TYPE-CERTIFICATE
DATA SHEET

NO. EASA.A.594

for
PC-24

Type Certificate Holder
PILATUS Aircraft Ltd.

Pilatusstrasse 1
6371 Stans
Switzerland

For models: PC-24
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SECTION A: PC-24

A.I General

1. Type: PC-24
   Variant:

2. Airworthiness Category: CS-23, Commuter category

3. Type Certificate Holder: PILATUS Aircraft Ltd.
   Pilatusstrasse 1, 6371 Stans
   Switzerland

4. Manufacturer: PILATUS Aircraft Ltd.
   Pilatusstrasse 1, 6371 Stans
   Switzerland

5. Certification Application Date: 9 July 2012

6. EASA Certification Date: 7 December 2017

A.II Certification Basis

1. Reference Date for determining the applicable requirements: 7 December, 2012

2. (reserved)

3. (reserved)

4. Certification Basis: EASA CRI A-01 (Refer further to Note 5)


   (Refer further to Note 5)

7. Special Conditions:
   - CRI B-01 Handling and Performance
   - CRI B-02 High Speed Characteristics
   - CRI B-03 Stall Speed Determination
   - CRI B-04 Contaminated Runways
   - CRI B-05 Stick Pusher
   - CRI B-152 Human Factors
   - CRI C-01 Sonic Fatigue
   - CRI C-02 Pressurisation into Non-Pressurized Areas
   - CRI C-05 Dynamic Response
   - CRI C-06 Out of Trim Conditions (Structures)
   - CRI C-07 Round-the-clock Gust
   - CRI D-01 Take-Off Warning System
   - CRI D-02 Extension and Retraction Systems
CRI D-03  Wheels
CRI D-04  Brakes and Braking Systems
CRI D-05  Doors
CRI D-06  Bird Strike
CRI D-09  Operation above 41.000 ft (see note 4)
CRI E-01  Fuel Tank Crashworthiness
CRI E-04  Lines, Fittings and Components
CRI E-06  Powerplant Fire Extinguishing Systems
CRI E-10  Fuel Tank Ignition Prevention
CRI E-11  Induction System Ice Protection - Cold Soaked Fuel
CRI E-59  Engine Installation (Rain Conditions)
CRI E-102 Single Point Defueling
CRI F-01  Battery Endurance Requirement
CRI F-03  Interaction of Systems and Structures
CRI F-07  Data Link Services Recording
CRI F-15  Airworthiness Information Security
CRI F-52  Protection from effect of HIRF
CRI F-54  Protection from the effects to lightning strike, indirect effects
CRI F-58  Lithium Battery Installations
CRI F-62  Flight Instrument External Probes – Qualification in extended icing conditions
CRI F-110  Auto-throttle
CRI G-02  Approval process of digital AFM
CRI O-01  Steep Approach
CRI O-04  Towbarless towing loads
CRI AWO-101 CAT II requirements for CS 23 aeroplane

8. Exemptions:
   None.

9. Equivalent Safety Findings:
   CRI E-56  Powerplant System Indications
   CRI F-05  IMA Individual Circuit Protection
   CRI F-90  ASI Flaps Markings on PFD
   CRI F-108 ESIS 3rd ATT Indicator (ESIS) Compliance to CS 23.1303
   CRI F-111 Mechanical Magnetic Compass - Flight Deck without Whisky Compass
   CRI F-112 Pressurization and Pneumatic systems – bleed air level compliance

10. Environmental Standards:
    CRI N-01 Noise Standards.
    CRI N-02 Reference T/O-speed for Part 23 Jet Noise Certification.
    CRI N-03 Use of NTO vs. MTO

11. Operational Suitability Certification
   Basis:
   MMEL: CS-MMEL, Initial Issue.
   Simulator Validation Data: CS-SIMD, Initial Issue.


