TYPE CERTIFICATE DATA SHEET

No. EASA.R.509

for
AW169

Type Certificate Holder
Leonardo S.p.A.

Helicopters
Piazza Monte Grappa, 4
00195 Roma
Italy

For Model: AW169
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SECTION 1: AW169

I. General

1. Type/ Model
   1.1 Type
   AW169
   1.2 Model
   AW169

2. Airworthiness Category
   Large Rotorcraft, Category A and B

3. Type Certificate Holder
   Leonardo S.p.A. Helicopters
   Piazza Monte Grappa, 4
   00195 Roma, Italy

4. Manufacturer
   See Note 2

5. Type Certification Application Date
   9 February 2011

6. State of Design Authority
   EASA

7. EASA Type Certification Date
   15 July 2015

II. Certification Basis

1. Reference Date for determining the applicable requirements
   For Airworthiness and Environmental Protection: 9 February 2011, for OSD elements: 7 October 2014.

2. Airworthiness Requirements
   CS-29 Amdt. 2, dated 17 November 2008
   CS-29 Amdt. 3, dated 11 December 2012 for the following installations and affected areas only:

   - Kit Single Rescue Hoist P/N 6F2591F00111
   - 50 meters Hoist P/N 6F2591F00211, except for CS 29.337 through 29.341, CS 29.571 and CS 29.29.865 (a), (f) of CS-29 Amdt. 6, dated 17 December 2018
   - CS 29.1465 Vibration health monitoring of CS-29 Amdt. 5, dated 14 June 2018
   - CS-29 Amdt. 6, dated 17 December 2018 for Kit Enhanced Performance (P/N 6F0000F00511, 6F0000F00611) and affected areas only

3. Special Conditions
   SC E-12 Loss of Oil from Gearboxes Utilising a Pressurised Lubrication System
   SC E-15 Extended Take-Off Power Duration
   SC F-1 ‘HIRF Protection’ in accordance with JAA Interim Policy INT/POL/27&29/1, Issue 3, dated 1 October 2003
   SC F-21 Lithium Battery Installation
   SC F-23 Non-Rechargeable Lithium Battery Installation

4. Equivalent Safety Findings
   ESF D-02 CS 29.813(c) – ‘Emergency Exit access’
   ESF D-03 CS 29.807(c)(1) – ‘Passenger Emergency Exits other than side-of-fuselage’
   ESF D-04 CS 29.811(d) ‘Emergency Exit signs’
   ESF D-05 CS 29.601, CS 29.603, CS 29.605, CS 29.865, CS 29.1301(d) – ‘Hoist installation’
   ESF D-07 CS 29.807(d)(2) – ‘Ditching Emergency Exits for passengers’
   ESF E-17 CS 29.923, CS 29.927 – ‘Rotor drive system and control mechanism tests: Endurance and additional tests by test rig’
5. Deviations none
6. Environmental Protection Requirements
   6.1 Noise Requirements CS-36 Amdt. 3, see TCDSN EASA.R.509
   6.2 Emission Requirements Chapter 2 of ICAO Annex 16 Volume II, Amdt. 6, Part II to
   Chicago Convention (as implemented in CS-34 Initial Issue)
7. Operational Suitability Data (OSD) (For OSD elements see SECTION 2 below)
   7.1 Master Minimum Equipment List (MMEL) Certification Specifications and Guidance Material for
   Master Minimum Equipment List, CS-MMEL, initial issue, dated 31 January 2014
   7.2 Flight Crew Data (FCD) Certification Specifications for Operational Suitability
   Data (OSD) Flight Crew Data, CS-FCD, initial issue, dated 31 January 2014
   7.3 Simulation Data (SIMD) Special Condition NPA 2013-17 (CS-SIMD),
dated 27 August 2013
   7.4 Maintenance Certifying Staff Data (MCSD) reserved

