RESTRICTED TYPE-CERTIFICATE
DATA SHEET

No. EASA.IM.A.643

for

CL-215

Type Certificate Holder:

Viking Air Limited

1959 de Havilland Way

Sidney, British Columbia V8L 5V5

Canada

Intentionally left blank
TABLE OF CONTENTS

SECTION 1: MODEL CL-215-1A10 ................................................................. 6

I. General .............................................................................................................. 6
   1. Type/Model/Variant ..................................................................................... 6
   2. Performance Class ...................................................................................... 6
   3. Certifying Authority .................................................................................. 6
   4. Manufacturer ............................................................................................ 6
   5. State of Design Authority Certification Application Date ......................... 6
   6. EASA Type Certification Application Date ................................................. 6
   7. State of Design Authority Type Certificate Date ........................................ 6
   8. EASA Type Certification Date ................................................................. 6

II. Certification Basis .......................................................................................... 7
   1. Reference Date for determining the applicable requirements .................. 7
   2. State of Design Airworthiness Authority Type Certification Data Sheet No. 7
   3. State of Design Airworthiness Authority Certification Basis .................... 7
   4. EASA Airworthiness Requirements ......................................................... 8
   5. Special Conditions ................................................................................... 8
   6. Exemptions ............................................................................................... 8
   7. Deviations ............................................................................................... 8
   8. Equivalent Safety Findings ..................................................................... 8
   9. Environmental Protection ....................................................................... 8

III. Technical Characteristics and Operational Limitations ............................... 9
   1. Type Design Definition ............................................................................ 9
   2. Description ............................................................................................... 9
   3. Equipment ............................................................................................... 9
   4. Dimensions ............................................................................................. 9
   5. Engines .................................................................................................. 9
   6. Auxiliary Power Unit .............................................................................. 9
   7. Propellers .............................................................................................. 9
   8. Fluids (Fuel, Oil, Additives, Hydraulics) ................................................. 10
   9. Fluid Capacities .................................................................................... 10
   10. Airspeed Limits ...................................................................................... 11
   11. Flight Envelope ..................................................................................... 12
   12. Operating Limitations ........................................................................... 12
   13. Maximum Certified Masses ................................................................. 13
   14. Centre of Gravity Range ...................................................................... 14
   15. Datum .................................................................................................. 14
   16. Mean Aerodynamic Chord (MAC) ...................................................... 14
   17. Levelling Means ................................................................................... 14
   18. Minimum Flight Crew ......................................................................... 14
   19. Minimum Cabin Crew .......................................................................... 14
   20. Maximum Seating Capacity ................................................................. 14
   21. Baggage/Cargo Compartment .............................................................. 14
   22. Wheels and Tyres ................................................................................. 14
   23. ETOPS ................................................................................................. 15
   Not applicable .......................................................................................... 15

IV. Operating and Service Instructions ............................................................... 15
   1. Airplane Flight Manual (AFM) ................................................................. 15
2. Instructions for Continued Airworthiness and Airworthiness Limitations .......................... 15
3. Weight and Balance Manual (WBM) .................................................................................. 15

V. Notes .................................................................................................................................. 15

SECTION 2: MODEL CL-215-6B11 ...................................................................................... 16

I. General .................................................................................................................................. 16
1. Type/ Model/ Variant ............................................................................................................. 16
2. Performance Class ................................................................................................................ 16
3. Certifying Authority ............................................................................................................ 16
4. Manufacturer ....................................................................................................................... 16
5. State of Design Authority Certification Application Date .................................................. 16
6. EASA Type Certification Application Date ......................................................................... 16
7. State of Design Authority Type Certificate Date ............................................................... 16
8. EASA Type Certification Date ............................................................................................ 16

II. Certification Basis .............................................................................................................. 17
1. Reference Date for determining the applicable requirements .......................................... 17
2. State of Design Airworthiness Authority Type Certification Data Sheet No. ................. 17
3. State of Design Airworthiness Authority Certification Basis ............................................. 17
4. EASA Airworthiness Requirements ................................................................................... 17
5. Special Conditions ............................................................................................................. 18
6. Exemptions ........................................................................................................................ 18
7. Deviations ......................................................................................................................... 18
8. Equivalent Safety Findings ................................................................................................ 18
9. Environmental Protection ................................................................................................ 18

III. Technical Characteristics and Operational Limitations .................................................... 18
1. Type Design Definition ..................................................................................................... 18
2. Description ......................................................................................................................... 19
3. Equipment ......................................................................................................................... 20
4. Dimensions ....................................................................................................................... 20
5. Engines ................................................................................................................................ 20
6. Auxiliary Power Unit ....................................................................................................... 20
7. Propellers ......................................................................................................................... 20
8. Fluids (Fuel, Oil, Additives, Hydraulics) .......................................................................... 21
9. Fluid Capacities ................................................................................................................. 22
10. Airspeed Limits ............................................................................................................... 22
11. Flight Envelope ............................................................................................................... 23
12. Operating Limitations ..................................................................................................... 23
13. Maximum Certified Masses ............................................................................................. 25
14. Centre of Gravity Range .................................................................................................. 26
15. Datum ................................................................................................................................ 26
16. Mean Aerodynamic Chord (MAC) ................................................................................ 26
17. Levelling Means ................................................................................................................ 26
18. Minimum Flight Crew ..................................................................................................... 26
19. Minimum Cabin Crew ...................................................................................................... 26
20. Maximum Seating Capacity ............................................................................................. 26
21. Baggage/ Cargo Compartment ......................................................................................... 26
22. Wheels and Tyres ............................................................................................................. 27
23. ETOPS ............................................................................................................................ 27
Not applicable ........................................................................................................................ 27

IV. Operating and Service Instructions ................................................................................... 27
1. Airplane Flight Manual (AFM) .......................................................................................... 27
2. Instructions for Continued Airworthiness and Airworthiness Limitations ............................... 27
3. Weight and Balance Manual (WBM) ........................................................................................ 27
   Canadair Weight and Balance Reports RAW-215T-xxx, for each individual aeroplane .......... 27
   Canadair Weight and balance Reports RAW-415-xxx, for each individual aeroplane ........... 27

V. Notes ........................................................................................................................................... 28

SECTION: ADMINISTRATIVE ........................................................................................................ 29

I. Acronyms and Abbreviations ............................................................................................... 29

II. Type Certificate Holder Record .......................................................................................... 29

III. Change Record ................................................................................................................... 29
SECTION 1: MODEL CL-215-1A10

I. General
This data sheet, which is part of the Restricted Type Certificate No. EASA.IM.A.643, prescribes conditions and limitations under which the product for which the restricted Type Certificate was issued meets the airworthiness requirements of the European Aviation Safety Agency.

