

TYPE-CERTIFICATE

DATA SHEET

NO. EASA.A.367

for DR 200, DR300, AND DR400 SERIES

> **Type Certificate Holder** C.E.A.P.R.

1b route de TROYES 21121, DAROIS FRANCE

For models:

DR 200 DR 221 DR 250 B-160	DR 220 DR 221 B DR 253	DR 220 A DR 250 DR 253 B	DR 220 B DR 250-160	DR 220 AB DR 250 B
DR 340 DR 300/180 R	DR 315 DR 300/140	DR 360 DR 300/125	DR 380 DR 300/120	DR 300/108
DR 400/125 DR 400/2+2 DR 400/160 D DR 400 NGL	DR 400/140 DR 400/120 DR 400/120 D DR 400/200 R	DR 400/160 DR 400/125i DR 400/180 S DR 400/500	DR 400/180 DR 400/140 B DR 400/100 DR 400/200 I	DR 400/180 R DR 400/120 A DR 400 RP



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Section A: DR 200

A.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 200
2.	Airworthiness Category:	Normal Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	April 06, 1965
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

A.II Certification Basis

1.	Reference Date for determining the applicable requirements:	30 October 1964
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None.

A.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to the CEAPR document	nt 1003343
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- 2. Description: Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
- 3. Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4.	Dimensions:	Span8.7	′2 m	(28.61 ft)
		Height 1.8	33 m	(6.00 ft)
		Length 6.6	68 m	(21.92 ft)
		Wing Area 14.1	15 m²	(152.31 ft ²)
5.	Engines:	POTEZ 4 E 20 B		

5.1 Engine Limits: Maximum Continuous Power:. 2750 rpm (105 HP, 77 kW)

6. Propellers:

		Manufacturer	М	odel	Ø	Number of blades	Minimum static RPM at sea level
		Légère	2102 RA (pitch - 0,5)		1.80 m	2	2400 rpm
		Ratier	FH-110R (pitch 62.5°)	1.75 m	2	2400 rpm
7.	Fluids:						
	7.1 Fue	el:		100 octane n	ninimum	aviation gas	oline grade.
	7.2 Enç	gine Oil:					SAE 30 (AERO 65) SAE 40 (AERO 80)
8.	Fluid ca	apacities:					
	8.1 Fue	el:		Wing tanks:.			2 x 40 liters
							55 liters usable during level flight
	8.2 Oil:			Oil sump cap	oacity		.5 liters (4.8 U.S. quarts)
9.	Air spe	eds:		V _{NO} V _A V _{FE}		2 	5 km/h (148.5 knots IAS) 40 km/h (130 knots IAS) 78 km/h (92 knots IAS) 50 km/h (81 knots IAS) 40 km/h (130 knots IAS)
10. Maximum Operating Altitude:			Refer to app	roved airc	craft flight m	anual.	
11.	Operati	onal Capability:		Refer to app	roved airo	craft flight m	anual.
12.	. Maximı	um Masses:					850 kg (1874 lb) 850 kg (1874 lb)
13.	Centre	of Gravity Range	e:				0.31 m aft of datum 0.55 m aft of datum
14.	. Datum:			Leading edge Chord length			part of the wing. : 1.71 m.
15.	. Load fa	ictor (n) at maxir	num weight:	Flaps retract Flaps retract			
16.	Levelin	g Means:		Horizontal re	ference u	ıpper fusela	ge spar
17. Minimum Flight Crew:		1 (pilot) at 0.42 ±0.05 m aft of datum					
18.	Maximu	um Passenger S	eating Capa	city: 1 at 0.42	±0.05 m :	and 2 at 1.1	6m aft of datum.
19.	Baggag	je/cargo compar	tment	Maximum ba datum, withir			t 20 kg at 1.85m aft of limits.



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TCDS No: EASA.A.367	DR 200, DR300, and DR400 s	
Issue: 05	DR 200	Date: 16 October 2023
20. Wheels and Tires	Wheel tire size	2.59 m (8.5ft) main gear wheel:380 x 150 tail wheel:6 x 2 Refer to the maintenance manual
21. Control surface movements:	Elevator:	up
	Ailerons:	up 12° ± 0.5° down 12° ± 0.5°
	Elevator trim tab (m	•
	Elevator nose	
		position:4° ± 1° sition:
	Elevator nose Tab down	
	Wing Flaps:	1^{st} notch20° \pm 3° 2^{nd} notch45° \pm 3°

22. (Reserved)

A.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

A.V Note:

Section B: DR 220

B.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 220
2.	Airworthiness Category:	Normal Category and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	June 24, 1966
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

B.II Certification Basis

1.	Reference Date for determining the applicable requirements:	15 November 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	EASA Environmental Standards:	None

B.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003343
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- 2. Description: Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
- Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

5. Engines:

Span	8.72 m	(28.6 ft)
Height	1.90 m	(6.2 ft)
Length	6.80 m	(22.3 ft)
Wing Area	13.60 m²	(146.4 ft ²)

Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm (100 HP, 74.5 kW)

6. Propellers:

Make	Model	Ø	Number	Minimum static RPM
IVIANE	Model		of blades	at sea level
Légère	2102 RA (pitch – 0.5)	1.80 m	2	2350 rpm
Légère	2102 RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Ratier	FH-110R (pitch-3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal-3)	1.85 m	2	2300 rpm

7. Fluids:

7.1 Fuel:

80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8	8.1 Fuel:	Main fuel tank capacity:110 liters Usable: the last 5 liters are only usable during level flight
		Supplemental fuel tank capacity:50 liters
8	8.2 Oil:	Oil sump capacity
9. /	Air speeds:	
		V _{NE} 270 km/h (146 knots IAS)
		V _{NO} 210 km/h (113 knots IAS)
		V _A 170 km/h (92 knots IAS)
		V _{FE} 150 km/h (81 knots IAS)
		Vc210 km/h (113 knots IAS)
10. I	Maximum Operating Altitude:	Refer to approved aircraft flight manual.
11. (Operational Capability:	Refer to approved aircraft flight manual. In Normal Category, all aerobatic maneuvers, including spins, are forbidden.



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12. Maximum

Masses:	Normal Category	Take-Off Landing	
	Utility category	T/O & Ldg	700 kg

13. Centre of Gravity Range:

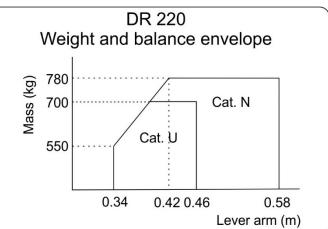
Normal category Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (24.5 % ref.):0.42 m aft of datum at 780 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg Utility categories

Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (22.8 % ref.):0.39 m aft of datum at 700

kg

Aft limit (26.9 % ref.): 0.46 m aft of datum at 700 kg



14.	Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight:	Normal Category: Flaps up+ 3.8 Flaps up 1.52 Flaps down+ 2
	Utility Category: Flaps up+ 4.4 Flaps up 1.76 Flaps down+ 2
16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 1 at 0.43 ±0.05 m aft of datum
18. Maximum Passenger Seating Capa	city:
	1 at 0.43 ± 0.05 m and 2 (maximum 90kg) at 1.22m aft of datum
	The rear seats can be used only if seat belts are provided and if weight and balance are respected.
	Rear seats must not be used in utility category.
19. Wheels and Tires	Main gear track2.59 m (ft)Wheel tire sizemain gear wheel:



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20. Control surface movements:

Elevator:	up 9°30' (+0°; -0°30') down 12° (+0°; -0°30')
Ailerons:	up 12° (+0°; -0°30') down 12° (+0°; -0°30') neutral: trailing edge aligned on
flaps	
	position: 28° sition: 6°30'
	n position: 12°30' sition: 16°30'
Wing Flaps:	1^{st} notch20° \pm 2° 2^{nd} notch60° \pm 2°

21. (Reserved)

B.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

B.V <u>Note:</u>

(Reserved)

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Section C: DR 220 A

C.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 220 A
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	January 4, 1967
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

C.II Certification Basis

1965

C.III Technical Characteristics and Operational Limitations

1. Type Design Definition Refer to the CEAPR document 1003343	3
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- 2. Description: Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
- 3. Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

5. Engines:

Span	8.72 m	(28.6 ft)
Height	1.90 m	(6.2 ft)
Length	6.80 m	(22.3 ft)
Wing Area	13.60 m²	(146.4 ft ²)

Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm (100 HP, 74.5 kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Légère	2102RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Ratier	FH-110R (pitch - 3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal - 3)	1.85 m	2	2300 rpm

- 7. Fluids:
 - 7.1 Fuel:

80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8.1 Fuel:	Main fuel tank capacity:110 liters Usable: the last 5 liters are only usable during level flight
	Supplemental fuel tank capacity:50 liters
8.2 Oil:	Oil sump capacity 5.7 liters (6 U.S. quarts)
9. Air speeds:	
	V _{NE}
	V _{NO}
	V _A 190 km/h (103 knots IAS)
	V _{FE}
	Vc
10. Maximum Operating Altitude:	Refer to approved aircraft flight manual.
11. Operational Capability:	Refer to approved aircraft flight manual. In Normal Category, all aerobatic maneuvers, including spins, are forbidden.



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Normal Category	Take-Off Landing	-
Litility category	T/O & I da	780 ka

13. Centre of Gravity Range:

12. Maximum Masses:

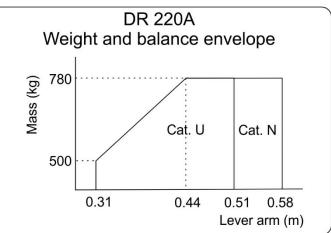
Utility category T/O & Ldg......780 kg

<u>Normal category</u> Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780 kg Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg

<u>Utility categories</u>

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780 kg

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight:	Normal Category:	
	Flaps up	+ 3.8
	Flaps up	1.52
	Flaps down	+ 2
	Utility Category:	

y Category.	
Flaps up	+ 4.4
Flaps up	
Flaps down	+ 2

16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 ± 0.05 m and 2 (maximum 110kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category.

19. Wheels and Tires	Main gear track	
	Wheel tire size	main gear wheel:
		tail wheel:6 x 2
	Tire pressure	Refer to the maintenance manual



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20. Control surface movements:

Elevator:	up
Ailerons:	up 12° (+0°; -0°30') down 12° (+0°; -0°30') neutral: trailing edge aligned on
flaps	
	position:
	n position: 12°30 sition: 16°30
Wing Flaps:	1^{st} notch20° \pm 2° 2^{nd} notch60° \pm 2°

21. (Reserved)

C.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

C.V Note:

1. This model is identical to DR220 except wing structure and landing gear.



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Section D: DR 220 B

D.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 220 B
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	July 20, 1968
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003
8.	The EASA type Certificates replace	sDGAC-France Type Certificate no. 40.
D.II <u>(</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	15 November 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965

- Requirements elected to comply: None
 EASA Special Conditions: None
- EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: None

D.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition	Refer to the CEAPR document 1003343
\mathbf{a}	Description:	Single engine four cost low wing airplane we

- 2. Description: Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
- Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

5. Engines:

Span	8.72 m	(28.6 ft)
Height	1.90 m	(6.2 ft)
Length	6.80 m	(22.3 ft)
Wing Area	13.60 m²	(146.4 ft ²)

Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power: 2750 rpm (100 HP, 74.5 kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Légère	2102 RA (pitch - 0,5)	1.80 m	2	2350 rpm
Légère	2102 RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Ratier	FH-110R (pitch-3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal-3)	1.85 m	2	2300 rpm

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

above 40°F (5°C)

80

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°E (5°C)	65	30	10W-30 / 15W-50 / 20W-50

50

80/87 octane, minimum aviation grade gasoline.

15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8.1 Fuel: Usable:.. the last 5 liters are only usable during level flight 8.2 Oil: Oil sump capacity 5.7 liters (6 U.S. quarts) 9. Air speeds: Vc210 km/h (113 knots IAS) 10. Maximum Operating Altitude: Refer to approved aircraft flight manual. Refer to approved aircraft flight manual. 11. Operational Capability: ... In Normal Category, all aerobatic maneuvers, including



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spins, are forbidden.

12. Maximum Masses:	Normal Category	Take-Off Landing	•
	Utility category	T/O & Ldg	.700 kg

13. Centre of Gravity Range:

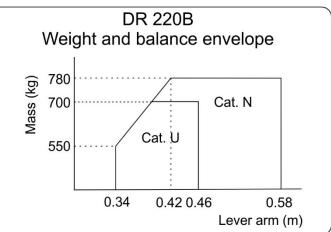
<u>Normal category</u> Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (25.6 % ref.):0.42 m aft of datum at 780 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg

Utility categories

Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (22.8 % ref.):0.39 m aft of datum at 700 kg

Aft limit (26.9 % ref.): 0.46 m aft of datum at 700 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight:	Flaps up+ 3.8
	Flaps up 1.52 Flaps down+ 2
	Utility Category:
	Flaps up+ 4.4
	Flaps up 1.76
	Flaps down+ 2
16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.43 ±0.05 m aft of datum
18. Maximum Passenger Seating Capa	city:
	1 at 0.43 \pm 0.05 m and 2 (maximum 90kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category.

19. Wheels and Tires	Main gear track	
		main gear wheel:
		tail wheel:6 x 2
	Tire pressure	Refer to the maintenance manual



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20. Control surface movements:

Elevator:	up
Ailerons:	up 12° (+0°; -0°30') down 12° (+0°; -0°30') neutral: trailing edge aligned on
flaps	
	position:
	ו position: 12°30 sition: 16°30
Wing Flaps:	1^{st} notch20° ± 2° 2 nd notch60° ± 2°

21. (Reserved)

D.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6

D.V Note:

1. This model is identical to DR220 except leading edge profile of trapezoidal wing part.



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Section E: DR 220 AB

General E.I

	1.	a) Type:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 220 AB
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
;	5. (Reserved)	
	6. C	OGAC Type Certification Date:	July 20, 1968
	7. E	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003
	8. 1	The EASA type Certificates replaces	DGAC-France Type Certificate no. 40.
	C	ertification Basis	
	1	Reference Date for determining	
	••	the applicable requirements:	15 November 1965
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
;	5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
(6.	Requirements elected to comply:	None

E.II

1.	Reference Date for determining the applicable requirements:	15 November 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	. EASA Environmental Standards:	None

E.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003343
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
3.	Equipment:	The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed. 4. Dimensions:

5. Engines:

Span	8.72 m	(28.6 ft)
Height	1.90 m	(6.2 ft)
Length	6.80 m	(22.3 ft)
Wing Area	13.60 m²	(146.4 ft ²)

Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm – 100 HP (74.5 kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Légère	2102 RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Ratier	FH-110R (pitch-3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal-3)	1.85 m	2	2300 rpm

7. Fluids:

80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

7.1 Fuel:

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8.1 Fuel:

8.2 Oil:

9. Air speeds:

V _{NE}	
V _{NO}	
VA	
V _{FE}	
Vc	
Refer to appro	ved aircraft flight manual.

Oil sump capacity 5.7 liters (6 U.S. quarts)

10.	Maximum	Operating	Altitude:	
10.	Maximann	oporating	/ littado:	

11. Operational Capability:

Refer to approved aircraft flight manual.

In Normal Category, all aerobatic maneuvers, including spins, are forbidden.



12. Maximum Masses:	Normal Category	780 kg 780 kg
	Litility category	780 ka

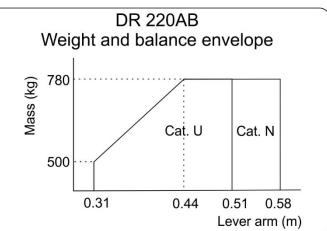
13. Centre of Gravity Range:

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg <u>Utility categories</u>

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780 kg

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight:	Normal Category:
	Flaps up

Flaps up	
Flaps up	1.52
Flaps down	.+2

Flaps up	.+ 4.4
Flaps up	1.76
Flaps down	

16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 ± 0.05 m and 2 (maximum 110kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category.

19. Wheels and Tires	Main gear track	
	Wheel tire size	main gear wheel:
		tail wheel:6 x 2
	Tire pressure	Refer to the maintenance manual



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20.

Control surface movements:

Elevator:	up
Ailerons:	up 12° (+0°; -0°30') down 12° (+0°; -0°30') neutral: trailing edge aligned on
flaps	
Elevator anti tab: Elevator up Tab down Tab up pos	position:
	n position: 12°30 sition: 16°30
Wing Flaps:	1^{st} notch20° ± 2° 2^{nd} notch60° ± 2°

21. (Reserved)

E.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

E.V Note:

1. This model is identical to DR220A except leading edge profile of trapezoidal wing part.



Section F: DR 221

F.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 221
2.	Airworthiness Category:	Normal Category and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	April 25, 1967
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

F.II Certification Basis

1.	Reference Date for determining the applicable requirements:	24 march 1967
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None

F.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition	Refer to the CEAPR document 1003343
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.
4.	Dimensions:	Span8.72 m (28.6 ft)
		Height1.90 m (6.2 ft) Length6.80 m (22.3 ft) Wing Area13.60 m² (146.4 ft²)
5.	Engines:	Lycoming O-235-C2A The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



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standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Takeoff (max 5 minutes): 2800 rpm – 115 HP (85 kW) Maximum Continuous power: 2600 rpm – 108 HP (79.5 kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1A105BCM7056	1.78 m	2	2400 rpm (Note 1)
Jodel Evra	88-75-34-F	1.76 m	2	2250 rpm

Note 1: Maximum RPM: 2600 rpm

7. Fluids:

7.1 Fuel:

80/87 octane, minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

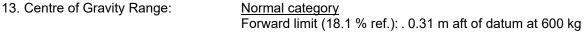
7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or	SAE30
	SAE20W40	
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

	1		
8.1	Fuel:	Main fuel tank capacity:110 liters Usable: the last 5 liters are only usable during level flight	
		Supplemental fuel	tank capacity:50 liters
8.2	2 Oil:		6 U.S. quarts (5.7 liters) 4 U.S. quarts (3.8 liters)
9. Air	speeds:	V _{NO} V _A V _{FE}	
10. Ma	ximum Operating Altitude:	Refer to approved	aircraft flight manual.
11. Op	erational Capability:		aircraft flight manual. gory, all aerobatic maneuvers, including m.
12. Ma	ximum Masses:	Normal Category	Take-Off840 kg Landing840 kg
		Utility category	T/O & Ldg780 kg





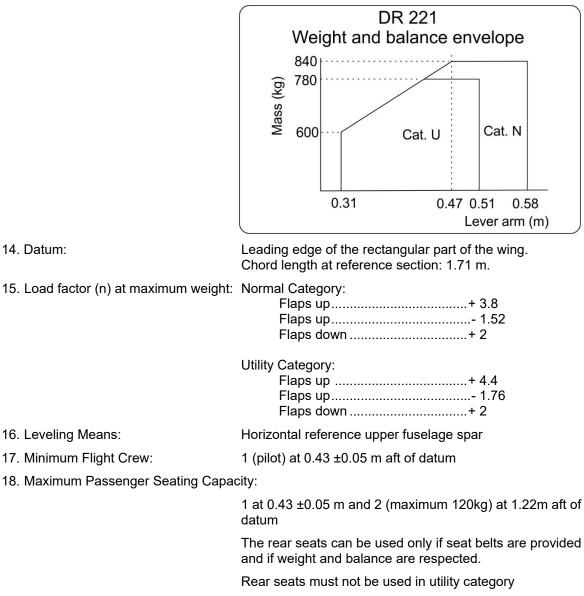
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Intermediate limit (27.5 % ref.):0.47 m aft of datum at 840 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 840 kg

<u>Utility categories</u> Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg Intermediate limit (25.1 % ref.):0.43 m aft of datum at 780 kg

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



	Real Seals must i	Iot be used in utility cate	JOLA
19. Wheels and Tires:	Main gear track		
	Wheel tire size	main gear wheel:	
		tail wheel:	6 x 2
	Tire pressure	Refer to the mainte	enance manual

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20. Control surface movements:

Elevator:	up
Ailerons:	up
flaps	5 5 5
	n position:
	/n n position: 8°30 osition:
Wing Flaps:	1^{st} notch20° \pm 2° 2^{nd} notch60° \pm 2°

21. (Reserved)

F.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Minor inspection Schedule	Refer to latest amendment of service letter n°6
Airplane Major inspection Schedule	Refer to latest amendment of service letter n°6

F.V Note:

1. This model is identical to DR220A except power plant and pitch tab deflection.



Section G: DR 221 B

G.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 221 B
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	July 20, 1968
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

G.II Certification Basis

1.	Reference Date for determining the applicable requirements:	24 march 1967
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052, amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None

G.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition	Refer to the CEAPR document 1003343
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.
4.	Dimensions:	
		Span
5.	Engines:	Lycoming O-235-C2A
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



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Landing840 kg

September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 5.1 Engine Limits: Takeoff (max 5 minutes): 2800 rpm - 115 HP (85 kW) Maximum Continuous Power: 2600 rpm - 108 HP (79.5 kW)
- 6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1A105BCM7056	1.78 m	2	2400 rpm (Note 1)
Jodel Evra	88-75-34-F	1.76 m	2	2250 rpm

Note 1: Maximum RPM: 2600 rpm

7. Fluids:

7.1 Fuel:

80/87 octane, minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

	8.1 Fuel:	Main fuel tank capacity:
	8.2 Oil:	Oil sump capacity
9.	Air speeds:	VNE
10.	Maximum Operating Altitude:	Refer to approved aircraft flight manual.
11. Operational Capability:		Refer to approved aircraft flight manual.
		In Normal Category, all aerobatic maneuvers, including spins, are forbidden.
12.	Maximum Masses:	Normal Category Take-Off



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Utility category T/O & Ldg......780 kg

13. Centre of Gravity Range:

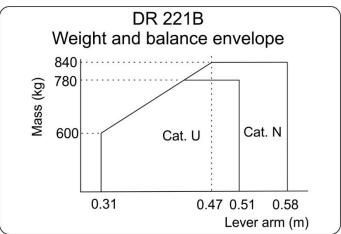
<u>Normal category</u> Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg Intermediate limit (27.5 % ref.):0.47 m aft of datum at 840 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 840 kg

Utility categories

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg Intermediate limit (25.1 % ref.):0.43 m aft of datum at 780 kg

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up+	3.8
Flaps up	1.52
Flaps down+	2

Utility Category:

Flaps	up	+ 4.4
	up	
	down	

Horizontal reference upper fuselage spar

16. Leveling Means:

17. Minimum Flight Crew:

1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 ± 0.05 m and 2 (maximum 120kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category

19. Wheels and Tires:

Main gear track		2.59 m (ft)
Wheel tire size	main gear wheel:	380 x 150
	tail wheel:	6 x 2
Tire pressure	Refer to the mainte	enance manual

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20. Control surface movements:

Elevator:	up
Ailerons:	up 12° (+0°; -0°30') down 12° (+0°; -0°30') neutral: trailing edge aligned on
flaps	
	position:
	position: 8°30 sition: 13°30
Wing Flaps:	1^{st} notch20° \pm 2° 2^{nd} notch60° \pm 2°

21. (Reserved)

G.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
Airplane Major inspection Schedule	Refer to latest amendment of service letter n°6

G.V Note:

1. This model is identical to DR221 except leading edge profile of trapezoidal wing part.



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Section H: DR 250

H.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series	
	b) Model:	DR 250	
2.	Airworthiness Category:	Normal Category	
3.	Type Certificate Holder:	Refer to Note 2 Section PP	
4.	Manufacturer:	Refer to Note 3 Section PP	
5.	(Reserved)		
6.	DGAC Type Certification Date:	May 25, 1965	
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003	

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

H.II Certification Basis

1.	Reference Date for determining the applicable requirements:	18 May 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None.

