



TYPE-CERTIFICATE DATA SHEET

Do 28 Series

Type Certificate Holder:
General Atomics AeroTec Systems GmbH
Claude-Dornier-Strasse 1
D-82234 Wessling
Germany

For Variants: Do 28 A-1
Do 28 A-1(R)
Do 28 B-1
Do 28 D
Do 28 D-1
Do 28 D-2
Do 28 D-6
Dornier 128-6

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General for all Variants

1. Data Sheet No.: TCDS A.360

2. Type: **Do 28**

3. Type Certificate Holder: Dornier Luftfahrt GmbH
LBA Approved Design Organisation
Certificate No.: LBA.JA.002
D-82230 Wessling
Federal Republic of Germany

*01 June 2000 –
27 July 2003:* Fairchild Dornier GmbH
LBA DOA Certificate No.: LBA.JA.002
D-82230 Wessling
Federal Republic of Germany

*28 July 2003 –
14 March 2021:* RUAG Aerospace Services GmbH
DOA Certificate No: EASA.21J.038
Oberpfaffenhofen Airfield
P.O. Box 1253
D-82231 Wessling
Federal Republic of Germany

Since 15 March 2021: [General Atomics AeroTec Systems GmbH](#)
[DOA Certificate No: EASA.21J.038](#)
[Claude-Dornier-Strasse 1](#)
[D-82234 Wessling](#)
[Federal Republic of Germany](#)

4. Manufacturer: Dornier Luftfahrt GmbH
LBA Approved Production Organisation
Certificate No.: LBA.G.002
D-82230 Wessling
Federal Republic of Germany

*01 June 2000 –
30 June 2003* Fairchild Dornier GmbH
LBA Approved Production Organisation
Certificate No.: LBA.G.002
D-82230 Wessling
Federal Republic of Germany

*27 August 2004 –
15 March 2021:* RUAG Aerospace Services GmbH
Oberpfaffenhofen Airfield
POA Certificate Holder No.: DE.21G.0176
P.O. Box 1253
D-82231 Wessling
Federal Republic of Germany

Since 16 March 2021: [General Atomics AeroTec Systems GmbH](#)
[POA Certificate Holder No.: DE.21G.0176](#)
[Claude-Dornier-Strasse 1](#)
[D-82234 Wessling](#)
[Federal Republic of Germany](#)

SECTION A: Variant Do 28 A-1

A.I. General

- | | |
|---|-------------|
| 1. a) Type | Do 28 |
| b) Variant | Do 28 A-1 |
| 2. Airworthiness Category: | Normal |
| 3. Certification Application Date: | --- |
| 4. LBA Certification Date: | 8 July 1960 |
| 5. The EASA TCDS is based on the LBA TCDS No. 613/SA for Do28 A-1 (at Issue 11, dated 29 July 2005) | |

A.II Certification Basis

- | | |
|--|---------------|
| 1. Reference Date for determining the applicable requirements: | -- |
| 2. (reserved) | |
| 3. (reserved) | |
| 4. Airworthiness Requirements: | |
| - CAR 3 dated 15 May 1956, including Amendment 3-1 through 3-5 | |
| 5. Requirements elected to comply: | None |
| 6. EASA Special Conditions: | None |
| 7. EASA Exemptions: | None |
| 8. EASA Equivalent Safety Findings: | None |
| 9. EASA Environmental Standards: | ICAO Annex 16 |

A.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents KZ-28.01 and KZ-28.02.

2. Description:

Twin engine, high wing monoplane made of metal; engine installed on stubbed wing cowlings; fixed main landing gear; tail wheel.

3. Equipment:

Minimum equipment according to airworthiness requirements CAR 3; including stall warning system.

Approved installation kits and special equipment

Denomination	Document
a) rotary seat	UZ 28.803
b) worktable	UZ 28.804
c) camera Wild RC 8	UZ 28.807
d) camera Zeiss RMK 15/23	UZ 28.808
e) Oxygen System	UZ 28.809
f) forward seat bench (three seats)	UZ 28.1.24
g) rear seat bench (three seats)	UZ 28.1.25
h) stretcher supports	UZ 27.116-20
i) Stollenwerk-handbarrow light metal	120.001
j) Stollenwerk-handbarrow steel	110.001

Note:

c) and d): only in combination with installed bottom flap according to UZ 28.1.9 and UZ 28.1.13

i) and k): only in combination with Stollenwerk tie-down belt according to Drawing No. 125.001

4. Dimensions:

Wing Span	15,55 m (51,0 ft)
Length	11,71 m (38,4 ft)
Height	4,01 m (13,2 ft)

5. Engines:

Number	2
Manufacturer	Lycoming
Type	O-540-A1D
LBA TCDS	4561/EN

Operating Limits:

Takeoff power, sea level, ISA	250 HP(186 KW) at 2575 RPM
Continuous power, sea level, ISA	250 HP(186 KW) at 2575 RPM
Max. rotary speed	2575 RPM

6. Propellers:

Number	2
Manufacturer	Hartzell
Type	HC-A2XK-2
LBA TCDS	32.130/006/PR
Blades:	2
Nominal Diameter:	2083 mm (82 inch)

7. Fluids (Fuel/Oil/Additives):

Fuel	Aircraft fuel, min. grade 91/96 Octane (alternative) grade 100/130 Octane
Oil	
below -12°C:	SAE 20
-12°C up to +4°C:	SAE 30
above +4°C:	SAE 50

8. Fluid capacities

Fuel:	2 twin tanks:	2 x 110 ltr = 220 ltr
	and	2 x 50 ltr = 100 ltr
	2 secondary tanks	2 x 72 ltr = 144 ltr
Lubricant:		2 x 11,4 ltr = 22,8 ltr (remaining quantity 2,5 ltr)

9. Air Speeds:

VNE (maximum speed)	177 KCAS (328 km/h)
VNO (maximum cruise speed)	139 KCAS (257 km/h)
VA (maneuvering speed)	102 KCAS (189 km/h)
Minimum air speed for different flaps position and power settings:	
a) idling power:	
flap: 0°	45 KCAS (83 km/h)
flap: 15° up to 45°	43 KCAS (80 km/h)
a) max. continuous power:	
flap: 0°	39 KCAS (73 km/h)
flap: 15° up to 45°	37 KCAS (69 km/h)