1. Type/ Model/ Variant
   CL-215-1A10 Amphibious Flying Boat

   Aircraft configured such that water or Chemical/Water Mix may be loaded and promptly jettisoned.

   The CL-215-1A10 has been produced in five groups: Serial Numbers 1001 to 1030, 1031 to 1050, 1051 to 1065, 1066 to 1080, and 1081 and subsequent. Data in this TCDS that contains no specific reference to any group of serial numbers applies to all groups. (See NOTE 4).

2. Performance Class
   A

3. Certifying Authority
   TCCA

4. Manufacturer
   Viking Air Limited
   1959 de Havilland Way
   Sidney, British Columbia V8L 5V5
   Canada

5. State of Design Authority Certification Application Date
   12 March 1965

6. EASA Type Certification Application Date
   Unknown

7. State of Design Authority Type Certificate Date
   Model CL-215-1A10 Approved 7 March 1969, by the Canadian Department of Transport

8. EASA Type Certification Date
   9 September 2003, by ENAC
SECTION 1: MODEL CL-215-1A10- continued

II. Certification Basis

[Note: in the case that there is an Explanatory Note to this TCDS, please insert the following text]
“Non-proprietary data contained in selected Special Conditions that are part of the Certification Basis are published in an Explanatory Note to the TCDS with the number: 01. The document is not exhaustive and will be gradually updated. An update of the Explanatory Note will not cause an update of the TCDS.”

1. Reference Date for determining the applicable requirements
15 January 1966

2. State of Design Airworthiness Authority Type Certification Data Sheet No.
Transport Canada A-86

3. State of Design Airworthiness Authority Certification Basis
CL-215-1A10 S/N 1001 to 1030

FAR Part 25, dated February 1, 1965 plus amendment 25-18 dated September 29, 1968 with the deviations recorded in the following documents.


CL-215-1A10 S/N 1031 to 1125*


* For Aircraft S/N 1058, 1059, 1062 and 1063, following note applies.

Aircraft S/N 1058 and 1059 are not eligible for Canadian Certificate of Airworthiness until compliance has been shown with requirements FAR 25.813(a) and 25.815. (Reference DOT Telex LIAE 120 dated 17 June 1978).

Aircraft S/N 1062 and 1063 are not eligible for Canadian Certificate of Airworthiness until compliance has been shown with requirements FAR 25.803, 25.809, 25.813(a) and 25.875(b). (Reference DOT Telex LIAE 10 dated 22 January 1979).
4. EASA Airworthiness Requirements

**CL-215-1A10 S/N from 1001 to 1030**
FAR Part 25, dated February 1, 1965 plus amendment 25-18 dated September 29, 1968 and including the deviations recorded in, the following document:


Compliance with the following requirements has been established:

Ditching provisions of FAR Part 25.801(b) through (e), and 25.807(d). (The requirements of 25.1415(a) through (d) are not applicable, per Report RAO-215-100, Appendix 11, item RU.801).

**CL-215-1A10 S/N from 1031 to 1125**
FAR Part 25, dated February 1, 1965 plus amendment 25-18 dated September 29, 1968 and including the deviations recorded in, the following document:


Compliance with the following requirements has been established:

Ditching provisions of FAR Part 25.801(b) through (e), and 25.807(d). (The requirements of 25.1415(a) through (d) are not applicable, per Report RAO-215-100, Appendix 11, item RU.801).

5. Special Conditions
   See state of origin Airworthiness Authority Certification Basis, Transport Canada TCDS A-86.

6. Exemptions
   See state of origin Airworthiness Authority Certification Basis, Transport Canada TCDS A-86.

7. Deviations
   See state of origin Airworthiness Authority Certification Basis, Transport Canada TCDS A-86.

8. Equivalent Safety Findings
   See state of origin Airworthiness Authority Certification Basis, Transport Canada TCDS A-86.

9. Environmental Protection
   ICAO Annex 16, Volume I. For details, see TCDSN EASA.IM.A.643
SECTION 1: MODEL CL-215-1A10- continued

III. Technical Characteristics and Operational Limitations

1. Type Design Definition


Eligible Serial Number   S/N 1001 to 1125 with S/N 1058, 1059, 1062, 1063 excluded.
Placards

Placards are listed in the following Canadair Drawings:

Approved Publications


D.O.T. Approved Loading Instructions. (See NOTE 1).

2. Description

High wing Amphibious flying boat.

3. Equipment

All Equipment required by the airworthiness requirements as reported in the certification basis must be installed on each delivered aircraft.
In addition it is required to bring on board the approved AFM

4. Dimensions

<table>
<thead>
<tr>
<th>Model CL-215</th>
</tr>
</thead>
<tbody>
<tr>
<td>Span</td>
</tr>
<tr>
<td>Length</td>
</tr>
<tr>
<td>Height</td>
</tr>
<tr>
<td>Wing Area</td>
</tr>
</tbody>
</table>

5. Engines

Two Pratt & Whitney Canada Double Wasp CA3

6. Auxiliary Power Unit

GPU-2: Andover Motors, Model 204

7. Propellers

TE.CERT.00051-001 © European Aviation Safety Agency, 2017. All rights reserved. ISO9001 Certified. Proprietary document. Copies are not controlled. Confirm revision status through the EASA-Internet/Intranet.
SECTION 1: MODEL CL-215-1A10- continued

Manufacturer Hamilton Standard

On aircraft S/N 1001 to 1030 and 1039 to 1125: Intermix in any combination of four types listed below:

Propeller Types 43E60-581 P1
43E60-581 P2
43E60-701
43E60-583 S/N 1001-1125

Blades - Number and Type
Three 6093A-10S (for -581P1 or P2)
or Three 6901S-10 (for -701)
or Three 6903A-10 (for -583) S/N 1001-1125

Diameter Limits Maximum 14 ft. 3 in (4.34 m)
Minimum 13 ft. 11 1/2 in. (4.25 m)

Pitch setting at 72-inch station

For Propeller Type 43E60-581P1, 43E60-581P2, or 43E60-701:
Low pitch stop 9.5 degrees (±0.2 degrees)
Constant speed range 9.5 degrees to 30 degrees
Feathered 81 degrees (±0.5 degrees)

For Propeller Type 43E60-583:
Low pitch stop 9.0 degrees (±0.2 degrees)
Constant speed range 9.0 degrees to 30 degrees
Feathered 81 degrees (±0.5 degrees)

Except for transients, propeller must not be operated in the 1550 to 1750 RPM range.