H.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition	Refer to the CEAPR document 1003344
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.
4.	Dimensions:	
		Span
		Height1.86 m (6.10 ft) Length6.98 m (22.90 ft)
		Wing Area
5.	Engines:	Lycoming O-320 E2A (150 HP)
		The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state

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acceptance or certification of this standard prior to 28

September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:2700 rpm (152 HP, 112 kW)

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
HOFFMANN Croisière	FH2/LC23 180 155-6.5R	1.80 m	2	2150 rpm
HOFFMANN	FH2/LC23 180-140-6,5 R	1.80 m	2	2250 rpm
JODEL EVRA	91-78-34	1.84 m	2	2250 rpm
SENSENICH	M74 DMS-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YL-1 7663-4	1.83 m	2	Hartzell H1	Constant speed

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

80/87 octane minimum aviation gasoline grade. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant	Mineral
	(AD)grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or	SAE30
	SAE20W40	
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:



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sue: 05	DR 250	Date: 16 October 2023
		ity:70 liters ters are only usable during level flight
	Supplemental fuel ta	nk capacity:50 liters
8.2 Oil:		
9. Air speeds:		
	V _{NO} Va V _{FE}	
10. Maximum Operating Altitude:	Refer to approved ai	rcraft flight manual.
11. Operational Capability:	Refer to approved ai	rcraft flight manual.
12. Maximum Masses:		
13. Centre of Gravity Range:		
		ref.): 0.29 m aft of datum 0.565 m aft of datum
14. Datum:		rectangular part of the wing. ence section: 1.71 m.
15. Load factor (n) at maximum weigh		
		d positive n+ 3.8 d negative n 1.52
16. Levelling Means:	Horizontal reference	upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.42 ±0.0	5 m aft of datum
18. Maximum Passenger Seating Cap	acity: 1 at 0.42 ±0.05	5 m and 2 at 1.16m aft of datum.
19. Baggage/cargo compartment:		compartment 40 kg at 1.90m aft of nt and balance limits. See note 1. fuel tank".
20. Wheels and Tires:	Wheel tire size	
	Tire pressure	Refer to the maintenance manual
21. Control surface movements:		
	Elevator:	up
	Ailerons:	up
	Tab up pos Elevator nose Tab down j	down position:4° ± 1° sition:

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Wing Flaps:

1 st notch	$\dots 20^{\circ} \pm 3^{\circ}$
2 nd notch	$\dots 60^{\circ} \pm 3^{\circ}$

22. (Reserved)

H.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

H.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 liters of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR200 with Lycoming 0-320 E 2 A (150 HP) engine, longer and larger fuselage, flap defection up to 60°, fuel capacity increased (rear and supplemental tank).



Section I: DR 250 - 160

I.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 250 - 160
2.	Airworthiness Category:	Normal Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	September 09, 1965
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003
8.	The EASA Type Certificates replace	es DGAC-France Type Certificate no. 34.

I.II Certification Basis

1.	Reference Date for determining the applicable requirements:	May 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None.

I.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition	Refer to the CEAPR document 1003344
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.
4.	Dimensions:	
		Span
		Height1.86 m (6.10 ft) Length6.98 m (22.90 ft)
		Wing Area
5.	Engines:	Lycoming O-320 D2A (160 HP)
		The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state



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DR 250 - 160

acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:2700 rpm (162 HP, 119 kW)

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
HOFFMANN	FH2/LC23 180-155-6,5R	1.80 m	2	2250 rpm
HOFFMANN	FH2/LC23 180-140-6,5R	1.80 m	2	2350 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-2-66	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-66	1.83 m (*)	2	2150 rpm
JODEL EVRA	91-86-34 F	1.82 m	2	2250 rpm
JODEL EVRA	91-78-34 F	1.84 m	2	2300 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

91/96 octane minimum aviation gasoline grade. Refer to latest revision of Service Instruction Lycoming No. 1070. Refer to latest revision of Service Instruction Lycoming

No. 1014.		modiad		9
	Ashless dispersant (AD)		Mineral	
	arades		arades	

Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:	Wing tanks:2 x 40 liters Main fuel tank capacity:70 liters Usable: the last 7 liters are only usable during level flight
	Supplemental fuel tank capacity:50 liters
8.2 Oil:	Oil sump capacity
9. Air speeds:	V _{NE}
10 Maximum Operating Altitude:	Refer to approved aircraft flight manual

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.



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TCDS No: EASA.A.367	DR 200, DR300, and DR40	0 series
Issue: 05	DR 250 - 160	Date: 16 October 2023
11. Operational Capability:	Refer to approve	d aircraft flight manual.
12. Maximum Masses:		
13. Centre of Gravity Range:		
		% ref.): 0.29 m aft of datum f.): 0.565 m aft of datum
14. Datum:		the rectangular part of the wing. eference section: 1.71 m.
15. Load factor (n) at maximum weig		
		cted positive n+ 3.8 cted negative n 1.52
16. Leveling Means:	Horizontal refere	nce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.42 ±	0.05 m aft of datum
18. Maximum Passenger Seating Ca	apacity: 1 at 0.42 ±0.0	5 m aft and 2 at 1.16m aft of datum
19. Baggage/cargo compartment		ge compartment 40 kg at 1.90m aft of eight and balance limits. See note 1. rear fuel tank".
20. Wheels and Tires	Main gear track Wheel tire size	2.59 m (ft) main gear wheel:380 x 150 tail wheel:6 x 2
	Tire pressure	Refer to the maintenance manual
21. Control surface movements:		
	Elevator:	up
	Ailerons:	up 12° ± 0.5° down 12° ± 0.5°
	Rudder L & R: befo	
	Elevator trim tab (manual):	
	Elevator nose down Tab down position:4°	
	Tab up position:	
	Elevator nose up Tab down position:	
		position: 16° ± 1°
	Wing Flaps: 1 st notch20	
		2^{nd} notch60° ± 3°
22 (Reserved)		

22. (Reserved)



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I.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

I.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 liters of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR250 with Lycoming 0-320 D 2 A (160 HP) engine.



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Section J: DR 250 B

J.I <u>General</u>

1.	а) Туре:	DR 200, DR 300, and DR 400 series		
	b) Model:	DR 250 B		
2.	Airworthiness Category:	Normal Category		
3.	Type Certificate Holder:	Refer to Note 2 Section PP		
4.	Manufacturer:	Refer to Note 3 Section PP		
5.	(Reserved)			
6.	DGAC Type Certification Date:	July 20, 1968		
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003		

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

J.II Certification Basis

1.	Reference Date for determining the applicable requirements:	May 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None.

J.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition	Refer to the CEAPR document 1003344		
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.		
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.		
4.	Dimensions:	Span 8.72 m (28.61 ft) Height 1.86 m (6.10 ft) Length 6.98 m (22.90 ft) Wing Area 14.15 m² (152.31 ft²)		
5.	Engines:	Lycoming O-320 E2A (150 HP)		
		The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28		



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Maximum Continuous Power: 2700 rpm (152 HP, 112 kW)

5.1 Engine Limits:

6. Propellers:

Number Minimum static RPM Manufacturer Model Ø of blades at sea level HOFFMANN FH2/LC23 180 155-6.5R 2 1.80 m 2150 rpm Croisière HOFFMANN FH2/LC23 180-140-6,5 R 1.80 m 2 2250 rpm JODEL 2 91-78-34 1.84 m 2250 rpm EVRA 1.83 m (*) SENSENICH M74 DMS-2-64 2 2150 rpm 2 SENSENICH 74 DM 6S5-2-64 1.83 m (*) 2150 rpm 2 2200 rpm SENSENICH M74 DMS-0-64 1.88 m SENSENICH 74 DM 6S5-0-64 1.88 m 2 2200 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YL-1 7663-4	1.83 m	2	Hartzell H1	Constant speed

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

80/87 octane minimum aviation gasoline grade. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
•	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à	SAE40	SAE40
+30°C)		
0°F to 70°F (-15°C à	SAE30, SAE40 or	SAE30
+20°C)	SAE20W40	
0°F to 90°F (-15°C à	SAE20W50 or SAE15W50	SAE20W50
+30°C)		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:



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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR400 s DR 250 B	series Date: 16 October 2023	
8.1 Fuel:	Main fuel tank capa		
		ank capacity:50 liters	
8.2 Oil:		8 U.S. quarts (7.6 liters) 	
9. Air speeds:	V _{NO} V _A V _{FE}		
10. Maximum Operating Altitude:	Refer to approved a	aircraft flight manual.	
11. Operational Capability:	Refer to approved a	aircraft flight manual.	
12. Maximum Masses:			
13. Centre of Gravity Range:		ref.): 0.29 m aft of datum 0.565 m aft of datum	
14. Datum:		e rectangular part of the wing. erence section: 1.71 m.	
15. Load factor (n) at maximum weig	Flaps retracted	ed positive n+ 3.8 ed negative n 1.52	
16. Leveling Means:	Horizontal reference	e upper fuselage spar	
17. Minimum Flight Crew:	1 (pilot) at 0.42 ±0.0	05 m aft of datum	
18. Maximum Passenger Seating Ca	pacity: 1 at 0.42 ±0.05	m and 2 at 1.16m aft of datum.	
19. Baggage/cargo compartment		e compartment 40 kg at 1.90m aft of ght and balance limits. See note 1. ar fuel tank".	
20. Wheels and Tires	Wheel tire size		
21. Control surface movements:	Flovetor		
	Elevator:	up 9.5° ± 0.5° down 12° ± 0.5°	
	Ailerons:	up	
	Elevator trim tab (manual): Elevator nose down Tab down position:		
	Tab up position:		
	Tab down	position: 11° ± 1° psition: 16° ± 1°	

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Wing Flaps:

1 st notch	20° ± 3°
2 nd notch	$\dots 60^{\circ} \pm 3^{\circ}$

22. (Reserved)

J.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

J.V <u>Note:</u>

1. Supplementary rear fuel tank operation:

Always use first 50 liters of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR250 except the leading edge profile of trapezoidal wing part.



Section K: DR 250 B - 160

K.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series	
	b) Model:	DR 250 B - 160	
2.	Airworthiness Category:	Normal Category	
3.	Type Certificate Holder:	Refer to Note 2 Section PP	
4.	Manufacturer:	Refer to Note 3 Section PP	
5.	(Reserved)		
6.	DGAC Type Certification Date:	July 20, 1968	
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003	

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

K.II Certification Basis

1.	Reference Date for determining the applicable requirements:	May 1965
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment May 1 st 1965
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	None

K.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003344
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed conventional landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.
4.	Dimensions:	
		Span8.72 m (28.61 ft) Height1.86 m (6.10 ft)
		Length
5.	Engines:	Lycoming O-320 D2A (160 HP)
		The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:2700 rpm (162 HP, 119 kW)

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
HOFFMANN	FH2/LC23 180-155-6,5R	1.80 m	2	2250 rpm
HOFFMANN	FH2/LC23 180-140-6,5R	1.80 m	2	2350 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-2-66	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-66	1.83 m (*)	2	2150 rpm
JODEL EVRA	91-86-34 F	1.82 m	2	2250 rpm
JODEL EVRA	91-78-34 F	1.84 m	2	2300 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

91/96 octane minimum aviation gasoline grade. Refer to latest revision of Service Instruction Lycoming No. 1070.

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:	
8.1 Fuel:	Wing tanks:2 x 40 liters Main fuel tank capacity:70 liters Usable: the last 7 liters are only usable during level flight
	Supplemental fuel tank capacity:50 liters
8.2 Oil:	Oil sump capacity
9. Air speeds:	V _{NE}
10. Maximum Operating Altitude:	Refer to approved aircraft flight manual.
11. Operational Capability:	Refer to approved aircraft flight manual.
12. Maximum Masses:	Take-Off:
13. Centre of Gravity Range:	
	Forward limit (17 % ref.):0.29 m aft of datum Aft limit (33 % ref.):0.565 m aft of datum
14. Datum:	Leading edge of the rectangular part of the wing.
15. Load factor (n) at maximum weight	: Flaps retracted positive n+ 3.8 Flaps retracted negative n 1.52
16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.42 ±0.05 m aft of datum
18. Maximum Passenger Seating Capa	acity: 1 at 0.42 \pm 0.05 m and 2 at 1.16m aft of datum.
19. Baggage/cargo compartment:	Maximum baggage compartment 40 kg at 1.90m aft of datum, within weight and balance limits. See note 1. "Supplementary rear fuel tank".
20. Wheels and Tires:	Main gear track
	Tire pressureRefer to the maintenance manual

21. Control surface movements:

Elevator:	up
Ailerons:	up $12^{\circ} \pm 0.5^{\circ}$ down $12^{\circ} \pm 0.5^{\circ}$
	,



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Elevator nose up		
Tab down position: 11°:		11° ± 1°
	position:	
Wing Flaps:	1 st notch	20° ± 3°
	2 nd notch	$\dots 60^{\circ} \pm 3^{\circ}$

22. (Reserved)

K.IV Operating and Service Instructions

Airplane Flight Manual	.Refer to latest amendment of service letter n°6
Airplane Minor inspection Schedule	.Refer to latest amendment of service letter n°6
Airplane Major inspection Schedule	.Refer to latest amendment of service letter n°6

K.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 liters of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR250/160 except the leading edge profile of trapezoidal wing part.

Section L: DR 253

L.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 253
2.	Airworthiness Category:	Normal Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	July 11, 1967
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no 42

L.II Certification Basis

1.	Reference Date for determining the applicable requirements:	June 1966
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052.
5.	Airworthiness Requirements:	France AIR2052 amendment June 6 th , 1966.
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	EASA Environmental Standards:	None.

L.III Technical Characteristics and Operational Limitations

- Type Design Definition
 Refer to the CEAPR document 1003346
 Description:
 Single-engine, four-seat, low-wing airplane, wooden construction, fixed tricycle landing gear.
- 3. Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning Safe Flight n°164



5. Engines:

4.	Dimensions:	Span	8.72 m	(28.6 ft)
		Height	2.38 m	(7.8 ft)
		Length	7.00 m	(22.9 ft)
		Wing Area	14.20 m ²	(152.8 ft ²)

Lycoming O-360 A1A (variable-pitch propeller) Lycoming O-360 A3A (Sensenich propeller)

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:2700 rpm (133 kW, 183 HP)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
SENSENICH	M 76 EMMS-0-64 76 EM8S5-0-64	1,93 m (76 in.)	2	2300 rpm

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YK 7666-2	1,88 m (74 in.)	2	Hartzell D 16 or F3	Constant speed low pitch: 12° high: 28° 8 (*)

Remark: (*) Continuous operation between 2000 and 2250 rpm must be avoided.

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



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8.	Fluid capacities:	
	8.1 Fuel:	Wing tank: 40 liters in each tank
		Fuselage tank:
		Auxiliary tank: (see note 1)50 or 60 liters
	8.2 Oil:	Oil sump capacity
9.	Air speeds:	VNE
10.	Maximum Operating Altitude:	Refer to approved aircraft flight manual.
11.	Operational Capability:	Refer to approved aircraft flight manual.
12.	Maximum Mass:	take-off1100 kg landing1045 kg
13.	Centre of Gravity Range:	Normal category Forward limit (14.6 % ref.): 0.250 m aft of datum at 800 kg Intermediate limit (25 % ref.): 0.430 m aft of datum at 1100 kg Aft limit (33 % ref.): 0.565 m aft of datum at 1100 kg DR 253 Weight and balance envelope (5) (5) (1100) (25)
14.	Datum:	Wing leading edge of the rectangular part of wing. Cord length at reference section: 1.71 m.
15.	Load factor (n) at maximum weight:	Normal Category: Flaps up+ 3.8 Flaps up 1.52
16.	Leveling Means:	Horizontal reference upper fuselage spar.
17.	Minimum Flight Crew:	1 (pilot) at 0.47 ±0.05 m aft of datum
18.	Maximum Passenger Seating Capac	city: 1 at 0.47 ±0.05 m and 2 at 1.25m aft of datum

*** * * * * **

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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR400 series DR 253 Dat	e: 16 October 2023
19. Baggage / Cargo Compartment:	: 40 kg within the limits of weight and balar	nce authorized.
	Lever arm:+ 2.1	
20. Wheels and tires:	Main gear track Base width	()
	Wheel tire size	420 x 150
		0-5 (See note 2)
	PressureRefer to the mair	ntenance manual
	Front wheel movements (left and right):	25° ^{+2°} _{-0°}
21. Control surface movements:	Elevator:nc	ose up: 13° $^{+0,5^{\circ}}_{-0^{\circ}}$
	nos	e down: 6° $^{+0,5^{\circ}}_{-0^{\circ}}$
	Ailerons:	up: 12° ^{-0°} _{+0,5°}
		down: 12° $^{-0^{\circ}}_{+0,5^{\circ}}$
	Rudder:	L & R: 25° +0° _3°
	minimum before differential bra	king (L & R): 15°
	Wing Flaps: 1 st notch (T/O)	15° ^{+0°} _{-5°}
	2 nd notch (Ldg)	60° ^{+0°}
	Elevator tab:	Tab movements

22.	(Reserved)	

L.IV Operating and service instructions

Airplane Flight Manual	.Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	.Refer to latest amendment of service letter n°6
Airplane Major inspection schedule	.Refer to latest amendment of service letter n°6

Maximum "Nose up"

Maximum "Nose down"

Up

10°5

12°

Down

31°

3°



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L.V Note:

1. Supplementary rear fuel tank operation:

Always use first 60 liters of fuel from rear main tank (more or less 1h30min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. Alternative Oleo Strut and Tires:

Tires size 380x150 or 5.00-5 can only be installed on the aircraft if the DR400 landing gear has been installed in accordance with the SB 160403.



