 (reissue for EASA)

H.II EASA Certification Basis

1. Reference Date for determining the applicable requirements

2.	Airworthiness Requirements	JAR-22, Change 5, issued 28-Oct-1995
_		JAR-1, Change 5, Issued 15-Jul-1996
3.	Special Conditions	CRI E-1 Propeller Feathering Control
		CRI G-1 Engine Operating Limitation
		CRI O-1 Use as Tow-plane
4.	Exemptions	None
5.	(Reserved) Deviations	None
6.	Equivalent Safety Findings	CRI E-2 Propeller Type Definition
		CRI D-1 Middle Air brake stop
		CRI E-3 Fuel System
7.	Environmental Protection	Zivilluftfahrzeug-Lärmzulässigkeitverordnung
		BGBI. 738/1993

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H.III Technical Characteristics and Operational Limitations

1.	Type De	sign Definition	Drawing List H 1996 including	K 36 T** Doc. 3.08.01 dated 20-Dec- g Design Changes up to 57 and of Design Changes (ÄM) HK 36 T**	
2.	Descript	ion	Single engine, airplane, GFRP seating configu landing gear, a tanks	two-seated cantilever low wing construction, T-tail, side by side uration, fixed two-legged tri cycle ir brakes on upper wing surface, wing	
3.	Equipme	ent	Minimum Equi	ipment:	
			1 airspeed indicator (range up to 300 km/h)		
			1 altimeter w	vith mbar barometric dial	
			1 magnetic c	ompass with deviation table	
			1 RPM indica	tor (Showing engine RPM)	
			1 running tim	ne meter	
			1 oil pressure	e indicator	
			1 oil tempera	ature indicator	
			1 cylinder head temperature or coolant temperature gauge (MÄM 36-450 installed)		
			2 fuel quantity indicators		
			1 "Low Fuel"	caution light	
			1 manifold p	ressure indicator	
			1 fuel pressure warning light		
			1 ammeter		
			1 4-piece har	ness for each seat	
			1 temperatu	re control light (EGT, airbox)	
			1 generator v	warning light	
			1 TCU warnir	ng light	
			1 TCU contro	l light	
4.	Dimensi	ons	Span	16.33 m including Winglet	
			Length	7.28 m	
			Height	1.78 m	
_			Wing Area	15.3 m²	
5.	Engine				
		Model	Rotax 914 F3 c	or Rotax 914 F4	
		Type Certificate	EASA.E.122		
		Limitations May take-off Power (5 min)	5800 RDN1 / 20	A inHg or 39 0 inHg may 20 0 inHg	
		Maximum Continuous Power	5500 RENI / 50	1.0 in Hg or 34.0 in Hg max. 35.5 in Hg	
		Maximum Continuous FOWER	JJUU INFINI / 34	no ming of 54.5 ming max. 55.4 ming	

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C-F/CF 175-05	
16.5°±0.2°	
28°±1°	
19°±1°	
83°±1°	
32.5°±1°	
1: 2.4286	
ter system fuel	
eed V _A :	176 km/h
ed V _{RA} :	210 km/h
:	261 km/h
V _{abf} :	150 km/h
	770 kg
ifting parts	610 kg
318 mm behind	Datum
430 mm benind	Datum
tal on fusalara	tubo
ital oli luselage	lube
30 kg	
"0	
	C-F/CF 175-05 16.5°±0.2° 28°±1° 19°±1° 33°±1° 32.5°±1° 1: 2.4286 ter system fuel eed V_A : ed V_{RA} : : V_{abf} : ifting parts 318 mm behind 0.6 m otal on fuselage 30 kg

H.IV Operating and Service Instructions

1.	Flight Manual	Airplane Flight Manual, HK 36 TTC-ECO,
		Doc. No. 3.01.25, ACG approved, issued 10-July-1998
2.	Maintenance Manual (incl. Airworthiness Limitations)	
		Airplane Maintenance Manual, HK 36 "SUPER
		DIMONA", Doc. No. 3.02.21 or Doc. No.
		3.02.04 (German Version), see Note 5
		Service Informations and Service Bulletins

