



# European Aviation Safety Agency

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**EASA**

**TYPE-CERTIFICATE  
DATA SHEET**

**EASA.A. 586**

**F260**

Type Certificate Holder:

Alenia Aermacchi S.p.A.  
Via Paolo Foresio, 1  
21040 Venegono Superiore (VA)  
Italy

For variants:      F260  
                          F260B  
                          F260C  
                          F260D  
                          F260E  
                          F260F  
                          SF260TP

Issue 01, 21 December 2011

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## F260

### I. General

1. **Data Sheet No:** EASA.A. 586

2. **Type / Variant or Model**

- (a) **Type:** F260
- (b) **Variant or Model:** F260

3. **Airworthiness Category:** Acrobatic & Utility Category

4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*

5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*

6. **National Certification Date:** 16 March 1966 (ENAC C.O. A132)

7. **ENAC Application Date:** 19 June 1963 (ENAC C.O. A132)

8. **ENAC Recommendation Date:** N.A.

9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 19 June 1963

2. **ENAC Certification Date:** 16 March 1966

3. **ENAC Type Certificate Data Sheet No** A 132 Rev. 15

4. **ENAC Certification Basis:** Civil Air regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8.

5. **Airworthiness Requirements:** Civil Air regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8.

6. **Special Conditions:** N.A.

7. **Reversion and Exemptions:** N.A.

8. **Equivalent Safety Findings:** N.A.

9. **Environmental Standards including Noise:** N.A.

### III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel.

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.  
Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM

4. **Dimensions:**

- Span: 8,350 m
- Length: 7,100 m
- Height: 2,410 m

5. **Engines:** Lycoming O-540-E4A5  
State of Design Engine TCDS No. FAA n°E-295  
EASA Engine TCDS No: N.A

5.1 Installed Engine Limits: For all operations, 2700 rpm (260 hp).

6. **Propeller/s:** Hartzell HC-C2YK-1B/8467-8R  
State of Design Propeller TCDS No. FAA P-920  
EASA Propeller TCDS No: N.A

6.1 Propeller Limits: Diameter: Max. 1,930 m (76 in.), Minimum allowable for repairs 1,905 m (75 in.)  
Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15°30'

**7. Fluids (Fuel/Oil/Additives):**

7.1 Fuel Aviation Grade fuel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N°1070.

7.2 Oil Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N°1014.

**8. Fluid capacities (See Note 2):**

8.1 Fuel Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total Usable fuel 235 lt (62 US Gal)

8.2 Oil 11,4 lt (3 US Gal) at 0,800 m  
Usable 8,75 lt (9,25 qt.)

**9. Airspeed limits (See Note 4):**

– Never Exceed Speed VNE: 236 Kts (272 mph ) IAS  
For detailed information see approved Aircraft Flight Manual.

10. **Maximum Operating Altitude and Temperature:** N.A.

**11. Operating Limitations:**

11.1 General: In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

11.2 Additional limitations for take-off and landing: N.A.

12. **Maximum Certified Weights (See Note 4):** Maximum gross weight 1102 Kg (2430 lbs) Utility  
1000 Kg (2205 lbs) Acrobatic

13. **Centre of Gravity Range:** See Flight Manual for CG envelope.

14. **Datum:** Reference Datum: 1,50m forward of firewall.

15. **Levelling Means:** Two screws on the left side of fuselage.

16. **Minimum Flight Crew:** 1 pilot seated in the RH front seat.

17. **Maximum Passenger Seating Capacity (See Note 4):** 2 (two passengers )(1 at +2,550m 1 at +3,300m) – Utility Category,  
1 (one passenger) (1 at +2,550m) – Acrobatic Category, (See Note a pag. 24).

18. **Passenger Emergency Exit:** N.A. (Optional emergency canopy release)

**19. Maximum Baggage/Cargo Loads:**

| Location             | Max Load/Loading |
|----------------------|------------------|
| Baggage compartments | 41Kg at +3,800m. |

20. **Auxiliary Power Unit (APU):** N.A.

21. **Life-limited parts:** Refer Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1  
RAI Approved No 251.669/T dated February 7, 1989.

**IV. Operating and Service Instructions**

1. **Flight Manual, Document No(See Note 4):** F.260 and F.260 B RAI-Approved Flight Manual 65.665/T dated April 18, 1968, Revision 1 through 7 and subsequent approved revisions.