A.III  Technical Characteristics and Operational Limitations

1. Type Design Definition: 500.00.24.001

2. Description: The PC-24 is a low-wing Business aircraft, powered by two rear-mounted Williams FJ44-4A-QPM twin spool turbofan engines of 3,420 lbs take-off thrust rating, with a T-tail configuration and a retractable undercarriage.

   The PC-24 is pressurised with an 8’000 ft cabin altitude at its maximum operating altitude of 45’000ft.

   It has a maximum seating capacity of up to 10 passengers in the cabin and 1 or 2 pilots. Standard seating configuration is a 6-seat executive arrangement with forward lavatory and aft galley. The aircraft may be flown with one or two pilots.

   A unique feature of the PC-24 shall be the capability of transporting a mixture of passengers and cargo, using the two doors. The PC-24 has a passenger door on the left hand side behind the cockpit, a large cargo door at the back of the cabin on the left hand side behind the wing and two over wing emergency exits, one on each side of the cabin.

   The PC-24 aircraft is designed to be able to take-off and land in short airfields (<2’650ft)

3. Dimensions:
   Main Wing Span: 17’000 mm (55 ft 9 in)
   Length: 16’800 mm (55 ft 1 in)
   Height: 5’400 mm (17 ft 4 in)
   Total Wing Area: 30.91 m2 (332.7 ft2)
4. Engines:
   Model: 2 Williams International FJ44-4A-QPM Turbofan engines of 3,420 lbf maximum take-off thrust each, situated in nacelles on each side of the rear fuselage.
   Type Certificate: The FJ44-4-QPM is certified by EASA under Type Data Sheet number TCDS IM.E.016 issue 10 dated 4 August 2017.

5. Engine Limits:
   Refer to latest revision TCDS No. IM.E.016 Williams International Engine FJ44-4A-QPM.
   Oil Temperature: Refer to latest revision TCDS No. IM.E.016 Williams International Engine FJ44-4A-QPM.

6. Fluids:
               • Fuel Anti-Ice Additives are not required.

   6.2. Oil: • Refer to the latest revision Williams International Engine Installation and Operating Instructions 110675-201 FJ-44-4A-QPM (73200-201) (including Mobil Jet II, Mobil 254)

7. Fuel capacities
   7.1. Fuel:
   Total: 3,389 lt (894 US Gal) 2,721 kg (6,000 lb)
   Usable: 3,369 lt (890 US Gal) 2,705 kg (5,964 lb)
   Unusable: 20 lt (5.3 US Gal) 16 kg (35 lb)

   7.2. Oil:
   Total: 5.5 lt (5.85 qts)
   Usable quantity: 4.3 lt (4.63 qts)
8. Air Speeds:

VMO (maximum operating speed) 290 KEAS
MMO (maximum operating Mach number) 0.74
VD (maximum diving speed) 360 KEAS
MD (maximum diving Mach number) 0.81
VA (manoeuvring speed) at MTOW 185 KEAS
VC (design cruising speed) 290 KEAS
VFE (max. flap extended speed) 8° (Take-Off) Flap 200 KEAS
15° (Approach) Flap 200 KEAS
33° (Landing) Flap 175 KEAS
VLO (maximum landing gear operating speed) Extension 250 KEAS
Retraction 200 KEAS
VLE (maximum landing gear extended speed) 250 KEAS
VSO (stall speed, ISA, sea level, max landing weight, landing
configuration) 82 KCAS

9. Maximum Operating Altitude:

13'716 m / 45'000 ft
(see note 4)

10. Operational Capabilities:

IFR Day/Night; VFR Day/Night, FIKI (Note 6)

11. Maximum Weight:

For aircraft 101 - 130 Post SB 42-002, and 131 - Up

<table>
<thead>
<tr>
<th>Category</th>
<th>Weight (kg)</th>
<th>Weight (lbs)</th>
</tr>
</thead>
<tbody>
<tr>
<td>Taxi and ramp</td>
<td>8'345</td>
<td>18'400</td>
</tr>
<tr>
<td>Take-off</td>
<td>8'300</td>
<td>18'300</td>
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<tr>
<td>Landing</td>
<td>7'665</td>
<td>16'900</td>
</tr>
<tr>
<td>Zero fuel</td>
<td>6'450</td>
<td>14'220</td>
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</table>

