European Aviation Safety Agency

EASA
TYPE-CERTIFICATE
DATA SHEET

EASA.A.374

ATL series

Type Certificate Holder:
C.E.A.P.R.
1 route de Troyes
21121 DAROIS
FRANCE

Manufacturer:
Avions Pierre Robin
ROBIN S.A.
DAROIS
21121 FONTAINE LES DIJON
FRANCE

For variants:
ATL
ATL “S”
ATL “L”

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SECTION A: ATL

A.I. General

1. a) Type: ATL
   b) Variant: Not applicable

2. Airworthiness Category: Normal Category

3. Type Certificate Holder: C.E.A.P.R.
   1 route de Troyes
   21121 DAROIS
   FRANCE

4. Manufacturer: Avions Pierre Robin
   ROBIN S.A.
   DAROIS
   21121 Fontaine les Dijon

5. (reserved)

6. DGAC Type Certification Date: 15 January 1986

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA Type Certificates replaces DGAC-France Type Certificate no 178

A.II. Certification Basis

1. Reference Date for determining the applicable requirements: December 20, 1984

2. (Reserved)

3. (Reserved)

4. Certification Basis: FAR 23


6. Requirements elected to comply: None

7. EASA Special Conditions: DGAC France CTC23 issue 1 dated 20/12/84.

8. EASA Exemptions: None

9. EASA Equivalent Safety Findings: None


A.III. Technical Characteristics and Operational Limitations

1. (Reserved)

2. Description: Single-engine, two-seat, low-wing airplane, wood and composite construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
   Stall warning system “Safe Flight” n°164 or APR 798800 or other approved must be installed
4. Dimensions:
   - Span .................. 10.25 m (33.6 ft)
   - Height .................. 2.00 m (6.2 ft)
   - Length .................. 6.72 m (22.0 ft)
   - Wing Area ............. 12.00 m² (129.2 ft²)

5. Engines:
   - JPX-4T60/A or JPX-4T60/B
   5.1 Engine Limits:
   - Maximum Continuous Power: 3080 rpm (44.2 kW, 60 HP)

6. Propellers:

<table>
<thead>
<tr>
<th>Manufacturer</th>
<th>Model</th>
<th>∅</th>
<th>Number of blades</th>
<th>Minimum static RPM at sea level</th>
</tr>
</thead>
<tbody>
<tr>
<td>EVRA</td>
<td>EVRA-160-81-11</td>
<td>1.60 m</td>
<td>2</td>
<td>2600 rpm</td>
</tr>
<tr>
<td></td>
<td>EVRA-160-81-11-I</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td></td>
<td>EVRA-160-81-11-I-B</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>MUHLBAUER</td>
<td>MT150L-85-1A</td>
<td>1.50 m</td>
<td>2</td>
<td>2800 rpm</td>
</tr>
</tbody>
</table>

7. Fluids:
   7.1 Fuel: 100LL minimum aviation grade gasoline or automobile leaded fuel Super grade
   7.2 Engine Oil: 20W50

8. Fluid capacities:
   8.1 Fuel: Two structural wing tanks
   - Total capacity: ........................................ 45 liters
   - Usable: .................................................. 1 liter
   - Unusable: .............................................. 1 liter
   8.2 Oil: Total capacity: .................................... 2.5 liters
   - Usable: .................................................. 1 liter

9. Air speeds:
   - $V_{NE}$: 200 km/h (108 knots IAS)
   - $V_{NO}$: 155 km/h (84 knots IAS)
   - $V_A$: 155 km/h (84 knots IAS)
   - $V_{FE}$: 153 km/h (83 knots IAS)
   - $V_{C}$: 155 km/h (84 knots IAS)
   - $V_D$: 217 km/h (117 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.
12. Maximum Mass: Take-off and landing: 580 kg
13. Centre of Gravity Range: Normal category
   - Forward limit (21.5 % ref.): 0.269 m aft of datum at 470 kg
   - Intermediate limit (22 % ref.): 0.275 m aft of datum at 580 kg
   - Aft limit (24 % ref.): 0.300 m aft of datum at 580 kg
14. Datum: Wing leading edge at the wing/fuselage junction panel. Length of the reference chord: 1.25 m