III. Technical Characteristics and Operational Limitations
1. Type Design Definition Doc. No. 169F0272N002
2. Description Large twin-engine helicopter, conventional configuration,
   5-blade fully articulated interblade main rotor, 3-blade
   fully articulated tail rotor, retractable tricycle landing gear.
3. Equipment As per compliance with certification basis and included in
   Type Design Definition Document
4. Dimensions
   4.1 Fuselage Length: 12.19 m
   Width hull: 2.15 m
   Height: 3.88 m
   4.2 Main Rotor Diameter: 12.12 m
   4.3 Tail Rotor Diameter: 2.40 m
5. Engine
   5.1 Model Pratt & Whitney Canada
   2 x Model PW210A
   or,
   2 x Model PW210A1 (helicopters with Kit Enhanced
   Performance installed)
   5.2 Type Certificate TCCA TC/TCDS: E-36
   EASA TC/TCDS: EASA IM.E.126
5.3 Limitations

In accordance with PW210A Pratt & Whitney Canada Installation Manual (Ref. to 30L2374)

5.3.1 Installed Engine Limits

### PW210A

<table>
<thead>
<tr>
<th>Rating</th>
<th>Max Torque [% (Nm)]</th>
<th>Max ITT [°C]</th>
<th>Max NG [% (rpm)]</th>
<th>Max NF [% (rpm)]</th>
</tr>
</thead>
<tbody>
<tr>
<td>AEO</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Continuous</td>
<td>118.6 (395.9)</td>
<td>868</td>
<td>96.5 (49 200)</td>
<td>107 (28 120)</td>
</tr>
<tr>
<td>Take-off 5 min</td>
<td>125.9 (420.3)</td>
<td>930</td>
<td>98.2 (50 100)</td>
<td></td>
</tr>
<tr>
<td>Take-off 30 min(*)</td>
<td>148.3 (494.9)</td>
<td>941</td>
<td>98.9 (50 430)</td>
<td></td>
</tr>
<tr>
<td>OEI</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Continuous</td>
<td>148.3 (494.9)</td>
<td>941</td>
<td>98.9 (50 430)</td>
<td>107 (28 120)</td>
</tr>
<tr>
<td>2.5 min</td>
<td>174.7 (583)</td>
<td>1 020</td>
<td>100.7 (51 360)</td>
<td></td>
</tr>
</tbody>
</table>

(*) if Core Avionic SW phase 4.0 P/N 6F4600A00114, or later, is installed.

### PW210A1 (helicopters with Kit Enhanced Performance installed)

<table>
<thead>
<tr>
<th>Rating</th>
<th>Max Torque [% (Nm)]</th>
<th>Max ITT [°C]</th>
<th>Max NG [% (rpm)]</th>
<th>Max NF [% (rpm)]</th>
</tr>
</thead>
<tbody>
<tr>
<td>AEO</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Continuous</td>
<td>118.6 (395.9)</td>
<td>868</td>
<td>96.5 (49 200)</td>
<td>107 (28 120)</td>
</tr>
<tr>
<td>Take-off 5 min</td>
<td>125.9 (420.3)</td>
<td>937</td>
<td>98.2 (50 100)</td>
<td></td>
</tr>
<tr>
<td>Take-off 30 min</td>
<td>148.3 (494.9)</td>
<td>937</td>
<td>98.8 (50 400)</td>
<td>107 (28 120)</td>
</tr>
<tr>
<td>OEI</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Continuous</td>
<td>148.3 (494.9)</td>
<td>937</td>
<td>98.8 (50 400)</td>
<td>107 (28 120)</td>
</tr>
<tr>
<td>2.5 min</td>
<td>185 (618.3)</td>
<td>1 020</td>
<td>100.7 (51 360)</td>
<td></td>
</tr>
</tbody>
</table>