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Section M: DR 253 B

M.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 253 B
2.	Airworthiness Category:	Normal Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification Date:	July 20, 1968
7.	EASA Type Certification Date:	Transferred by Commission Regulation (EC) No. 1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no 42

M.II Certification Basis

1.	Reference date for determining the applicable requirements:	June 1966
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6 th , 1966.
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	. EASA Environmental Standards:	None

M.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003346
2.	Description:	Single-engine, four-seat, low-wing airplane, wooden construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning Safe Flight n°164

5. Engines:

4.	Dimensions:	Span	8.72 m	(28.6 ft)
		Height	2.38 m	(7.8 ft)
		Length	7.00 m	(22.9 ft)
		Wing Area	14.20 m ²	(152.8 ft ²)

Lycoming O-360 A1A (variable-pitch propeller) Lycoming O-360 A3A (Sensenich propeller)

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine limits:

Maximum Continuous Power:2700 rpm (133 kW, 183 HP)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
SENSENICH	M 76 EMMS-0-64 76 EM8S5-0-64	1,93 m (76 in.)	2	2300 rpm

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YK 7666-2	1,88 m (74 in.)	2	Hartzell D 16 or F3	Constant speed low pitch: 12° high: 28° 8 (*)

Remark: (*) Continuous operation between 2000 and 2250 rpm must be avoided.

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



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8.	Fluid capacities:	
	8.1 Fuel:	Wing tank: 40 liters in each tank
		Fuselage tank:
		Auxiliary tank: (see note 1)50 or 60 liters
	8.2 Oil:	Oil sump capacity
	9. Air speeds:	VNE
	10. Maximum operating altitude:	Refer to approved aircraft flight manual.
	11. Operational capability:	Refer to approved aircraft flight manual.
	12. Maximum mass:	take-off1100 kg landing1045 kg
	13. Centre of gravity range:	Normal Category Forward limit (14.6 % ref.): 0.250 m aft of datum at 800 kg Intermediate limit (25 % ref.): 0.430 m aft of datum at 1100 kg Aft limit (33 % ref.): 0.565 m aft of datum at 1100 kg DR 253B Weight and balance envelope (5) 1100 (5) (5) $(1100)(5)$ $(1100)(5)$ $(1100)(5)$ $(1100)(1100)(1100)(1100)(25)$ (25)
	14. Datum:	Wing leading edge of the rectangular part of wing. Cord length at reference section: 1.71 m.
	15. Load factor (n) at maximum weight:	Normal Category: Flaps up+ 3.8 Flaps up 1.52
	16. Levelling means:	Horizontal reference upper fuselage spar.
	17. Minimum flight crew:	1 (pilot) at 0.47 ±0.05 m aft of datum
	18. Maximum passenger seating capaci	ty: 1 at 0.47 ±0.05 m and 2 at 1.25m aft of datum
	19. Baggage / Cargo compartment	40 kg within the limits of weight and balance authorized. Lever arm:+ 2.1 m aft of datum



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Up

10°5

12°

Maximum "Nose up"

Maximum "Nose down"

Down

31°

3°

20. Wheels and tires		2.58 m (8.46 ft) 1.61 m (5.28 ft)
	Wheel tire size	
	Pressure	
	Front wheel move	ements (left and right):
21. Control surface movements:	Elevator:	nose up: 13° $^{+0,5^{\circ}}_{-0^{\circ}}$
		nose down: 6° $^{+0,5^{\circ}}_{-0^{\circ}}$
	Ailerons:	up: 12° ^{-0°} _{+0,5°}
		down: 12° $^{-0^{\circ}}_{+0,5^{\circ}}$
	Rudder:	L & R: 25° ^{+0°} _{-3°}
	minimu	m before differential braking (L & R): 15°
	Wing Flaps:	1 st notch (T/O)15° $^{+0^{\circ}}_{-5^{\circ}}$
		2 nd notch (Ldg)60° $^{+0^{\circ}}_{-5^{\circ}}$
	Elevator tab:	
		Tab
		movements

22.	(Reserved)	

M.IV Operating and Service Instructions

M.V Note:

1. Supplementary rear fuel tank operation:

Always use first 60 liters of fuel from rear main tank (more or less 1h30min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH



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- 2. This model is identical to the DR 253 except the leading edge profile of trapezoidal wing part.
- 3. Alternative Oleo Strut and Tires:

Tires size 380x150 or 5.00-5 can only be installed on the aircraft if the DR400 landing gear has been installed in accordance with the SB 160403.



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Section N: <u>DR 340</u>

N.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 340
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	May 21, 1968
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	es DGAC-France Type Certificate no. 45
I <u>C</u>	Certification Basis	
1.	Reference Date for determining	
	the applicable requirements:	22 December 1967
2.	•	22 December 1967
2. 3.	the applicable requirements:	22 December 1967
3.	the applicable requirements: (Reserved)	22 December 1967 France AIR2052
3.	the applicable requirements: (Reserved) (Reserved)	
3. 4.	the applicable requirements: (Reserved) (Reserved) Certification Basis:	France AIR2052

N.II

1.	Reference Date for determining the applicable requirements:	22 December 1967
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6 th , 1966
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	. EASA Environmental Standards:	None

N.III Technical Characteristics and Operational Limitations

1. Type Design Definition	Refer to the CEAPR document 1003349
2. Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
	Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4. Dimensions:	Span
5. Engines:	Lycoming O-320-E2A
	The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level	
Sensenich	M 74 DMS-2-64 or M 74 DM-6S5-2-64	1.83 m (*)	2	2200	
Sensenich	M 74 DMS-0-64 or M 74 DM-6S5-0-64	1.88 m	2	2200 rpm	

Remarks: (*) No acceptable diameter reduction for repair. The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

7.2 Engine Oil:

80/87 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

- 8. Fluid capacities:
 - 8.1 Fuel:

Main tank		RH wir	ng tank	LH wing tank Auxiliary		ry tank	
(lit	ers)	(lite	ers)	(lite	ers)	(optiona	I) (liters)
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
75	65	40	40	40	40	50	50

8.2 Oil:

9. Air speeds:

VNF	
Vc	
VA	
V_{FE}	

Refer to approved aircraft flight manual.

11. Operational Capability:

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

12. Maximum Masses:



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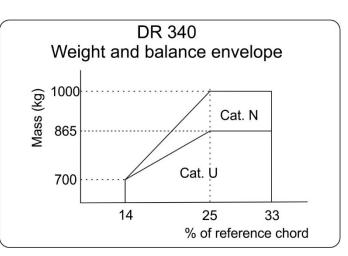
14. Datum:

DR 200, DR300, and DR400 series

DR 340

"N" Cate	"U" Category	
Take-off	Landing	
1000 kg (2205 lb)	950 kg (2094 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



Normal Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg

Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor (n) at maximum weight: Normal Category:

Flaps up n	+ 3.8
Flaps up n	- 1.9
Flaps down n	+ 2
Flaps down n	0

Utility Category:

	Flaps up n+ 4.4
	Flaps up n 2.2
	Flaps down n + 2
	Flaps down n0
16. Levelling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
19. Baggage / Cargo Compartment	Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR400 series DR 340	Date: 16 October 2023	
15540.05	51340		
20. Wheels and Tires	Main gear track Wheel tire size Front gear angular movement Tire pressure refer Oleo strut pressure refer		
21. Control surface movements:			
	Elevator:	+0° up 9°30' -30' +0° down 12° -30'	
		trailing edge of the wings	
	u		
	16° 15		
	Elevator tab: Elevator do	ator up: 25°30' ± 1°, 6° ± 1° wn: 10°30' ± 1°, 16°30' ± 1°	
	Flaps:		
	Rudder:	25° ^{+3°} 0°	

22. (Reserved)

N.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	Refer to latest amendment of service letter n°6

N.V <u>Note:</u>

1. This plane is identical to DR315 except:

powerplant

-

- addition of leading edge fuel tanks and 75 liters rear fuel tank
- wings profile at rectangular part

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Section O: <u>DR 315</u>

O.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 315
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	June 24, 1968
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
0.II <u>0</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	22 December 1967
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	. EASA Environmental Standards:	None
0.III <u>T</u>	echnical Characteristics and Oper	rational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003348
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Height
		Length6.96 m (22.83 ft) Wing Area13.60 m² (146.39 foot²)
5.	Engines:	Lycoming O-235-C2A or O-235-C2C
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state



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gasoline.

acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:

Propeller Manufacturer	Model	Maximum Continuous Power RPM 2800	
Evra	88-75-34 F		
	1 A 105 BCM 70-60	2600	
Mac Cauley	1 A 105 BCM 70-56	2600	
	1 B 90 ECM 72-50	2800	

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Evra	88-75-34 F	1.76 m	2	2250
	1 A 105 BCM 70-60	1.78 m	2	2250
Mac Cauley	1 A 105 BCM 70-56	1.78 m	2	2250
	1 B 90 ECM 72-50	1.83 m	2	2300

The EASA type certification standard includes that of FAA TC P-918 and FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:
 - 7.2 Engine Oil:

No. 1070. Refer to latest revision of Service Instruction Lycoming

octane minimum aviation grade Refer to latest revision of Service Instruction Lycoming

No. 1014.		, ,
Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

80/87



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- 8. Fluid capacities:
 - 8.1 Fuel:

8.2 Oil:

9. Air speeds:

Maii	n tank	Auxiliary tank		
(liters)		(optional) (liters)		
Capacity Usable		Capacity	Usable	
110	100	50	50	

Oil sump capacity	
Usable	

V_{NE}	
V_{NO}	
Vc	
VA	
V_{FE}	

10. Maximum Operating Altitude:

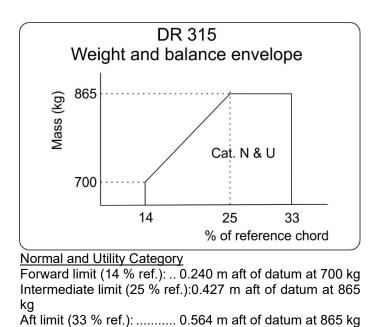
- 11. Operational Capability:
- 12. Maximum Masses:

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

"N" Cat	"U" Category	
Take-off	Landing	
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



Wing leading edge of the rectangular part of the wings.

14. Datum:

15. Load factor at maximum weight:

Normal Category:

	0,	
	Flaps up n	+ 3.8
	Flaps up n	- 1.9
	Flaps down n	+ 2
	Flaps down n	0
Utility	Category:	
	Flaps up n	+ 4.4
	Flaps up n	- 2.2
	Flaps down n	
	-	

Cord length at reference section: 1.71 m (5.61 ft)



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	Flaps	down n		0	
16. Levelling Means:	6. Levelling Means: Horizontal reference upper fuselage spar				
17. Minimum Flight Crew:	1 (pilot) at 0.	41±0.05m a	aft of datum	l	
18. Maximum Passenger Seating Capa		±0.05m aft at 1.19m aft		nd 2 (maxir	mum 120kg
19. Baggage / Cargo Compartment	Maximum ba aft of datum	ggage com	partment:	40 kg (88 l	b) at 1.90m
20. Wheels and Tires					
	Main gear tra	ick			8 m (8.46 ft)
	Wheel tire size	ze		380 x 1	50 or 500-5
	Front gear ar	ngular move	ement		left: 27°
					right: 27°
	Tire pressure		. refer to th	e maintena	nce manual
	Oleo strut pre	essure	. refer to th	e maintena	nce manual
21. Control surface movements:					
	Elevator:			d	+0°
	Ailerons:				
	Allei 0115		up	neutral	down
		-	16°30'	2°30'	11°10'
		l	15°	1°45'	9°45'
	Elevator tab:	Elovatoru	ID:	25°20' ⊥ 1°	6° ± 1°
	Elevator up:25°30' ± 1° 6° ± 1° Elevator down:10°30' ± 1°16°30' ± 1°				
	Flaps:	1st notch:		$15^{\circ} \pm 5^{\circ}$	
		and notes	1:	$+0^{\circ}$	
	Rudder:				25 ° +3°
22 (Reserved)					20 =•

22. (Reserved)

O.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	Refer to latest amendment of service letter n°6

O.V <u>Note:</u>



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Section P: DR 360

P.I General

	1.	a) Type:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 360
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	July 19, 1968
	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
P.II	<u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	22 December 1967
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966

6. Requirements elected to comply: None

- EASA Special Conditions: None
 EASA Exemptions: None
- 9. EASA Equivalent Safety Findings: None
- 10. EASA Environmental Standards: None

P.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003350		
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.		
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.		
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.		
4.	Dimensions:			
		Span8.72 m (28.61 ft)		
		Height2.23 m (7.32 ft)		
		Length		
		Wing Area14.20 m ² (152.85 foot ²)		



5. Engines:

Lycoming O-320-D2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm

6. Propellers:

-	Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
	Sensenich	M 74 DMS-2-66 or M 74 DM-6S5-2-66	1.83 m (*)	2	2150 rpm
		repair. The EASA typ TC P-886, b acceptance of September 20	e certification based on i r certification 03. Other sta ificated by in	n standard ndividual n of this s andards co dividual El	meter reduction for includes that of FAA EU member state standard prior to 28 nfirming to TC/TCDS U member state prior ptable.
Fluids:					
7.1 Fuel:		gasoline.			um aviation grade
72 Engin	e Oil [.]	Refer to lates	t revision of	f Service I	Instruction I vcoming

7.2 Engine Oil:

7. Fluids:

Refer to latest revision of Service Instruction Lycoming No. 1014.

NO. 1014.				
Air temperature	Ashless dispersant (AD)	Mineral		
	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

	0.11401	•							
		Main tank		RH wing tank		LH wing tank		Auxilia	ry tank
		(lit	ers)	(lite	ers)	(liters)		(optional) (liters)	
		Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
		75	65	40	40	40	40	50	50
	8.2 Oil:					ity			
9.	Air spee	ds:		V _{NE} V _{NO} V _C V _A V _{FE}				km/h (140 km/h (108	knots IAS) knots IAS) knots IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

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11. Operational Capability:

12. Maximum Masses:

Refer to approved aircraft flight manual.

"N" Ca	"U" Category	
Take-off	Landing	
1000 kg (2205 lb)	950 kg (2094 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:

	DR 360
	Weight and balance envelope
	D 1000 SE 865 700 Cat. N 14 25 33 % of reference chord 33
	Normal Category Forward limit (14 % ref.): 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 1000 kg Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg
	<u>Utility category</u> Forward limit (14 % ref.): 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865 kg Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg
14. Datum:	Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)
15. Load factor at maximum weight:	Normal Category: Flaps up n + 3.8 Flaps up n - 1.9 Flaps down n + 2 Flaps down n 0 Utility Category: Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n - 2.2 Flaps down n - 2.2 Flaps down n - 0
16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
19. Baggage / Cargo Compartment	Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum



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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR 360	DR400 series		Date: 16 C	october 2023
20. Wheels and Tires	Main gear tra Wheel tire siz Front gear ar Tire pressure	e igular mover	nent	380 x 15	50 or 500-5 left: 27° right: 27°
21. Control surface movements	Oleo strut pre	essure r	refer to the	e maintenar	nce manual
	Elevator:				••
	Ailerons:	Relativ wing <u>s.</u>		trailing eo	dge of the
			up	neutral	down
			16°30'	2°30'	11°10'
			15°	1°45'	9°45'
	Elevator tab:				
				25°30' ± 1° 10°30' ± 1°	
	Flaps:	1st notch:			15° ± 5°
		2nd notch:			
	Rudder:				25° $^{+3^{\circ}}_{-0^{\circ}}$

22. (Reserved)

P.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6

P.V Note:

1. This plane is identical to DR 340 except powerplant.



Section Q: DR 380

Q.I General

1	. а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 380
2	2. Airworthiness Category:	Normal Category
3	3. Type Certificate Holder:	Refer to Note 2 Section PP
4	. Manufacturer:	Refer to Note 3 Section PP
5	. (Reserved)	
6	 DGAC Type Certification date: 	May 29, 1969
7	2. EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8	3. The EASA type Certificates replace	es DGAC-France Type Certificate no. 45
Q.II	Certification Basis	
1	. Reference Date for determining the applicable requirements:	22 December 1967
2	. (Reserved)	
3	. (Reserved)	
4	. Certification Basis:	France AIR2052
Б	Airworthingga Paguiramonta:	France AIR2052 amondment lung 6th 1066

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

- 6. Requirements elected to comply: None
- 7. EASA Special Conditions: None
 8. EASA Exemptions: None
- 9. EASA Equivalent Safety Findings: None
- 10. EASA Environmental Standards: None

Q.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003350
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 7.08 m (23.23 ft) Wing Area 14.20 m² (152.85 foot²)
5.	Engines:	Lycoming O-360-A3A
		The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	76 EM 8S5-0-64 76 EM 8S5-0-68	1.93 m (1)	2	2250 (2)

Remarks:

(1) No acceptable diameter reduction for repair.

(2) Do not continuous operate between 2025 rpm and 2325 rpm.

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Auxiliary tank	LH tank		RH tank		Main tank	
ptional) (liters)	(liters)		ers)	(lite	(liters)	
pacity Usable	Usable	Capacity	Usable	Capacity	Usable	Capacity
50 50	40	40	40	40	65	75
K	Usable	Capacity	Usable	Capacity	Usable	Capacity

8.2 Oil:

9. Air speeds:

V _{NE}	305 km/h (165 knots IAS)
V _{NO}	270 km/h (146 knots IAS)
Vc	270 km/h (146 knots IAS)
V _A	200 km/h (108 knots IAS)
V _{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

- 11. Operational Capability:
- Refer to approved aircraft flight manual.
- 12. Maximum Masses:



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"N" Category				
Take-off	Landing			
1070 kg (2359 lb)	1020 kg (2249 lb)			

DR 380 Weight and balance envelope Image: Second S	13. Centre of Gravity Range:	
image: second		
Yest		Weight and balance envelope
Y Y		
At limit (33 % ref.):		s (k
700 12 24 33 % of reference chord Normal category Forward limit (12 % ref.): 0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.): 0.410 m aft of datum at 1070 kg Aft limit (33 % ref.):		Was
12 24 33 % of reference chord Normal category Forward limit (12 % ref.): 0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.):0.205 m aft of datum at 1070 kg Aft limit (33 % ref.):		Cat. N
% of reference chord Normal category Forward limit (12 % ref.): 0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.): 0.410 m aft of datum at 1070 kg Aft limit (33 % ref.):		700
Normal category Forward limit (12 % ref.):0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.): 0.410 m aft of datum at 1070 kg Aft limit (33 % ref.):		
Forward limit (12 % ref.): 0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.):0.410 m aft of datum at 1070 kg Aft limit (33 % ref.): 0.564 m aft of datum at 1070 kg 14. Datum: Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft) 15. Load factor at maximum weight: Normal Category: Flaps up n		
14. Datum: Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft) 15. Load factor at maximum weight: Normal Category: 14. Datum: Flaps up n 15. Load factor at maximum weight: Normal Category: 16. Leveling means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track 2.58 m (8.46 ft) Wheel tire size 380 x 150 or 5.00-5 Front gear angular movement left: 27°		Forward limit (12 % ref.): 0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.):0.410 m aft of datum at 1070 kg
Cord length at reference section: 1.71 m (5.61 ft) 15. Load factor at maximum weight: Normal Category: Flaps up n + 3.8 Flaps up n + 1.9 Flaps down n - 1.9 Flaps down n + 2 Flaps down n - 0 16. Leveling means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum. 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track 2.58 m (8.46 ft) Wheel tire size 380 x 150 or 5.00-5 Front gear angular movement left: 27°	14 Datum:	
Flaps up n + 3.8 Flaps up n - 1.9 Flaps down n + 2 Flaps down n - 0 16. Leveling means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track 2.58 m (8.46 ft) Wheel tire size 380 x 150 or 5.00-5 Front gear angular movement left: 27°		
Flaps up n - 1.9 Flaps down n + 2 Flaps down n 0 16. Leveling means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum. 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track 2.58 m (8.46 ft) Wheel tire size Main gear track	15. Load factor at maximum weight:	
Flaps down n0 16. Leveling means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum. 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track 2.58 m (8.46 ft) Wheel tire size Main gear angular movement		Flaps up n 1.9
 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum. 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track		
 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum. 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track	16. Leveling means:	Horizontal reference upper fuselage spar
datum. 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum 20. Wheels and Tires Main gear track	17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum
20. Wheels and Tires Main gear track	18. Maximum Passenger Seating Capa	
Main gear track	19. Baggage / Cargo Compartment	
Wheel tire size	20. Wheels and Tires	M is seen to all (0.40 ft)
right: 27°		
Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual		Tire pressure refer to the maintenance manual
21. Control surface movements	21. Control surface movements	
+0° Elevator: up		
+0°		+0°
down		Ailerons:down12° -30'Ailerons:Relative to the trailing edge of the
wings		wings
up neutral down 16°30' 2°30' 11°10'		