15. Design Limit Load Factors:
   - Flaps up: +3.8, -1.52
   - Flaps down: +2, -0

16. Levelling Means: Fuselage rear upper part (dorsal fin) with 1° setting.

17. Minimum Flight Crew: 1 (pilot) at 0.26m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.26m aft of datum

19. Baggage / Cargo Compartment: Maximum baggage compartment 10 kg at +0.65m aft of datum

20. Wheels and Tires:
   - Main gear track: 3.2 m (10.5 ft)
   - Wheel tire size:
     - Rear: 330 x 130
     - Front: 270 x 100
   - Tire pressure:
     - Rear: 2.2 bar
     - Front: 1.6 bar

21. Control surface movements:
   - Elevator: up 15° ± 2°, down 15° ± 2°
   - Ailerons: up 20° ± 2°, down 15° ± 2°
   - Rudder: L & R 17° ± 2°
   - Wing Flaps: 1st notch: 0° ± 2°, 2nd notch: 35° ± 2°

22. (Reserved)

A.IV. Operating and Service Instructions

Airplane Flight Manual: Refer to latest amendment of service letter n°6
Airplane Maintenance Manual: Refer to latest amendment of service letter n°6

A.V. Note:
SECTION B: ATL “S”

B.I. General

1. a) Type: ATL “S”
   b) Variant: Not applicable

2. Airworthiness Category: Normal Category

3. Type Certificate Holder: C.E.A.P.R.
   1 route de Troyes
   21121 DAROIS
   FRANCE

4. Manufacturer: Avions Pierre Robin
   ROBIN S.A.
   DAROIS
   21121 Fontaine les Dijon

5. (reserved)

6. DGAC Type Certification Date: 17 December 1986

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no 178

B.II. Certification Basis

1. Reference Date for determining the applicable requirements: December 20, 1984

2. (Reserved)

3. (Reserved)


5. Airworthiness Requirements: FAR 23 as amended by amendment 23-1 through 1-28 dated 28 April 1992

6. Requirements elected to comply: None

7. EASA Special Conditions: DGAC France CTC23 issue 1 dated 20/12/84

8. EASA Exemptions: None

9. EASA Equivalent Safety Findings: None


B.III. Technical Characteristics and Operational Limitations

1. (Reserved)

2. Description: Single-engine, two-seat, low-wing airplane, wood and composite construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

   Stall warning system “Safe Flight” n°164 or APR 798800 or other approved must be installed
4. Dimensions:  
   Span .................. 10.25 m (33.6 ft)  
   Height ................ 2.00 m (6.2 ft)  
   Length ............... 6.72 m (22.0 ft)  
   Wing Area .......... 12.00 m² (129.2 ft²)

5. Engines:  
   J PX-4T60/A or J PX-4T60/B  
   5.1 Engine Limits:  
   Maximum Continuous Power:  
   With EVRA propeller .......... 2900 rpm (41.6 kW, 57 HP)  
   With MUHLBAUER propeller 3080 rpm (44.2 kW, 60 HP)

6. Propellers:  
<table>
<thead>
<tr>
<th>Manufacturer</th>
<th>Model</th>
<th>∅</th>
<th>Number of blades</th>
<th>Minimum static RPM at sea level</th>
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<td>MT150L-85-1A</td>
<td>1.50 m</td>
<td>2</td>
<td>2800 rpm</td>
</tr>
</tbody>
</table>

7. Fluids:  
   7.1 Fuel:  
   100LL minimum aviation grade gasoline or automobile leaded fuel Super grade
   7.2 Engine Oil:  
   20W50

8. Fluid capacities:  
   8.1 Fuel:  
   Two structural wing tanks  
   Total capacity: ................................ 45 liters  
   Usable: .............................................. 1 liter
   8.2 Oil:  
   Total capacity: ..................................... 2.5 liters  
   Usable: ................................................. 1 liter