2. **Maintenance Manual, Document No:** Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989.  
F260-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
4. **Required Equipment:** Refer to equipment list in Flight Manual.

**V. Notes**

1. Applicable Serial Number: Aviamilano NC 502 and 503  
SIAI Marchetti from S/N 1-01 to 2-75.
2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:  
Unusable fuel: Wing tanks 4 lbs at +105,5 in.  
Tip tanks 8,6 lbs at +95,3 in.  
Trapped engine oil: 5,07 lbs at +31,5 in.
3. Aircraft Model F.260 with Serial Numbers 2-54 and up incorporate wings modified outline P/N 260-01-76, as model F.260B.
4. For Aviamilano NC 502 and 503, approved on March 16, 1966, the above mentioned limitations shall be replaced with the following ones:  
Air Speed:  
– Never Exceed Speed VNE: 220 Kts (252 mph)  
Maximum weight: 850 Kg Acrobatic Category  
Number of seats: 1 at +100 in. – Acrobatic Category  
Minimum Equipment: See Flight Manual  
Flight Manual: RAI-Approved Flight Manual 49.785/T dated March 15, 1966.

As of 1969, the certificate holder changed its name from "Aviamilano – Costruzioni Aeronautiche, via M. Melloni 70, Milano" to "SIAI-Marchetti S.p.A. – Sesto Calende (VA)".

## F260B

### I. General

1. **Data Sheet No:** EASA.A. 586
2. **Type / Variant or Model**
  - (a) **Type:** F260
  - (b) **Variant or Model:** F260B
3. **Airworthiness Category:** Acrobatic & Utility Category
4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
6. **National Certification Date:** 24.March.1971 (ENAC C.O. A132)
7. **ENAC Application Date:** 05.February.1971 (ENAC C.O. A132)
8. **ENAC Recommendation Date:** N.A.
9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 05.February.1971
2. **ENAC Certification Date:** 24.March.1971
3. **ENAC Type Certificate Data Sheet No** A 132 rev. 15
4. **ENAC Certification Basis:** Civil Air regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8.
5. **Airworthiness Requirements:** Civil Air regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8.
6. **Special Conditions:** N.A.
7. **Reversion and Exemptions:** N.A.
8. **Equivalent Safety Findings:** N.A.
9. **Environmental Standards including Noise:** N.A.

### III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.
2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260 except for vertical empennage of increased area and airfoil sections of wing leading edge).
3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.  
Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM
4. **Dimensions:**
  - Span: 8,350 m
  - Length: 7,100 m
  - Height: 2,410 m
5. **Engines:** Lycoming O-540-E4A5  
State of Design Engine TCDS No. FAA n°E-295  
EASA Engine TCDS No: N.A
- 5.1 **Installed Engine Limits:** For all operations, 2700 rpm (260 hp).

- 6. Propeller/s:** Hartzell HC-C2YK-1B/8467-8R  
State of Design Propeller TCDS No. FAA P-920  
EASA Propeller TCDS No: N.A
- 6.1 Propeller Limits: Diameter: Max. 1,930 m (76 in.), Minimum allowable for repairs 1,905 m (75 in.)  
Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15°30'
- 7. Fluids (Fuel/Oil/Additives):**
- 7.1 Fuel Aviation Grade fuel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N°1070.
- 7.2 Oil Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N°1014.
- 8. Fluid capacities (See Note 2):**
- 8.1 Fuel Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total Usable fuel 235 lt (62 US Gal)
- 8.2 Oil 11,4 lt (3 US Gal) at 0,800 m  
Usable 8,8 lt (9,25 qt.)
- 9. Airspeed limits (See Note 4):**  
– Never Exceed Speed VNE: 236 Kts (272 mph ) IAS  
For detailed information see approved Aircraft Flight Manual.
- 10. Maximum Operating Altitude and Temperature:** N.A.
- 11. Operating Limitations:**
- 11.1 General: In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.
- 11.2 Additional limitations for take-off and landing: N.A.
- 12. Maximum Certified Weights (See Note 4):** Maximum gross weight 1102 Kg (2430 lbs) Utility  
1000 Kg (2205 lbs) Acrobatic
- 13. Centre of Gravity Range:** See Flight Manual for CG envelope.
- 14. Datum:** Reference Datum: 1,50m forward of firewall.
- 15. Levelling Means:** Two screws on the left side of fuselage.
- 16. Minimum Flight Crew:** 1 pilot seated in the RH front seat.
- 17 Maximum Passenger Seating Capacity :** 2 (two passengers ) (1 at +2,550m 1 at +3,300m) – Utility Category,  
1 (one passenger) (1 at +2,550m) – Acrobatic Category, (See Note a pag. 24).
- 18. Passenger Emergency Exit:** N.A. (Optional emergency canopy release)
- 19. Maximum Baggage/Cargo Loads:**
- | Location             | Max Load/Loading |
|----------------------|------------------|
| Baggage compartments | 41Kg at +3,800m. |
- 20. Auxiliary Power Unit (APU):** N.A.
- 21. Life-limited parts:** : Refer Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1  
RAI Approved No 251.669/T dated February 7, 1989.