For aircraft 101 - 130 Pre SB 42-002 (see Note 13)

<table>
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<tr>
<th>Category</th>
<th>Weight (kg)</th>
<th>Weight (lbs)</th>
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<td>Taxi and ramp</td>
<td>8'050</td>
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<td>Take-off</td>
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<td>17'650</td>
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<tr>
<td>Landing</td>
<td>7'370</td>
<td>16'250</td>
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<tr>
<td>Zero fuel</td>
<td>6'100</td>
<td>13'450</td>
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</table>
12. Centre of Gravity Range:

The diagram shows the PC-24 Centre of Gravity (CG) limits, which accommodate all of the foreseen passenger and cargo loadings. For aircraft 101 - 130 Post SB 42-002, and 131 - Up.

For aircraft 101 - 130 Pre SB 42-002 (see Note 13)

13. Mean Aerodynamic Chord (MAC):

1.997 m (6ft 6”)

14. Levelling Means:

Refer to the PC-24 Airplane Flight Manual, Section 6
15. Minimum Flight Crew: 1 Pilot
(see Note 8))

16. Maximum Passenger Seating Capacity:
Executive Interiors Configuration: 8 PAX excluding pilot seats (see Note 9, 10 and 11). An optional fit allows two additional infants to be carried at the first seating row on the left and right sides.

Commuter Interior Configuration: 10 PAX excluding pilot seats. (see Note 9, 10 and 11).

Refer to the PC-24 Airplane Flight Manual, Section 6, for passengers and flight crew loading instructions and approved configurations.

17. Exit: Nb. and Type: 3 exits (fwd cabin LH passenger door and two over wing emergency exits, one on each side of the cabin) and 1 cargo door (LH rear cabin)

18. Baggage / Cargo Loading:
Refer to the PC-24 Airplane Flight Manual, Section 6

19. Wheels and Tyres:
19.1. Wheels: Nose Landing Gear: Parker 40-479
Main Landing Gear: Parker 40-478

19.2. Tyres:
<table>
<thead>
<tr>
<th>Nose Landing Gear:</th>
<th>Dimensions</th>
<th>Ply Rating</th>
<th>Speed Rating</th>
</tr>
</thead>
<tbody>
<tr>
<td>450x190-5</td>
<td>8 (PR)</td>
<td>190 (MPH)</td>
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<tr>
<td>Main Landing Gear:</td>
<td>24x7.7</td>
<td>10 (PR)</td>
<td>190 (MPH)</td>
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</tbody>
</table>

A.IV  Operating and Service Instructions

1. Aircraft Flight Manual (AFM):
   Airplane operation must be in accordance with the EASA approved PC-24 Airplane Flight Manual and AFM supplements as define below:
   S/N P03, 101 and up  Pilatus Report No. 02371

2. Aircraft Maintenance Manual (AMM):
   Airplane maintenance must be in accordance with the document as defined below:
   S/N P03, 101 and up  Pilatus Report No. 02378

   Airplane Repairs must be in accordance with the document as define below:
   S/N P03, 101 and up  Pilatus Report No. 02379

   S/N P03, 101 and up  Pilatus Report PC-24 No. 02383
5. Service Bulletins (SBs):
   All Pilatus PC-24 Bulletin are listed in the following document:
   S/N P03, 101 and up Pilatus Report No. 02430

6. All Pilatus PC-24 Service Letters are listed in the following document:
   S/N P03, 101 and up Pilatus Report No. 02431

7. RVSM capability for PC-24 S/N P03, 101 and subsequent:
   All airplanes equipped with Honeywell APEX system are RVSM capable, provided the operator follows the AFM Issue 003 Revision 1 (or later revisions) and the AMM Issue 005 Revision 00, or later EASA approved revisions.