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DR 200, DR300, and DR400 series DR 380

		up	neutral	down
		15°	1°45'	9°45'
Elevator tab:				
	Elevator	up:	.25°30' ± 1°	6° ± 1°
	Elevator	down:	.10°30' ± 1°	16°30' ± 1°
Flaps:	1st notch	:		$\dots 15^{\circ} \pm 5^{\circ}$
				+0°
	2nd notch	า:		$+0^{\circ}$ 60° - 5°
Rudder:				25° ^{+3°} 0°

22. (Reserved)

Q.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6

Q.V Note:

1. This plane is identical to DR 340 except:

- Powerplant
- Structure
- Landing gears



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Section R: DR 300/108

R.I General

	1.	a) Type:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 300/108
:	2.	Airworthiness Category:	Normal and Utility Category
;	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
:	5.	(Reserved)	
	6.	DGAC Type Certification date:	June 18, 1970
	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
;	8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
R.II	<u>C</u>	Certification Basis	
	1.	Reference Date for determining the applicable requirements:	22 December 1967
:	2.	(Reserved)	
:	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
:	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	None
	8.	EASA Exemptions:	None
9	9.	EASA Equivalent Safety Findings:	None
	10	. EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.
	_		

R.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003347
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Wing Area13.60 m ² (146.39 foot ²)
5.	Engines:	Lycoming O-235-C2A or Lycoming O-235-C2C
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power: 2600 rpm Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufactur er	Model	Ø	Number of blades	Minimum static RPM at sea level
-	1 A 105 BCM 70-56	1.78 m	2	
Mac Cauley	1 A 105 BCM 70-60	1.78 m	2	2250 mana (*)
	1 B 90 ECM 72-50	1.83 m	2	2250 rpm (*)
Evra	88-75-34 F	1.76 m	2	

Remarks: (*) Maximum authorized RPM: 2600 rpm The EASA type certification standard includes that of FAA TC P-918 and FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids: 7.1 Fuel:

80/87 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank	
(lit	ters)	(optiona	I) (liters)
Capacity	Usable	Capacity	Usable
110	100	50	50

8.2 Oil:



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9. Air speeds:

VNE	
Vc	
Va .	
V_{FE} .	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

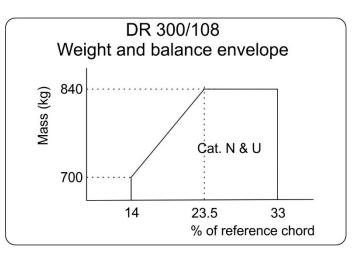
11. Operational Capability:

13. Centre of Gravity Range:

12. Maximum Masses:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Ca	tegory	"U" Category
Take-off	Landing	
840 kg (1852 lb)	840 kg (1852 lb)	840 kg (1852 lb)



Normal and Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (23.5 % ref.):0.401 m aft of datum at 840 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 840 kg

14. Datum:

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:		
J	Normal Category:	Flaps up n + 3.8 Flaps up n - 1.9 Flaps down n + 2 Flaps down n 0
	Utility Category:	Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n + 2 Flaps down n 0
16. Leveling Means:	Horizontal referen	ce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.	05m aft of datum
10 Maximum Dessences Conting Cone	aity 1 at 0 11 0 05	m off of dotum and 2 (maximum 100kg

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 (maximum 100kg (220lb)) at 1.19m aft of datum.

19. Baggage / Cargo Compartment: Not applicable



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20. Wheels and Tires:

Main gear track		
Wheel tire size		
Front gear angular	movement left: 27°	
	right: 27°	
Tire pressure	refer to the maintenance manual	
Oleo strut pressure	refer to the maintenance manual	

21. Control surface movements:

Elevator:	up			9°30', -30' +0°
Ailerons: wings		n itive to the		$12^{\circ} - 30'$ dge of the
Ū		up	neutral	down
		16°30'	2°30'	11°10'
		15°	1°45'	9°45'
Elevator tab:	Elev	ator up:	25°30' ± 1°	6° ± 1°
	Elev	ator down:1	0°30' ± 1°	16°30' ± 1°
Flaps:	1st r	notch:		$\dots 15^{\circ} \pm 5^{\circ}$
	2nd	notch:		$60^{\circ} + 0^{\circ} - 5^{\circ}$
Rudder:				$25^{\circ {}^{+3^{\circ}}_{-0^{\circ}}}$

22. (Reserved)

R.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6

R.V Note:

1. This plane is identical to DR 315 except cabin layout



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Section S: <u>DR 300/180 R</u>

S.I General

	1.	a) Type:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 300/180 R
	2.	Airworthiness Category:	Normal Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	July 24, 1970
	7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA Type Certificates replace	es DGAC-France Type Certificate no. 45
S.II	<u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	22 December 1967
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966
	5. 6.	Airworthiness Requirements: Requirements elected to comply:	France AIR2052 amendment June 6th, 1966 None
		•	
	6.	Requirements elected to comply:	None

10. EASA Environmental Standards: None

S.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003347
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Height2.23 m (7.32 ft) Length7.08 m (23.23 ft)
		Wing Area 13.60 m ² (146.39 foot ²)
5.	Engines:	Lycoming O-360-A3A
		The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:

2700 rpm

6. Propellers:

Manufactur	Model	Ø	Number	Minimum static RPM
er	Woder	Ø	of blades	at sea level
	76 EM 8S5-058	1.93 m (1)	2	2450 (2)
Sensenich	76 EM 8S5-064		2	2250 (2)
	76 EM 8S5-054		2	2500 (2)
Hoffmann	HO4-27HM-170-128	1.70 m	4	2240 (2)

Remarks:

(1) No acceptable diameter reduction for repair.

(2) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA and P6NE, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Mai	n tank	Auxiliary tank	
(lit	ers)	(optional) (liters)	
Capacity Usable		Capacity	Usable
110	100	50	50

8.2 Oil:

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14. Datum:

15. Load factor at maximum weight:

9. Air speeds:

V _{NO} V _C VA	
V_{FE}	170 km/h (92 knots IAS)
Vfe	170 km/h (92 knots IAS)

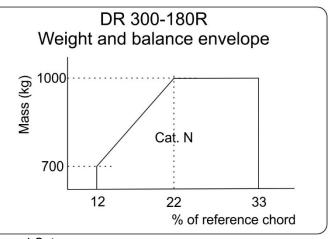
Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

10. Maximum Operating Altitude:

- 11. Operational Capability:
- 12. Maximum Masses:
- 13. Centre of Gravity Range:

"N" Category		
Take-off	Landing	
1000 kg (2205 lb)	950 kg (2094 lb)	



Normal Category Forward limit (12 % ref.):

Forward limit (12 % ref.): .. 0.205 m aft of datum at 700 kg Intermediate limit (22 % ref.):0.376 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

Normal Category:	Flaps up n	. + 3.8
	Flaps up n	1.9
	Flaps down n	+ 2
	Flaps down n	0

16. Leveling Means:	Horizontal reference upper fuselage spar	
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum	
18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.		
19. Baggage / Cargo Compartment	Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum	

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20. Wheels and Tires

Wheel tire size	2.58 m (8.46 ft)
FIOR year anyular	right: 27°
	refer to the maintenance manual e refer to the maintenance manual

21. Control surface movements

Elevator:	up			9°30' ^{+0°} +0°
Ailerons: wings		n itive to the		12° ^{-30'}
0		up	neutral	down
		16°30'	2°30'	11°10'
		15°	1°45'	9°45'
Elevator tab:		ator up:		
	Elev	ator down:	10°30' ± 1°	16°30' ± 1°
Flaps:	1st r	otch:		15° ± 5°
				$+0^{\circ}$
	2nd	notch:		60° – 5°
Rudder:				25° $^{+3°}_{-0°}$

22. (Reserved)

S.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6

S.V Note:

- 1. This plane is identical to DR 315 except:
 - powerplant
 - structure
 - landing gears
 - towing ability

2. Glider and banner towing Refer to approved flight manual

Takeoff maximum mass: 720kg Seaplane maximum mass towed:

550kg

Propeller approved for these operations: Sensenich 76 EM 8S5-058 Hoffmann H04-27HM-170-128

Section T: DR 300/140

T.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
1.		
	b) Model:	DR 300/140
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	February 22, 1971
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
т.II <u>с</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	22 December 1967
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	None
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	. EASA Environmental Standards:	None
т.ш т	echnical Characteristics and Oper	ational Limitations

T.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003347
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span8.72 m (28.61 ft) Height2.23 m (7.32 ft)
		Length
		Wing Area
5.	Engines:	Lycoming O-320-E2A
		The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum continuous power:

2700 rpm

6. Propellers:

Manufactur	Model	a	Number	Minimum static RPM
er	Model	Ø	of blades	at sea level
	M 74 DMS-2-64	1.83 m (*)	2	
Sanaaniah	74 DM 6S5-2-64		2	2200 rpm
Sensenich	M 74 DMS-0-64	1.00 m	2	2200 rpm
	74 DM 6S5-0-64	1.88 m	2	

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

80/87 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

- 8. Fluid capacities:
 - 8.1 Fuel:

		Main tank (liters)		Auxiliary tank (optional) (liters)		
		Capacity	Usable	Capacity	Usable	
		110	100	50	50	
8	3.2 Oil:			8 U.S. qua 6 U.S. qua	· · · · ·	
9. A	Air speeds:					
	•	V _{NE}		295 km/h	(159 knots IAS)	
				260 km/h		
		V _c		260 km/h	(140 knots IAS)	
		Va			(108 knots IAS)	
		V _{FE}		170 km/h	(92 knots IAS)	
10. Maximum Operating Altitude:		Refer to approved aircraft flight manual.				
11. Operational Capability:		Refer to approved aircraft flight manual.				

12. Maximum Masses:



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DR 200, DR300, and DR400 series

DR 300/140

"N" Ca	"U" Category	
Take-off	Landing	
1000 kg (2205 lb)	950 kg (2094 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:

13. Centre of Gravity Range:		
		DR 300-140
	Weigh	t and balance envelope
	1000 ھ	
	(b) see 865	Cat. N
	₩ 865	
	700	Cat. U
		14 25 33 % of reference chord
	Normal Catagory	
	Normal Category Forward limit (14	% ref.): 0.240 m aft of datum at 700 kg
	Intermediate limit	(25 % ref.):0.427 m aft of datum at 1000
	kg Aft limit (33 % ref	.): 0.564 m aft of datum at 1000 kg
	Utility Category	
	Forward limit (14	% ref.): 0.240 m aft of datum at 700 kg
	Intermediate limit	(25 % ref.):0.427 m aft of datum at 865
		.): 0.564 m aft of datum at 865 kg
14. Datum:		ge of the rectangular part of the wings. ference section: 1.71 m (5.61 ft)
15. Load factor at maximum weight:		
	Normal Category	: Flaps up n + 3.8
		Flaps up n 1.9 Flaps down n + 2
		Flaps down n0
	Utility Category:	Flaps up n+ 4.4
	·) · 3)	Flaps up n 2.2
		Flaps down n+ 2 Flaps down n0
16. Leveling Means:	Horizontal referen	nce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0).05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.0 datum.	05m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum baggag aft of datum	ge compartment: 40 kg (88 lb) at 1.90m
20. Wheels and Tires		
	Main gear track Wheel tire size	2.58 m (8.46 ft)
		ar movement left: 27°
	-	right: 27°

.....right: 27°



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Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual

21. Control surface movements

110					+0°
	Elevator:	up			9°30' - ^{30'}
					+0°
			า		
	Ailerons:	Rela	tive to the	trailing e	dge of the
	wings				
			up	neutral	down
			16°30'	2°30'	11°10'
			15°	1°45'	9°45'
	Elevator tab:	Eleva	ator up:	25°30' ± 1°	6° ± 1°
		Eleva	ator down:	10°30' ± 1°	16°30' ± 1°
	Flaps:	1st n	otch:		15° ± 5°
	•				$+0^{\circ}$
		2nd	notch:		60° - 5°
	Rudder:				+3°

22. (Reserved)

T.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6

T.V Note:

1. This plane is identical to DR 315 except powerplant



Section U: DR 300/125

U.I General

-	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 300/125
2	2.	Airworthiness Category:	Normal and Utility Category
3	3.	Type Certificate Holder:	Refer to Note 2 Section PP
2	4.	Manufacturer:	Refer to Note 3 Section PP
Ę	5.	(Reserved)	
6	6.	DGAC Type Certification date:	May 11, 1971
7	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
U.II	<u>c</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	22 December 1967
4	2.	(Reserved)	
	2. 3.	(Reserved) (Reserved)	
3	3.	. ,	France AIR2052
3	3. 4.	(Reserved)	France AIR2052 France AIR2052 amendment June 6th, 1966
3	3. 4.	(Reserved) Certification Basis:	

- 8. EASA Exemptions: None
- 9. EASA Equivalent Safety Findings: None
- 10. EASA Environmental Standards: None

U.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003347
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Height2.23 m (7.32 ft) Length6.96 m (22.83 ft)
		Wing Area 13.60 m ² (146.39 foot ²)
5.	Engines:	Lycoming O-235-F2B or O-235-F2A or O-235-J2A
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:

2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-54	1.80 m	2	2300 rpm (*)

Remarks: (*) Do not continuous operate between 2025 rpm and 2325 rpm.

The EASA type certification standard includes that of FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

No. 1070 Refer to latest revision of Service Instruction Lycoming No. 1014

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming

110. 1014.		
Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

0.11 del.					
		Main tank (liters)		Auxiliary tank (optional) (liters)	
		(111	615)	(optional) (liters)	
		Capacity	Usable	Capacity	Usable
		110	100	50	50
8.2 Oil:	Oil sump cap Usable				
9. Air speeds:					
·	V _{NE}				knots IAS)
	V _{NO}		260	km/h (140	knots IAS)
	Vc		260	km/h (140	knots IAS)
	V _A		200	km/h (108	knots IAS)
	Vfe		170	km/h (92	knots IAS)
10. Maximum Operating Altitude:	Refer to appr	oved aircra	ft flight man	ual.	
11. Operational Capability:	Refer to appr	oved aircra	ft flight man	ual.	

12. Maximum Masses:

"N" Category		"U" Category
Take-off	Landing	



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865 kg (1907 lb)

865 kg (1907 lb)

865 kg (1907 lb)

13. Centre of Gravity Range:	
	DR 300/125
	Weight and balance envelope
	(b) 865 see Cat. N & U 700 700
	14 25 33
	% of reference chord
	Normal and Utility Category Forward limit (14 % ref.): 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865 kg Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg
14. Datum:	Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)
15. Load factor at maximum weight:	Normal Category: Flaps up n+ 3.8 Flaps up n 1.9 Flaps down n+ 2 Flaps down n0
	Utility Category: Flaps up n+ 4.4 Flaps up n 2.2 Flaps down n+ 2 Flaps down n0
16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum
18. Maximum Passenger Seating Capa	acity: 1 at 0.41±0.05m aft of datum and 2 (maximum 120kg (265lb)) at 1.19m aft of datum.
19. Baggage / Cargo Compartment	Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum



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TCDS No: EASA.A.367	DR 200, DR300, and DR40	00 series
Issue: 05	DR 300/125	Date: 16 October 2023
20. Wheels and Tires		
	Main gear track	2.58 m (8.46 ft)
	Front gear angui	ar movement left: 27° right: 27°
		refer to the maintenance manual are refer to the maintenance manual
21. Control surface movements		
		+0°
	Elevator:	up
		down
	Ailerons:	down12° ^{-30′} Relative to the trailing edge of the
	wings	Relative to the training edge of the
	0	up neutral down
		16°30' 2°30' 11°10'
		15° 1°45' 9°45'
	Elevator tab:	Elevator up: $25^{\circ}30' \pm 1^{\circ}$ $6^{\circ} \pm 1^{\circ}$
	Flana	Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:15° ± 5°
		$^{+0^{\circ}}_{2nd notch:60^{\circ} - 5^{\circ}}$
	Rudder:	

22. (Reserved)

U.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6

U.V Note:

1. This plane is identical to DR 315 except powerplant

Section V: DR 300/120

V.I General

	1.	a) Type:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 300/120
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	February 11, 1975
	7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
V.II	<u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	3 February 1975
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	None
	8.	EASA Exemptions:	None

- 9. EASA Equivalent Safety Findings: None
- 10. EASA Environmental Standards: None

V.III Technical Characteristics and Operational Limitations

1.	Type Design Definition	Refer to the CEAPR document 1003347
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Height2.23 m (7.32 ft) Length6.96 m (22.83 ft)
		Wing Area 13.60 m ² (146.39 foot ²)
5.	Engines:	Lycoming O-235-L2A or O-235-K2A or O-235-K2B
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



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September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:

2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-47	1.80 m 1.77 m (1)	2	2200 (2)
Hoffmann	HO 14-178/115	1.78 m 1.73 m (1)	2	2250
Sensenich	72 CKS6-0-56	1.83 m (3)	2	2220
Sensenich	72 CKS5-0-56	1.03 III (3)	2	2220

Remarks:

No. 1070

(1) Minimum diameter after repair.

(2) Do not continuous operate between 2025 rpm and 2325 rpm.

(3) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-842 and FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

Refer to latest revision of Service Instruction Lycoming No. 1014.

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming

Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank	
(liters)		(optional) (liters)	
Capacity	Usable	Capacity	Usable
110	100	50	50

8.2 Oil:	Oil sump capacity6 U.S. quarts (5.68 liters) Usable4 U.S. quarts (3.79 liters)
9. Air speeds:	V _{NE}

VNE	
V_{NO}	
Vc	
VA	
V_{FE}	170 km/h `(92 knots IAS)

* * * * * * *

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^{7.2} Engine Oil:

10. Maximum Operating Altitude:

11. Operational Capability:

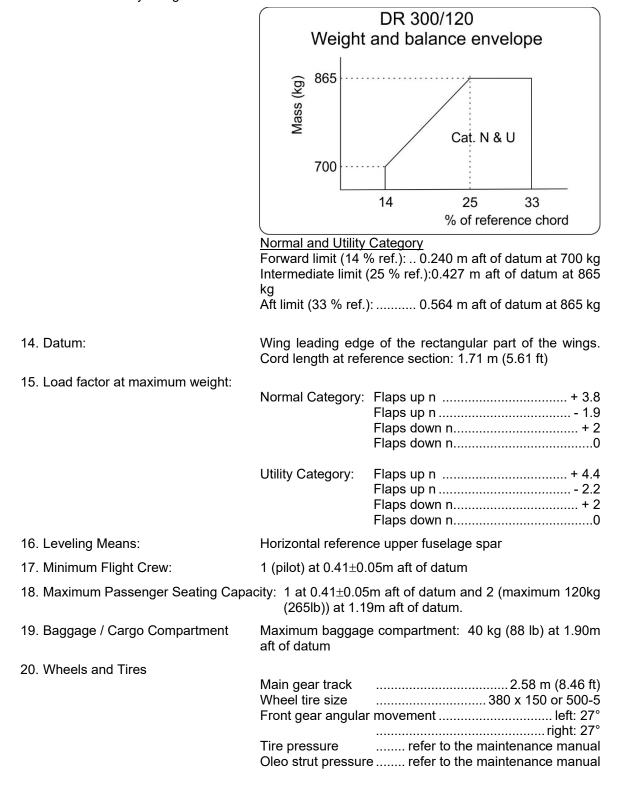
12. Maximum Masses:

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

"N" Ca	tegory	"U" Category
Take-off	Landing	
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:





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21. Control surface movements

up		
up	neutral	down
16°30'	2°30'	11°10'
15°	1°45'	9°45'
Elevator up:.	25°30' ± ′	1°6° ± 1°
Elevator dow	/n: 10°30' ± ′	1° 16°30' ± 1°
1st notch:		15° ± 5°
2nd notch:		+0° 60° -5°
		25° ^{+3°} 0°
	down Relative to <u>up</u> 16°30' 15° Elevator up:. Elevator dow 1st notch: 2nd notch:	16°30' 2°30'

22. (Reserved)

V.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6

V.V <u>Note:</u>

1. This plane is identical to DR 315 except powerplant



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Section W: <u>DR 400/125</u>

W.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/125
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	May 10, 1972
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
W.II <u>C</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	21 March 1971
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052

-	•	
5	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6	Requirements elected to comply:	None
7	EASA Special Conditions:	Canopy emergency release system
8	EASA Exemptions:	None
9	EASA Equivalent Safety Findings:	None
1	0. EASA Environmental Standards:	None

W.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed
4.	Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft) Wing Area 13.60 m² (146.39 foot²)
5.	Engines:	Lycoming O-235-F2B or O-235-J2A
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power: 2800 rpm