10. Maximum Operating Altitude: not defined

11. Maximum Weight:
- Max Takeoff weight 2450 kg (5401 lb)

12. Centre of Gravity Range:
forward c.g. limit: 3265 mm
rear c.g. limit: 3493 mm

13. Datum:
Reference plane: 3000 mm in front of slat leading edge
Aircraft position: Horizontal reference plane
(refer to Operating Manual)

14. Minimum Flight Crew: 1 (Pilot)

15. Occupants
Maximum Number: 6 to 8 (Cockpit 2)
(Cabin forward seat bench: 2 to 3*)
(Cabin rearward seat bench: 2 to 3*)
* three seats, if seat bench UZ 28.1.24/25 installed.
16. Baggage / Cargo Compartments
Maximum loading weight 60 kg (132 lb)

A.IV. Operating and Service Instructions

1. Operating Instructions:
- a. Flight Manual Do 28 A-1, dated February 1961
 - b. Placards according to Flight Manual page 5, chapter 1.11
2. Service Instructions:
- a. Operating Manual Do 28 A-1, Description, Operation, Maintenance
 - b. Operator's Manual, Lycoming Model O-540-A Series, Aircraft engines
 - c. Overhaul Manual Lycoming O-360 Series, O-540 Series Aircraft Engines
 - d. Owner's Propeller Manual, Model HC-A2XK-2, Hartzell Propeller, Inc., Piqua, Ohio/USA
 - e. Operator's Manual one-axis heading control FRG 5-3, Bodenseewerke Perkin, Elmer & Co.

A.V. Notes

1. Eligible Serial Numbers
Without limitations
2. Other
- a. Installation of one-axis heading control FRG 5-3 (Equipment No. 30.321/3) according UZ 28.9.9 approved. For this Flight Manual Supplement DO 28 A-1 "Kursregler FRG 5-3" required.
 - b. Usage of snow-skid landing gear UZ 28.802 approved. For this take into consideration Flight Manual Supplement DO 28 A-1 and DO 28 A-1(R), dated 12 October 1961 "Kursregler FRG 5-3".
 - c. Usage of Stollenwerk seat bench:
 - a) made of light metal according to Stollenwerk drawing no. 120.001
 - b) made of steel according to Stollenwerk drawing no. 110.001 approved. For this take into consideration addendum No. 10 to type investigation report Do28-A-1, dated 2 December 1963.
 - d. Approved for "aerotowing of flexible towed parts". For this take into consideration Flight Manual DO 28 A-1, dated 14 July 1964.

SECTION B: Variant Do 28 A-1(R)

B.I. General

- | | |
|--|--------------|
| 1. a) Type | Do 28 |
| b) Variant | Do 28 A-1(R) |
| 2. Airworthiness Category: | Normal |
| 3. Certification Application Date: | --- |
| 4. LBA Certification Date: | 22 July 1960 |
| 5. The EASA TCDS is based on the LBA TCDS No. 613/SA for Do28 A-1(R) (at Issue 11, dated 29 July 2005) | |

B.II Certification Basis

- | | |
|---|---------------|
| 1. Reference Date for determining the applicable requirements: | -- |
| 2. (reserved) | |
| 3. (reserved) | |
| 4. Airworthiness Requirements: | |
| - CAR 3 dated 15 May 1956, additionally Amendment 3-1 through 3-5 including CAR 8 | |
| 5. Requirements elected to comply: | None |
| 6. EASA Special Conditions: | None |
| 7. EASA Exemptions: | None |
| 8. EASA Equivalent Safety Findings: | None |
| 9. EASA Environmental Standards: | ICAO Annex 16 |

B.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents KZ-28.01 and KZ-28.02.

2. Description:

Twin engine, high wing monoplane made of metal; engine installed on stubbed wing cowlings; fixed main landing gear; tail wheel.

3. Equipment:

Minimum equipment according to airworthiness requirements CAR 3; including stall warning system.

Approved installation kits and special equipment:

Denomination	Document
a) rotary seat	UZ 28.803
b) worktable	UZ 28.804
c) camera Wild RC 8	UZ 28.807
d) camera Zeiss RMK 15/23	UZ 28.808
e) Oxygen System	UZ 28.809
f) forward seat bench (three seats)	UZ 28.1.24
g) rear seat bench (three seats)	UZ 28.1.25
h) stretcher supports	UZ 27.116-20
i) Stollenwerk-handbarrow light metal	120.001
j) Stollenwerk-handbarrow steel	110.001

Note:

c) and d): only in combination with installed bottom flap according to UZ 28.1.9 and UZ 28.1.13

i) and k): only in combination with Stollenwerk tie-down belt according to Drawing No. 125.001

Approved installation kits and special equipment:

Denomination	Document
rotary seat	UZ 28.803
worktable	UZ 28.804
additional fuselage fuel tank	UZ 28.806

After installation of the above named installation kits the aircraft must be operated as a "Special Mission aircraft".

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	4,01 m	(13,2 ft)

5. Engines:

Number	2
Manufacturer	Lycoming
Type	O-540-A1D
LBA TCDS	4561/EN

Operating Limits:	
Takeoff power, sea level, ISA	250 HP(216 KW) at 2575 RPM
Continuous power, sea level, ISA	250 HP(216 KW) at 2575 RPM
Max. rotary speed	2575 RPM

6. Propellers:

Number	2
Manufacturer	Hartzell
Type	HC-A2XK-1, -2
LBA TCDS	32.130/006/PR
Blades:	2
Nominal Diameter:	2083 mm (820 inch)

7. Fluids (Fuel/Oil/Additives):

Fuel	Aircraft fuel, min. grade 91/96 Octane (alternative) grade 100/130 Octane
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Oil	
below -12°C:	SAE 20
-17°C up to +21°C:	SAE 30
above +15°C:	SAE 50

8. Fluid capacities

Fuel:	2 twin tank:	2 x 110 ltr = 220 ltr
	and	2 x 50 ltr = 100 ltr
	2 secondary tank	2 x 72 ltr = 144 ltr
	2 additional fuselage tanks	2 x 160 ltr = <u>320 ltr</u>
		totaling <u>784 ltr</u>

Lubricant	2 x 11,4 ltr = 22,8 ltr (remaining quantity 2,5 ltr)
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9. Air Speeds:

Normal Category	
VNE (maximum speed)	177 KCAS (328 km/h)
VNO (maximum cruise speed)	139 KCAS (257 km/h)
VA (maneuvering speed)	102 KCAS (189 km/h)

Special Aircraft	
VNE (maximum speed)	160 KCAS (286 km/h)

10. Maximum Operating Altitude:	not defined
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11. Maximum Weight:	
- Max Takeoff weight	2700 kg (5952 lb)
- Max Landing weight	2500 kg (5512 lb)

12. Centre of Gravity Range:
forward c.g. limit: 3265 mm from 1830 kg up to 2450 Kg
straight line up to: 3292 mm at 2700 kg
rear c.g. limit: 3500 mm
13. Datum:
Reference plane: 3000 mm in front of slat leading edge
Aircraft position: Horizontal reference plane
(refer to Operating Manual)
14. Minimum Flight Crew: 1 (Pilot)
15. Occupants
Maximum Number: 2
16. Baggage / Cargo Compartments
Maximum loading weight 60 kg (132 lb)

B.IV. Operating and Service Instructions

1. Operating Instructions:
- Flight Manual Do 28 A-1, dated June 1961 with Supplement dated June 1961 for operation as special mission aircraft
 - Placards according to Flight Manual Do 28 A-1, page 5, chapter 1.11 and Supplement to Flight Manual (Sonderflugzeuge) page III, chapter 1.11a
2. Service Instructions:
- Operating Manual Do 28 A-1, Description, Operation, Maintenance
 - Operator's Manual, Lycoming Model O-540-A Series, Aircraft engines
 - Overhaul Manual Lycoming O-360 Series, O-540 Series Aircraft Engines
 - Owner's Propeller Manual, Model HC-A2XK-2, Hartzell Propeller, Inc., Piqua, Ohio/USA
 - Operator's Manual one-axis heading control FRG 5-3, Bodenseewerke Perkin, Elmer & Co.

B.V. Notes

1. Eligible Serial Numbers
Without limitations
2. Other
- Installation of one-axis heading control FRG 5-3 (Equipment No. 30.321/3) according UZ 28.9.9 approved. For this Flight Manual Supplement DO 28 A-1 "Kursregler FRG 5-3" required.
 - Usage of snow-skid landing gear UZ 28.802 approved. For this take into consideration Flight Manual Supplement DO 28 A-1 and DO 28 A-1(R), dated 12 October 19661 "Kursregler FRG 5-3".
- c. Usage of Stollenwerk seat bench:
- made of light metal according to Stollenwerk drawing no. 120.001
 - made of steel according to Stollenwerk drawing no. 110.001 approved. For this

take into consideration addendum No. 10 to type investigation report Do28-A-1, dated 2 December 1963.

- d. Approved for "aero towing of flexible towed parts". For this take into consideration Flight Manual DO 28 A-1, dated 14 July 1964.

SECTION C: Variant Do 28 B-1

C.I. General

1. a) Type Do 28
b) Variant Do 28 B-1
2. Airworthiness Category: Normal
3. Certification Application Date: ---
4. LBA Certification Date: 8 July 1960
5. The EASA TCDS is based on the LBA TCDS No. 613/SA for Do28 B-1 (at Issue 11, dated 29 July 2005)

C.II Certification Basis

1. Reference Date for determining the applicable requirements: --
2. (reserved)
3. (reserved)
4. Airworthiness Requirements:
CAR 3 dated 15 May 1956, including Amendment 3-1 through 3-7
5. Requirements elected to comply: None
6. EASA Special Conditions: None
7. EASA Exemptions: None
8. EASA Equivalent Safety Findings: None
9. EASA Environmental Standards: ICAO Annex 16

C.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents KZ-28.01 and KZ-28.02.

2. Description:

Twin engine, high wing monoplane made of metal; engine installed on stubbed wing cowlings; fixed main landing gear; tail wheel.

3. Equipment:

Minimum equipment according to airworthiness requirements CAR 3; including stall warning system with visual and audible indication.

Approved installation kits and special equipment:

- optional installations according to Flight Manual Do 28 B-1

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	4,01 m	(13,2 ft)

Flight control

5. Engines:

Number	2
Manufacturer	Lycoming
Type	IO-540-A1A5
LBA TCDS	4535/EN

Operating Limits:

Takeoff power, sea level, ISA	290 HP(186 KW) at 2575 RPM
Continuous power, sea level, ISA	290 HP(186 KW) at 2575 RPM
Max. rotary speed	2575 RPM

6. Propellers:

Number	2
Manufacturer	Hartzell
Type	HC-A3VK-2
LBA TCDS	32.130/008/PR
Blades:	3 blades variable-pitch
Nominal Diameter:	2083 mm (820 inch)

7. Fluids (Fuel/Oil/Additives):

Fuel	Aircraft fuel, grade 100/130 Octane
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Oil	
below -12°C:	SAE 20
-17°C up to +21°C:	SAE 30
-1° up to +32°C	SAE 40
above +15°C:	SAE 50

8. Fluid capacities

Fuel:	2 triplet tank:	2 x 232 ltr =	464 ltr
	thereof usable fuel		457 ltr
	2 additional tank (optional)	2 x 76 ltr =	152 ltr
	thereof usable fuel		150 ltr
Lubricant:	each engine		11,4 ltr

9. Air Speeds:

VNE (maximum speed)	180 KCAS (333 km/h)
VNO (maximum cruise speed)	145 KCAS (268 km/h)
VA (maneuvering speed)	105 KCAS (194 km/h)
Maximum flap operating speed	95 KCAS (176 km/h)

10. Maximum Operating Altitude: not defined

11. Maximum Weight:
- Max Takeoff weight 2720 kg (5996 lb)

12. Centre of Gravity Range:
forward c.g. limit: 3282 mm from 2300 kg up to
3380 mm at 2720 kg

rear c.g. limit: 3493 mm at all weights

13. Datum:
Reference plane: 3000 mm in front of slat leading edge
Aircraft position: bottom edge Cockpit door entrance horizontally

14. Minimum Flight Crew: 1 (Pilot)

15. Occupants
Maximum Number: 6 to 8 (Cockpit 2)
(Cabin forward seat bench: 2 to 3*)
(Cabin rearward seat bench: 2 to 3*)
* three seats, if seat bench UZ 28.1.24/25 installed.