8. Fluids (Fuel, Oil, Additives, Hydraulics)
Fuel CL-215-1A10
Avgas Grade 100/130 as per CAN 2-3.25-M77 or MIL-G-5572
Avgas Grade 100 LL as per ASTM D910

GPU-2 Same as Aircraft

Oil CL-215-1A10
Engine: All oils reported in PW service Bulletin N. 1183 rev P or subsequent
Auxiliary Unit: MIL-0-6082 grade 1065 or SAE 10W30 Automotive Oil

GPU-2 MIL-0-6082 grade 1065 or SAE 10W30 Automotive Oil

9. Fluid Capacities

Fuel Capacity Usable Fuel
4337 L (1146 US Gal) S/N 1001 to 1030
# SECTION 1: MODEL CL-215-1A10- continued

5914 L (1562 US Gal) S/N 1031 and subs

### Oil Capacity Engines (Each)

<table>
<thead>
<tr>
<th></th>
<th>L</th>
<th>US Gal</th>
</tr>
</thead>
<tbody>
<tr>
<td>Total</td>
<td>136,4</td>
<td>36,0</td>
</tr>
<tr>
<td>Usable*</td>
<td>116,4</td>
<td>30,7</td>
</tr>
</tbody>
</table>

* Excluding propeller feathering reserve of 7,7 L (2 US Gal.) each engine.

### Oil Capacity GPU-2

<table>
<thead>
<tr>
<th></th>
<th>L</th>
<th>Imp gal</th>
<th>US Gal</th>
</tr>
</thead>
<tbody>
<tr>
<td>Total</td>
<td>3,4</td>
<td>0,75</td>
<td>0,9</td>
</tr>
</tbody>
</table>

10. Airspeed Limits

<table>
<thead>
<tr>
<th></th>
<th>IAS (knots)</th>
<th>CAS (knots)</th>
</tr>
</thead>
<tbody>
<tr>
<td>VMO (maximum Operating)</td>
<td></td>
<td></td>
</tr>
<tr>
<td>S.L. to 12,000 ft</td>
<td>188</td>
<td>190</td>
</tr>
<tr>
<td>12,000 feet to 20,000 ft</td>
<td>158</td>
<td>160</td>
</tr>
<tr>
<td>VFE (Flaps Extended at 10°)</td>
<td>138</td>
<td>140</td>
</tr>
<tr>
<td>(Flaps Extended at 25° for land operation and overwater operation)</td>
<td>110</td>
<td>112</td>
</tr>
<tr>
<td>(Flaps Extended at 25°* for water operation)</td>
<td>114</td>
<td>116</td>
</tr>
</tbody>
</table>

* Applicable to A/C 1056 and subsequent

**VA (Maneuvering Speed)**

<table>
<thead>
<tr>
<th></th>
<th>IAS (knots)</th>
<th>CAS (knots)</th>
</tr>
</thead>
<tbody>
<tr>
<td>For MTOW = 16329 kg (36000 lb) water operation only</td>
<td>126</td>
<td>127</td>
</tr>
<tr>
<td>For MTOW = 17100 kg (37700 lb) water operation only**</td>
<td>128</td>
<td>129</td>
</tr>
<tr>
<td>For MTOW = 19731 kg (43500 lb) land operation only</td>
<td>149</td>
<td>150</td>
</tr>
<tr>
<td>For MTOW = 19731 kg (43500 lb) Lift off and flap extended at 15° water operation only</td>
<td>131</td>
<td>133</td>
</tr>
</tbody>
</table>

**Refers to Serial Numbers 1003, 1007, 1008, 1009, 1012, 1017, 1018, 1020, 1031 and subsequent, and aircraft Serial Number 1001 through 1030 fitted with additional buoyancy compartment in accordance with Canadair Service Bulletin Number CL215-124.

<table>
<thead>
<tr>
<th></th>
<th>IAS (knots)</th>
<th>CAS (knots)</th>
</tr>
</thead>
<tbody>
<tr>
<td>VLE (Maximum Speed-Landing Gear Extended)</td>
<td>129</td>
<td>130</td>
</tr>
<tr>
<td>VLO (Maximum Speed-Landing Gear Operation)</td>
<td>129</td>
<td>130</td>
</tr>
</tbody>
</table>
SECTION 1: MODEL CL-215-1A10- continued

<table>
<thead>
<tr>
<th>Parameter</th>
<th>Value</th>
<th>Value</th>
</tr>
</thead>
<tbody>
<tr>
<td>VMCA (Minimum Control Speed in the air with Automatic Propeller Feathering Operative)</td>
<td>86</td>
<td>85</td>
</tr>
<tr>
<td>VLL (Maximum speed at which Landing lights may be extended or used)***</td>
<td>129</td>
<td>130</td>
</tr>
<tr>
<td>VWD (Maximum Speed at which water doors may be opened or operated in flight)</td>
<td>129</td>
<td>130</td>
</tr>
<tr>
<td>Maximum speed on Water with Probes extended (water speed) ****</td>
<td>80</td>
<td>90</td>
</tr>
</tbody>
</table>

*** not applicable to A/C 1081 and subsequent

**** Refers to aircraft S/N 1051 and subsequent and aircraft which embody Service bulletin CL215-203.

See AFM as listed in Approved Publications.

11. Flight Envelope
   See Approved Airplane Flight Manual

12. Operating Limitations

12.1 Approved Operations

<table>
<thead>
<tr>
<th>Maximum Operating Altitude (Pressure Altitude)</th>
<th>Take-off and Landing: 8000 feet</th>
<th>Enroute: 20000 feet</th>
<th>Water Pick-up: 5000 feet</th>
</tr>
</thead>
</table>
| (*)Refers to aircraft S/N 1051 and subsequent and aircraft which embody Canadair Service bulletin CL215-203.

Control Surface Movements

Controls to be rigged in accordance with the following Canadair Drawings:
(a) P215-90014 Diagram Aileron Controls.
(b) P215-90015 Diagram Elevator Controls.
(c) P215-90016 Diagram Rudder Controls.

Engine limitation: see approved AFM

12.2 Other Limitations

Dispatch into known icing conditions is prohibited
SECTION 1: MODEL CL-215-1A10- continued

13. Maximum Certified Masses

<table>
<thead>
<tr>
<th></th>
<th>CL 215-1A010</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>Kg</td>
</tr>
<tr>
<td>Ramp (Land Operation)</td>
<td>19731</td>
</tr>
<tr>
<td>Ramp (Water Operation)</td>
<td>16465</td>
</tr>
<tr>
<td>Take off (Land Operation)</td>
<td>17236*</td>
</tr>
<tr>
<td>Take off (Water Operation)</td>
<td>19731</td>
</tr>
<tr>
<td>Landing (Land Operation)</td>
<td>16329</td>
</tr>
<tr>
<td>Landing (Water Operation)</td>
<td>17100</td>
</tr>
<tr>
<td>Zero fuel (Land Operation)</td>
<td>15604</td>
</tr>
<tr>
<td>Zero fuel (Water Operation)</td>
<td>16783</td>
</tr>
<tr>
<td></td>
<td>16103</td>
</tr>
<tr>
<td></td>
<td>16783</td>
</tr>
<tr>
<td>Zero fuel</td>
<td>18597</td>
</tr>
<tr>
<td></td>
<td>18597</td>
</tr>
</tbody>
</table>

[a] See AFM, maximum weights vary with serial numbers and modifications.