6. Propellers:

	N	lanufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
	Μ	lac Cauley	1 A 135 JCM 71-54	1.80 m	2	2300 rpm (*)
			Remarks: (*) Do not c	ontinuous d	operate between 2025
			rpm and 23	25 rpm.		
	The EASA type certification standard includes that of FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.					I EU member state standard prior to 28 confirming to TC/TCDS EU member state prior
7.	Fluids:					
	7.1 Fuel:					n grade gasoline. nstruction Lycoming
	7.2 Engine	Oil:	Refer to lat No. 1014.	test revision	of Service	e Instruction Lycoming

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

		Main tank		Auxiliary tank	
		(liters)		(optional) (liters)	
		Capacity	Usable	Capacity	Usable
		110	100	50	50
8.2 Oil:	Oil sump capa Usable				
	USable			S. quaits (5.79 mers)
9. Air speeds:					
	V _{NE}			km/h (166	
	V _{NO}				
	Vc				
	Va		215	km/h (116	knots IAS)
	V _{FE}		170	km/h (92	knots IAS)
10. Maximum Operating Altitude:	Refer to appre	oved aircra	ft flight manu	ual.	
11. Operational Capability:	Refer to appre	oved aircra	ft flight manu	ual.	
12. Maximum Masses:					
	"N"	Category		"U" Ca	tegory



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DR 200, DR300, and DR400 series

DR 400/125

Date: 16 October 2023

Take-off	Landing	
900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)

13. Centre of Gravity Range:

	(j) 900 See 750	DR 400/125 and balance envelope Cat. N & U 12 25 33	
		% of reference chord	
	Intermediate limit kg	<u>Category</u> % ref.): 0.205 m aft of datum at 750 kg (25 % ref.):0.428 m aft of datum at 900): 0.564 m aft of datum at 900 kg	
14. Datum:		e of the rectangular part of the wings. erence section: 1.71 m (5.61 ft)	
15. Load factor at maximum weight:	Normal Category:	Flaps up n	
	Category Utility:	Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n + 2 Flaps down n 0	
16. Leveling Means:	Horizontal referen	ce upper fuselage spar	
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum		
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.0 datum.	5m aft of datum and 2 at 1.19m aft of	
19. Baggage / Cargo Compartment	Maximum baggag aft of datum	e compartment: 40 kg (88 lb) at 1.90m	



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20. Wheels and Tires:		
	Main gear track	2.58 m (8.46 ft)
	Wheel tire size	
	Front gear angula	ar movement left: 27°
		right: 27°
	Tire pressure	refer to the maintenance manual
		re refer to the maintenance manual

21. Control surface movements

E	levator:	up down		
	ilerons: ings			edge of the
		up	neutral	down
		15°± 1°	2° ± 1°	10°± 1°
E	levator tab:	Elevator up: Elevator dow		
F	laps:			15° ± 5°
		2nd notch:		
R	udder:			25° ^{+3°} _{-0°} (1)
) For planes fitte edals:			
	16° (-0°, +2°) b	petore oberati	na arum brak	es

 16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

W.IV Operating and Service Instructions

W.V Note:

- 1. This plane is identical to DR 300/180R except:
 - powerplant
 - centre and front parts of the fuselage
 - forward sliding canopy
 - fuel circuit



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Section X: DR 400/140

X.I General

1	. a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/140
2	Airworthiness Category:	Normal and Utility Category
3	. Type Certificate Holder:	Refer to Note 2 Section PP
4	Manufacturer:	Refer to Note 3 Section PP
5	(Reserved)	
6	DGAC Type Certification date:	December 01, 1972
7	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
X.II	Certification Basis	
1	. Reference Date for determining the applicable requirements:	21 March 1971
2	(Reserved)	
3	(Reserved)	
4	Certification Basis:	France AIR2052
5	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6	Requirements elected to comply:	None
7	EASA Special Conditions:	Canopy emergency release system
8	EASA Exemptions:	None
9	EASA Equivalent Safety Findings:	None
1	0. EASA Environmental Standards:	None

X.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft) Wing Area 13.60 m² (146.39 foot²)
5.	Engines:	Lycoming O-320-E2A

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The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:

2700 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
	M 74 DMS-2-64	1.83 m (*)	2	2200 rpm
Sensenich	74 DM 6S5-2-64		2	
Sensenich	M 74 DMS-0-64	1.00 m	2	2200 rpm
	74 DM 6S5-0-64	1.88 m	2	

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

80/87 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Ashless dispersant (AD)	Mineral			
grades	grades			
SAE15W50 or SAE20W50				
SAE60	SAE60			
SAE40 or SAE50	SAE50			
SAE40	SAE40			
SAE30, SAE40 or SAE20W40	SAE30			
SAE20W50 or SAE15W50	SAE20W50			
SAE30 or SAE20W30	SAE20			
	grades SAE15W50 or SAE20W50 SAE60 SAE40 or SAE50 SAE40 SAE30, SAE40 or SAE20W40 SAE20W50 or SAE15W50			

8. Fluid capacities:

8.1 Fuel:

Main tank (liters) Capacity Usable		Auxiliary tank (optional) (liters)		
		Capacity	Usable	
110	100	50	50	
Oil sump capa Usable	acity		uarts (7.57 liters uarts (5.68 liters	
V _{NE}			n (166 knots IAS	
V _{NO}		260 km/ł		
V _{NO}			n (140 knots IAS n (140 knots IAS	
		260 km/ł	n (140 knots IAS	

8.2 Oil:

9. Air speeds:

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10. Maximum Operating Altitude:

11. Operational Capability:

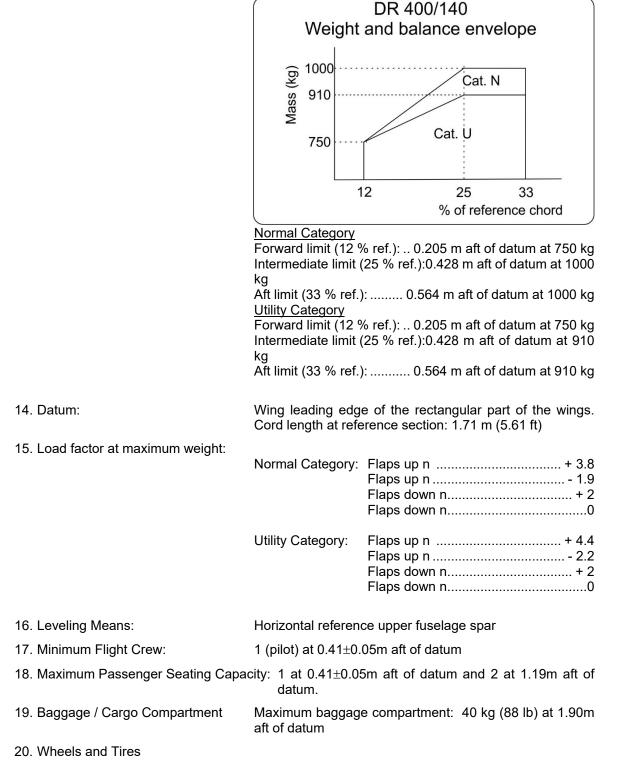
12. Maximum Masses:

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

"N" Category		"U" Category
Take-off	Landing	
1000 kg (2205 lb)	1000 kg (2205 lb)	910 kg (2006 lb)

13. Centre of Gravity Range:





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Main gear track	2.58 m (8.46 ft)
Wheel tire size	
Front gear angul	ar movement left: 27°
	right: 27°
	refer to the maintenance manual
Oleo strut pressu	re refer to the maintenance manual
Elevator:	up

21. Control surface movements

Elevator:	up		9°30' ± 30'
	down		12° ± 30'
Ailerons:	Relative to	the trailing	edge of the
wings		_	-
-	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	Elevator up:	25°30' ± 7	1°6° ± 1°
	Elevator dow	/n: 10°30' ± ′	1° 16°30' ± 1°
Flaps:	1st notch:		15° ± 5°
			$+0^{\circ}$
	2nd notch:		60° – 5°
Rudder:			25° ^{+3°} 0° (1)

(1) For planes fitted with brakes controlled with rudder pedals:

 16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

X.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Schedule....... Refer to the latest amendment of Service Letter no. 6

X.V Note:

1. This plane is identical to DR 400/125 except powerplant



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Section Y: DR 400/160

Y.I General

	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400/160
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	September 06, 1972
	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	BOGAC-France Type Certificate no. 45
Y.I	<u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	21 March 1971
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	5. 6.	-	France AIR2052 amendment June 6th, 1966
	6.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6. 7.	Airworthiness Requirements: Requirements elected to comply:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7 None
	6. 7.	Airworthiness Requirements: Requirements elected to comply: EASA Special Conditions:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7 None Canopy emergency release system

Y.III Technical Characteristics and Operational Limitations

1. Type Design Definition:	Refer to CEAPR document n°1001131 for DR400 "STANDARD 88" (Refer to note 2)
	Refer to CEAPR document n°1001130 for DR400 "STANDARD 92" (Refer to note 2)
2. Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
	Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed
4. Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft) - Round spinner 7.10 m (23.29 ft) - Sharp spinner



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5. Engines:

Wing Area 14.20 m ²	(152.85 foot ²)

Lycoming O-320-D2A The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm Remarks: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufactur er	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	74 DM 6S5-2-66	1.83 m (*)	2	2150 rpm
Sensenich	74 DM 6S5-2-64	1.03 m ()	2	2250 rpm

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

91/96 or 100/130 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

-		
Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank RH tank		LH tank		Auxiliary tank			
(lit	(liters) (liters)		(liters)		(optional) (liters)		
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial no. 2210, unusable quantity of fuel reduced from 10 liters to 1 liter, (Refer to note 2).

8.2 Oil:

Oil sump capacity 8 U.S. quarts (7.57 liters)



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9. Air speeds:

V _{NO}	308 km/h (166 knots IAS) 260 km/h (140 knots IAS) 260 km/h (140 knots IAS)
VA	215 km/h (116 knots IAS) 170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

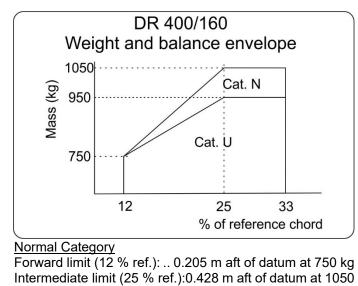
Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

12. Maximum Masses:

11. Operational Capability:

"N" Ca	"U" Category	
Take-off	Landing	
1050 kg (2315 lb)	1045 kg (2304 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:



kg Aft limit (33 % ref.): 0.564 m aft of datum at 1050 kg Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 950 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:		
Ũ	Normal Category:	Flaps up n+ 3.8
		Flaps up n 1.9
		Flaps down n+ 2
		Flaps down n0
	Litility Cotogony	
	Utility Category:	Flaps up n+ 4.4
		Flaps up n 2.2
		Flaps down n+ 2
		Flaps down n0

16. Leveling Means:

14. Datum:

Horizontal reference upper fuselage spar

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17. Minimum Flight Crew:	1 (pilot) at 0.41±0).05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.0 datum.	05m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum bagga aft of datum	ge compartment: 40 kg (88 lb) at 1.90m
20. Wheels and Tires		
	Wheel tire size Front gear angula	2.58 m (8.46 ft)
	Tire pressure	
21. Control surface movements		
	Elevator:	up
	Ailerons: wings	Relative to the trailing edge of the
		up neutral down
		15°± 1° 2° ± 1° 10°± 1°
	Elevator tab:	Elevator up:25°30' ± 1°6° ± 1°
	Flame	Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:15° ± 5° +0°
		2nd notch:
	16° (-0°, +2	2°) before operating drum brakes 3°) before operating disk brakes

22. (Reserved)

Y.IV Operating and Service Instructions

Airplane Flight Manual Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual. Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance ScheduleRefer to the latest amendment of Service Letter no. 6

Y.V <u>Note:</u>

1. This plane is identical to DR 400/125 except:

- powerplant
- leading edge of centre part of the wings
- leading edge fuel tanks
- luggage compartment door
- 2. "Standard 92" models: Since February 1994 (from serial nr. 2220 included)

"Standard 88" models: Before February 1994 (before serial nr.2220 excluded)



Section Z: <u>DR 400/180</u>

Z.I General

Z.II

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/180
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	May 10, 1972
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
I <u>C</u>	ertification Basis	
1.	Reference Date for determining the applicable requirements:	21 March 1971
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	Canopy emergency release system
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.

Z.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition:	Refer to CEAPR document n°1001131 for DR400 "STANDARD 88" (Refer to note 2)
	Refer to CEAPR document n°1001130 for DR400 "STANDARD 92" (Refer to note 2)
2. Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4. Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft) - Round spinner 7.10 m (23.29 ft) - Sharp spinner



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5. Engines:

Wing Area 14.20 m² (152.85 foot²)

Lycoming O-360-A3A or O-360-A1A or O-360-A1P (*)

(*) from serial nr 2207 included

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power: 2700 rpm

6. Propellers:

Manufactur	lanufactur Model		Number	Minimum static RPM	
er	INIOUEI	Ø	of blades	at sea level	
	76 EM 8S5-0-54	1.02 m (1)	2	2500 rpm (3)	
Sensenich	76 EM 8S5-0-58		2	2500 rpm (3)	
Sensenich	76 EM 8S5-0-64	1.93 m (1)	2	2200 rpm (3)	
	76 EM 8S5-0-68		2	2250 rpm (3)	
Hoffmann	HO 27 HM/180/160	1.80 m (2)	2	2350 rpm	

Remarks:

(1) No acceptable diameter reduction for repair.

(2) When Hoffmann HO 27 installed, major change nr 35 must be applied.

(3) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA and FAA TC P3EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level		
Hoffmann	HO V 123 K/180R	1.80 m	3	Woodward B 210- 689	Constant speed (4)		

Remarks:

(4) Modification of engine from O-360-A3A to O-360-A1A The EASA type certification standard includes that of FAA TC P5EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

Lycoming O-360-A3A or Lycoming O-360-A1A:

100/100LL octane minimum aviation grade gasoline. Lycoming O-360-A1P:

91/96 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.



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7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming

NO. 1014.					
Air temperature	Ashless dispersant (AD) grades	Mineral grades			
All temperature	SAE15W50 or SAE20W50				
Above 80°F (+25°C)	SAE60	SAE60			
Above 60°F (+15°C)	SAE40 or SAE50	SAE50			
30°F to 90°F (O°C à +30°C)	SAE40	SAE40			
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30			
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50			
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20			

8. Fluid capacities:

8.1 Fuel:

•							
Main tank RH tank		tank	LH tank		Auxiliary tank		
(lit	(liters) (liters)		(liters)		(optional) (liters)		
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial no. 2210, unusable quantity of fuel reduced from 10 liters to 1 liter, (refer to note 2).

8.2 Oil:

9. Air speeds:

V _{NO}	308 km/h (166 knots IAS) 260 km/h (140 knots IAS)
Vc V _A V _{FE}	215 km/h (116 knots IAS)

10. Maximum Operating Altitude:

11. Operational Capability:

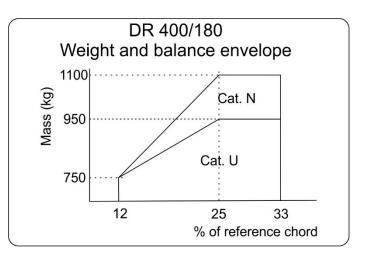
12. Maximum Masses:

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

"N" Category		"U" Category
Take-off	Landing	
1100 kg (2425 lb)	1045 kg (2304 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:





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	,,	
	Intermediate limit kg Aft limit (33 % ref. <u>Utility Category</u> Forward limit (12 ° Intermediate limit	% ref.): 0.205 m aft of datum at 750 kg (25 % ref.):0.428 m aft of datum at 1100): 0.564 m aft of datum at 1100 kg % ref.): 0.205 m aft of datum at 750 kg (25 % ref.):0.428 m aft of datum at 950
	kg Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg
14. Datum:		e of the rectangular part of the wings. erence section: 1.71 m (5.61 ft)
15. Load factor at maximum weight:	Normal Category:	Flaps up n
	Utility Category:	Flaps up n+ 4.4 Flaps up n 2.2 Flaps down n+ 2 Flaps down n0
16. Leveling means:	Horizontal referen	ce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.	.05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.0 datum.	5m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum baggag aft of datum	e compartment: 60 kg (132 lb) at 1.90m
20. Wheels and Tires:	Main gear track Wheel tire size Front gear angula	
	Tire pressure Oleo strut pressur	right: 27° refer to the maintenance manual re refer to the maintenance manual
21. Control surface movements:	Elevator: Ailerons:	up
	wings	up neutral down
	Elevator tab:	$15^{\circ}\pm 1^{\circ}$ $2^{\circ}\pm 1^{\circ}$ $10^{\circ}\pm 1^{\circ}$ Elevator up:25°30' ± 1°6° ± 1°Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:15° \pm 5° $+0^{\circ}$
	Duddor	2nd notch:
Rudder:		
	pedals: 16° (-0°, +2	ted with brakes controlled with rudder (°) before operating drum brakes (°) before operating disk brakes

**** ****
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22. (Reserved)

Z.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

Z.V Note:

1. This plane is identical to DR 400/160 except:

Powerplant

- towing ability (if equipped with towing hook)

2. "Standard 92" models: Since October 1993 (serial nr. 2207 and from serial nr. 2216 included)

"Standard 88" models: Before October 1993 (serial nr.2207 excluded and before serial nr.2216 excluded)



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Section AA: DR 400/180 R

AA.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/180 R
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	November 28, 1972
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
AA.II (Certification Basis	
_		
1.	Reference Date for determining the applicable requirements:	3 August 1972
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	Canopy emergency release system
		Airplane and towed sailplane maximum masses are limited considering the minimum climb performances required.
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10	. EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.
AA.III	Technical Characteristics and Op	erational Limitations

 Type Design Definition: Refer to CEAPR document n°1001131 for DR400 "STANDARD 88" Refer to CEAPR document n°1001130 for DR400 "STANDARD 92"
 Description: Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
 Equipment: The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or

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APR 79.88.00 or approved equivalent must be installed.

(*) from serial nr 2207 included

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2600 rpm Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number	Minimum static RPM
Manufacturer			of blades	at sea level
	76 EM 8S5-0-54		2	2500
Sensenich	76 EM 8S5-0-58	1.93 m (*)	2	2400
	76 EM 8S5-0-64		2	2300
Hoffmann	HO 27 HM/180/138	1.00 m	2	2400
Evra	TR5-180-102-140	1.80 m	3	

The EASA type certification standard includes that of FAA TC P4EA and FAA TC P3EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

Lycoming O-360-A3A: 100/100LL octane minimum aviation

grade gasoline.

Lycoming O-360-A1P: 91/96 octane minimum aviation grade

gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



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- 8. Fluid capacities:
 - 8.1 Fuel:

Mai	n tank	Auxiliary tank		
(liters)		(optional) (liters)		
Capacity	Usable	Capacity	Usable	
110	100/109 (1)	50	50	

(1) New standard called "Standard 92" from serial no.
2210, unusable quantity of fuel reduced from 10 liters to
1liter, (refer to note 2).

8.2 Oil:	Oil sump capacity	8 U.S.	quarts	(7.57 liters)
	Usable	6 U.S.	quarts	5.68 liters)

9. Air speeds:

V_{NE}	
Vc	
VA	
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

11. Operational Capability:

12. Maximum Masses:

"N" Category		"U" Category
Take-off	Landing	
1000 kg (2205 lb)	1000 kg (2205 lb)	910 kg (2006 lb)

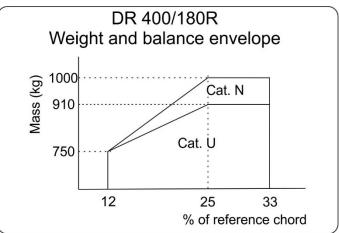
Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

12.1 Towing mass limitations:

13. Centre of Gravity Range:

Each maximum mass of the tug and of the towed glider is limited by the minimum climb performance.



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg <u>Utility Category</u>

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 910 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 910 kg



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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR40 DR 400/180 R	0 series Date: 16 October 2023
14. Datum:		ge of the rectangular part of the wings. Ference section: 1.71 m (5.61 ft)
15. Load factor at maximum weight		: Flaps up n + 3.8 Flaps up n 1.9 Flaps down n
	Category Utility:	Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n + 2 Flaps down n 0
16. Leveling Means:	Horizontal referer	nce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0	0.05m aft of datum
18. Maximum Passenger Seating C	apacity: 1 at 0.41±0.0 datum.)5m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum baggag aft of datum	ge compartment: 60 kg (132 lb) at 1.90m
20. Wheels and Tires:	Tire pressure	2.58 m (8.46 ft)
21. Control surface movements:	Elevator: Ailerons: wings	up $9^{\circ}30' \pm 30'$ down $12^{\circ} \pm 30'$ Relative to the trailing edge of the up neutral 15^{\circ} \pm 1^{\circ} $2^{\circ} \pm 1^{\circ}$ 10^{\circ} \pm 1^{\circ} Elevator up: $25^{\circ}30' \pm 1^{\circ}6^{\circ} \pm 1^{\circ}$
	Elevator tab:	Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$ 1st notch:
	Flaps:	$+0^{\circ}$
		2nd notch:
	Rudder:	
22. (Reserved)	pedals: 16° (-0°, +2	tted with brakes controlled with rudder 2°) before operating drum brakes 3°) before operating disk brakes



AA.IV Operating and Service Instructions

AA.V Note:

1. This plane is identical to DR 400/125 except:

- powerplant
- towing ability
- landing gears
- rearview mirror and rear panoramic windows
- 2. "Standard 92" models: Since October 1993 (serial nr.2207 and from serial nr. 2216 included)

"Standard 88" models: Before October 1993 (serial nr. 2207 excluded and before serial nr.2216 excluded)

3. Glider and banner towing:

Refer to approved flight manual.