16. Baggage / Cargo Compartments
Maximum loading weight 60 kg (132 lb)

C.IV. Operating and Service Instructions

1. Operating Instructions:
 - a. Flight Manual Do 28 B1, dated February 1964
 - b. Placards according to Flight Manual page 6 and 6a
2. Service Instructions:
 - a. Operating Manual Do 28 B-1, dated April 1964
 - b. Operator's Manual, Lycoming Model IO-540-B, dated February 1961, with amendment for IO-540-A1A5, dated May 1962
 - c. Overhaul Instruction Handbook (Manual 114) for Hartzell Propeller HC-A3VK-2
 - e. Operating Instructions optional to Special equipment

C.V. Notes

1. Eligible Serial Numbers
Without limitations
2. Other
 - a. Approved for "aerotowing of flexible towed parts". For this take into consideration Appendix to Flight Manual DO 28 B-1 banner towing UZ 28.838, dated March 1966.
 - b. Aircrafts Do28 B-1 Serial Number 3111, 3112 and 3120 are approved as "Special Mission Aircrafts". MTOW is 2900 kg.

Operation of these Aircrafts must be operated in accordance to Appendix Nr. 1 to the Flight Manual EO 28 B-1 Edition 1 April 1971, and Flight Manual DO 28 B-1, dated 15. April 1971

- a. Approved for "aerotowing of flexible towed parts". For this take into consideration Appendix to Flight Manual DO 28 B-1 banner towing UZ 28.838, dated March 1966.

SECTION D: Variant Do 28 D

D.I. General

1. a) Type Do 28
b) Variant Do 28 D
2. Airworthiness Category: Normal
3. Certification Application Date: 29 October 1964
4. LBA Certification Date: 24 February 1967
5. The EASA TCDS is based on the LBA TCDS No. 2031/SA for Do28 D (at Issue 13, dated 19 January 2006)

D.II Certification Basis

1. Reference Date for determining the applicable requirements: --
2. (reserved)
3. (reserved)
4. Airworthiness Requirements:
 - CAR 3 dated 01 March 1963, including Amendment 3-1 through 3-8
5. Requirements elected to comply: None
6. EASA Special Conditions: None
7. EASA Exemptions: None
8. EASA Equivalent Safety Findings: None
9. EASA Environmental Standards: ICAO Annex 16

D.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents KZ-28.01 and KZ-28.02.

2. Description:

Twin engine, high wing monoplane made of metal; engine installed on stubbed wing cowlings; fixed main landing gear; tail wheel.

3. Equipment:

If aircraft certification is to be granted, the required basic equipment resulting from the applicable airworthiness standards (D.II.4) including stall warning system must be installed in every aircraft. The required minimum equipment as well as the equipment approved for optional installation are listed in the following document:
- Equipment list Do 28 D "Skyservant".

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	4,01 m	(13,2 ft)
Total Wing Area	29,0 m ²	(312 ft ²)

5. Engines:

Number	2
Manufacturer	Lycoming
Type	Lycoming IGSO-540-A1E
LBA TCDS	4539/EN
Operating Limits:	

Takeoff power, sea level under standard conditions (5 min.)	380 HP(289 KW) at 3400 RPM 47 inch HG manifold pressure
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Continuous power, sea level under standard conditions:	360 HP(268 KW) at 3200 RPM 45 inch HG manifold pressure
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6. Propellers:

Number	2
Manufacturer	Hartzell
Type	HC-B3W30-2 B/W 10151-8R
LBA TCDS	32.130/10/PR
Nominal Diameter:	2360 mm (929 inch) reduction not permitted climbing range at 760 mm (299 inch) radius: 16° 50' up to 85°

7. Fluids (Fuel/Oil/Additives):

Fuel	Aircraft fuel, grade 100/130 min
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8. Fluid capacities

Fuel:	total capacity: 2 x 455 ltr = 910 ltr
Unusable fuel:	32 liters
Lubricant	2 x 16,5 ltr = 33 ltr.

9. Air Speeds:

VNE (maximum speed)	180 KCAS
VNO (maximum cruise speed)	143 KCAS
VA (maneuvering speed)	115 KCAS
VFE (max. flap extended speed)	102 KCAS

10. Maximum Operating Altitude: 25000 ft

11. Maximum Weight:
- Max Takeoff weight 3500 kg (7716 lb)

12. Centre of Gravity Range:
for 3500 kg: 3310 mm through 3580 mm
for 2700 kg and less: 3200 mm through 3580 mm with linear course
between these values

13. Datum:
Reference plane: 3000 mm in front of slat leading edge
Aircraft position: Cabin floor horizontally

14. Minimum Flight Crew: 1 (Pilot)

15. Maximum Passenger Seating Capacity
Maximum Number 15
13 Passenger and 2 Crew seats

16. Baggage / Cargo Compartments
Forward baggage compartment:
- Maximum loading weight 50 kg (110 lb)
Rear baggage compartment:
- Maximum loading weight 80 kg (176 lb)
(behind cabin door)
See Flight Manual for max. floor load.

D.IV. Operating and Service Instructions

1. Operating Instructions:

- Flight Manual Do 28 D and D-1, with the appendices Weight and Balance Report, Loading Plan and Equipment List.
- Pilot's checklist Do 28 D and D-1
- Operating Manual Do 28 D and D-1
- Placards according to Flight Manual

2. Service Instructions:

- Maintenance Manual Do 28 D and D-1

- b. Parts Catalogue Do 28 D and D-1
- c. Operator's Manual, Lycoming Model IGSO-540-A, Series Aircraft engines
- d. Overhaul Manual, Lycoming Geared and Supercharged models
- e. Owner's Propeller Manual, Model HC-B3W Series, Hartzell Propeller, Inc.