For water Bomber configuration, including Chemical Foam

<table>
<thead>
<tr>
<th></th>
<th>Kg</th>
<th>Lbs</th>
</tr>
</thead>
<tbody>
<tr>
<td>Touch down for water pick-up</td>
<td>15195</td>
<td>33500</td>
</tr>
<tr>
<td></td>
<td>16103</td>
<td>35500<em>4</em></td>
</tr>
<tr>
<td>Lift off following water pick-up</td>
<td>19731</td>
<td>43500</td>
</tr>
</tbody>
</table>

[b] See AFM, maximum weights vary with serial numbers and modifications.

*1* Refers to Serial Numbers 1003, 1007, 1008, 1009, 1012, 1017, 1018, 1020, 1031 and subsequent, and aircraft Serial Number 1001 through 1030 fitted with additional buoyancy compartment in accordance with Canadair Service Bulletin Number CL215-124.

*2* Refers to S/N 1056 to 1125 which incorporate Canadair Service Bulletin CL215-376

*3* Refers to aircraft 1056 and subsequent.

*4* Refers to aircraft Serial Numbers 1051 and subsequent, and aircraft which incorporate New Probe System to Canadair Service Bulletin CL215-203.
SECTION 1: MODEL CL-215-1A10- continued

14. Centre of Gravity Range
   See approved Airplane Flight Manual

15. Datum
   The reference datum is located 300 inches (762 cm) forward of the keyhole slot in the chine angle on both sides of the fuselage at station 300.0.

16. Mean Aerodynamic Chord (MAC)
   The leading edge of the MAC is 366.57 inches (931.08 cm) aft of the reference datum. The length of the MAC is 139.4 inches (354.07 cm).

17. Levelling Means
   Longitudinal: Lugs on left hand nose wheel well sidewall at stations 170.0 and 182.5.
   Lateral: Lugs on front face of nose wheel well rear bulkhead, station 222.50.

18. Minimum Flight Crew
   2 (Pilot and Co-Pilot)

19. Minimum Cabin Crew
   (in accordance with the emergency evacuation test)
   0

20. Maximum Seating Capacity
   10 (including two crew) see Note 4

21. Baggage/ Cargo Compartment
   General Cargo
   Load distribution must not exceed 150 lb./sq. ft. (732.36 kg/m²), nor 500 lbs. (226.8 kg) per running foot.
   For compartment limitations, refer to the following Canadair Reports:
   RAW-215-110 for S/N 1001 to 1030.
   RAW-215-146 for S/N 1031 to 1055.
   RAW-215-145 for SAR Aircraft S/N 1031 to 1038.
   RAW-215-190 for S/N 1056 and subsequent.

   Jettisonable Liquid Cargo:
   Water Bomber Configuration: Two tanks at Station 403.8, maximum water load of 6000 lb. (2722 kg) each. Volume 706 gallons (2673 L) each.

   Additional Cargo Limitations
   Carriage of general cargo in the cabin is prohibited when the water or chemical concentrate or chemical/water mix tanks are in use.
   (b) Landing with jettisonable water or chemical/water load in the tanks is prohibited.

22. Wheels and Tyres
SECTION 1: MODEL CL-215-1A10- continued

Nose Wheel Tyre: 6.50x10 Type III 10 ply,
Main Wheel Tyre: 15.00x16 Type III 16 ply,

23. ETOPS
    Not applicable

IV. Operating and Service Instructions

1. Airplane Flight Manual (AFM)
   for S/N 1001 to 1030 and Publication No. 291 for S/N 1031 and subsequent.

2. Instructions for Continued Airworthiness and Airworthiness Limitations
   No. 295.

3. Weight and Balance Manual (WBM)
   Canadair Weight and Balance Reports RAW-215-xxx, for each individual aeroplane

V. Notes

   See notes under section 2
SECTION 2: MODEL CL-215-6B11

1. General

1. Type/ Model/ Variant
   Model CL-215-6B11 (CL-215T Variant) (Restricted Category)
   The data in this Type Approval apply to Aircraft model CL-215-1A10 retrofitted with the
   Canadair Modification Kit described in Report RAD-215T-103 (Kit Specification for the
   Retrofit of CL-215-piston aircraft with turboprop engines). After embodiment of the kit,
   affected A/C shall be redesignated as model CL-215-6B11 (CL-215T Variant).

   Model CL-215-6B11 (CL-415T Variant) (Restricted Category)
   Turboprop engines since initial built
   Restricted Category Aircraft configured such that water or Chemical/Water Mix may be
   loaded and promptly jettisoned.

2. Performance Class
   A

3. Certifying Authority
   TCCA

4. Manufacturer
   Viking Air Limited
   1959 de Havilland Way
   Sidney, British Columbia V8L 5V5
   Canada

5. State of Design Authority Certification Application Date
   Unknown

6. EASA Type Certification Application Date
   March 16, 1994 for CL-215-6B11 (CL-215T and CL-415 variant) to ENAC

7. State of Design Authority Type Certificate Date
   Model CL-215-6B11 (CL-215T Variant), Restricted Category, Approved March 28, 1991 by the
   Canadian Department of Transport (DOT)

   Model CL-215-6B11 (CL-415T Variant), Approved June 24, 1994 by the Canadian Department
   of Transport (DOT)

8. EASA Type Certification Date
SECTION 2: MODEL CL-215-6B11- continued

II. Certification Basis

[Note: in the case that there is an Explanatory Note to this TCDS, please insert the following text]
“Non-proprietary data contained in selected Special Conditions that are part of the Certification Basis are published in an Explanatory Note to the TCDS with the number: 01. The document is not exhaustive and will be gradually updated. An update of the Explanatory Note will not cause an update of the TCDS.”

1. Reference Date for determining the applicable requirements
   Unknown

2. State of Design Airworthiness Authority Type Certification Data Sheet No.
   TCCA A-86

3. State of Design Airworthiness Authority Certification Basis

   Compliance with the following requirements has been established:

   Ditching provisions of FAR Part 25.801(b) through (e), and 25.807(d). (The requirements of 25.1415(a) through (d) are not applicable, per Report RAO-215-100, Appendix II, item RU.801).