A.V Operational Suitability Data (OSD)

1. Master Minimum Equipment List (MMEL) Pilatus Report PC-24 No 02384, latest approved revision
2. Flight Crew Data Pilatus Report PC-24 No 02423, latest approved revision, as per the table below:

<table>
<thead>
<tr>
<th>Manufacturer</th>
<th>Aircraft model/name</th>
<th>License endorsement</th>
<th>Variants</th>
<th>Complex</th>
<th>SP/SP HPA / MP</th>
<th>OE GM / OEB / OSD FC available</th>
<th>Remarks</th>
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<tr>
<td>Pilatus Aircraft Ltd.</td>
<td>PC-24</td>
<td>PC-24</td>
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<td>OSD FC PC-24</td>
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<td>PC-24 AYT</td>
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<tr>
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<td>PC-24 TF</td>
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<td></td>
<td>PC-24 AYT/TF</td>
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</tr>
</tbody>
</table>

3. Simulator Data Validation Data Roadmap (VDR) report ER-24-001168, latest approved revision
A.VI Notes

1. Requirements for the issue of the C. of A.
   - The minimum required equipment as prescribed in the applicable airworthiness regulations must be installed on the individual aircraft for certification.

   - Current weight and balance data, a list of equipment included in the certification empty weight and loading information when necessary must be provided for each aircraft when the C.o.A. will be issued.

   The certification empty weight and balance data shall include the unusable fuel and the total engine oil as specified:

   - Airplane Flight Manual is required.

2. Placards
   All required placards as listed in the Pilatus Aircraft Flight Manual, and subsequent approved revisions, must be installed in the appropriate locations.

3. Continued Airworthiness
   - Airworthiness Limitations are contained in Chapter 4 of the Pilatus AMM. These Limitations may not be changed without EASA approval.

   - Current weight and balance data together with a list of equipment included in the certificated empty weight, and loading instructions, when necessary, must be provided for each airplane at the time of original certification.

   - Only interior configurations described in the official Pilatus AFM are approved for installation in the PC-24 aircraft.

4. High altitude operations
   PC-24 airplanes have been approved for high altitude operations (altitudes above 41,000 feet), by Special Conditions. Any modifications to the pressure vessel must be approved in accordance with the requirements as shown in the certification basis. This includes modifications which could result in a pressure vessel opening, either through crack-growth or antenna loss, greater than 2.65 sq.in.

5. To support European operators and ensure that the PC-24 aircraft complies to the airspace rules, at initial TC, Pilatus elected to comply with Airborne Communications, Navigation and Surveillance (ACNS), reference CS ACNS initial issue dated 17 December 2013.

   For the Major Change project EASA n°60077318 introducing RNP AR, Pilatus elected to comply with the following sections of CS-ACNS issue 2 dated 26 April 2019,

   Subpart C — Navigation (NAV):
   - Subsection 6 — Supplementary specifications for RNP authorisation required (RNP AR)
   - Subsection 7 — Supplementary specifications for applications for advanced RNP (A-RNP)
   - Subsection 8 — Supplementary specifications supporting radius to fix (RF)
   - Subsection 10 — Supplementary specifications supporting parallel offset

   As CS-ACNS is continuously developed and updated, Pilatus may choose in the future to elect to comply with new sections or amendments.


For Special Conditions (SC) and Equivalent Safety Findings (ESF), which are listed in the CRI A-01 and are part of the applicable certification basis refer further to the Annex to EASA.A.594

6. The PC-24 is approved for flight into known or forecasted icing. Compliance has been shown i.a.w. CS-23.1419 and SC F-62.

7. The PC-24 S/N P03, 101 and subsequent equipped with Honeywell APEX system are RVSM capable. The commercial designation of the Honeywell APEX system as installed on the PC-24 is the PC-24 Advanced Cockpit Environment (ACE™).