D.V. Notes

- 1. Eligible Serial Numbers
 - All civil Serial Numbers
- 2. Other
 - a. Movements of control surfaces and flaps:

Flaps down:	$20^{\circ} \pm 1^{\circ}$	and	$52^{\circ} \pm 1^{\circ}$
Aileron up:	$20^{\circ} \pm 45'$	down	$25^{\circ} \pm 45'$
Rudder left	$27,5^{\circ} \pm 1^{\circ}$	right	$27,5^{\circ} \pm 1^{\circ}$
Elevator up	$22^{\circ} \pm 30'$	down	$17^{\circ} \pm 30'$
 - b. Aircraft of this type may be equipped according to the instrument flight regulations (IFR) for flights.
 - c. Dropping of parachutists with parachutes with manual and automatic release permitted.
For this purpose appendix no. 11 to Flight Manual Do 28 D and D-1, dated 11 August 1970 must be followed.
 - d. Customized Cabin Interior and Seating Configuration must be approved.

SECTION E: Variant Do 28 D-1

E.I. General

1. a) Type Do 28
b) Variant Do 28 D-1
2. Airworthiness Category: Normal
3. Certification Application Date: 29 October 1964
4. LBA Certification Date: 06 November 1967
5. The EASA TCDS is based on the LBA TCDS No. 2031/SA for Do28 D-1 (at Issue 13, dated 19 January 2006)

E.II Certification Basis

1. Reference Date for determining the applicable requirements: --
2. (reserved)
3. (reserved)
4. Airworthiness Requirements:
 - FAR 3 dated 01 February 1965, including Amendment 23-1 through 23-3
5. Requirements elected to comply: None
6. EASA Special Conditions: None
7. EASA Exemptions: None
8. EASA Equivalent Safety Findings: None
9. EASA Environmental Standards: ICAO Annex 16

E.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents KZ-28.01 and KZ-28.02.

2. Description:

Twin engine, high wing monoplane made of metal; engine installed on stubbed wing cowlings; fixed main landing gear; tail wheel.

3. Equipment:

If aircraft certification is to be granted, the required basic equipment resulting from the applicable airworthiness standards (E.II.4) including stall warning system must be installed in every aircraft. The required minimum equipment as well as the equipment approved for optional installation are listed in the following document:
- Equipment list Do 38 D-1 "Skyservant".

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	4,01 m	(13,2 ft)
Total Wing Area	29,0 m ²	(312 ft ²)

5. Engines:

Number	2
Manufacturer	Lycoming
Type	Lycoming IGSO-540-A1E
LBA TCDS	4539/EN
Operating Limits:	

Takeoff power, sea level under standard conditions (5 min.)	380 HP(289 KW) at 3400 RPM 47 inch HG manifold pressure
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Continuous power, sea level under standard conditions:	360 HP(268 KW) at 3200 RPM 45 inch HG manifold pressure
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6. Propellers:

Number	2
Manufacturer	Hartzell
Type	HC-B3W30-2 B/W 10151-8R
LBA TCDS	32.130/10/PR
Nominal Diameter:	2360 mm (929 inch) reduction not permitted climbing range at 760 mm (299 inch) radius: 16° 50' up to 85°

7. Fluids (Fuel/Oil/Additives):

Fuel	Aircraft fuel, grade 100/130 min
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8. Fluid capacities

Fuel: total capacity: 2 x 415 ltr = 830 ltr
Unusable fuel: 8 liters

Lubricant 2 x 16,5 ltr = 33 ltr.

9. Air Speeds:

VNE (maximum speed) 180 KCAS

VNO (maximum cruise speed) 143 KCAS

VA (maneuvering speed) 115 KCAS

VFE (max. flap extended speed) 104 KCAS

10. Maximum Operating Altitude: 25000 ft

11. Maximum Weight:
- Max Takeoff weight 3650 kg (8047 lb)

12. Centre of Gravity Range:
for 3650 kg: 3310 mm through 3580 mm
for 2750 kg and less: 3200 mm through 3580 mm with linear course
between these values

13. Datum:
Reference plane: 3000 mm in front of slat leading edge
Aircraft position: Cabin floor horizontally

14. Minimum Flight Crew: 1 (Pilot)

15. Maximum Passenger Seating Capacity
Maximum Number 15
13 Passenger and 2 Crew seats

16. Baggage / Cargo Compartments
Forward baggage compartment:
- Maximum loading weight 50 kg (110 lb)
Rear baggage compartment:
- Maximum loading weight 80 kg (176 lb)
(behind cabin door)
See Flight Manual for max. floor load.

E.IV. Operating and Service Instructions

1. Operating Instructions:

- a. Flight Manual Do 28 D and D-1, with the appendices Weight and Balance Report, Loading Plan and Equipment List.
- b. Pilot's checklist Do 28 D and D-1
- c. Operating Manual Do 28 D and D-1
- e. Placards according to Flight Manual

2. Service Instructions:

- a. Maintenance Manual Do 28 D and D-1
- b. Parts Catalogue Do 28 D and D-1
- c. Operator's Manual, Lycoming Model IGSO-540-A, Series Aircraft engines
- d. Overhaul Manual, Lycoming Geared and Supercharged models
- e. Owner's Propeller Manual, Model HC-B3W Series, Hartzell Propeller, Inc.

E.V. Notes

1. Eligible Serial Numbers

All civil Serial Numbers

2. Other

- a. Movements of control surfaces and flaps:

Flaps down:	$20^{\circ} \pm 1^{\circ}$	and	$52^{\circ} \pm 1^{\circ}$
Aileron up:	$20^{\circ} \pm 45'$	down	$25^{\circ} \pm 45'$
Rudder left	$27,5^{\circ} \pm 1^{\circ}$	right	$27,5^{\circ} \pm 1^{\circ}$
Elevator up	$22^{\circ} \pm 30'$	down	$17^{\circ} \pm 30'$
- b. Aircraft of this type may be equipped according to the instrument flight regulations (IFR) for flights.
- c. Dropping of parachutists with parachutes with manual and automatic release permitted.
For this purpose appendix no. 11 to Flight Manual Do 28 D and D-1, dated 11 August 1970 must be observed.
- d. The use of individual aircraft serial numbers for models D-1 with a take-off weight of 3700 kg is permitted. The prerequisites and the aircraft serial numbers affected are listed in appendix no.12 to Flight Manual. Operation of the aircraft is to be carried out according to the information in the Flight Manual, revision dated August 1, 1970 and later.
- e. Customized Cabin Interior and Seating Configuration must be approved.
- f. Operating Instructions and Technical Notes may be obtained [General Atomics AeroTec Systems GmbH, Claude-Dornier-Strasse 1, D-82234 Wessling](#).
- g. The installation of external fuel tanks in model Do 28 D-1 according to drawing lists UZ 28.02.826 is permitted.
Permissible fuel quantity 246 ltr per external tank
Usable fuel quantity 236 ltr per external tank
Corresponding operating instructions for model Do 28 D-1:
Appendix no. 17 flight manual Do28 D and D-1