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Section BB: DR 400/2+2

BB.I General

1.	a) Type:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/2+2
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	December 19, 1972
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
BB.II	Certification Basis	
1.	Reference Date for determining the applicable requirements:	3 August 1972
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	Canopy emergency release system
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
1(). EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.
BB.III	Technical Characteristics and Op	perational Limitations
	Tura Dasian Dafinitian	

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.
		Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed
4.	Dimensions:	
		Span8.72 m (28.61 ft)
		Height2.23 m (7.32 ft)
		Length6.96 m (22.83 ft) Wing Area13.60 m² (146.39 foot²)
5.	Engines:	Lycoming O-235-H2C or O-235-C2C
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2600 rpm Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
	1 A 105 BCM 70-56	1.78 m	2	
Mac Cauley	1 A 105 BCM 70-60	1.70 m	2	2250
	1 A 90 ECM 72-50	1.83 m	2	2250
Evra	88-75-34 F	1.76 m	2	

The EASA type certification standard includes that of FAA TC P-918 and FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

80/87 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

100.1014.		
Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

- 8. Fluid capacities:
 - 8.1 Fuel:

Mai	n tank	Auxiliary tank		
(liters)		(optional) (liters)		
Capacity	Usable	Capacity	Usable	
110	100	50	50	

8.2 Oil:

Oil sump capacity	6 U.S.	quarts	(5.68 liters))
Usable	4 U.S.	quarts	(3.79 liters)

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^{7.1} Fuel:

9. Air speeds:

V_{NE}	
V_{NO}	
Vc	
VA	215 km/h (116 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

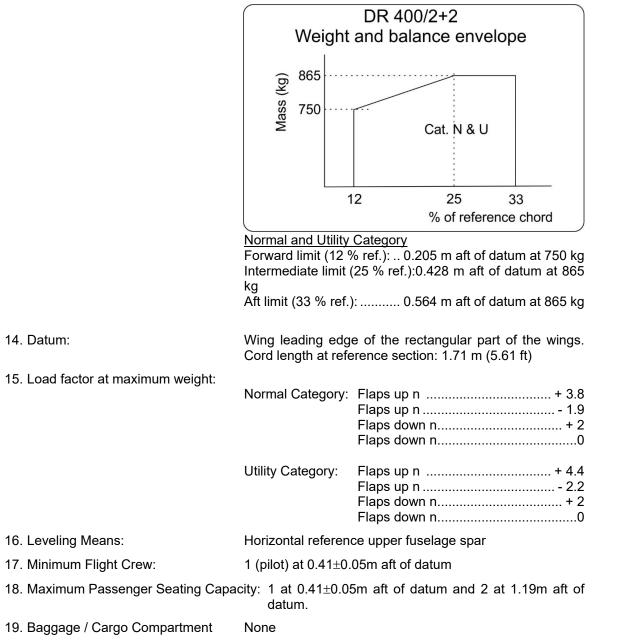
11. Operational Capability:

12. Maximum Masses:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Category		"U" Category
Take-off Landing		
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR40 DR 400/2+2	00 series Date: 16 October 2023
20. Wheels and Tires:	Wheel tire size Front gear angula	
		are refer to the maintenance manual
21. Control surface movements:		
	Elevator:	up 9°30' ± 30'
		down 12° ± 30'
	Ailerons: wings	Relative to the trailing edge of the
	5	up neutral down
		15°± 1° 2° ± 1° 10°± 1°
	Elevator tab:	Elevator up:25°30' \pm 1°6° \pm 1°
		Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:15° \pm 5°
		+0°
		2nd notch:
	Rudder:	
	(1) For planes f pedals:	itted with brakes controlled with rudder 2°) before operating drum brakes

 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

BB.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Schedule...... Refer to the latest amendment of Service Letter no. 6

BB.V Note:

- 1. This plane is identical to DR 400/125 except:
 - powerplant
 - luggage compartment removed
 - rear seat

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Section CC: DR 400/120

CC.I General

CC

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/120
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	February 11, 1975
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
C.II <u>C</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	18 September 1974
1. 2.	•	18 September 1974
	the applicable requirements:	18 September 1974
2.	the applicable requirements: (Reserved)	18 September 1974 France AIR2052
2. 3. 4.	the applicable requirements: (Reserved) (Reserved)	
2. 3. 4.	the applicable requirements: (Reserved) (Reserved) Certification Basis:	France AIR2052 France AIR2052 amendment June 6th, 1966
2. 3. 4. 5.	the applicable requirements: (Reserved) (Reserved) Certification Basis: Airworthiness Requirements:	France AIR2052 France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
2. 3. 4. 5.	the applicable requirements: (Reserved) (Reserved) Certification Basis: Airworthiness Requirements: Requirements elected to comply:	France AIR2052 France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7 None
2. 3. 4. 5. 6. 7.	the applicable requirements: (Reserved) (Reserved) Certification Basis: Airworthiness Requirements: Requirements elected to comply: EASA Special Conditions:	France AIR2052 France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7 None Canopy emergency release system

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

CC.III Technical Characteristics and Operational Limitations

1. Type Design Definition:	Refer to CEAPR document n°1001131 for DR400 "STANDARD 88" (Refer to note 2)
	Refer to CEAPR document n°1001130 for DR400 "STANDARD 92" (Refer to note 2)
2. Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed
4. Dimensions:	Span



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5. Engines: Lycoming O-235-L2A or O-235-K2A or O-235-K2B The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28

September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum continuous Power: 2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-47	1.80 m	2	2220
Hoffmann	HO-14-178/115	1.78 m	2	2250
	72CK-S6-0-56	1.83 m (*)	2	2220
Sensenich	72CK-S5-0-56	1.05 m ()	2	2220
	72CK-S6-0-54	1.83 m	2	2300

Remark: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-842 and FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014

W50 or SAE20W50	Mineral grades SAE60
W50 or SAE20W50	
	 SAE60
	SAE60
or SAE50	SAE50
	SAE40
, SAE40 or SAE20W40	SAE30
W50 or SAE15W50	SAE20W50
or SAE20W30	SAE20
)	, SAE40 or SAE20W40 W50 or SAE15W50

8. Fluid capacities:

8.1 Fuel:

8.2 Oil:

Main tank (liters)			Auxiliary tank (optional) (liters)	
Capa	Capacity Úsable		Capacity	Úsable
11	0	100/109 (1)	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 liters to 1liter. (Refer to note 2)

Oil sump capacity.	6 U.S. quarts (5.68 liters)
Usable	

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9. Air speeds:

V _{NE}	308 km/h (166 knots IAS)
V _{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
VA	215 km/h (116 knots IAS)
V _{FE}	170 km/h (92 knots IAS)

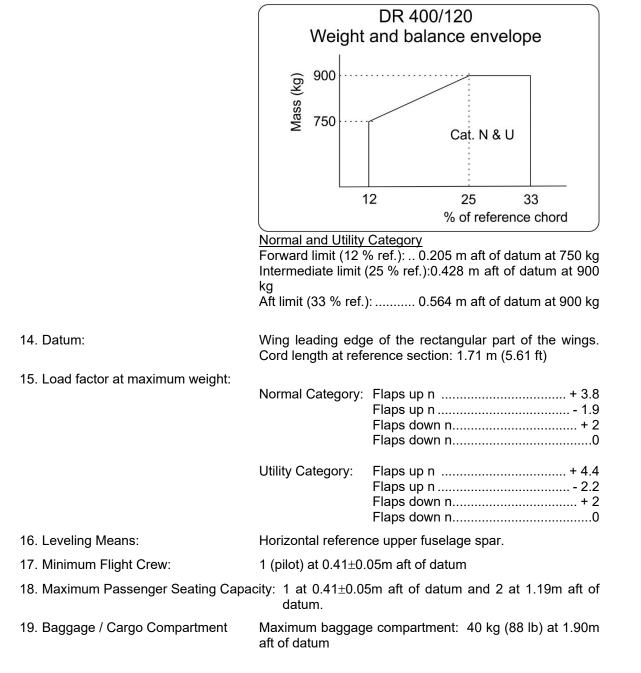
10. Maximum Operating Altitude:

Operational Capability:
 Maximum Masses:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Ca	"U" Category	
Take-off	Landing	
900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)

13. Centre of Gravity Range:





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20. Wheels and Tires:		
	Wheel tire size	2.58 m (8.46 ft)
	i ioni geai angu	right: 27°
	Tire pressure	•
		ure refer to the maintenance manual
21. Control surface movements:		
	Elevator:	up
	Ailerons:	down12° ± 30' Refer to following table
	Allerons.	
		upneutraldown $15^{\circ} \pm 1^{\circ}$ $2^{\circ} \pm 1^{\circ}$ $10^{\circ} \pm 1^{\circ}$
	Elevator tab:	Elevator up: $25^{\circ}30' \pm 1^{\circ}$ $6^{\circ} \pm 1^{\circ}$
		Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:
	Tiaps.	+0°
		2nd notch:
	Rudder:	
	pedals:	fitted with brakes controlled with rudder) before operating drum brakes

 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

CC.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Schedule...... Refer to the latest amendment of Service Letter no. 6

CC.V Note:

- 1. This plane is identical to DR 400/125 except powerplant
- 2. "Standard 92" models: Since July 1993 (from serial nr. 2212 included)

"Standard 88" models: Before July 1993 (before serial nr.2212 excluded)



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Section DD: <u>DR 400/125i</u>

DD.I General

1	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/125i
2	Airworthiness Category:	Normal and Utility Category
3	Type Certificate Holder:	Refer to Note 2 Section PP
4	Manufacturer:	Refer to Note 3 Section PP
5	(Reserved)	
6	DGAC Type Certification date:	September 25, 1975
7	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
8	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
DD.II	Certification Basis	
1	Reference Date for determining the applicable requirements:	18 September 1974
2	(Reserved)	
3	(Reserved)	
4	Certification Basis:	France AIR2052
5	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6	Requirements elected to comply:	None
7	EASA Special Conditions:	Canopy emergency release system
8	EASA Exemptions:	None
9	EASA Equivalent Safety Findings:	None
1	0. EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.
ווו חח	Technical Characteristics and Or	perational Limitations

DD.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1001131		
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.		
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.		
4.	Dimensions:			
		Span		
		Height2.23 m (7.32 ft) Length6.96 m (22.83 ft)		
		Wing Area		
5.	Engines:	Continental IO-240 A, B		
		The EASA type certification standard includes that of FAA TC E-7SO, based on individual EU member state		



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum continuous power: 2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
MT Propeller	MTV-7-D/170-09	1.70 m	3	Electrical variable pitch	Constant speed

The EASA type certification standard includes that of FAA TC P20BO, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

91/96 or 100/130 octane minimum aviation grade gasoline

7.2 Engine Oil:

Teledyne Continental engine IO-240-B

(Refer to Continental specifications MHS24 or MHS-25 and SB M87-12R1)

Oil	Ashless dispersant (AD)	Straight mineral
All temperatures	SAE15W50 or 20W50	
Above +4°C (40°F)	SAE15W50 or 20W60	SAE50
Below +4°C (40°F)	10W30, 15W30, 20W50	SAE30

7.3 Coolant:

Not Applicable

- 8. Fluid capacities:
 - 8.1 Fuel:

8.2 Oil:

Mai	n tank	Auxiliary tank		
(lit	ers)	(optional) (liters)		
Capacity Usable		Capacity	Usable	
110 109		50	50	

maximum	minimum
5.7 liters	2.9 liters

		5.7 IIIEIS	2.9 111015
9. Air speeds:			
			3 km/h (166 knots IAS)
	V _{NO}) km/h (140 knots IAS)
	V _C		0 km/h (140 knots IAS)
	VA		5 km/h (116 knots IAS)
	V _{FE}	170) km/h (92 knots IAS)
10. Maximum Operating Altitude:	Refer to approve	ed aircraft flight mar	nual.
11. Operational Capability:	Refer to approve	ed aircraft flight mar	nual.

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12. Maximum Masses:

["N" Category		"U" Category
	Take-off	Landing	
	900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)

13. Centre of Gravity Range:

	DR 400/125i		
	Weight and balance envelope		
	(b) 900 Sep 750 Cat. N & U		
	12 25 33		
	% of reference chord		
	Normal and Utility Category Forward limit (12 % ref.): 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 900 kg Aft limit (33 % ref.): 0.564 m aft of datum at 900 kg		
4. Datum:	Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)		
5. Load factor at maximum weight:			
	Normal Category: Flaps up n + 3.8 Flaps up n 1.9 Flaps down n + 2 Flaps down n		
	Utility Category: Flaps up n + 4.4 Flaps up n 2.2 Flaps down n + 2 Flaps down n		
6. Leveling Means:	Horizontal reference upper fuselage spar		
7. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum		
8. Maximum Passenger Seating Capa	city: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.		
9. Baggage / Cargo Compartment	Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum		
0. Wheels and Tires	Main gear track2.58 m (8.46 ft)Wheel tire size380 x 150 or 5.00-5Front gear angular movementleft: 27°right: 27°		
	Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual		



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21. Control surface movements

Elevator:	up down			
Ailerons: wings			edge of the	
Ū	up	neutral	down	
	15°± 1°	2° ± 1°	10°± 1°	
Elevator tab:	Elevator up:.	25°30' ± 1	l°6° ± 1°	
	Elevator dow	/n: 10°30' ± 1	l°16°30' ± 1°	
Flaps:	1st notch:		15° ± 5°	
			$+0^{\circ}$	
	2nd notch:			
Rudder:			25° ^{+3°} _{-0°} (1)	
 (1) For planes fitted with brakes controlled with rudder pedals: 16° (-0°, +2°) before operating drum brakes 				
20° (-0°, +3°)	before operati	ng disk brake	s	

22. (Reserved)

DD.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

DD.V Note:

1. This plane is identical to DR 400/120 except powerplant



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Section EE: DR 400/140 B

EE.I General

	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400/140 B
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	November 09, 1975
	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
EE.	ш <u>с</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	29 August 1975
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	Canopy emergency release system
	8.	EASA Exemptions:	None
	9.	EASA Equivalent Safety Findings:	None
	10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.

EE.III Technical Characteristics and Operational Limitations

1. Type Design Definition:	Refer to CEAPR document n°1001131 for DR40 "STANDARD 88" (Refer to note 2)				
	Refer to CEAPR document n°1001130 for DR400 "STANDARD 92" (Refer to note 2)				
2. Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.				
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.				
4. Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft) - Round spinner 7.10 m (23.29 ft) - Sharp spinner				



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Wing Area 13.60 m² (146.39 foot²)

5. Engines: Lycoming O-320-D2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Propeller Manufacturer	Model	Maximum Continuous Power RPM	
Sensenich	74 DM 6S5-2-64	2700 rpm (1)	
Sensenich	74 DM 6S5-2-60	2500 rpm (1)	

Remarks: (1) Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Manufacturer Model		Number of blades	Minimum static RPM at sea level
Sensenich	74 DM 6S5-2-64	1.02 m (1)	2	2200 rpm
Sensenich	74 DM 6S5-2-60	1.83 m (1)	2	2300 rpm

Remarks: (1) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

91/96 or 100/130 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



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- 8. Fluid capacities:
 - 8.1 Fuel:

Mai	n tank	Auxiliary tank			
(liters)		(optional) (liters)			
Capacity Usable		Capacity	Usable		
110	100/109 (1)	50	50		

(1) New standard called "Standard 92" from serial number
2210, unusable quantity of fuel reduced from 10 liters to 1
liter, (refer to note 2).

8.2 Oil:	Oil sump capacity	8 U.S.	quarts	(7.57 liters)
	Usable	6 U.S.	quarts	(5.68 liters)

9. Air speeds:

V_{NE}	
V_{NO}	
Vc.	
V_{FE}	

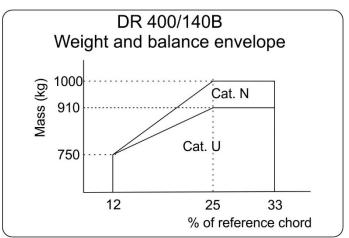
10. Maximum Operating Altitude:

11. Operational Capability:

- 12. Maximum Masses:
- 13. Centre of Gravity Range:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Ca	"U" Category	
Take-off	Landing	
1000 kg (2205 lb)	1000 kg (2205 lb)	910 kg (2006 lb)



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 910 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 910 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)



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14. Datum:

15. Load factor at maximum weight:	Normal Category:	Flaps up n
	Utility Category:	Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n + 2 Flaps down n 0
16. Leveling Means:	Horizontal referen	ce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.	.05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.0 datum.	5m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum baggag aft of datum	e compartment: 40 kg (88 lb) at 1.90m
20. Wheels and Tires:		
	Wheel tire size	2.58 m (8.46 ft)
	Tire pressure Oleo strut pressur	
21. Control surface movements:		
	Elevator:	up
	Ailerons: wings	Relative to the trailing edge of the
		up neutral down
	Floweter tob	$15^{\circ} \pm 1^{\circ} 2^{\circ} \pm 1^{\circ} 10^{\circ} \pm 1^{\circ}$
	Elevator tab:	Elevator up:25°30' ± 1°6° ± 1° Elevator down: 10°30' ± 1°16°30' ± 1°
	Flaps:	1st notch:15° \pm 5°
		2nd notch:
	Rudder:	
	(1) For planes fit pedals: 16° (-0°, +2°)	ted with brakes controlled with rudder before operating drum brakes before operating disk brakes

22. (Reserved)

EE.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6



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EE.V Note:

- 1. This plane is identical to DR 400/140 except powerplant
- 2. "Standard 92" models: Since June 1993 (from serial nr 2211 included)"Standard 88" models: Before June 1993 (before serial nr.2211 excluded)



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Section FF: <u>DR 400/120A</u>

FF.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/120 A
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	November 15, 1976
7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
FF.II	Certification Basis	
1.	Reference Date for determining the applicable requirements:	28 June 1976
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
6.	Requirements elected to comply:	None
7.	EASA Special Conditions:	Canopy emergency release system
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
1(). EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.

FF.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed
4.	Dimensions:	
		Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft)
		Wing Area
5.	Engines:	Lycoming O-235-L2A or O-235-K2A or O-235-K2B
		The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum continuous power: 2800 rpm

6. Propellers:

Manufactur er	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-50 1 A 135 JCM 71-47	1.80 m	2	2200 rpm (2)
		1.77 m (1)	2	
		1.80 m	2	
		1.77 m (1)	2	
Hoffmann	HO-14-178/115	1.78 m	2	2250 rpm
Tiolimann		1.73 m (1)	2	2230 Ipin

Remarks:

No. 1070

(1) Minimum diameter after repair.

(2) Do not continuous operate between 2025 rpm and 2325 rpm.