5.3.2 Transmission Torque Limits

### PW210A

<table>
<thead>
<tr>
<th>Rating</th>
<th>Max Torque [% (Nm)]</th>
<th>Input speed [rpm]</th>
<th>Input power [hp]</th>
</tr>
</thead>
<tbody>
<tr>
<td>AEO Max</td>
<td>2 x 100 (334)</td>
<td>14 400</td>
<td>1 350 (675 x 2)</td>
</tr>
<tr>
<td>5 min</td>
<td>2 x 111 (371)</td>
<td></td>
<td>1 500 (750 x 2)</td>
</tr>
<tr>
<td>30 min(*)</td>
<td></td>
<td></td>
<td>1 500 (750 x 2)</td>
</tr>
<tr>
<td>OEI Max</td>
<td>140 (470)</td>
<td>14 400</td>
<td>950</td>
</tr>
<tr>
<td>2.5 min</td>
<td>174 (583)</td>
<td></td>
<td>1 180</td>
</tr>
</tbody>
</table>

(*) if Core Avionic SW phase 4.0 P/n 6F4600A00114, or later, is installed.

### PW210A1 (helicopters with Kit Enhanced Performance installed)

<table>
<thead>
<tr>
<th>Rating</th>
<th>Max Torque [% (Nm)]</th>
<th>Input speed [rpm]</th>
<th>Input power [hp]</th>
</tr>
</thead>
<tbody>
<tr>
<td>AEO Max</td>
<td>2 x 100 (334)</td>
<td>14 400</td>
<td>1 350 (675 x 2)</td>
</tr>
<tr>
<td>5 min</td>
<td>2 x 111 (371)</td>
<td></td>
<td>1 650 (825 x 2)</td>
</tr>
<tr>
<td>30 min</td>
<td></td>
<td></td>
<td>1 650 (825 x 2)</td>
</tr>
<tr>
<td>OEI Max</td>
<td>148 (493)</td>
<td>14 400</td>
<td>1000</td>
</tr>
<tr>
<td>2.5 min</td>
<td>185 (618)</td>
<td></td>
<td>1 250</td>
</tr>
</tbody>
</table>

6. Fluids

6.1 Fuel

JET A, JET A1, JP8, JP8+100, No. 3 Jet Fuel (for code no. specification and more details refer to approved RFM)

6.2 Oil

Transmissions: AeroShell Turbo Oil 555 (DoD-L-85734).

No different specification or brand allowed.

Engine: Refer to approved RFM
Hydraulics: MIL-PRF-83282, MIL-PRF-87257 (as alternative)

6.3 Additives

Refer to approved RFM

6.4 Coolant

R134a

7. Fluid capacities

7.1 Fuel

<table>
<thead>
<tr>
<th>Total capacity [litres (kg(*)]]</th>
<th>Unusable [litres (kg(*)]]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Two main fuel tanks (LH and RH)</td>
<td>1130 (904) 20 (16)</td>
</tr>
</tbody>
</table>

(*)Fuel mass defined assuming a standard fuel density of 0.8 kg/litre

7.2 Oil

<table>
<thead>
<tr>
<th>Engine (each)</th>
<th>min 5.25 (4.948) - max 5.78 (5.448)</th>
</tr>
</thead>
<tbody>
<tr>
<td>Main gearbox (min/max)</td>
<td>min 17 (16.968) - max 19 (18.964)</td>
</tr>
<tr>
<td></td>
<td>(16.8 + 2.2 for oil cooler, oil ducts and filter)</td>
</tr>
<tr>
<td>Intermediate gearbox</td>
<td>0.77 (0.768)</td>
</tr>
<tr>
<td>Tail gearbox</td>
<td>1.10 (1.098)</td>
</tr>
<tr>
<td>Hydraulic (per each Power Control Module)</td>
<td>1.3 (1.1)</td>
</tr>
</tbody>
</table>

(*) Oil mass at 80°C

7.3 Coolant System Capacity

2.1 kg

8. Air Speed Limitations

V_NE Power On AEO*: 165 KIAS
V_NE Power On AEO*: 160 KIAS
V_NE Power On OEI*: 135 KIAS
V_Ne Power Off*: 125 KIAS

For reduction of the V_Ne with density altitude (HP/OAT), refer to approved RFM.