SECTION F: Variant Do 28 D-2

F.I. General

1. a) Type Do 28
b) Variant Do 28 D-2
2. Airworthiness Category: Normal
3. Certification Application Date: 03 November 1967
4. LBA Certification Date: 30 April 1971
5. The EASA TCDS is based on the LBA TCDS No. 2031/SA for Do28 D-2 (at Issue 13, dated 19 January 2006)

F.II Certification Basis

1. Reference Date for determining the applicable requirements: --
2. (reserved)
3. (reserved)
4. Airworthiness Requirements:
 - FAR 23 dated 01 February 1965, including Amendment 23-1 through 23-6
5. Requirements elected to comply: None
6. EASA Special Conditions: None
7. EASA Exemptions: None
8. EASA Equivalent Safety Findings: None
9. EASA Environmental Standards: ICAO Annex 16

F.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents KZ-28.04.
(refer also to F.V.2 "Other" item g)

2. Description:

Twin engine, high wing monoplane made of metal; engine installed on stubbed wing cowlings; fixed main landing gear; tail wheel.

3. Equipment:

If aircraft certification is to be granted, the required basic equipment resulting from the applicable airworthiness standards (F.II.4) including stall warning system must be installed in every aircraft. The required minimum equipment as well as the equipment approved for optional installation are listed in the following document:
- Equipment list Do 38 D-2 "Skyservant".

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	4,01 m	(13,2 ft)
Total Wing Area	29,0 m ²	(312 ft ²)

5. Engines:

Number	2
Manufacturer	Lycoming
Type	Lycoming IGSO-540-A1E
LBA TCDS	4539/EN

Operating Limits:

Takeoff power, sea level under standard conditions (5 min.)	380 HP(289 KW) at 3400 RPM 47 inch HG manifold pressure
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Continuous power, sea level under standard conditions:	360 HP(268 KW) at 3200 RPM 45 inch HG manifold pressure
--	--

6. Propellers:

Number	2
Manufacturer	Hartzell
Type	HC-B3W30-2 B/W 10151-8R
LBA TCDS	32.130/10/PR
Nominal Diameter:	2360 mm (930 inch) reduction not permitted climbing range at 760 mm (299 inch) radius: 16° 50' up to 85°

7. Fluids (Fuel/Oil/Additives):

Fuel	Aircraft fuel, grade 100/130 min
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Oil 2 x 16,5 ltr = 33

8. Fluid capacities

Fuel: total capacity: 2 x 452 ltr = 904 ltr
Unusable fuel: 10 liters

Lubricant 2 x 16,5 ltr = 33 ltr.

9. Air Speeds:

VNE (maximum speed)	185 KCAS
VNO (maximum cruise speed)	148 KCAS
VA (maneuvering speed)	125 KCAS
VFE (max. flap extended speed)	99 KCAS
VFO (max flap actuating speed)	89 KCAS
VB (max gust intensity speed)	110 KCAS

10. Maximum Operating Altitude: 25000 ft

11. Maximum Weight:

- Max weight taxi /ramp	3862 kg (8514 lb)
- Max Takeoff weight	3842 kg (8470 lb)
- Max Landing weight	3650 kg (8047 lb)

12. Centre of Gravity Range:

for 3842 kg:	3789 mm through 4036 mm
for 2750 kg and less:	3656 mm through 4036 mm with linear course between these values

13. Datum:

Reference plane:	3456 mm in front of slat leading edge
Aircraft position:	Cabin floor horizontally

14. Minimum Flight Crew: 1 (Pilot)

Maximum Passenger Seating Capacity	
Maximum Number	15

13 Passenger and 2 Crew seats

15. Baggage / Cargo Compartments

Maximum loading weight (behind cabin door)	80 kg (176 lb)
See Flight Manual for max. floor load.	

F.IV. Operating and Service Instructions

1. Operating Instructions:

- Flight Manual Do28 D-2, with the appendices Weight and Balance Report, Loading Plan and Standard Equipment List.
- Pilot's checklist Do 28 D-2
- Operating Manual Do 28 D-2

- e. Placards according to Flight Manual
2. Service Instructions:
- a. Maintenance Manual Do 28 D-2 with Time change Item List
 - b. Illustrated Part Breakdown
 - c. Operator's Manual, Lycoming Manual IGSO-540-A, Series Aircraft engines
 - d. Overhaul Manual, Lycoming Geared and Supercharged models
 - e. Owner's Propeller Manual, Model HC-B3W Series, Hartzell Propeller, Inc.

F.V. Notes

1. Eligible Serial Numbers
All civil Serial Numbers except the aircraft serial numbers listed under F.V.2.g.
2. Other
- a. Movements of control surfaces and flaps:

Outer Flaps:	$11,5^{\circ} \pm 1^{\circ}$	and	$33,4^{\circ} \pm 1^{\circ}$
Center and inner Flaps:	$20^{\circ} \pm 1^{\circ}$	and	$52^{\circ} \pm 1^{\circ}$
Aileron up:	$22^{\circ} \pm 45'$	down	$29^{\circ} \pm 45'$
Rudder left	$28^{\circ} \pm 1^{\circ}$	right	$28^{\circ} \pm 1^{\circ}$
Elevator up	$19^{\circ} \pm 30'$	down	$17^{\circ} \pm 30'$
 - b. Aircraft of this type may be equipped according to the instrument flight regulations (IFR) for flights.
 - c. Dropping of parachutists with parachutes with manual and automatic release permitted.
For this purpose appendix no. 4 to Flight Manual Do 28 D-2, dated 22 August 1975 must be observed.
 - d. Customized Cabin Interior and Seating Configuration must be approved.
 - e. Operating Instructions and Technical Notes may be obtained from [General Atomics AeroTec Systems GmbH, Claude-Dornier-Strasse 1, D-82234 Wessling.](#)
 - f. The installation of external fuel tanks in model Do 28 D-2 according to drawing lists UZ 28.04.848 is permitted.
Permissible fuel quantity 246 ltr. per external tank
Usable fuel quantity 236 ltr. per external tank
Corresponding operating instructions for model Do 28 D-2:
Appendix no. 7 Flight Manual Do28 D-2.
 - g. Do28 D-2 "Military Version"
Based on simplified type certification as per §4 LuftGerPO, Do 28 D-2 series aircraft are also approved in accordance with the stipulations of the type certificate no. ML 1510-73 issue 8 of "Musterprüfleitstelle für Luftfahrtgerät der Bundeswehr (ML)" including the associated data sheet.