9. Air speeds:  
   \( V_{NE} \) Never Exceed ................... 200 km/h (108 knots IAS)  
   \( V_{NO} \) Maximum Normal Operation 155 km/h (84 knots IAS)  
   \( V_A \) Maneuvering ..................... 155 km/h (84 knots IAS)  
   \( V_{FE} \) Maximum Flap Extended .... 153 km/h (83 knots IAS)  
   \( V_C \) Design cruising .................. 155 km/h (84 knots IAS)  
   \( V_D \) Design dive ...................... 217 km/h (117 knots IAS)

10. Maximum Operating Altitude:  
    Refer to approved aircraft flight manual.

11. Operational Capability:  
    Day VFR in non-icing conditions.

12. Maximum Mass:  
    Take-off and landing: ................................. 580 kg

13. Centre of Gravity Range:  
    Normal category  
    Forward limit (21.5 % ref.): 0.269 m aft of datum at 470 kg  
    Intermediate limit (22 % ref.): 0.275 m aft of datum at 580 kg  
    Aft limit (24 % ref.): 0.300 m aft of datum at 580 kg
14. Datum: 
Wing leading edge at the wing/fuselage junction panel.
Length of the reference chord: 1.25 m

15. Design Limit Load Factors:
   - Flaps up: +3.8, -1.52
   - Flaps down: +2, -0

16. Levelling Means:
Fuselage upper part (dorsal fin) with 1° setting.

17. Minimum Flight Crew: 1 (pilot)

18. Maximum Passenger Seating Capacity: 1 at 0.26 m aft of datum

19. Baggage / Cargo Compartment: 
   Maximum baggage compartment 10 kg at +0.65 m aft of datum

20. Wheels and Tires:
   - Main gear track: 3.2 m (10.5 ft)
   - Wheel tire size:
     - Rear: 330 x 130
     - Front: 270 x 100
   - Tire pressure:
     - Rear: 2.2 bar
     - Front: 1.6 bar

21. Control surface movements:
   - Elevator:
     - Up: 15° ± 2°
     - Down: 15° ± 2°
   - Ailerons:
     - Up: 20° ± 2°
     - Down: 15° ± 2°
   - Rudder:
     - L & R: 17° ± 2°
   - Wing Flaps:
     - 1st notch: 0° ± 2°
     - 2nd notch: 35° ± 2°

22. (Reserved)

B.IV. Operating and Service Instructions

Airplane Flight Manual: Refer to latest amendment of service letter n°6
Airplane Maintenance Manual: Refer to latest amendment of service letter n°6

B.V. Note:
1. This model is identical to the ATL except the maximum continuous power of the engine.
SECTION C: ATL “L”

C.I. General

1. a) Type: ATL “L”
   b) Variant: Not applicable
2. Airworthiness Category: Normal Category
3. Type Certificate Holder: C.E.A.P.R.
   1 route de Troyes
   21121 DAROIS
   FRANCE
4. Manufacturer: Avions Pierre Robin
   ROBIN S.A.
   DAROIS
   21121 Fontaine les Dijon
5. (reserved)
6. DGAC Type Certification Date: 9 June 1989
7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No. 1702/2003
8. The EASA type Certificates replaces DGAC-France Type Certificate no 178

C.II. Certification Basis

1. Reference Date for determining the applicable requirements: December 20, 1984
2. (Reserved)
3. (Reserved)
5. Airworthiness Requirements: FAR 23 as amended by amendment 23-1 through 1-28 dated 28 April 1982
6. Requirements elected to comply: None
7. EASA Special Conditions: DGAC France CTC23 issue 1 dated 20/12/84
8. EASA Exemptions: None
9. EASA Equivalent Safety Findings: None

C.III. Technical Characteristics and Operational Limitations

1. (Reserved)
2. Description: Single-engine, two-seat, low-wing airplane, wood and composite construction, fixed tricycle landing gear.
3. Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
   Stall warning system “Safe Flight” n°164 or APR 798800 or other approved must be installed
4. Dimensions: Span ................. 10.25 m (33.6 ft)
   Height .................. 2.00 m (6.2 ft)
   Length .................. 6.83 m (22.4 ft)
   Wing Area ........... 12.00 m² (129.2 ft²)
5. Engines: Limbach L2000 DA2
   5.1 Engine Limits: Maximum Continuous Power: 3000 rpm (51 kW, 70 HP)