H.V Notes

- 1. Only industrial manufacturing is permitted.
- 2. All components exposed to direct sunlight, except for areas used for registration markings and warning marks, must basically have a white surface. Deviations, carried out in accordance with the Maintenance Manual, are permitted.
- 3. Certification is eligible for S/Ns 36.581 and subsequent except 36.713, 36.717, 36.719, 36.725 and 36.729.
- 4. Acrobatics, cloud flying, night VFR and intentional spinning are not permitted
- 5. The HK 36 Series AMM Doc. No. 3.02.21 and 3.02.04 replaces the former singular AMM Doc. No. 3.02.01 and 3.02.01E which will be no longer revised. Supplemental supplier manuals which are required for maintenance are listed in the HK 36 Series AMM.
- 6. The engine Rotax 914 F has to be modified in accordance with Rotax SB 914-01, ACG approved, with Propeller Governor WOODWARD A210790, or Rotax SB 912-24, ACG approved, with Propeller Governor McCauley DCFU290D17B/T2 and Rotax TM 914-06 exhaust muffer.
- 7. The installation and use of the type HK 36 TTC-ECO as a towing airplane, in accordance with DAI SB 40, latest revision, is permitted.
- 8. Use of different engine TCU-versions in accordance with DAI SB 66 is permitted.

SECTION I: ADMINISTRATIVE SECTION

I.I Acronyms & Abbreviations

- ACG Austrocontrol GmbH
- AFM Airplane Flight Manual
- AMM Airplane Mainenance Manual
- DAI Diamond Aircraft Industries GmbH
- HOAC Hoffmann Aircraft
- S/N Serial Number
- SB Bervice Bulletin

I.II <u>Type Certificate Holder Record</u>

Before 1996:

Hoffmann Flugzeugbau Friesach Gesellschaft mbH

9322 Hirth/Friesach

Austria

Hoffmann Aircraft Flugzeugproduktion und Entwicklung GmbH, Richard Neutra-Gasse 5 1214 Wien Austria

Hoffmann Aircraft GesmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

HOAC Austria GesmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

Since 1996: Diamond Aircraft Industries GmbH N.A. Otto-Str. 5 2700 Wiener Neustadt Austria

I.III Change Record

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Issue	Date	Changes	TC Issue No. & Date
Issue 01	21-Dec-2005	Transfer fromACG TCDS SF 3/82 issue 15 to the EASA Type	Initial Issue,
		Design	21-Dec-2005
Issue 02	06-Jul-2009	Corrections	Initial Issue,
		B.III.5 engine shall be L2400 Inclusion of EASA engine and	21-Dec-2005
		Propeller TC Numbers, issue Nr for that changes remain	
		unchanged	
Issue 03	24-Aug-2015	C.III.7.1. Propeller designation corrected.MÄM 36-450,	22-Apr-2013
		EASA Project No. 0010037087;C. III.3 to H.III.3: "1 cylinder	
		head temperature or coolant temperature gauge	
		(MÄM36-450 installed)"	
Issue 04	03-Aug-2016	MÄM 36-396, EASA 0010008901B.IV; C.IV; D.IV; E.IV; F.IV;	22-Apr-2013
		G.IV; H.IV AMM document number, applicable manuals	
		included in the AMMB.V.; C.V; D.V; E.V;F.V; G.V; H.V -	
		standard wording in all notes for color and marking	
		limitations	
		Sections renumbered to alphanumeric (A to H), separate	
		section issue dates removed and replaced by TCDS Issue	
		and change record.	
Issue 05	30-MAR-2021	Transfer to new TCDS template, editorial/typo changes	22-Apr-2013
		and corrections.	

-END-