**IV. Operating and Service Instructions**

1. **Flight Manual, Document No:** F.260 and F.260 B RAI-Approved Flight Manual 65.665/T dated April 18, 1968, Revision 1 through 7 and subsequent approved revisions.
2. **Maintenance Manual, Document No:** Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989.  
F260-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
4. **Required Equipment:** Refer to equipment list in Flight Manual.

**V. Notes**

1. Applicable Serial Number: SIAI Marchetti from S/N 3-76 and Subsequent.
2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:

|                     |                                |
|---------------------|--------------------------------|
| Unusable fuel:      | Wing tanks 4 lbs at +105,5 in. |
|                     | Tip tanks 8,6 lbs at +95,3 in. |
| Trapped engine oil: | 5,07 lbs at +31,5 in.          |

## F260C

### I. General

1. **Data Sheet No:** EASA.A. 586
2. **Type / Variant or Model**
  - (a) **Type:** F260
  - (b) **Variant or Model:** F260C
3. **Airworthiness Category:** Acrobatic & Utility Category
4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
6. **National Certification Date:** 23.October.1976 (ENAC C.O. A132)
7. **ENAC Application Date:** 09.February.1976 (ENAC C.O. A132)
8. **ENAC Recommendation Date:** N.A.
9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 09.February.1976
2. **ENAC Certification Date:** 23.October.1976
3. **ENAC Type Certificate Data Sheet No** A 132 rev. 15
4. **ENAC Certification Basis:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendments 23-7 for the paragraph 23-729, 23-735; EASA CS Part 23 Amendment 1, dated February 12, 2009, for para. 23.1353 (h).
5. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendments 23-7 for the paragraph 23-729, 23-735; EASA CS Part 23 Amendment 1, dated February 12, 2009, for para. 23.1353 (h).
6. **Special Conditions:** N.A.
7. **Reversion and Exemptions:** N.A.
8. **Equivalent Safety Findings:** N.A.
9. **Environmental Standards including Noise:** FAR 36 Appendix F

### III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.
2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260B except for lowered seats, ailerons with servotab, battery relocated forward and radio equipment).
3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release. Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM
4. **Dimensions:**
  - Span: 8,350 m
  - Length: 7,100 m
  - Height: 2,410 m

5. **Engines:** Lycoming O-540-E4A5  
State of Design Engine TCDS No. FAA n°E-295  
EASA Engine TCDS No: N.A
- Lycoming AEIO-540-D4A5  
State of Design Engine TCDS No. FAA n°1E4  
EASA Engine TCDS No: N.A

5.1 Installed Engine Limits: For all operations, 2700 rpm (260 hp).

6. **Propeller/s:** Hartzell HC-C2YK-1B / 8467-8R or  
Hartzell HC-C2YK-4F / FC8477-8R or  
Hartzell HC-C2YK-1BF / F8477-8R  
State of Design Propeller TCDS No. FAA P-920 for all propellers  
EASA Propeller TCDS No: N.A

6.1 Propeller Limits: Diameter: Max. 1,930 m (76 in.), No reduction permitted.  
Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15°30'

7. **Fluids (Fuel/Oil/Additives):**

7.1 Fuel Aviation Grade dual minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N°1070.

7.2 Oil Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N°1014.

8. **Fluid capacities** (See Note 2):

8.1 Fuel Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total Usable fuel 235 lt (62 US Gal) with O-540-E4A5  
Total Usable fuel 220,6 lt (58,3 US Gal) with AEIO-540-D4A5

8.2 Oil 11,4 lt (3 US Gal) at 0,800 m  
Usable 8,75 lt (9,25 qt.) with O-540-E4A5  
Usable 5,7 lt (6 qt.) with AEIO-540-D4A5

9. **Airspeed limits :**

- Never Exceed Speed VNE: 236 Kts (272 mph ) IAS  
For detailed information see approved Aircraft Flight Manual.