(*) if Core Avionics SW Phase 6.0, or later is installed

9. Rotor Speed Limitations

<table>
<thead>
<tr>
<th>Power On AEO*</th>
<th>Condition</th>
<th>[rpm]</th>
<th>[%]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Minimum Continuous</td>
<td>317.56</td>
<td>96.0</td>
<td></td>
</tr>
<tr>
<td>Maximum Continuous</td>
<td>354.72</td>
<td>103.0</td>
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</tr>
</tbody>
</table>

<table>
<thead>
<tr>
<th>Power On OEI</th>
<th>Condition</th>
<th>[rpm]</th>
<th>[%]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Minimum Cautionary</td>
<td>304.05</td>
<td>90.0</td>
<td></td>
</tr>
<tr>
<td>Minimum Continuous</td>
<td>341.21</td>
<td>101.0</td>
<td></td>
</tr>
<tr>
<td>Maximum Continuous</td>
<td>354.72</td>
<td>105.0</td>
<td></td>
</tr>
</tbody>
</table>

<table>
<thead>
<tr>
<th>Power Off</th>
<th>Condition</th>
<th>[rpm]</th>
<th>[%]</th>
</tr>
</thead>
<tbody>
<tr>
<td>Minimum Continuous</td>
<td>304.05</td>
<td>90.0</td>
<td></td>
</tr>
<tr>
<td>Maximum Continuous</td>
<td>371.61</td>
<td>110.0</td>
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</tr>
</tbody>
</table>

(*) Maximum and minimum continuous values of the flight envelope. AVSR provides a governing of the rotor speed
at different values depending on airspeed (TAS/IAS**) and density altitude. As the NR datum is variable, NR green band is variable as well (±2% across the datum value).

(**) IAS if Core Avionics SW Phase 6.0, or later is installed

Refer to approved RFM for additional rotor speed limitations

10. Maximum Operating Altitude and Temperature

10.1 Altitude

Maximum operating altitude
15 000 ft* for operation at gross mass up to 4 600 kg, and,
10 000 ft* for operation at gross mass above 4 600 kg without Kit Enhanced Performance installed, or,
15 000 ft* for operation at gross mass above 4 600 kg with Kit Enhanced Performance is installed

Maximum Take-off and Landing altitude
15 000 ft* for operation at gross mass up to 4 600 kg, and,
10 000 ft* for operation at gross mass above 4 600 kg without Kit Enhanced Performance installed, or,
15 000 ft* for operation at gross mass above 4 600 kg with Kit Enhanced Performance is installed

* altitude in PA/DA (whichever occurs first)

10.2 Temperature

-40°C ± +50°C (ISA+35°C)
-40°C ± +50°C (ISA+35°C) for Cat A operations

For variation of temperature limitations with altitude refer to approved RFM and applicable supplement

11. Operating Limitations

VFR day and night and IFR operations in non-icing conditions

12. Maximum Mass

Take-off and landing: 4 600 kg, or,
4 800 kg, if P/N 6F0000F00211 is installed

Taxi and Towing: 4 650 kg, or,
4 850 kg, if P/N 6F0000F00211 is installed

13. Centre of Gravity Range

Refer to approved RFM

14. Datum

Longitudinal:
The datum plane (STA 0) is located at 3 528 mm forward to the front jack point

Lateral:
The datum plane (B.L. 0) is located at ±225 mm inboard of LH/RH front jack points.

15. Levelling Means

Plumb line from ceiling reference point to index plate on floor of baggage compartment; clinometer.

16. Minimum Flight Crew

One (1) pilot for VFR day and night and IFR.