8. Approval for operation with a minimum crew of one pilot is based upon the cockpit equipment installation and arrangement evaluated during certification testing. No significant changes may be made to the installed cockpit equipment or arrangement (EFIS, autopilot, avionics, etc.), except as permitted by the approved MMEL, without prior approval.

9. All replacement seats (crew and passenger), although they may comply with TSO C127, must also be demonstrated to comply with CS 23.321, 23.395, 23.561, 23.562, and 23.785.

10. The foam cushion build up of all seats (crew and passenger) may not be altered. Any deviations in the foam construction or stiffness must be demonstrated by test to comply with the listed CS 23 paragraphs

11. During single pilot operation, the pilot occupies the left hand cockpit seat and an additional passenger may occupy the right hand cockpit seat

12. Aircraft Classification

In accordance with the EU Regulation (EC) 216/2008 definitions, the PC-24 is a "complex motor powered aircraft".

In accordance with the EU "Air Ops" Regulation (EC) 965/2012 definitions, the PC-24 is a "Performance Class A aeroplane".

In addition, the type certification basis of the baseline PC-24 product is CS-23 Amdt. 3 in the commuter category. However, considering the new definitions introduced in CS-23 Amdt. 5 §23.2005, the PC-24 is considered as equivalent to an aeroplane in the new normal category with:

- Aeroplane certification level: "Level 4"
- Aeroplane performance level "High speed"

13. All aircraft MSN 101-130 have been retrofitted with SB 42-002.
ADMINISTRATIVE SECTION

I. Acronyms & Abbreviations
A.C. – Advisory Circular
A.D. – Airworthiness Directive
AMM – Aircraft Maintenance Manual
C.o.A. – Certificate of Airworthiness
CRI – Certification Review Item
CS – Certification Specification
EASA – European Union Aviation Safety Agency
EFIS – Electronic Flight Information System
FADEC – Full Authority Digital Engine Control
FIKI – Flight Into Known Icing
FOCA – Swiss Federal Office of Civil Aviation
IAS – Indicated Airspeed
ICAO – International Civil Aviation Organization
IFR – Instrument Flight Rules
KCAS – Calibrated Airspeed [knots]
KEAS – Equivalent Airspeed [knots]
KIAS – Indicated Airspeed [knots]
Lt – Litres
MAC – Mean Aerodynamic Chord
MMEL – Master Minimum Equipment List
N.A.A. – National Aviation Authority
OSD – Operational Suitability Data
RVSM – Reduced Vertical Separation Minimum
TCDS – Type Certificate Data Sheet
VFR – Visual Flight Rules

II. Type Certificate Holder Record
PILATUS Aircraft Ltd.
Pilatusstrasse 1
6371 Stans
Switzerland
### III. Change Record

<table>
<thead>
<tr>
<th>Issue</th>
<th>Date</th>
<th>Changes</th>
</tr>
</thead>
<tbody>
<tr>
<td>Issue 1</td>
<td>7 Dec 2017</td>
<td>Initial Issue</td>
</tr>
<tr>
<td>Issue 2</td>
<td>17 Apr 2018</td>
<td>Update to include OSD-FCD and OSD-SIMD</td>
</tr>
<tr>
<td>Issue 3</td>
<td>9 Oct 2018</td>
<td>Update for MTOW increase</td>
</tr>
<tr>
<td>Issue 4</td>
<td>5 Nov 2020</td>
<td>Addition of CRI F-07 (special condition for Data Link Services Recording) and CRI O-01 (special condition for Steep Approach); addition of 10 seat commuter configuration.</td>
</tr>
<tr>
<td>Issue 5</td>
<td>12 Nov 2020</td>
<td>Correction of Note 11</td>
</tr>
<tr>
<td>Issue 6</td>
<td>15 Mar 2021</td>
<td>OSD FCD table added.</td>
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