10. **Maximum Operating Altitude and Temperature:** N.A.

11. **Operating Limitations:**

11.1 General: In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

11.2 Additional limitations for take-off and landing: N.A.

12. **Maximum Certified Weights :** Maximum gross weight 1102 Kg (2430 lbs) Utility  
1000 Kg (2205 lbs) Acrobatic

13. **Centre of Gravity Range:** See Flight Manual for CG envelope.

14. **Datum:** Reference Datum: 1,50m forward of firewall.

15. **Levelling Means:** Two screws on the left side of fuselage.

16. **Minimum Flight Crew:** 1 pilot seated in the RH front seat.

17. **Maximum Passenger Seating Capacity:** 2 (two passengers )(1 at +2,550m 1 at +3,300m) – Utility Category,  
1 (one passenger) (1 at +2,550m) – Acrobatic Category, (See Note a pag. 24).

18. **Passenger Emergency Exit:** Emergency canopy release

19. **Maximum Baggage/Cargo Loads:**

| Location             | Max Load/Loading |
|----------------------|------------------|
| Baggage compartments | 41Kg at +3,800m. |

20. **Auxiliary Power Unit (APU):** N.A.
21. **Life-limited parts:** : Refer Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1  
RAI Approved No 251.669/T dated February 7, 1989.

**IV. Operating and Service Instructions**

1. **Flight Manual, Document No:** F.260C RAI-Approved Airplane Flight Manual 135.212/T dated October 20, 1976 and subsequent approved revisions.  
F.260C with Lycoming AEIO-540-D4A5 engine RAI-Approved Airplane Flight Manual 181.907/T dated May 11, 1983 and subsequent approved revisions.  
F.260C EASA - AFM Approval n°10033166 dated 21-12 -2010.
2. **Maintenance Manual, Document No:** Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989.  
SF260C-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
4. **Required Equipment:** Refer to equipment list in Flight Manual.

**V. Notes**

1. Applicable Serial Number: SIAI Marchetti from S/N 266 and subsequent.
2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:
- |                     |  |
|---------------------|--|
| Unusable fuel:      | Wing tanks 4 lbs at +105,5 in.<br>Wing tanks 26,9 lbs at +105,5 in. with AEIO-540-D4A5 engine only<br>Tip tanks 8,6 lbs at +95,3 in. |
| Trapped engine oil: | 5,07 lbs at +31,5 in.  |

## F260D

### I. General

1. **Data Sheet No:** EASA.A. 586
2. **Type / Variant or Model**
  - (a) **Type:** F260
  - (b) **Variant or Model:** F260D
3. **Airworthiness Category:** Acrobatic & Utility Category
4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
6. **National Certification Date:** 14.December.1985 (ENAC C.O. A132)
7. **ENAC Application Date:** 31.July.1980 (ENAC C.O. A132)
8. **ENAC Recommendation Date:** N.A.
9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 31.July.1980
2. **ENAC Certification Date:** 14.December.1985
3. **ENAC Type Certificate Data Sheet No** A 132 rev. 15
4. **ENAC Certification Basis:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendments 23-7 for the paragraph 23-729, 23-735; EASA CS Part 23 Amendment 1, dated February 12, 2009, for parag. 23.1353 (h).
5. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendments 23-7 for the paragraph 23-729, 23-735; EASA CS Part 23 Amendment 1, dated February 12, 2009, for parag. 23.1353 (h).
6. **Special Conditions:** N.A.
7. **Reversion and Exemptions:** N.A.
8. **Equivalent Safety Findings:** N.A.
9. **Environmental Standards including Noise:** FAR 36 Appendix F

### III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.
2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260C except for reinforced wing main spar).
3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release. Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM
4. **Dimensions:**
  - Span: 8,350 m
  - Length: 7,193 m
  - Height: 2,680 m

- 5. Engines:**  
Lycoming O-540-E4A5  
State of Design Engine TCDS No. FAA n°E-295  
EASA Engine TCDS No: N.A  
  