The EASA type certification standard includes that of FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming

Ashless dispersant (AD)	Mineral	
grades	grades	
SAE15W50 or SAE20W50		
SAE60	SAE60	
SAE40 or SAE50	SAE50	
SAE40	SAE40	
SAE30, SAE40 or SAE20W40	SAE30	
SAE20W50 or SAE15W50	SAE20W50	
SAE30 or SAE20W30	SAE20	
	grades SAE15W50 or SAE20W50 SAE60 SAE40 or SAE50 SAE40 SAE30, SAE40 or SAE20W40 SAE20W50 or SAE15W50	

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank	
(liters)		(optional) (liters)	
Capacity	Usable	Capacity	Usable
110	100	50	50

8.2 Oil:



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9. Air speeds:

V_{NE}	
V_{NO}	
Vc	
VA	
V_{FE}	170 km/h (92 knots IAS)

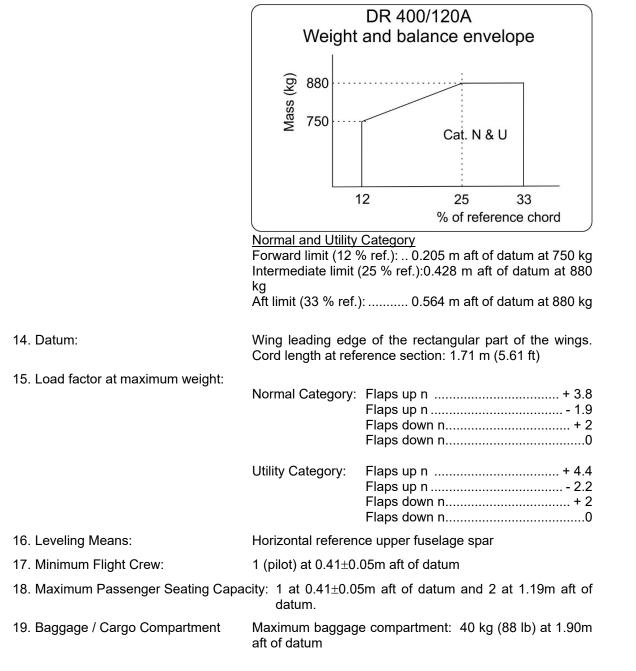
10. Maximum Operating Altitude:

Operational Capability:
 Maximum Masses:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Category		"U" Category
Take-off	Landing	
880 kg (1940 lb)	880 kg (1940 lb)	880 kg (1940 lb)

13. Centre of Gravity Range:





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20. Wheels and Tires:		
	Wheel tire size	2.58 m (8.46 ft)
	r tont gear angul	right: 27°
	Tire pressure Oleo strut pressu	
21. Control surface movements	Elevator:	up 0°30' ± 30'
		up
	Ailerons: wings:	Relative to the trailing edge of the
	wings.	up neutral down
		15°± 1° 2° ± 1° 10°± 1°
	Elevator tab:	Elevator up:25°30' ± 1°6° ± 1°
		Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:15° \pm 5°
		$+0^{\circ}$
		2nd notch:
	Rudder:	
	(1) For planes fi pedals:	tted with brakes controlled with rudder

22. (Reserved)

FF.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Schedule....... Refer to the latest amendment of Service Letter no. 6

 16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

FF.V Note:

1. This plane is identical to DR 400/120 except:

- propeller
- maximum mass



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Section GG: DR 400/160D

GG.I General

	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400/160 D
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	April 27, 1981
	7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
GG.	.II <u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	21 March 1971
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	Canopy emergency release system
	8.	EASA Exemptions:	None
	9.	EASA Equivalent Safety Findings:	None
	10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.

GG.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed
4.	Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 6.96 m (22.83 ft) - Round spinner 7.10 m (23.29 ft) - Sharp spinner Wing Area 14.2 m² (152.85 foot²)
5.	Engines:	Lycoming O-320-D2A
		The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Maximum continuous Power: 2600 rpm Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

5.1 Engine Limits:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sanaaniah	M74 DMS-2-66	1.83 m (1)	2	2150 rpm
Sensenich	74 DM6S5-2-64		2	2250 rpm

Remark: (1) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

91/96 or 100/130 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

	n tank œrs)	RH (lite	tank ers)	LH t (lite		Auxilia (optiona	ry tank I) (liters)
Capacity	Usable	Capacity	/	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 liters to 1 liter (refer to note 2).

8.2 Oil:

9. Air speeds:



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V _{NE}	308 km/h (166 knots IAS)
V _{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
VA	215 km/h (116 knots IAS)
VFE	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

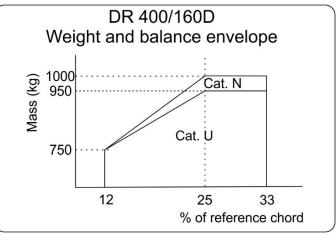
- 11. Operational Capability:
- 12. Maximum Masses:

"N" Ca	"U" Category	
Take-off		
1000 kg (2205 lb)	1000 kg (2205 lb)	950 kg (2094 lb)

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

13. Centre of Gravity Range:



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg <u>Utility Category</u>

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 950 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg Wing leading edge of the rectangular part of the wings.

Cord length at reference section: 1.71 m (5.61 ft)

14. Datum:

15. Load factor at maximum weight:

16. Leveling Means:

Normal Category:	Flaps up n	+ 3.8
	Flaps up n	1.9
	Flaps down n	+ 2
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	+ 2
	Flaps down n	
Horizontal referen	ce upper fuselage spar	

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.



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Main gear track Wheel tire size	2.58 m (8.46 ft) 380 x 150 or 5.00-5
	movement left: 27° right: 27°
•	refer to the maintenance manual

21. Control surface movements:

Elevator:	up		9°30' ± 30'
	down		12° ± 30'
Ailerons:	Refer to follo	wing table	
	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	Elevator up:	25°30' ± 7	1°6° ± 1°
	Elevator dow	/n: 10°30' ± ′	1° 16°30' ± 1°
Flaps:	1st notch:		15° ± 5°
-			$+0^{\circ}$
	2nd notch:		60° – 5°
Rudder:	Rudder:		
(1) For planes fit pedals:	ted with brak		

 16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

GG.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6



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TCDS No: EASA.A.367 Issue: 05

GG.VNote:

1. This plane is identical to DR 400/160 except:

maximum continuous power rpm

maximum mass

-

2. "Standard 92" model: since November 1993



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Section HH: DR 400/120 D

HH.I General

	1.	a) Type:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400/120 D
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	April 28, 1981
	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
HH.	.II <u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	28 June 1976
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	Canopy emergency release system
	8.	EASA Exemptions:	None
	9.	EASA Equivalent Safety Findings:	None
	10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.

HH.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1001131		
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.		
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed		
4.	Dimensions:	Span		
5.	Engines:	Lycoming O-235-L2A or O-235-K2A or O-235-K2B		

-

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The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power:

Propeller Manufacturer	Propeller model	Maximum Continuous Power RPM
Sensenich	72 CKS6-0-56	2600 (1)
Hoffmann	HO-14-178/115	2583 (1)

Remarks: (1) Maximum continuous power limited by noise regulation.

6. Propellers:

	Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
	Sensenich	72 CK-S6-0-56	1.83 m (1)	2	2220
	Hoffmann	HO-14-178/115	1.78 m	2	2220
	поппапп		1.73 m (2)	2	2250

Remarks:

(1) No acceptable diameter reduction for repair.

(2) Minimum diameter after repair.

The EASA type certification standard includes that of FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1070.

Refer to latest revision of Service Instruction Lycoming No. 1014.

100/100LL octane minimum aviation grade gasoline.

Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



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- 8. Fluid capacities:
 - 8.1 Fuel:

8.2 Oil:

9. Air speeds:

Main tank		Auxiliary tank	
(liters)		(optional) (liters)	
Capacity	Usable	Capacity	Usable
110	100	50	50

10.	Maximum	Operating	Altitude:

- 11. Operational Capability:
- 12. Maximum Masses:
- 13. Centre of Gravity Range:

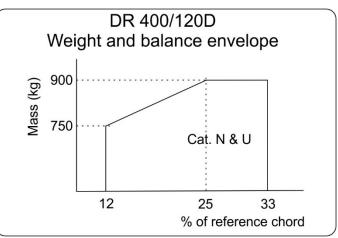
Usable		
V _{NE}		
V _{NO}		
Vc		
VA		
V _{FE}	170 km/h (92 knots IAS)	

Oil sump capacity 6 U.S. quarts (5.68 liters)

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

"N" Ca	"U" Category	
Take-off	Landing	
900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 900 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 900 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

Normal Category:	Flaps up n	
	Flaps up n	1.9
	Flaps down n	+ 2
	Flaps down n	0
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	+ 2

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14. Datum:

15. Load factor at maximum weight:

	Flaps down n0
16. Leveling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum
18. Maximum Passenger Seating Capa	acity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
19. Baggage / Cargo Compartment	Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum
20. Wheels and Tires	
	Main gear track 2.58 m (8.46 ft) Wheel tire size 380 x 150 or 5.00-5 Front gear angular movement left: 27°
21. Control surface movements	
	Elevator:
	down
	up neutral down
	$15^{\circ} \pm 1^{\circ}$ $2^{\circ} \pm 1^{\circ}$ $10^{\circ} \pm 1^{\circ}$ Elevator tab:Elevator up: $25^{\circ}30' \pm 1^{\circ}6^{\circ} \pm 1^{\circ}$
	Elevator tab
	Flaps:
	2nd notch:
	Rudder:
	1) For planes fitted with brakes controlled with rudder pedals:
	16° (-0°, +2°) before operating drum brakes
	20° (-0°, +3°) before operating disk brakes

22. (Reserved)

HH.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Schedule....... Refer to the latest amendment of Service Letter no. 6

HH.V Note:

1. This plane is identical to DR 400/120 except maximum continuous power rpm



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Section II: DR 400/180 S

II.I General

11.11

1.	a) Type:	DR 200, DR 300, and DR 400 series					
	b) Model:	DR 400/180S					
2.	Airworthiness Category: Normal and Utility Category						
3.	Type Certificate Holder:	Refer to Note 2 Section PP					
4.	Manufacturer:	Refer to Note 3 Section PP					
5.	(Reserved)						
6.	DGAC Type Certification date:	February 11, 1985					
7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)					
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45					
<u>C</u>	Certification Basis						
1.	Reference Date for determining the applicable requirements:	31 January 1985					
2.	(Reserved)						
3.	(Reserved)						
4.	Certification Basis:	France AIR2052					
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7					
6.	Requirements elected to comply:	None					
7.	EASA Special Conditions:	Canopy emergency release system					
8.	EASA Exemptions:	None					
9.	EASA Equivalent Safety Findings:	None					
10	. EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 6.					

II.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	Span
5.	Engines:	Lycoming O-360-A3A
		The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum continuous power: 2600 rpm Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number	Minimum static RPM
Manulacturer	Woder	Ø	of blades	at sea level
Sensenich	76 EM8S5-0-64	1.93 m (1)	2	2250 (2)
		•		on for repair. etween 2150 rpm and
	TC P4EA, acceptance September	based or or certifica 2003. Other	n individua tion of this standards	rd includes that of FAA I EU member state s standard prior to 28 confirming to TC/TCDS EU member state prior

No. 1070.

- 7. Fluids:
 - 7.1 Fuel:

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

to 28 September 2003 are also acceptable.

8. Fluid capacities:

8.1 Fuel:

Main tank RH tank		tank	LH tank		Auxiliary tank		
(lit	ers)	(liters)		(liters)		(optional) (liters)	
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 liters to 1 liter, (refer to note 2).

8.2 Oil:

9. Air speeds:

V_{NE}	



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Vc	
V _{FE}	170 km/h (92 knots IAS)

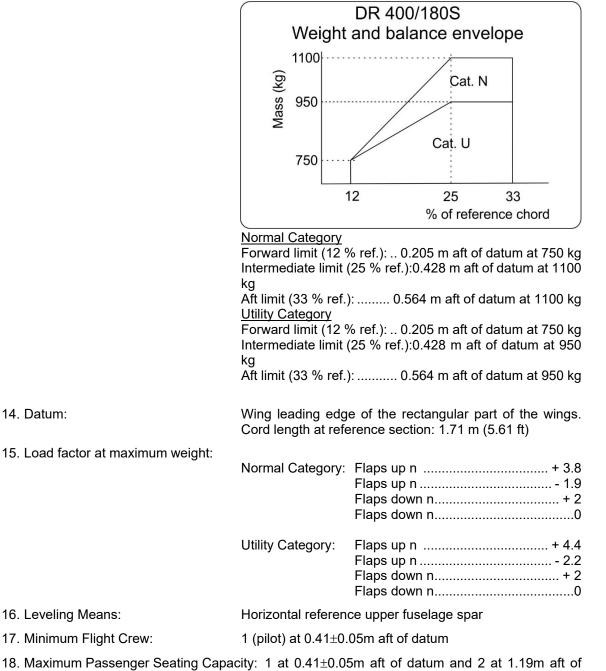
- 10. Maximum Operating Altitude:
- 11. Operational Capability:
- 12. Maximum Masses:

Ref	er	to	appro	oved	aircraf	t fligh	nt manua	al.
-----	----	----	-------	------	---------	---------	----------	-----

Refer to approved aircraft flight manual.

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1045 kg (2304 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:



datum.

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TCDS No: EASA.A.367 Issue: 05	DR 200, DR300, and DR400 series DR 400/180 S	Date: 16 October 2023
19. Baggage / Cargo Compartment	Maximum baggage compartment: aft of datum)	: 60 kg (132 lb) at 1.90m
20. Wheels and Tires:	Front gear angular movement	380 x 150 or 5.00-5 left: 27° right: 27° the maintenance manual
21. Control surface movements	down	9°30' ± 30'
	Elevator down Flaps: 1st notch:	25°30' ± 1°6° ± 1° i: 10°30' ± 1°16°30' ± 1° 15° ± 5° $+0^{\circ}$
	Rudder: (1) For planes fitted with brakes pedals:	s controlled with rudder
	16° (-0°, +2°) before operating 20° (-0°, +3°) before operating	

22. (Reserved)

II.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

II.V Note:

- 1. This plane is identical to DR 400/180 except:

 - maximum continuous power rpm
 Sensenich 76 FM8S5_0 64 procession Sensenich 76 EM8S5-0-64 propeller only

2. "Standard 92" model



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Section JJ: <u>DR 400/100</u>

JJ.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR 400/100
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	November 06, 1987
7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
JJ.II <u>C</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	13 April 1987
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
	Certification Basis: Airworthiness Requirements:	France AIR2052 France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
		France AIR2052 amendment June 6th, 1966
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7

- 9. EASA Equivalent Safety Findings: None
- 10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

JJ.III Technical Characteristics and Operational Limitations

1. Type Design Definition:	Refer to CEAPR document n°1001131 for DR400 "STANDARD 88" (Refer to note 2)
	Refer to CEAPR document n°1001130 for DR400 "STANDARD 92" (Refer to note 2)
2. Description:	Single-engine, two-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4. Dimensions:	Span



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5.1 Engine Limits: Maximum continuous power: 2600 rpm Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	72 CKS6-0-56	1.83 m (*)	2	2220 rpm

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

72	Engine Oil:	
1.2	Ligine Oil.	

Refer to latest revision of Service Instruction Lycoming No. 1014.

Ashless dispersant (AD) grades	Mineral grades	
<u> </u>	gradoo	
SAE15W50 or SAE20W50		
SAE60	SAE60	
SAE40 or SAE50	SAE50	
SAE40	SAE40	
SAE30, SAE40 or SAE20W40	SAE30	
SAE20W50 or SAE15W50	SAE20W50	
SAE30 or SAE20W30	SAE20	
	grades SAE15W50 or SAE20W50 SAE60 SAE40 or SAE50 SAE40 SAE30, SAE40 or SAE20W40 SAE20W50 or SAE15W50	

8. Fluid capacities:

8.1 Fuel:

8.2 Oil:

9. Air speeds:

Main tank		Auxiliary tank	
(liters)		(optional) (liters)	
Capacity	Usable	Capacity	Usable
110	100/109 (1)	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 liters to 1 liter, (refer to note 2).

Oil sump capacity 6 U.S. quarts (5.68 liters) Usable...... 4 U.S. quarts (3.79 liters)



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14. Datum:

V _{NO}	
Vc	
	170 km/h `(92 knots IAS)

- 10. Maximum Operating Altitude:
- 11. Operational Capability:

13. Centre of Gravity Range:

12. Maximum Masses:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Category		"U" Category
Take-off	Landing	
800 kg (1764 lb)	800 kg (1764 lb)	800 kg (1764 lb)

DR 400/100 Weight and balance envelope 800 Mass (kg) 700 Cat. N & U 12 25 33 % of reference chord Normal and Utility Category Forward limit (12 % ref.): .. 0.205 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 800 kg Aft limit (33 % ref.): 0.564 m aft of datum at 800 kg Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:		
-	Normal Category:	Flaps up n+ 3.8
		Flaps up n 1.9
		Flaps down n+ 2
		Flaps down n0
	Utility Category:	Flaps up n+ 4.4
		Flaps up n 2.2
		Flaps down n+ 2
		Flaps down n0
16. Leveling Means:	Horizontal referen	ce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.	05m aft of datum

- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.10m aft of datum

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20. Wheels and Tires:

Wheel tire size	2.58 m (8.46 	-5
Front gear angular	movement left: 2	7°
0 0	right: 2	
Tire pressure	refer to the maintenance manu	ıal
	refer to the maintenance manu	ıal

21. Control surface movements

Elevator:			
Ailerons:			12° ± 30'
wings		the training	cuge of the
	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	.Elevator up:	25°30' ± '	1°6° ± 1°
	Elevator dow	/n: 10°30' ± ′	1° 16°30' ± 1°
Flaps:	.1st notch:		15° ± 5°
			$+0^{\circ}$
	2nd notch:		
Rudder:			25° ^{+3°} 0° (1)
(4)	بالمسط ببالأقلم المسماد		م ما ما ما ما م

(1) For planes fitted with brakes controlled with rudder pedals:

 16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

JJ.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

JJ.V Note:

1. This plane is identical to DR 400/120 D except:

- rear seats removed
- luggage compartment layout
- maximum weight
- brakes
- new instrument panel

2. "Standard 92" models: Since April 2017 (from serial nr 2703 included)

"Standard 88" models: Before April 2017 (before serial nr.2703 excluded)

Section KK: DR 400 RP

KK.I General

1.	а) Туре:	DR 200, DR 300, and DR 400 series
	b) Model:	DR400 RP
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	(Reserved)	
6.	DGAC Type Certification date:	August 11, 1988
7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
KK.II <u>(</u>	Certification Basis	
1.	Reference Date for determining the applicable requirements:	January 1986
2.	(Reserved)	
3.	(Reserved)	
4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 32
6.	Requirements elected to comply:	None
-		

- 7. EASA Special Conditions: Canopy emergency release system
- EASA Exemptions: None
 EASA Equivalent Safety Findings: None
- 10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

KK.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1001131
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	Span 8.72 m (28.61 ft) Height 2.23 m (7.32 ft) Length 7.45 m (24.44 ft) Wing Area 13.60 m² (146.39 foot²)
5.	Engines:	Porsche PFM 3200 N01
	5.1 Engine Limits:	Maximum Continuous Power:

6. Propellers:

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Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
Hoffmann	HO V 123 F1/200 CQ	2.00 m	3	Woodward B 2109-681	Constant speed

The EASA type certification standard includes that of FAA TC P5EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:	91/96 or 100/130 octane	e minimum aviation grade gasoline
7.2 Engine Oil:	Only automotive type	SAE 5W 30
		SAE 5W 50
		SAE 10W 30
		SAE 15W 50 (*)

(*) Do not use below -5°C (25°F) external on ground temperature

SAE 20W 50 (*)

8. Fluid capacities:

8.1 Fuel:

_				
	Main tank		Auxiliary tank	
	(liters)		(optional) (liters)	
	Capacity	Usable	Capacity	Usable
	115	108	50	50

8.2 Oil:

9. Air speeds:

Refer to approved flight manual

V _{NE}	308 km/h (166 knots IAS)
V _{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
VA	215 km/h (116 knots IAS)
V _{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

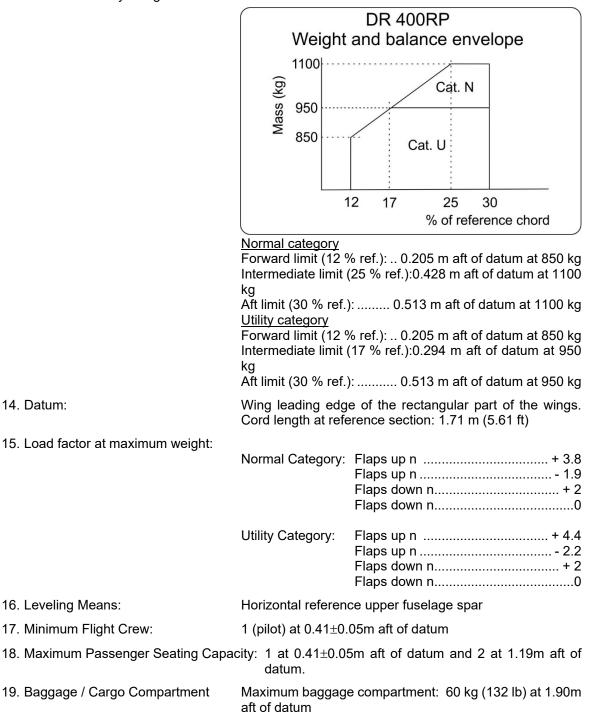
11. Operational Capability:

12. Maximum Masses:

Refer to approved aircraft flight manual. Refer to approved aircraft flight manual.