   CL-215-6B11 (CL-415 Variant)
   Transport Canada Special Conditions (Airworthiness)
   a) SCA 93-4 High Intensity Radiated Fields (HIRF)
   b) SCA 93-5 Lightning Protection

4. EASA Airworthiness Requirements

   CL-215-6B11 (CL-215T Variant), and CL-215-6B11 (CL-415 Variant)
   FAR Part 25, dated February 1, 1965 plus amendment 25-18 dated September 29, 1968 and selected requirements of FAR Part 25 including amendments 25-1 through 25-61 and of Airworthiness Manual Chapters 525 and 516, as specified in, and including the deviations recorded in, the following document:


   Canadair Report RAO-215-100, Revision I-1, for CL-215-6B11 (CL-215T Variant) with S/N 1081 to 1090, 1092 to 1108, 1110 to 1112, 1123.

   Compliance with the following requirements has been established:

   SFAR 27-2, Environmental Protection Agency Final Venting and Exhaust Emission
SECTION 2: MODEL CL-215-6B11- continued

Requirements For Turbine Powered Aircraft.

Noise requirements of FAR Part 36 with Amendments 36-1 through 36-17, change 22 Appendices A, B, and C and ICAO Annex 16, Chapter 3.

Findings of Equivalent Safety:
a) FAR Part 25.901(b)(1)(i) Installation
b) FAR Part 25.1045(e) Cooling Test Procedures

Ditching provisions of FAR Part 25.801(b) through (e), and 25.807(d). (The requirements of 25.1415(a) through (d) are not applicable, per Report RAO-215-100, Appendix II, item RU.801).

CL-215-6B11 (CL-415 Variant)
Transport Canada Special Conditions (Airworthiness)
SCA 93-4 High Intensity Radiated Fields (HIRF)
SCA 93-5 Lighting Protection

5. Special Conditions

6. Exemptions

7. Deviations

8. Equivalent Safety Findings

9. Environmental Protection
   ICAO Annex 16, Volume I. For details, see TCDSN EASA.IM.A.643

III. Technical Characteristics and Operational Limitations

1. Type Design Definition
   CL-215-6B11 (CL-215T Variant) Restricted Category

   D.O.T.Approved Drawing List, Canadair Report No. RAL-215-xxx (for each individual aircraft) (CL-215-1A10) in addition to the following Canadair Modification Summaries
   (to achieve model CL-215-6B11):
SECTION 2: MODEL CL-215-6B11- continued

Eligible Serial Number  S/N 1056 to 1125 with retrofitted turboprop engines
Placards  Substitution and/or removal of existing placards on CL-215-1A10 in addition to new placards, as specified in Canadair Report MBS-215T-111 required to achieve model CL-215-6B11 aircraft (refer to Approved Publications).


D.O.T. Approved Drawing List, Canadair Report No. RAL-215-xxx (for each individual aircraft) (CL-215-1A10) in addition to the following Canadair Modification Summaries (to achieve model CL-215-6B11):

D.O.T. Approved Loading Instructions (see NOTE 1).

CL-415 Variant

Eligible Serial Number  Serial Number 2001 to 2999
Placards  Required placards are specified in Canadair report MBS-215T-105


D.O.T. Approved Airworthiness limitations, scheduled inspections and maintenance intervals sections of Product Support Publication No. 495.


D.O.T. Approved Loading Instructions (see NOTE 1).

2. Description
SECTION 2: MODEL CL-215-6B11- continued

High wing Amphibious flying boat.

3. Equipment

   All Equipment required by the airworthiness requirements as reported in the certification basis must be installed on each delivered aircraft.
   In addition it is required to bring on board the approved AFM

4. Dimensions

<table>
<thead>
<tr>
<th>CL-415 Variant</th>
</tr>
</thead>
<tbody>
<tr>
<td>Span</td>
</tr>
<tr>
<td>Length</td>
</tr>
<tr>
<td>Height</td>
</tr>
<tr>
<td>Wing Area</td>
</tr>
</tbody>
</table>

5. Engines

   Model CL-215-6B11 (CL-215T and CL-415 variants) Restricted Category

   Two Pratt & Whitney Canada
   PW123AF (Turboprop) with P&WC SB 21211 incorporated. (see Note 5)

6. Auxiliary Power Unit

   Not Applicable

7. Propellers

   CL-215T Variant

   Manufacturer: Hamilton Standard
   Propeller Type: Two 14SF-17 (four-bladed) or Two 14SF-19 (four-bladed)
   Diameter: 13 feet, 1/4 inch (3.97 m)
   Pitch setting at 42-inch station (1.06 m):
      Reverse: -10° ± 1.17°
      Feathered: 86.0°

   CL-415 Variant

   Manufacturer: Hamilton Standard
   Propeller Type: Two 14SF-17 (four-bladed) or Two 14SF-19 (four-bladed)
   Diameter: 13 feet, 1/4 inch (3.97 m)
SECTION 2: MODEL CL-215-6B11- continued

Pitch setting at 42-inch station (1.06 m):

Reverse: -10° ± 1.17° (dynamic conditions)
-13.6° to -15.6° (reverse pitch stop position)

Feathered: 86.0°

8. Fluids (Fuel, Oil, Additives, Hydraulics)

CL-215-6B11 (CL-215T and CL-415 variants) Restricted Category

Fuels conforming to any of the following specifications are approved for use.

Mixing of fuels is permitted.

<table>
<thead>
<tr>
<th>Tipo</th>
<th>Specifiches</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>CANADA</td>
</tr>
<tr>
<td></td>
<td>U.S.A.</td>
</tr>
<tr>
<td></td>
<td>U.K.</td>
</tr>
<tr>
<td>Kerosene</td>
<td></td>
</tr>
<tr>
<td>Jet A, A1</td>
<td>3-GP-23</td>
</tr>
<tr>
<td>JP8</td>
<td>ASTM D 1655</td>
</tr>
<tr>
<td></td>
<td>MIL-T-83133</td>
</tr>
<tr>
<td></td>
<td>D. Eng RD2494</td>
</tr>
<tr>
<td></td>
<td></td>
</tr>
<tr>
<td>Wide Cut*</td>
<td>CGSB 3.22</td>
</tr>
<tr>
<td>Jet B</td>
<td>ASTM D 1655</td>
</tr>
<tr>
<td>JP4</td>
<td>MIL-T-5624</td>
</tr>
<tr>
<td></td>
<td>D. Eng RD2486</td>
</tr>
<tr>
<td>High Flash</td>
<td>3-GP-24</td>
</tr>
<tr>
<td>JP5</td>
<td>MIL-T-5624</td>
</tr>
<tr>
<td></td>
<td>D. Eng RD2452</td>
</tr>
</tbody>
</table>


Oil

CL-215-6B11 (CL-215T and CL-415 variants)

Engine: All MIL-L-23699, type II oils and Castrol 4000
SECTION 2: MODEL CL-215-6B11- continued