6. Propellers:

<table>
<thead>
<tr>
<th>Manufacturer</th>
<th>Model</th>
<th>( \varnothing )</th>
<th>Number of blades</th>
<th>Minimum static RPM at sea level</th>
</tr>
</thead>
<tbody>
<tr>
<td>MT Propeller</td>
<td>155L80-1-</td>
<td>1.55 m</td>
<td>2</td>
<td>2750 rpm</td>
</tr>
</tbody>
</table>

7. Fluids:
   7.1 Fuel: 100LL minimum aviation grade gasoline or automobile leaded fuel Superior grade

7.2 Engine Oil: Refer to Airplane Flight Manual.

8. Fluid capacities:
   8.1 Fuel: Two structural wing tanks
             Total capacity: 45 liters
             Usable: 1 liter
             Optional two structural wing tanks
             Total capacity: 70 liters
             Usable: 2 liters

8.2 Oil: Refer to Airplane Flight Manual

9. Air speeds:
   \( V_{\text{NE}} \) ............................................ 200 km/h (108 knots IAS)
   \( V_{\text{NO}} \) ............................................ 155 km/h (84 knots IAS)
   \( V_{\text{A}} \) ............................................ 155 km/h (84 knots IAS)
   \( V_{\text{FE}} \) ............................................ 153 km/h (83 knots IAS)
   \( V_{\text{C}} \) ............................................ 155 km/h (84 knots IAS)
   \( V_{\text{D}} \) ............................................ 217 km/h (117 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.
12. Maximum Mass: Take-off and landing: 580 kg
13. Centre of Gravity Range: Normal category
    Forward limit (20.5 % ref.): 0.256 m aft of datum at 530 kg
    Intermediate limit (21 % ref.): 0.263 m aft of datum at 580 kg
    Aft limit (24 % ref.): 0.300 m aft of datum at 580 kg

14. Datum: Wing leading edge at the wing/fuselage junction panel.
Length of the reference chord: 1.25 m
15. Design Limit Load Factors:
   Flaps up +3.8
   - 1.52
   Flaps down +2
   -0

16. Levelling Means: Fuselage upper part (dorsal fin) with 1° setting.

17. Minimum Flight Crew: 1 (pilot) at 0.26m aft of datum.

18. Maximum Passenger Seating Capacity: 1 at 0.26m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment 10 kg at +0.65 m aft of datum

20. Wheels and Tires Main gear track .............................................. 3.2 m (10.5 ft)
    Wheel tire size
    rear: ............................................. 330 x 130
    front: ........................................... 270 x 100
    Tire pressure
    rear: .......................................... 2.2 bar
    front: .......................................... 1.6 bar

21. Control surface movements:
    Elevator: ........................................... up 15° ± 2°
    down 15° ± 2°
    Ailerons: ........................................... up 20° ± 2°
    down 15° ± 2°
    Rudder: ........................................... L & R 17° ± 2°
    Wing Flaps: 1st notch ............................................. 0° ± 2°
                 2nd notch...................................... 35° ± 2°

22. (Reserved)

C.IV. Operating and Service Instructions

Airplane Flight Manual........ Refer to latest amendment of service letter n°6
Airplane Maintenance Manual Refer to latest amendment of service letter n°6

C.V. Note:

1. This model is identical to the ATL except the LIMBACH L2000 DA2 engine.
ADMINISTRATIVE SECTION

I. Acronyms

II. Type Certificate Holder Record

ROBIN S.A.  
Société Avions Robin  
ROBIN Aviation  
APEX Aircraft

III. Change Record

<table>
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<tr>
<th>Issue</th>
<th>Date</th>
<th>Description</th>
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<tbody>
<tr>
<td>Issue 1</td>
<td>10 May 2013</td>
<td>Initial issue on transfer of this Type Certificate to CEAPR</td>
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