"N" Category		"U" Category
Take-off	Landing	
1100 kg (2425 lb)	1100 kg (2425 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:





20. Wheels and Tires

20. Wheels and Tires				
	Main gear track			
	Wheel tire size			
	Front gear angula	r movement		
	Tire pressure			
	Oleo strut pressur			
21. Control surface movements				
	Elevator:	up		9°30' ± 30'
		down		12° ± 30'
	Ailerons:	.Relative to	the trailing	edge of the
	wings		Ū	U
	Ū	up	neutral	down
		15°± 1°	2° ± 1°	10°± 1°
	Elevator tab:	Elevator up:	25°30' ±	1°6° ± 1°
		Elevator dow	/n: 10°30' ± '	1° 16°30' ± 1°
	Flaps:	1st notch:		15° ± 5°
				$+0^{\circ}$
		2nd notch:		60° – 5°
	Rudder:			25° ^{+3°} 0°

22. (Reserved)

KK.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

KK.V Note:

- 1. This plane is identical to DR 400/180 R except:
 - powerplant
 - maximum weight -



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Section LL: DR 400 NGL

LL.I General

	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400 NGL
	2.	Airworthiness Category:	Normal Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	February 19, 1991
	7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replace	s DGAC-France Type Certificate no. 45
LL.I	I <u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	21 March 1971
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	Canopy emergency release system
	8.	EASA Exemptions:	None
	9.	EASA Equivalent Safety Findings:	None
	10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 10.

LL.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1002197
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Height2.23 m (7.32 ft) Length7.10 m (23.29 ft)
		Wing Area 14.20 m ² (152.85 foot ²)
5.	Engines:	Lycoming O-360-A3A
		The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2600 rpm

Remark: Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	76 EM8S5-0-64	1.93 m (1)	2	2180 rpm (2)

Remarks:

No. 1070.

(1) No acceptable diameter reduction for repair.

(2) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

	Main tank (liters)		RH tank (liters)		LH tank (liters)		Auxiliary tank (optional) (liters)	
(11.613)		· · ·		(11013)				
	Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
	110	100	40	40	40	40	50	50
8.2 Oil:					ity			7.57 liters) 5.68 liters)



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9. Air speeds:

V_{NE}	
V_{NO}	
Vc	
VA	
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

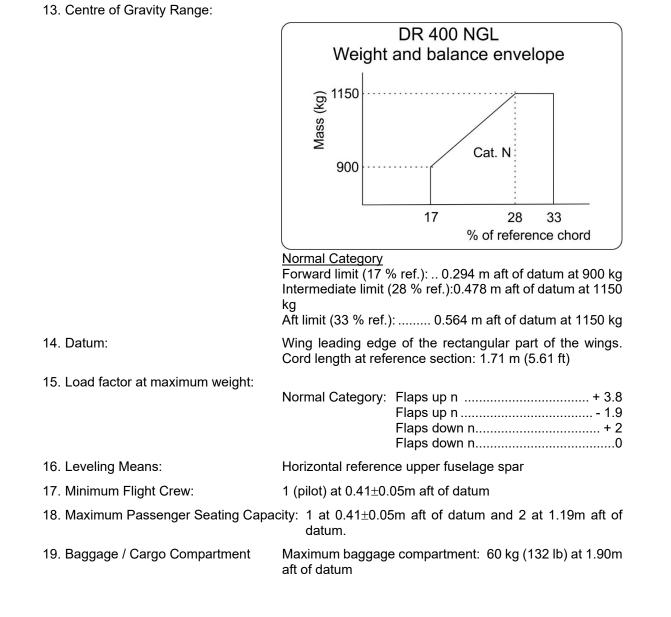
11. Operational Capability:

12. Maximum Masses:

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

"N" Category				
Take-off	Landing			
1150 kg (2535 lb)	1150 kg (2535 lb)			



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20. Wheels and Tires:

Main gear track	2.58 m (8.46 ft)
Wheel tire size	
Front gear angular	movement left: 27°
0 0	right: 27°
Tire pressure	refer to the maintenance manual
	e refer to the maintenance manual

21. Control surface movements

Elevator:	.up		9°30' ± 30'
	down		12° ± 30'
Ailerons: wings	.Relative to	the trailing	edge of the
	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	Elevator up:	25°30' ± '	1°6° ± 1°
	Elevator dow	/n: 10°30' ± ′	1° 16°30' ± 1°
Flaps:	.1st notch:		15° ± 5°
			$+0^{\circ}$
	2nd notch:		60° – 5°
Rudder:			25° ^{+3°} 0°

22. (Reserved)

LL.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

LL.V Note:

- 1. This plane is identical to DR 400/180 except:
 - larger cabin
 - maximum weight
 - Sensenich 76 EM8S5-0-64 propeller only



Section MM: DR 400/200R

MM.I General

	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400/200 R
	2.	Airworthiness Category:	Normal and Utility Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	December 11, 1992
	7.	EASA Type Certification Date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
ММ	.II <u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	3 August 1972
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	Canopy emergency release system
			Airplane and towed sailplane maximum masses are limited considering the minimum climb performances required.
	8.	EASA Exemptions:	None
	9.	EASA Equivalent Safety Findings:	None
	10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 10.

MM.III Technical Characteristics and Operational Limitations

1. Type Design Definition:	Refer to CEAPR document n°1001131 for DR400 "STANDARD 88"
	Refer to CEAPR document n°1001130 for DR400 "STANDARD 92"
2. Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3. Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.



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4. Dimensions:

Span	8.72 m	(28.61 ft)
Height		`(7.32 ft)
Length	7.22 m	(23.69 ft)
Wing Area	.13.60 m²	(146.39 foot ²)

5. Engines: Lycoming IO-360-A1 B6

The EASA type certification standard includes that of FAA TC 1E10, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

6 Propellers

Maximum Continuous Power: 2700 rpm

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
Hartzell	HC-C2YK-1BF/F7666A-2	1.88 m	2	Woodward B 2109-681	Constant speed (*)

Remarks: (*) Variable pitch from 14° to 29.2°

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

100/130 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Maii	n tank	Auxilia	ry tank
(lit	ers)	(optiona	I) (liters)
Capacity	Usable	Capacity	Usable
110	109	50	50

8.2 Oil:



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9. Air speeds:

V _{NE}	
V _{NO}	
Vc	
Va	215 km/h (116 knots IAS)
Vfe	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

12. Maximum Masses:

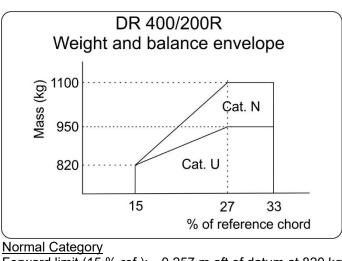
11. Operational Capability:

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1100 kg (2425 lb)	950 kg (2094 lb)

12.1 Towing mass limitations:

Each maximum mass of the tug and of the towed glider is limited by the minimum climb performance.

13. Centre of Gravity Range:



Forward limit (15 % ref.): .. 0.257 m aft of datum at 820 kg Intermediate limit (27 % ref.):0.462 m aft of datum at 1100 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1100 kg Utility Category

Forward limit (15 % ref.): .. 0.257 m aft of datum at 820 kg Intermediate limit (27 % ref.):0.462 m aft of datum at 950 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)

 Normal Category:
 Flaps up n
 + 3.8

 Flaps up n
 - 1.9

 Flaps down n
 + 2

 Flaps down n
 - 0

 Utility Category:
 Flaps up n
 + 4.4

 Flaps up n
 - 2.2

 Flaps down n
 - 2.2

 Flaps down n
 - 2.2

 Flaps down n
 - 0



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14. Datum:

15. Load factor at maximum weight:

TCDS No: EASA.A.367	DR 200, DR300, and DR400 series	
Issue: 05	DR 400/200R	Date: 16 October 2023
16. Leveling Means:	Horizontal reference upper fuse	lage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of dat	um
18. Maximum Passenger Seating C	Capacity: 1 at 0.41±0.05m aft of datu datum.	um and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum baggage compartmer aft of datum	nt: 60 kg (132 lb) at 1.90m
20. Wheels and Tires		
	Front gear angular movement	
21. Control surface movements		
	Elevator:up down Ailerons:Relative to wings	12° ± 30'
	up 15°± 1°	$\begin{array}{c c} neutral & down \\ 2^{\circ} \pm 1^{\circ} & 10^{\circ} \pm 1^{\circ} \end{array}$
	Elevator tab:Elevator up:	
		/n: 10°30' ± 1° 16°30' ± 1°
	Flaps:1st notch:	15° ± 5° +0°
	2nd notch:	
	Rudder:	

22. (Reserved)

MM.IV Operating and Service Instructions

Airplane Flight Manual	Refer t	to the	latest	amendment	of Service	e Letter	no. 6
Airplane Maintenance Manual	Refer t	to the	latest	amendment	of Service	e Letter	no. 6
Airplane Maintenance Schedule	Refer t	to the	latest	amendment	of Service	e Letter	no. 6

MM.V Note:

- 1. This plane is identical to DR 400/180 R except:
 - Powerplant
 - maximum weight
- 2. Glider and Banner towing:

Refer to approved flight manual.

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Section NN: DR 400/500

NN.I General

	1.	а) Туре:	DR 200, DR 300, and DR 400 series
		b) Model:	DR 400/500
	2.	Airworthiness Category:	Normal Category
	3.	Type Certificate Holder:	Refer to Note 2 Section PP
	4.	Manufacturer:	Refer to Note 3 Section PP
	5.	(Reserved)	
	6.	DGAC Type Certification date:	March 10, 1998
	7.	EASA Type Certification date:	January28, 2013 (Type Certificate transfer)
	8.	The EASA type Certificates replaces	s DGAC-France Type Certificate no. 45
NN.	II <u>C</u>	ertification Basis	
	1.	Reference Date for determining the applicable requirements:	21 March 1971
	2.	(Reserved)	
	3.	(Reserved)	
	4.	Certification Basis:	France AIR2052
	5.	Airworthiness Requirements:	France AIR2052 amendment June 6th, 1966 FAR part 23 as amended by amendment 7
	6.	Requirements elected to comply:	None
	7.	EASA Special Conditions:	Canopy emergency release system
	8.	EASA Exemptions:	None
	9.	EASA Equivalent Safety Findings:	None
	10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 10.

NN.III Technical Characteristics and Operational Limitations

1.	Type Design Definition:	Refer to CEAPR document n°1002197
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.
4.	Dimensions:	
		Span
		Height2.23 m (7.32 ft) Length7.22 m (23.69 ft)
		Wing Area 14.20 m ² (152.85 foot ²)
5.	Engines:	Lycoming IO-360-A1 B6
		The EASA type certification standard includes that of FAA TC 1E10, based on individual EU member state



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acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power: 2700 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
Hartzell	HC-C2YK-1BF/F7666A-2	1.88 m	2	Woodward B 2109-681	Constant speed (*)

Remarks: (*) variable pitch from 14° to 29.2°

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

- 7. Fluids:
 - 7.1 Fuel:

7.2 Engine Oil:

100/130 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

•							
Main tank RH tank		LH tank		Auxiliary tank			
(lit	ers)	(liters)		(liters)		(optional) (liters)	
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
105	104	40	40	40	40	50	50
105	104	40	40	40	40	50	5

8.2 Oil:

9. Air speeds:

V _{NO}	
Va	215 km/h (116 knots IAS)
VFE	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

11. Operational Capability:

Refer to approved aircraft flight manual.

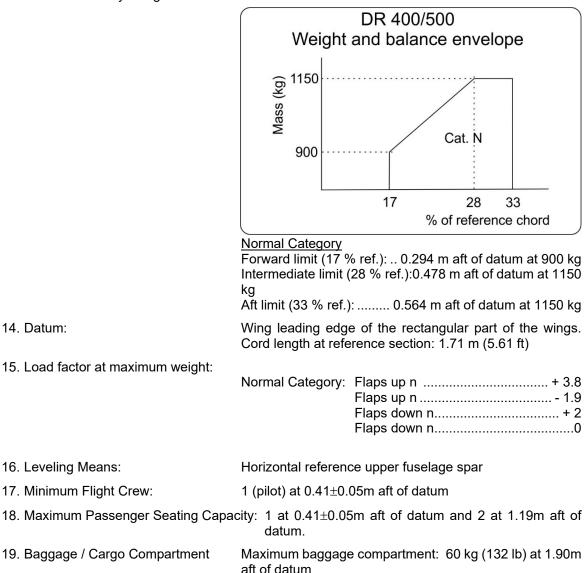


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12. Maximum Masses:

13.	Centre	of	Gravity	Rang	e:
10.	OCHUC	UI.	Oravity	rung	ς.

"N" Category			
Take-off Landing			
1150 kg (2535 lb)	1150 kg (2535 lb)		





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20. Wheels and Tires:		
	Wheel tire size	2.58 m (8.46 ft) 380 x 150 or 5.00-5
	Front gear angula	ar movement left: 27° right: 27°
	Tire pressure Oleo strut pressu	refer to the maintenance manual re refer to the maintenance manual
21. Control surface movements		
	Elevator:	up 9°30' ± 30'
		down 12° ± 30'
	Ailerons: wings	Relative to the trailing edge of the
		up neutral down
		15°± 1° 2° ± 1° 10°± 1°
	Elevator tab:	Elevator up:25°30' ± 1°6° ± 1°
		Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$
	Flaps:	1st notch:15° \pm 5°
		2nd notch:
	Rudder:	25° ^{+3°}

22. (Reserved)

NN.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

NN.V Note:

1. This plane is identical to DR 400 NGL except powerplant



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Section OO: DR400 / 200 I

OO.I General

1.	a) Type :	DR 200, DR 300, and DR 400 series
	b) Model:	DR400 / 200 I
2.	Airworthiness Category:	Normal and Utility Category
3.	Type Certificate Holder:	Refer to Note 2 Section PP
4.	Manufacturer:	Refer to Note 3 Section PP
5.	EASA Type Certification Application Date :	26 April 2016
6.	(Reserved)	
7.	(Reserved)	
8.	EASA Type Certification Date:	25 September 2017
00	0.II EASA Certification Basis	
1.	Reference date for determining the applicable requirements:	3 August 1972
2. 3.	(Reserved) (Reserved)	
3. 4.	Certification Basis:	France AIR2052
5.	Airworthiness Requirements :	France AIR2052 amendment June 6th, 1966
	·	FAR part 23 as amended by amendment 7
6.	Requirement elected to comply:	None
7.	Special Conditions:	Canopy emergency release system
8.	EASA Exemptions:	None
9.	EASA Equivalent Safety Findings:	None
10.	EASA Environmental Standards:	ICAO Annex 16, Vol.1. Chap 10

OO.III <u>Technical Characteristics and Operational Limitations</u>

1.	Type Design Definition:	Refer to C.E.A.P.R. document 1001130
2.	Description:	Single-engine, four-seat, low-wing airplane, wood construction, fixed tricycle landing gear.
3.	Equipment:	The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

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4. Dimensions

Span	8.72 m	(28.61 ft)
Height	2.23 m	(7.32 ft)
Length	7.10 m	(23.29 ft)
Wing Area	14.20 m²	(152.85 foot ²)

5. Engine:

Lycoming IO-360-A1 B6

The EASA type certification standard includes that of FAA TC 1E10, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

2700 rpm

Maximum Continuous Power:

6. Propellers:

5.1 Engine Limits:

Manufacturer	Model	Ø	Number of blades	Type Certificate	Sense of rotation
MT Propeller	MTV-12B/188-59b	1.88 m	3	EASA TC P 013	Clockwise (viewed in flight direction)

7. Fluids:

7.1 Fuel:

100/130 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Mai	n tank	RH	tank	LH t	tank	Auxilia	ry tank
(lit	ters)	(lite	ers)	(lite	ers)	(optiona	l) (liters)
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	109	40	40	40	40	50	50

8.2 Oil:



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^{7.2} Engine Oil:

9. Air speeds:

V_{NE}	
Vc	
VA	
V_{FE}	170 km/h (92 knots IAS)

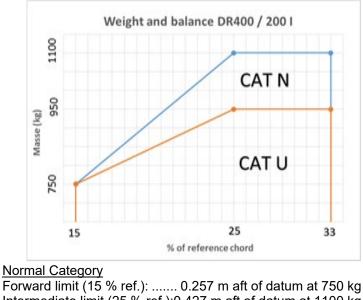
- 10. Maximum Operating Altitude:
- 11. Operational Capability:
- 12. Maximum Masses:

"N" Ca	"U" Category	
Take-off		
1100 kg (2425 lb)	1100 kg (2425 lb)	950 kg (2094 lb)

Refer to approved aircraft flight manual.

Refer to approved aircraft flight manual.

13. Centre of Gravity Range:



Forward limit (15 % ref.): 0.257 m aft of datum at 750 kg
Intermediate limit (25 % ref.):0.427 m aft of datum at 1100 kg
Aft limit (33 % ref.): 0.564 m aft of datum at 1100 kg
Utility Category
Forward limit (15 % ref.): 0.257 m aft of datum at 750 kg
Intermediate limit (25.9% ref.): 0.427 m off of datum at 050 kg

Intermediate limit (25 % ref.):. 0.427 m aft of datum at 950 kg Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:			
6	Normal Category:	Flaps up n+ 3.8	
		Flaps up n 1.9	
		Flaps down n+ 2	
		Flaps down n0	
	Utility Category:	Flaps up n+ 4.4	
	, , ,	Flaps up n 2.2	
		Flaps down n+ 2	
		Flaps down n0	
16. Leveling Means:	Horizontal reference upper fuselage spar		
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum		

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14. Datum:

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.						
19. Baggage / Cargo Compartment	Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum					
20. Wheels and Tires						
	Main gear track2.58 m (8.46 ft)Wheel tire size380 x 150 or 5.00-5Front gear angular movementleft: 27°right: 27°right: 27°Tire pressure refer to the maintenance manualOleo strut pressurerefer to the maintenance manual					
21. Control surface movements						
	Elevator:					
	down 12° ± 30' Ailerons: Relative to the trailing edge of the wings					
	up neutral down					
	15°± 1° 2° ± 1° 10°± 1°					
	Elevator tab:6° ± 1° Elevator up:25°30' ± 1°6° ± 1°					
	Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$					
	Flaps:1st notch:15° \pm 5°					
	2nd notch:					
	Rudder:					

22. (Reserved)

OO.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6
Airplane Structural Repair Manual	. Refer to the latest amendment of Service Letter no. 6
Illustrated Parts Catalogue	. Refer to the latest amendment of Service Letter no. 6

OO.VNotes

1. This plane is identical to DR 400/180 except for:

Powerplant installation which is identical to DR400/200R except for the propeller MT Propeller MTV-12B/188-59b

2. First model is serial number 2695.

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Section PP: <u>Common Notes</u>

1. <u>Type transformation :</u>

Type transformation are only possible by the manufacturer.

DR 400 RP:

It is not authorized to transform a DR400/180R to a DR400 RP.

DR 400/200 R and DR 400/200 I:

It is not authorized to transform a DR 400/180R to a DR 400/200 R or a DR 400/200 I.

2. Type Certificate Holder :

C.E.A.P.R. (Centre Est Aéronautique Pierre Robin) 1 route de Troyes 21121 DAROIS FRANCE

3. Manufacturer:

From October 1957 to August 1996

Centre Est Aéronautique Boite Postale 40 21 DIJON FRANCE

Avions P. Robin 21121 FONTAINE LES DIJON FRANCE

From September 1996 to December 2003 C.A.B (Construction Aéronautique de Bourgogne) 1 route de Troyes 21121 DAROIS FRANCE

From January 2004 to August 2008

APEX Industries 1 route de Troyes 21121 DAROIS FRANCE

2011 : DR400-140B serial number 2650 only Finch Aircraft 1 route de Troyes 21121 DAROIS FRANCE

Since May 2011

Robin Aircraft 1b route de Troyes 21121 DAROIS FRANCE



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ADMINISTRATIVE SECTION

- I. Acronyms & Abbreviations
- II. Type Certificate Holder Record

Société Avions Pierre Robin Société Avions Robin ROBIN Aviation APEX Aircraft C.E.A.P.R.

III. Change Record

Issue 1	January 28, 2013	Initial issue on transfer of this Type Certificate to CEAPR	
Issue 2	2017, September 25	New model DR400 / 200 I (refer to section AB)	
Issue 3	December 2018	Merger with EASA.A.510 (DR 200 series)	
Issue 4	November 2 nd , 2020	TC holder and manufacturer transferred to Section PP Note 2 and 3	
		Type Definition Design is added for each aircraft	
		DR300 & DR400 series : wheel dimension 5.00-5 added	
		DR400 series : Standard 92 and Standard 88 models definition added	
		Section JJ : DR400/100 - Update	
		Section NN : DR400/500 N.III.15 – Load Factor correction	
		Section OO : DR400/200I - editorial change	
		Section PP : Note 1;2 & 3 - update	
Issue 5	October 16 th , 2023	DR200, DR300 & DR400 series : Oleo strut and tire pressure suppression	
		DR253 series : Tire dimension 5.00-5 added + note addition.	

-END-

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