For NVG operations, two (2) pilots or one (1) pilot and one (1) crew member required. Both pilot and crew member must be equipped with NVGs (see Note 3).

17. Maximum Passenger Seating Capacity

8

10 (if the kit 10 Seats Internal Arrangement P/N 6F2520F00111 is installed)
18. Passenger Emergency Exit
2 on each side of the passenger cabin
1 on each side of the passenger cabin, if the kit Sliding Aft Passenger Windows P/N 6F5630F00411 is installed.

19. Maximum Baggage/Cargo Loads
250 kg located in the baggage/cargo compartment

20. Rotor Blade Control Movement
For rigging information, refer to RFM

21. Auxiliary Power Unit (APU)
none

22. Life-limited Parts
Refer to the Airworthiness Limitation Section (ALS) of the Maintenance Manual

23. Wheels and Tyres
MLG wheel assembly with 18x5.5 tubeless tyres
NMG wheel assembly with 5x5.5 tubeless tyres

IV. Operating and Service Instructions

1. Flight Manual
Doc. No. 169F0290X001, initial issue, dated 8 July 2015, EASA approved 15 July 2015, or later approved revisions
Doc. No. 169F0290X012, initial issue, dated 21 December 2021, EASA approved 21 December 2021, or later approved revisions, only for helicopters with Kit Enhanced Performance installed

‘AW169 Maintenance Planning Information’
Doc. No. 69-A-AMPI-00-P, EASA accepted 15 July 2015, or later revisions, including:
- Chapter 4 ALS, EASA approved dated 15 July 2015, or later approved revisions;
- Chapter 5 with Scheduled Maintenance Requirements
  ‘Maintenance Review Board Report AW169 Helicopter’
  Doc. No. 169F0000M005
  ‘AW169 Maintenance Publication’
  Doc. No. 69-A-AMP-00-X
  ‘AW169 Material Data Information’
  Doc. No. 69-A-AMDI-00-X
  ‘AW169 Corrosion Control Publication’
  Doc. No. 69-A-ACCP-00-X
  ‘AW169 Fault Isolation Publication’
  Doc. No. 69-A-AFIP-00-X
  ‘AW169 Wiring Data Publication’
  Doc. No. 69-A-AWDP-00-X

‘AW169 Structural Repair Publication’
Doc. No. 69-A-ASRP-00-X
‘AW169 Component Repair and Overhaul Publication’
Doc. No. 69-A-CR&OP-00-X

Refer to the Section 6 of the RFM and applicable RFMS

5. Illustrated Parts Catalogue
‘AW169 Illustrated Tool and Equipment Publication’
Doc. No. 69-A-ITEP-00-X
‘AW169 Illustrated Part Data’
Doc. No. 69-A-IPD-00-X

6. Service Letters and Service Bulletins
As published by AgustaWestland, Finmeccanica or Leonardo

7. Required equipment
As per compliance with certification basis and included in Type Design Definition standard.
V. Notes

1. Manufacturer’s eligible serial numbers:
   69005, and subsequent

2. Manufacturer:
   AgustaWestland S.p.A. in Italy(*)
   (*) Effective on 1 January 2016, AgustaWestland S.p.A. ownership was transferred to Finmeccanica S.p.A.;
   Effective on 28 July 2016, Finmeccanica S.p.A. name was changed into Leonardo S.p.A.

3. NVG Operations:
   Night Vision Goggle Operations are permitted according to RFM 169F0290X001 Supplement No. 16. The aircraft configuration involving internal/external emitting/reflecting equipment approved for use with NVG is described in the Report N. 169F3360A001 ‘AW169 NVG Compatibility Reference Handbook’.
   Subsequent modifications and deviations to the NVG helicopter configuration shall be managed in accordance with document 169F3360E001 ‘AW169 Helicopter NVG Policy’.