Affected are the serial nos. 4080, 4083, 4084, 4089, 4090, 4095, 4098, 4103-4105, 4107, 4111-4114, 4121, 4124-4125, 4127-4129, 4134-4137, 4141-4142, 4144, 4146-4149, 4151, 4152, 4153-4157, 4159-4162, 4164, 4167, 4169, 4173-4175, 4177-4180, 4182-4185, 4186, 4187-4188, 4190-4193, 4196, 4199 and 4200.

SECTION G: Variant Do 28 D-6

G.I. General

1. a) Type Do 28
b) Variant Do 28 D-6
2. Airworthiness Category: Normal
3. Certification Application Date: ---
4. LBA Certification Date: 25 February 1981
5. The EASA TCDS is based on the LBA TCDS No. 2031A/SA for Do28 D-6 (at Issue 23, dated 08 April 2005)

G.II Certification Basis

1. Reference Date for determining the applicable requirements: --
2. (reserved)
3. (reserved)
4. Airworthiness Requirements:
 - FAR 23 dated 01 February 1965, including Amendment 23-1 through 23-6
5. Requirements elected to comply: None
6. EASA Special Conditions:

From engine installation affected regulations additional airworthiness requirements according to FAR 23 Amendment 23-23 compliance have been shown:
23.3 to 23.253, 23.361 to 23.363, 23.367, 23.371, 23.507, 23.509, 23.629, 23.771, 23.777, 23.831, 23.863 to 23.999, 23.1011 to 23.1301, 23.1303(a) to (d), 23.1305 to 23.1323, 23.1331 to 23.1367, 23.1437 to 23.1545, 23.1549 to 23.1559, 23.1563 to 23.1589.
7. EASA Exemptions: None
8. EASA Equivalent Safety Findings:

Equivalent Safety Finding according to LBA-Letter I 22-2031a/81 dated 25 February 1981 (concerning single engine emergency power and stall warning)
9. EASA Environmental Standards: ICAO Annex 16

G.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents including later amendments as follows:
Aircraft Serial No. 4302: KZ D02-000-000 A00F.

2. Description:

Landplane for STOL-Operation with two turboprops. Cantilever high-wing aircraft of all-metal construction with external tanks under the wing and fixed main landing gear, tail wheel.

3. Equipment:

Refer to Equipment List of POH Do28 D-6

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	4,01 m	(13,2 ft)
Total Wing Area	29,0 m ²	(312 ft ²)

5. Engines:

Number	2
Manufacturer	Pratt & Whitney Canada
Type	PT6A-110
LBA TCDS	7020/EN

Power rating (Sea level, ISA):	
Takeoff power (5 min), max RPM	298 KW (400 HP)
Continuous power, max. RPM	298 KW (400 HP)

Single engine power, max. RPM	336 KW (450 HP)
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Maximum continuous rotary speed	1900 RPM
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Interstage Turbine Temperature (ITT)

- Takeoff (5 min)	685°C
- Continuous, Emergency	685°C
- Start (2 sec)	1085°C
- Transition(2 sec)	825°C

6. Propellers:

Number:	2
Manufacturer:	Hartzell
Type:	Propeller 1:HC-B3TN-3 D/T 102 82-9.5 Propeller 2:HC-B3TN-3 D/T 102 82 B-9.5
LBA TCDS	32.130/31/PR

Nominal Diameter:	2362 mm ± 0 mm (929 inch)
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Propellers installed at one aircraft must be of the same type.

7. Fluids (Fuel/Oil/Additives):

Max Fuel capacity	1514 ltr.
Usable fuel:	1470 ltr.
Oil	2 x 11,8 ltr = 23,6 ltr

8. Air Speeds:

VMO (maximum operating speed)	155 KCAS
VA (maneuvering speed)	129 KCAS
VFE (max. flap extended speed)	110 KCAS
Max. Tire Speed	105 KCAS
Max crosswind component proven	25 KCAS
VMCA (minimum control speed with critical engine inoperative)	65 KCAS

9. Maximum Operating Altitude: 20000 ft

10. Maximum Weight:

- Max weight taxi /ramp	4080 kg (8995 lb)
- Max Takeoff weight	4050 kg (8929 lb)
- Max Landing weight	3650 kg (8047 lb)
- Empty weight	3515 kg (7749 lb)

11. Centre of Gravity Range:

Refer to Pilot's Operating Handbook DO 28 D-6, Section 2

12. Datum:

Refer to Pilot's Operating Handbook DO 28 D-6, Section 2

13. Minimum Flight Crew: 1 (Pilot)

14. Maximum Passenger Seating Capacity

Maximum Number	13
11 Passenger and 2 Crew seats	

15. Baggage / Cargo Compartments

Maximum loading weight	100 kg (220 lb)
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16. Life limit Parts

Refer to Maintenance Manual DO 28 D-6

17. All Weather Operation

Category 1

G.IV. Operating and Service Instructions

1. Operating Instructions:
 - a. Pilot's Operating Handbook Do 28 D-6, dated 25 February 1981, with the appendices Weight and Balance Report, Placards and Equipment List.
2. Service Instructions:
 - a. Maintenance Manual Do 128 D-6
 - b. Part Catalogue Do 128 D-6
 - c. UACL-PT 6A-110 Maintenance Manual 302 1242 with Illustrated Part List
 - d. UACL-PT 6A-110 Overhaul Manual 302 1243
 - e. UACL-PT 6A-110 Parts Catalogue 302 1244
 - f. Hartzell Propeller Owner's Manual 106 L
 - g. Hartzell Overhaul Instructions Manual 118 D
 - h. Service Bulletins

G.V. Notes

1. Eligible Serial Numbers
Without limitations
2. Other
 - a. Data in this TCDS Section refers to Aircraft Type DO 28 D-6 in Standard Configuration
 - b. Customized Cabin Interior and Seating Configuration must be approved.