9. Fluid Capacities


<table>
<thead>
<tr>
<th>Fuel Capacity</th>
<th>Usable Fuel</th>
</tr>
</thead>
<tbody>
<tr>
<td>Pressure refueling:</td>
<td>5796 L (1530 US Gal) (Total Capacity)</td>
</tr>
<tr>
<td>Gravity refueling:</td>
<td>5914 L (1562 US Gal) (Total Capacity)</td>
</tr>
<tr>
<td>Maximum refuel pressure:</td>
<td>50 psig.</td>
</tr>
</tbody>
</table>

Oil Capacity Engines: (Each)

<table>
<thead>
<tr>
<th>L</th>
<th>US Gal</th>
</tr>
</thead>
<tbody>
<tr>
<td>Total</td>
<td>19,3</td>
</tr>
<tr>
<td>Usable</td>
<td>3,5</td>
</tr>
</tbody>
</table>

10. Airspeed Limits

**CL-215-6B11 (CL-215T and CL-415 variants) Restricted Category**

<table>
<thead>
<tr>
<th></th>
<th>IAS (knots)</th>
<th>CAS (knots)</th>
</tr>
</thead>
<tbody>
<tr>
<td>VMO (maximum Operating) S.L. to 20,000 feet</td>
<td>187</td>
<td>190</td>
</tr>
<tr>
<td>VFE (Flaps Extended at 10°)</td>
<td>138</td>
<td>140</td>
</tr>
<tr>
<td>(Flaps Extended at 15°)</td>
<td>138</td>
<td>140</td>
</tr>
<tr>
<td>(Flaps Extended at 25°)</td>
<td>114</td>
<td>116</td>
</tr>
<tr>
<td>(Flaps Extended at 25** )</td>
<td>116</td>
<td>117</td>
</tr>
<tr>
<td>(Flaps Extended at 25*** )</td>
<td>119</td>
<td>120</td>
</tr>
</tbody>
</table>

* For CL-215-6B11 (CL-215T Variant) Refer to A/C S/N 1056 and subsequent incorporating Canadair SB 215-376

** For CL-215-6B11 (CL-415 Variant).

VA (Maneuvering Speed)
See Flight Manual for variation of VA with aircraft weight.

| VLE (Maximum Speed-Landing Gear Extended) | 129 | 130 |
| VLO (Maximum Speed-Landing Gear Operation) | 129 | 130 |
| VMCA (Minimum Control Speed in the air with Automatic Propeller Feathering Operative) | 84 | 84 |
| VLL (Maximum speed at which Landing lights may be extended or used)*** | 129 | 130 |

*** not applicable to A/C S/N 1081 and subsequent

| VWD (Maximum Speed at which) | 129 | 130 |
### SECTION 2: MODEL CL-215-6B11- continued

<table>
<thead>
<tr>
<th>Water doors may be opened or operated in flight</th>
<th></th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum speed on Water with Probes extended (water speed)</td>
<td>80 ****90</td>
</tr>
</tbody>
</table>

**** Refers to aircraft S/N 1051 and subsequent and aircraft which embody Service Bulletin CL215-203.

See AFM as listed in Approved Publications.

11. Flight Envelope
   See Approved Airplane Flight Manual

12. Operating Limitations

12.1 Approved Operations
   CL-215-6B11 (CL-215T and CL-415 variants)

<table>
<thead>
<tr>
<th>Maximum Operating Altitude (Pressure Altitude)</th>
<th>Take-off and Landing: 10000 feet</th>
<th>Enroute: 20000 feet</th>
<th>Water Pick-up: 8000 feet*</th>
</tr>
</thead>
</table>

* A/C incorporating Canadair Service Bulletin CL215-203

#### Engine Limitations

**Model CL-215-6B11 (CL-215T and CL-415 variants) Restricted Category**

**CL-215T Variant**

<table>
<thead>
<tr>
<th>CL-215T Variant</th>
<th>Torque (1)</th>
<th>Np (2)</th>
<th>ITT (ºC)</th>
<th>NH (%)</th>
<th>Oil Temp (ºC)</th>
<th>Oil Press (psig)</th>
<th>NL (%)</th>
</tr>
</thead>
<tbody>
<tr>
<td>Take off</td>
<td>100 (5)</td>
<td>1200 (5)</td>
<td>800</td>
<td>102.7</td>
<td>0 to 115</td>
<td>55 to 65</td>
<td>104.0</td>
</tr>
<tr>
<td>Max Continuous</td>
<td>90 (5)</td>
<td>1200 (5)</td>
<td>800</td>
<td>102.7</td>
<td>0 to 115</td>
<td>55 to 65</td>
<td>104.0</td>
</tr>
<tr>
<td>Min in flight</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>(4)</td>
<td>-40 to 115</td>
<td>55 to 65</td>
<td>N/A</td>
</tr>
<tr>
<td>Min on ground</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>-40 to 115</td>
<td>55 to 65</td>
<td>N/A</td>
</tr>
<tr>
<td>Max reserve</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>0 to 115</td>
<td>55 to 65</td>
<td>N/A</td>
</tr>
<tr>
<td>Transient</td>
<td>115.2 (8)</td>
<td>1320 (20 sec)</td>
<td>840 (20 sec)</td>
<td>103.7 (20 sec)</td>
<td>125 (20 min)</td>
<td>200</td>
<td>104.3 (20 sec)</td>
</tr>
<tr>
<td>Start-up</td>
<td>N/A</td>
<td>N/A</td>
<td>950 (5 sec)</td>
<td>STARTER OFF BY 45.8</td>
<td>-40</td>
<td>40 to 200</td>
<td>N/A</td>
</tr>
</tbody>
</table>

**NOTE:**

1. Maximum torque with feathered propeller is 50%
   Torque meter gives a flickering value with control lever in IDLE/FEATHER position

TE.CERT.00051-001 © European Aviation Safety Agency, 2017. All rights reserved. ISO9001 Certified. Page 23 of 29
Proprietary document. Copies are not controlled. Confirm revision status through the EASA-Internet/Intranet.
### SECTION 2: MODEL CL-215-6B11- continued

2. Continuous operation with Np between 500 and 780 is prohibited
3. For correct operation of the de-ice air intake system oil temperature shall be kept above 45 °C.
   Oil temp must be above 0 °C for engine operation modes higher than FLIGHT-IDLE
4. The use of HIGH CAM is restricted to ground operation.
5. Take off and MCP performances are limited by OAT and altitude (See Performance section, Engine Torque limits of the applicable AFM)
6. Operational Rating that are limited to 5 minutes in normal operation can be extended to 10 minutes in OEI condition. Considering the low occurrence of the OEI operation, specific limitation or special inspection are not required.
7. Maximum torque is 96% for Np equal or below 1125 RPM
8. On airplanes incorporating Canadair Service Bulletin 215-3100: the transient torque limit is 122.4% (20 sec)