Lycoming AEIO-540-D4A5  
State of Design Engine TCDS No. FAA n°1E4  
EASA Engine TCDS No: N.A
- 5.1 Installed Engine Limits:** For all operations, 2700 rpm (260 hp).
- 6. Propeller/s:**  
Hartzell HC-C2YK-1B / 8467-8R or  
Hartzell HC-C2YK-4F / FC8477-8R or  
Hartzell HC-C2YK-1BF / F8477-8R  
State of Design Propeller TCDS No. FAA P-920 for all propellers  
EASA Propeller TCDS No: N.A
- 6.1 Propeller Limits:** Diameter: Max. 1,930 m (76 in.), No reduction permitted.  
Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15°30'
- 7. Fluids (Fuel/Oil/Additives):**
- 7.1 Fuel** Aviation Grade dual minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N°1070.
- 7.2 Oil** Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N°1014.
- 8. Fluid capacities (See Note 2):**
- 8.1 Fuel** Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total Usable fuel 235 lt (62 US Gal) with O-540-E4A5  
Total Usable fuel 220,6 lt (58,3 US Gal) with AEIO-540-D4A5
- 8.2 Oil** 11,4 lt (3 US Gal) at 0,800 m  
Usable 8,75 lt (9,25 qt.) with O-540-E4A5  
Usable 5,7 lt (6 qt.) with AEIO-540-D4A5
- 9. Airspeed limits :**  
– Never Exceed Speed VNE: 236 Kts (272 mph ) IAS  
For detailed information see approved
- 10. Maximum Operating Altitude and Temperature:** N.A.
- 11. Operating Limitations:**
- 11.1 General:** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.
- 11.2 Additional limitations for take-off and landing:** N.A.
- 12. Maximum Certified Weights: :** Maximum gross weight 1100 Kg (2425 lbs) Utility & Acrobatic  
–
- 13. Centre of Gravity Range:** See Flight Manual for CG envelope.
- 14. Datum:** Reference Datum: 1,50m forward of firewall.
- 15. Levelling Means:** Two screws on the left side of fuselage.
- 16. Minimum Flight Crew:** 1 pilot seated in the RH front seat.
- 17. Maximum Passenger Seating Capacity: :** 2 (two passengers )(1 at +2,550m 1 at +3,300m) – Utility Category,  
1 (one passenger) (1 at +2,550m) – Acrobatic Category, (See Note a pag. 24).
- 18. Passenger Emergency Exit:** Emergency canopy release
- 19. Maximum Baggage/Cargo Loads:**

Location

Max Load/Loading

Baggage compartments

41Kg at +3,800m.

20. **Auxiliary Power Unit (APU):**

N.A.

21. **Life-limited parts:** : Refer Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1  
RAI Approved No 251.669/T dated February 7, 1989.

#### **IV. Operating and Service Instructions**

1. **Flight Manual, Document No:** F.260D RAI-Approved Airplane Flight Manual 215.709/T dated December 14, 1985 and subsequent approved revisions.  
F.260D with Lycoming AEIO-540-D4A5 engine RAI-Approved Airplane Flight Manual 215.709/T dated December 14, 1985 and subsequent approved revisions.  
F.260D EASA - AFM Approval n°10033166 dated 21-12 -2010
2. **Maintenance Manual, Document No:** Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989.  
F260D/SF260D-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
4. **Required Equipment:** Refer to equipment list in Flight Manual.

#### **V. Notes**

1. Applicable Serial Number: SIAI Marchetti from S/N 563, 566, 568, 735, 739 to S/N 1999.
2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:  
Unusable fuel: Wing tanks 4 lbs at +105,5 in.  
Wing tanks 26,9 lbs at +105,5 in. with AEIO-540-D4A5 engine only  
Tip tanks 8,6 lbs at +95,3 in.  
Trapped engine oil: 5,07 lbs at +31,5 in.