4. Installation of the TSS-4100 system P/N 6F0630A03113, has been demonstrated compliant with Certification Specifications for Airborne Communications Navigation and Surveillance, CS-ACNS initial issue, dated 17 December 2013
SECTION 2: OPERATIONAL SUITABILITY DATA (OSD)


OSD Elements

1. MMEL
   AW169 Master Minimum Equipment List - MMEL, Doc.169F0270Q003, issue A dated 16 July 2015, EASA approved on 21 July 2015, or later EASA approved revisions

2. Flight Crew Data

3. SIM Data
   For Type Certificate Holder:
   AW169 FTD Validation Data Roadmap doc. THSS-169F1920U014, issue B, dated 7 May 2015, EASA approved on 19 January 2016, or later EASA approved revisions.
   AW169 FFS Validation Data Roadmap doc. 169F1920U01, issue A, dated 19 May 2016, EASA approved on 13 December 2016, or later EASA approved revisions.
   AW169 FFS Level D Flight Test Results Report doc. 169F1920N001, issue A, dated 25 May 2016, EASA approved on 13 December 2016, or later EASA approved revisions.

4. Maintenance Certifying Staff Data
   reserved
SECTION: ADMINISTRATIVE

I. Acronyms and Abbreviations

<table>
<thead>
<tr>
<th>AEO</th>
<th>All Engines Operative</th>
<th>NVG</th>
<th>Night Vision Goggle</th>
</tr>
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<tbody>
<tr>
<td>Amdt.</td>
<td>Amendment</td>
<td>OAT</td>
<td>Outside Air Temperature</td>
</tr>
<tr>
<td>AW</td>
<td>AgustaWestland</td>
<td>OEI</td>
<td>One Engine Inoperative</td>
</tr>
<tr>
<td>B.L.</td>
<td>Butt Line</td>
<td>OSD</td>
<td>Operational Suitability Data</td>
</tr>
<tr>
<td>C.G.</td>
<td>Centre of Gravity</td>
<td>P/N</td>
<td>Part number</td>
</tr>
<tr>
<td>CRI</td>
<td>Certification Review Item</td>
<td>PA</td>
<td>Pressure altitude</td>
</tr>
<tr>
<td>CS</td>
<td>Certification Specification</td>
<td>RFM</td>
<td>Rotorcraft Flight Manual</td>
</tr>
<tr>
<td>DA</td>
<td>Density altitude</td>
<td>RFMS</td>
<td>Rotorcraft Flight Manual Supplement</td>
</tr>
<tr>
<td>Doc.</td>
<td>Document</td>
<td>RH</td>
<td>Right Hand</td>
</tr>
<tr>
<td>HIRF</td>
<td>High Intensity Radiated Fields</td>
<td>s/n</td>
<td>Serial number</td>
</tr>
<tr>
<td>IFR</td>
<td>Instrument Flight Rules</td>
<td>STA</td>
<td>Station</td>
</tr>
<tr>
<td>ISA</td>
<td>International Standard Atmosphere</td>
<td>TCCA</td>
<td>Transport Canada Civil Aviation</td>
</tr>
<tr>
<td>JAA</td>
<td>Joint Aviation Authorities</td>
<td>TCDSN</td>
<td>Type Certificate Data Sheet Noise</td>
</tr>
<tr>
<td>LH</td>
<td>Left Hand</td>
<td>TOP</td>
<td>Take-off Power</td>
</tr>
<tr>
<td>MLG</td>
<td>Main Landing Gear</td>
<td>VFR</td>
<td>Visual Flight Rules</td>
</tr>
<tr>
<td>NLG</td>
<td>Nose Landing Gear</td>
<td>VNE</td>
<td>Velocity Never Exceed</td>
</tr>
<tr>
<td>No.</td>
<td>Number</td>
<td></td>
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II. Type Certificate Holder Record