For further limitations refer to the approved AFM

**CL-415 Variant**

<table>
<thead>
<tr>
<th>CL-415 Variant</th>
<th>Torque (1)</th>
<th>Np (2)</th>
<th>ITT (°C)</th>
<th>NH (%)</th>
<th>Oil Temp (°C)</th>
<th>Oil Press (psig)</th>
<th>NL (%)</th>
</tr>
</thead>
<tbody>
<tr>
<td>Take off</td>
<td>100 (5)</td>
<td>1210 (5)</td>
<td>800</td>
<td>102.7</td>
<td>0 to 115</td>
<td>55 to 65</td>
<td>104.0</td>
</tr>
<tr>
<td>Max Continuous</td>
<td>90 (5)</td>
<td>1200 (5)</td>
<td>800</td>
<td>102.7</td>
<td>0 to 115</td>
<td>55 to 65</td>
<td>104.0</td>
</tr>
<tr>
<td>Min in flight</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>(4)</td>
<td>-40 to 115</td>
<td>55 to 65</td>
<td>N/A</td>
</tr>
<tr>
<td>Min on ground</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>N/A</td>
<td>-40 to 115</td>
<td>55 to 65</td>
<td>N/A</td>
</tr>
<tr>
<td>Max reserve</td>
<td>N/A</td>
<td>1115</td>
<td>800</td>
<td>N/A</td>
<td>0 to 115</td>
<td>55 to 65</td>
<td>N/A</td>
</tr>
<tr>
<td>Transient</td>
<td>122 (20 sec)</td>
<td>1320</td>
<td>840 (20 sec)</td>
<td>103.7 (20 sec)</td>
<td>125 (20 min)</td>
<td>100</td>
<td>104.0 (20 sec)</td>
</tr>
<tr>
<td>Start-up</td>
<td>N/A</td>
<td>N/A</td>
<td>950 (5 sec)</td>
<td>STARTER OFF BY 48.0</td>
<td>-40</td>
<td>40 to100</td>
<td>N/A</td>
</tr>
</tbody>
</table>

**NOTE:**
1. Maximum torque with feathered propeller is 50%
   Torque meter gives a flickering value with control lever in IDLE/FEATHER position
2. Continuous operation with Np between 500 and 780 is prohibited
3. For correct operation of the de-ice air intake system oil temperature shall be kept above 45 °C.
   Oil temp must be above 0 °C for engine operation modes higher than FLIGHT-IDLE
4. The use of HIGH CAM is restricted to ground operation.
5. Take off and MCP performances are limited by OAT and altitude (See Performance section, Engine Torque limits of the applicable AFM)
SECTION 2: MODEL CL-215-6B11- continued

6. Operational Rating that are limited to 5 minutes in normal operation can be extended to 10 minutes in OEI condition. Considering the low occurrence of the OEI operation, specific limitation or special inspection are not required.

7. Maximum torque is 96% for Np equal or below 1125 RPM

For further limitations refer to the approved AFM

12.2 Other Limitations
Dispatch into known icing conditions is prohibited

13. Maximum Certified Masses

<table>
<thead>
<tr>
<th></th>
<th>CL 215 T Variant</th>
<th>CL 415 T Variant</th>
</tr>
</thead>
<tbody>
<tr>
<td>Ramp (Land Operation)</td>
<td>Kg (a)</td>
<td>Kg (a)</td>
</tr>
<tr>
<td>Ramp (Water Operation)</td>
<td>Kg (a)</td>
<td>Kg (a)</td>
</tr>
<tr>
<td>Take off (Land Operation)</td>
<td>19731 Kg</td>
<td>19890 Kg</td>
</tr>
<tr>
<td></td>
<td>43500 Lbs</td>
<td>43850 Lbs</td>
</tr>
<tr>
<td>Take-off (Water Operation)</td>
<td>17168 Kg</td>
<td>17168 Kg</td>
</tr>
<tr>
<td></td>
<td>37850 Lbs</td>
<td>37850 Lbs</td>
</tr>
<tr>
<td>Landing (Land Operation)</td>
<td>Kg (a)</td>
<td>Kg (a)</td>
</tr>
<tr>
<td>Landing (Water Operation)</td>
<td>Kg (a)</td>
<td>Kg (a)</td>
</tr>
<tr>
<td>Zero fuel f (Land Operation)</td>
<td>Kg (a)</td>
<td>Kg (a)</td>
</tr>
<tr>
<td>Zero fuel (Water Operation)</td>
<td>Kg (a)</td>
<td>Kg (a)</td>
</tr>
</tbody>
</table>

(a) See AFM, maximum weights vary with serial numbers and modifications.

For water Bomber configuration, including Chemical Foam

<table>
<thead>
<tr>
<th></th>
<th>CL 215 T Variant</th>
<th>CL 415 T Variant</th>
</tr>
</thead>
<tbody>
<tr>
<td>Touch down for water pick-up</td>
<td>Kg (b)</td>
<td>Kg (b)</td>
</tr>
<tr>
<td>Lift off following water pick-up</td>
<td>Kg (b)</td>
<td>Kg (b)</td>
</tr>
</tbody>
</table>

(b) See AFM, maximum weights vary with serial numbers and modifications.
SECTION 2: MODEL CL-215-6B11- continued

14. Centre of Gravity Range
   See approved Airplane Flight Manual

15. Datum
   The reference datum is located 300 inches (762 cm) forward of the keyhole slot in the chine angle on both sides of the fuselage at station 300.0.

16. Mean Aerodynamic Chord (MAC)
   The leading edge of the MAC is 366.57 inches (931.08 cm) aft of the reference datum.
   The length of the MAC is 139.4 inches (354.07 cm).

17. Levelling Means
   Longitudinal: Lugs on left hand nose wheel well sidewall at stations 170.0 and 182.5.
   Lateral: Lugs on front face of nose wheel well rear bulkhead, station 222.50.

18. Minimum Flight Crew
   2 (Pilot and Co-pilot)

19. Minimum Cabin Crew
   0

20. Maximum Seating Capacity
   Model CL-215-6B11 (CL-215T and CL-415 variants)
   10 (including two crew). Limited by approved seating arrangements. In particular, see NOTE 4.

21. Baggage/ Cargo Compartment
   Model CL-215-6B11 (CL-215T)

   General Cargo
   Load distribution must not exceed 150 lb./sq.ft. (732.36 kg/m2), nor 500 lbs. (226.8 kg) per running foot. For compartment limitations, refer to the following Canadair Report: RAW-215T-102

   Jettisonable Liquid Cargo:
   Water Bomber Configuration: Two tanks at Station 403.8, maximum water load of 6000 lb. (2722 kg) each. Volume 706 gallons (2673 L) each.