## F260E

### I. General

1. **Data Sheet No:** EASA.A. 586
2. **Type / Variant or Model**
  - (a) **Type:** F260
  - (b) **Variant or Model:** F260E
3. **Airworthiness Category:** Acrobatic & Utility Category
4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
6. **National Certification Date:** 21.January.1992 (ENAC C.O. A132)
7. **ENAC Application Date:** 17.May.1991 (ENAC C.O. A132)
8. **ENAC Recommendation Date:** N.A.
9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 17.May.1991
2. **ENAC Certification Date:** 21.January.1992
3. **ENAC Type Certificate Data Sheet No** A 132 rev. 15
4. **ENAC Certification Basis:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-961, 23-993, 23-1353 and including Amendment 23-7 for the paragraph 23-735, 23-955; Amendment 23-13 for the paragraph 23-1589; Amendment 23-18 for the paragraph 23-943, 23-959; Amendment 23-23 for the paragraph 23-345; Amendment 23-26 for the paragraph 23-729, 23-991; Amendment 23-29 for the paragraph 23-994, 23-995; Amendment 23-34 for the paragraph 23-333, 23-863, 23-1581, 23-1583, 23-1585, 23-1587; Amendment 23-36 for the paragraph 23-2; Amendment 23-38 for the paragraph 23-572; Amendment 23-40 for the paragraph 23-951; Amendment 23-41 for the paragraph 23-1309; Amendment 23-42 for the paragraph 23-221, 23-341, 23-425, 23-443; EASA CS Part 23 Amendment 1, dated February 12, 2009, for parag. 23.1353 (h).
5. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-961, 23-993, 23-1353 and including Amendment 23-7 for the paragraph 23-735, 23-955; Amendment 23-13 for the paragraph 23-1589; Amendment 23-18 for the paragraph 23-943, 23-959; Amendment 23-23 for the paragraph 23-345; Amendment 23-26 for the paragraph 23-729, 23-991; Amendment 23-29 for the paragraph 23-994, 23-995; Amendment 23-34 for the paragraph 23-333, 23-863, 23-1581, 23-1583, 23-1585, 23-1587; Amendment 23-36 for the paragraph 23-2; Amendment 23-38 for the paragraph 23-572; Amendment 23-40 for the paragraph 23-951; Amendment 23-41 for the paragraph 23-1309; Amendment 23-42 for the paragraph 23-221, 23-341, 23-425, 23-443; EASA CS Part 23 Amendment 1, dated February 12, 2009, for parag. 23.1353 (h).
6. **Special Conditions:** N.A.
7. **Reversion and Exemptions:** N.A.
8. **Equivalent Safety Findings:** N.A.
9. **Environmental Standards including Noise:** ICAO Annex 16, chapter 10.

### III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.
2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel.

(Same as Model F260D except for: reinforced wing with aerodynamic modifications for stall speed reduction, fuel system).

**3. Equipment:**

Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.  
Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM

**4. Dimensions:**

- Span: 8,350 m
- Length: 7,193 m
- Height: 2,680 m

**5. Engines:**

Lycoming IO-540-D4A5 or  
Lycoming AEIO-540-D4A5  
State of Design Engine TCDS No. FAA n°1E4  
EASA Engine TCDS No: N.A

**5.1 Installed Engine Limits:**

For all operations, 2700 rpm (260 hp).

**6. Propeller/s:**

Hartzell HC-C2YK-1BF / F8477-8R or  
Hartzell HC-C2YK-4F / FC8477-8R  
State of Design Propeller TCDS No. FAA P-920 for both propellers  
EASA Propeller TCDS No: N.A

**6.1 Propeller Limits:**

Diameter: Max. 1,930 m (76 in.), No reduction permitted.  
Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15°30'

**7. Fluids (Fuel/Oil/Additives):**

**7.1 Fuel**

Aviation Grade fuel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N°1070.

**7.2 Oil**

Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N°1014.

**8. Fluid capacities (See Note 2):**

**8.1 Fuel**

Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total usable fuel: 228,1 lt (60,26 US Gal) in:  
1 wing tank (left) of 41 lt (10,85 US Gal) at 2.420 m  
1 wing tank (right) of 48,5 lt (12,81 US Gal) at 2.420 m  
2 tip tank 69,3 lt (18,3 US Gal) each at 2.680 m

**8.2 Oil**

11,4 lt (3 US Gal) at 0,800 m  
Usable 8,8 lt (9,25 qt.) with IO-540-D4A5  
Usable 5,7 lt (6 qt.) with AEIO-540-D4A5

**9. Airspeed limits:**

- Never Exceed Speed VNE: 236 Kts (272 mph ) IAS  
For detailed information see approved

**10. Maximum Operating Altitude and Temperature:**

N.A.

**11. Operating Limitations:**

**11.1 General:**

In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

**11.2 Additional limitations for take-off and landing:** N.A.

**12. Maximum Certified Weights:** : Maximum gross weight 1200 Kg (2645 lbs) Utility & Acrobatic

**13. Centre of Gravity Range:** See Flight Manual for CG envelope.

**14. Datum:**

Reference Datum: 1,50m forward of firewall..