<table>
<thead>
<tr>
<th>Type Certificate Holder</th>
<th>Period</th>
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<tbody>
<tr>
<td>AgustaWestland S.p.A</td>
<td>from 15 July 2015 until 31 December 2015</td>
</tr>
<tr>
<td>Piazza Monte Grappa, 4, 00195 Roma, Italy</td>
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</tr>
<tr>
<td>Finmeccanica S.p.A.</td>
<td>From 1 January 2016 until 14 July 2016</td>
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<tr>
<td>Helicopter Division, Piazza Monte Grappa, 4, 00195 Roma, Italy</td>
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<tr>
<td>Leonardo S.p.A.</td>
<td>since 15 July 2016</td>
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<tr>
<td>Helicopters, Piazza Monte Grappa, 4, 00195 Roma, Italy</td>
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III. Change Record

<table>
<thead>
<tr>
<th>Issue</th>
<th>Date</th>
<th>Changes</th>
<th>TC issue</th>
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<tbody>
<tr>
<td>Issue 2</td>
<td>21 Jul 2015</td>
<td>AW legal office moved to Rome; OSD approvals for MMEL and FCD</td>
<td>- - -</td>
</tr>
<tr>
<td>Issue 3</td>
<td>13 Jan 2016</td>
<td>TCH company name changed and ownership transferred to Finmeccanica S.p.A.</td>
<td>Re-issued 13 January 2016</td>
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<tr>
<td>Issue 4</td>
<td>19 Jan 2016</td>
<td>OSD SIM added</td>
<td>- - -</td>
</tr>
<tr>
<td>Issue 6</td>
<td>18 Jan 2017</td>
<td>Introduction of ESF to CS 29.807 (d)(2) – “Ditching Emergency Exits for Passengers”; Take-off and landing</td>
<td>- - -</td>
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<tr>
<td>Issue</td>
<td>Date</td>
<td>Changes</td>
<td>TC issue</td>
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<td>altitude envelope extended; Introduction of Kit increased Gross Weight 4 800 kg; OSD SIM extended to FFS level D.</td>
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<tr>
<td>Issue 7</td>
<td>4 Oct 2017</td>
<td>Certification Basis: references to CRI are removed; Environmental Protection Requirements are condensed and make direct reference to TCDSN for Noise Requirements; maximum take-off and landing altitude changed for gross mass above 4 600 kg; introduction of clinometer as admissible levelling means; other minor corrections included.</td>
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<tr>
<td>Issue 8</td>
<td>30 Jan 2018</td>
<td>Introduction of Special Condition 'Non Rechargeable Lithium Battery Installation'; introduction of China No. 3 Jet Fuel</td>
<td></td>
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<tr>
<td>Issue 9</td>
<td>19 Dec 2018</td>
<td>II.3: Special Condition Extended TOP Duration added; II.7.: CS 29.1465 Amdt. 5 added; III.5.3: 'Extended TOP 30 min' added.</td>
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<tr>
<td>Issue 10</td>
<td>22 Dec 2020</td>
<td>II.3.,6.: SC/ESF references updated; II.6.: ESF ‘Never Exceed Speed – Power OFF’ introduced II.7.: Elect to comply for 50 m hoist kit P/N 6F2591F00211; III.8.: Vne added (see Core Avionics SW); III.9.: IAS added (see Core Avionics SW); IV.2.: AMPI initial release clarified; III.4.1., IV.1. V.3.: Correction of typos.</td>
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<td>Issue 11</td>
<td>21 Jul 2021</td>
<td>SECTION 1: II.2-II.7.: adapted to TCDS format policy; SECTION 2: OSD i.1-i.5.: moved to SECTION 1, II.7.; SIM Data for STC 10076972 added to 3.</td>
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<tr>
<td>Issue 12</td>
<td>2 Nov 2021</td>
<td>STC 10076972 removed, for its certification related information please refer to ‘List of EASA STC’; SECTION 1: II.7.3 and SECTION 2, 3.: reference removed</td>
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