   Additional Cargo Limitations
   (a) Carriage of general cargo in the cabin is prohibited when the water or chemical concentrate or chemical/water mix tanks are in use.
   (b) Landing with jettisonable water or chemical/water load in the tanks is prohibited.

   Model CL-215-6B11(CL-415 variants)

   General Cargo
   Load distribution must not exceed 150 lb./sq. ft. (732.36 kg/m2), nor 500 lbs. (226.8 kg) per running foot. For compartment limitations, refer to the following Canadair Report: RAW-415-102
SECTION 2: MODEL CL-215-6B11- continued

Jettisonable Liquid Cargo:

Water Bomber Configuration: water tank capacity

<table>
<thead>
<tr>
<th>Volume</th>
<th>Weight</th>
</tr>
</thead>
<tbody>
<tr>
<td>L</td>
<td>Imp Gal</td>
</tr>
<tr>
<td>Inboard tanks (2)</td>
<td>3173</td>
</tr>
<tr>
<td>Outboard tanks (2)</td>
<td>2964</td>
</tr>
<tr>
<td>All tanks (4)</td>
<td>6137</td>
</tr>
</tbody>
</table>

Additional Cargo Limitations
(a) Carriage of general cargo in the cabin is prohibited when the water or chemical concentrate or chemical/water mix tanks are in use.
(b) Landing with jettisonable water or chemical/water load in the tanks is prohibited.

22. Wheels and Tyres
- Nose Wheel Tyre: 6.50*10 Type III 10 ply,
- Main Wheel Tyre: 15.00*16 Type II 16 ply,

23. ETOPS
- Not applicable

IV. Operating and Service Instructions

1. Airplane Flight Manual (AFM)
   - CL-215T Variant
   - CL-415 Variant

2. Instructions for Continued Airworthiness and Airworthiness Limitations
   - CL-215T Variant
   - CL-415 Variant
     D.O.T. Approved Airworthiness limitations, scheduled inspections and maintenance intervals sections of Product Support Publication No. 495.

3. Weight and Balance Manual (WBM)
   - Canadair Weight and Balance Reports RAW-215T-xxx, for each individual aeroplane
   - Canadair Weight and balance Reports RAW-415-xxx, for each individual aeroplane
SECTION 2: MODEL CL-215-6B11- continued

V. Notes

Note 1: The current Weight and Balance Report, containing the list of equipment included in the approved empty weight and loading instructions, must be provided for each aircraft.

Note 2: All required placards must be installed in the specified locations.

Note 3: The aircraft must be operated in accordance with all sections of the Approved Flight Manual as listed in the Approved Publications.

Note 4 Carriage of Persons

The carriage of persons in the cabin of Restricted Category Aircraft is only permitted when:
   i. Such persons are Cargo Handlers or persons employed in support of the operation; and
   ii. The water tanks are not in use.

For CL-215T and CL-415
Usage of Mirabel Aero Service Inc. Model 430 Flight Engineer Seat is permitted only by personnel supporting firefighting operations.

Note 5 Every CL-415 manufactured and every CL215T conversion after June 1994 must have PW123AF engines with SB 21211 incorporated or later superseding SB. For CL215T converted before June 1994 if SB 21211 is not incorporated on both engines then both engines must have SB 21113 and the aircraft must have SB 215-A3030 until both engines incorporate SB 21211 or later superseding SB.

Note 6 Document RAO-215-100 describes the evolution of the certification basis with the following issues:
   RAO-215-100, issue NC, dated 12 March 1965, with revision letter C, Add. 1
   RAO-215-100, issue 2, revision H, dated September 1991, with supplement 1
   RAO-215-100 issue 2, revision I, dated June 1994, with supplement 2
SECTION: ADMINISTRATIVE

I. Acronyms and Abbreviations

<table>
<thead>
<tr>
<th>Acronym</th>
<th>Definition</th>
</tr>
</thead>
<tbody>
<tr>
<td>AFM</td>
<td>Airplane Flight Manual</td>
</tr>
<tr>
<td>APU</td>
<td>Auxiliary Power Unit</td>
</tr>
<tr>
<td>APR</td>
<td>Automatic Performance Reserve</td>
</tr>
<tr>
<td>AWO</td>
<td>All Weather Operation</td>
</tr>
<tr>
<td>CRI</td>
<td>Certification Review Item</td>
</tr>
<tr>
<td>CS</td>
<td>Certification Specification</td>
</tr>
<tr>
<td>EASA</td>
<td>European Aviation Safety Agency</td>
</tr>
<tr>
<td>ENAC</td>
<td>Ente Nazionale per l’Aviazione Civile</td>
</tr>
<tr>
<td>ESF</td>
<td>Equivalent Safety Finding</td>
</tr>
<tr>
<td>FAA</td>
<td>Federal Aviation Administration</td>
</tr>
<tr>
<td>GPU</td>
<td>Ground Power Unit</td>
</tr>
<tr>
<td>ICAO</td>
<td>International Civil Aviation Organization</td>
</tr>
<tr>
<td>JAR</td>
<td>Joint Aviation Requirement</td>
</tr>
<tr>
<td>MMEL</td>
<td>Master Minimum Equipment List</td>
</tr>
<tr>
<td>MEL</td>
<td>Minimum Equipment List</td>
</tr>
<tr>
<td>NPA</td>
<td>Notice of Proposed Amendment</td>
</tr>
<tr>
<td>OSD</td>
<td>Operational Suitability Data</td>
</tr>
<tr>
<td>INT/POL</td>
<td>JAA Interim Policy</td>
</tr>
<tr>
<td>RVSM</td>
<td>Reduced Vertical Separation Minima</td>
</tr>
<tr>
<td>SB</td>
<td>Service Bulletin</td>
</tr>
<tr>
<td>SC</td>
<td>Special Condition</td>
</tr>
<tr>
<td>S/N</td>
<td>Serial Number</td>
</tr>
<tr>
<td>TCCA</td>
<td>Transport Canada Civil Aviation</td>
</tr>
<tr>
<td>TCDS</td>
<td>Type Certificate Data Sheet</td>
</tr>
<tr>
<td>TCDSN</td>
<td>Type Certificate Data Sheet for Noise</td>
</tr>
</tbody>
</table>

II. Type Certificate Holder Record

Prior to 1st October 2016: Bombardier Inc.

III. Change Record

<table>
<thead>
<tr>
<th>Issue</th>
<th>Date</th>
<th>Changes</th>
<th>TC Issue</th>
</tr>
</thead>
<tbody>
<tr>
<td>Issue 01</td>
<td>Date: 29 June 2017</td>
<td>Initial Issue</td>
<td>Initial Issue, Date: 29 June 2017</td>
</tr>
</tbody>
</table>

-END-