SECTION H: Variant Dornier 128-6

H.I. General

1. a) Type Do 28
b) Variant Do 128-6
2. Airworthiness Category: Normal
3. Certification Application Date: ---
4. LBA Certification Date: 16 July 1981
5. The EASA TCDS is based on the LBA TCDS No. 2031A/SA for Do128-6 (at Issue 23, dated 08 April 2005)

H.II Certification Basis

1. Reference Date for determining the applicable requirements: --
2. (reserved)
3. (reserved)
4. Airworthiness Requirements:
 - FAR 23 dated 01 February 1965, including Amendment 23-1 through 23-6
5. Requirements elected to comply: None
6. EASA Special Conditions:

From engine installation affected regulations additional airworthiness requirements according to FAR 23 Amendment 23-23 compliance have been shown:
23.3 to 23.253, 23.361 to 23.363, 23.367, 23.371, 23.471, 23.511, 23.629, 23.723 to 23.727, 23.731 to 23.735, 23.771, 23.777, 23.831, 23.863 to 23.999, 23.1011 to 23.1301, 23.1303(a) to (d), 23.1305 to 23.1323, 23.1331 to 23.1367, 23.1437 to 23.1545, 23.1549 to 23.1559, 23.1563 to 23.1589.
7. EASA Exemptions: None
8. EASA Equivalent Safety Findings:

Equivalent Safety Finding according to LBA-Letter I 22-2031a/81 dated 25 February 1981 (concerning single engine emergency power and stall warning).
9. EASA Environmental Standards: ICAO Annex 16

H.III. Technical Characteristics and Operational Limitations

1. Type Design Definition:

The aircraft model is defined in documents including later amendments as follows:
Aircraft Serial No. 6001 up to 6099 and 4302 after modification: KZ D02-000-000 B00F.

2. Description:

Landplane for STOL-Operation with two turboprops. Cantilever high-wing aircraft of all-metal construction with external tanks under the wing and fixed main landing gear, tail wheel.

3. Equipment:

Refer to Equipment List of POH
"Summary of Basic Aircraft Modifications Dornier 228 MZ6" as amended

4. Dimensions:

Wing Span	15,55 m	(51,0 ft)
Length	11,71 m	(38,4 ft)
Height	3,90 m	(12,8 ft)
Total Wing Area	29,0 m ²	(312 ft ²)

5. Engines:

Number	2
Manufacturer	Pratt & Whitney Canada
Type	PT6A-110
LBA TCDS	7020/EN

Power rating (Sea level, ISA):

Takeoff power (5 min), max RPM	298 KW (400 HP)
Continuous power, max. RPM	298 KW (400 HP)

Single engine continuous power 336 KW (450 HP)
max RPM

Maximum continuous rotary speed 1900 RPM

Interstage Turbine Temperature (ITT)

- Takeoff (5 min)	685°C
- Continuous, Emergency	685°C
- Start (2 sec)	1085°C
- Transition(2 sec)	825°C

6. Propellers:
Number: 2
Manufacturer: Hartzell
Type: Propeller 1:HC-B3TN-3 D/T 102 82-9.5
Propeller 2:HC-B3TN-3 D/T 102 82 B-9.5
LBA TCDS 32.130/31/PR
Nominal Diameter: 2362 mm ± 0 mm (929 inch)

Propellers installed at one aircraft must be of the same type.

7. Fluids (Fuel/Oil/Additives):
Max Fuel capacity 1514 ltr.
Usable fuel: 1470 ltr.
- Oil 2 x 11,8 ltr = 23,6 ltr

8. Air Speeds:
- | | |
|---|----------|
| VMO (maximum operating speed) | 155 KCAS |
| VA (maneuvering speed) | 129 KCAS |
| VFE (max. flap extended speed) | 110 KCAS |
| Max. Tire Speed | 105 KCAS |
| Max crosswind component proven | 25 KCAS |
| VMCA (minimum control speed with critical engine inoperative) | 65 KCAS |

9. Maximum Operating Altitude: 20000 ft

10. Maximum Weight:
- | | |
|-------------------------|-------------------|
| - Max weight taxi /ramp | 4380 kg (9656 lb) |
| - Max Takeoff weight | 4350 kg (9590 lb) |
| - Max Landing weight | 4140 kg (9127 lb) |
| - Empty weight | 3824 kg (8430 lb) |

11. Centre of Gravity Range:
Refer to Pilot's Operating Handbook Dornier 128-6, Section 2

12. Datum:
Refer to Pilot's Operating Handbook Dornier 128-6, Section 2

13. Minimum Flight Crew: 1 (Pilot)

14. Maximum Passenger Seating Capacity
Maximum Number 14
12 Passenger and 2 Crew seats

15. Baggage / Cargo Compartments
Maximum loading weight (behind cabin door) 100 kg (220 lb)

16. Life limit Parts
Refer to Maintenance Manual Dornier 128-6

17. All Weather Operation
Category 1

H.IV. Operating and Service Instructions

1. Operating Instructions:
 - a. Pilot's Operating Handbook Dornier 128-6, dated 01 July 1981, with the appendices Weight and Balance Report, Placards and Equipment List. Additionally take into consideration for respective Aircrafts Serial No. POH Revision according to Service Bulletin No.: 1100-3000.
2. Service Instructions:
 - a. Maintenance Manual Do 128 D-6
 - b. Part Catalogue Do 128 D-6
 - c. UACL-PT 6A-110 Maintenance Manual 302 1242 with Illustrated Part List
 - d. UACL-PT 6A-110 Overhaul Manual 302 1243
 - e. UACL-PT 6A-110 Parts Catalogue 302 1244
 - f. Hartzell Propeller Owner's Manual 106 L
 - g. Hartzell Overhaul Instructions Manual 118 D
 - h. Service Bulletins

H.V. Notes

1. Eligible Serial Numbers
Without limitations
2. Other
 - a. Data in this TCDS Section refers to Aircraft Type Dornier 128-6 in Standard Configuration
 - b. Customized Cabin Interior and Seating Configuration must be approved.

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SECTION I: Change Record

Issue 1	Initial issue dated 24 July 2009
Issue 2	25.05.2021 New TC Holder, name changed to General Atomics AeroTec Systems GmbH

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