**15. Levelling Means:**

Two screws on the left side of fuselage.

**16. Minimum Flight Crew:**

1 pilot seated in the RH front seat.

17. **Maximum Passenger Seating Capacity:** : 2 (two passengers )(1 at +2,550m 1 at +3,300m) – Utility Category,  
1 (one passenger) (1 at +2,550m) – Acrobatic Category, (See Note a pag. 24).

18. **Passenger Emergency Exit:** Emergency canopy release

19. **Maximum Baggage/Cargo Loads:**

| Location             | Max Load/Loading |
|----------------------|------------------|
| Baggage compartments | 41Kg at +3,800m. |
| Cabin compartment    | N.A.             |

20. **Auxiliary Power Unit (APU):** N.A.

21. **Life-limited parts:** F.260E RAI-Approved Airplane Maintenance Manual – Chap. 4 and Chap. 5 97/2262/MAE.

#### **IV. Operating and Service Instructions**

1. **Flight Manual, Document No:** F.260E RAI-Approved Airplane Flight Manual 281.522/SCMA dated May 29, 1992 and subsequent approved revisions.  
F.260E EASA - AFM Approval n°10033166 dated 21-12-2010
2. **Maintenance Manual, Document No:** F.260E RAI-Approved Airplane Maintenance Manual – Chap. 4 and Chap. 5 97/2262/MAE.  
F260D/SF260D-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
4. **Required Equipment:** Refer to equipment list in Flight Manual.

#### **V. Notes**

1. Applicable Serial Number: Agusta S/N 784  
Aermacchi S/N 2001 to S/N 2999.
2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:  
Unusable fuel: Wing tanks 15 lbs at +105,5 in.  
Tip tanks 8,6 lbs at +95,3 in.  
Trapped engine oil: 5,07 lbs at +31,5 in.

## F260F

### I. General

1. **Data Sheet No:** EASA.A. 586
2. **Type / Variant or Model**
  - (a) **Type:** F260
  - (b) **Variant or Model:** F260F
3. **Airworthiness Category:** Acrobatic & Utility Category
4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
6. **National Certification Date:** 21.Jenuary.1992 (ENAC C.O. A132)
7. **ENAC Application Date:** 17.May.1991 (ENAC C.O. A132)
8. **ENAC Recommendation Date:** N.A.
9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 17.May.1991
2. **ENAC Certification Date:** 21.Jenuary.1992
3. **ENAC Type Certificate Data Sheet No** A 132 rev. 15
4. **ENAC Certification Basis:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendment 23-7 for the paragraph 23-735; Amendment 23-13 for the paragraph 23-1589; Amendment 23-23 for the paragraph 23-345; Amendment 23-26 for the paragraph 23-729; Amendment 23-34 for the paragraph 23-333, 23-1581, 23-1583,23-1585, 23-1587; Amendment 23-36 for the paragraph 23-2; Amendment 23-38 for the paragraph 23-572; Amendment 23-42 for the paragraph 23-221, 23-341, 23-425, 23-443; EASA CS Part 23 Amendment 1, dated February 12, 2009, for para. 23.1353 (h).
5. **Airworthiness Requirements:** Civil Air Regulation Part 23, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendment 23-7 for the paragraph 23-735; Amendment 23-13 for the paragraph 23-1589; Amendment 23-23 for the paragraph 23-345; Amendment 23-26 for the paragraph 23-729; Amendment 23-34 for the paragraph 23-333, 23-1581, 23-1583,23-1585, 23-1587; Amendment 23-36 for the paragraph 23-2; Amendment 23-38 for the paragraph 23-572; Amendment 23-42 for the paragraph 23-221, 23-341, 23-425, 23-443; EASA CS Part 23 Amendment 1, dated February 12, 2009, for para. 23.1353 (h).
6. **Special Conditions:** N.A.
7. **Reversion and Exemptions:** N.A.
8. **Equivalent Safety Findings:** N.A.
9. **Environmental Standards including Noise:** FAR Part 36 effective December 1, 1969 including Amendments 36-1 through 36-20, ICAO Annex 16, chapter 10..

### III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.
2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260D except for: reinforced wing with aerodynamic modifications for stall speed reduction).

- 3. Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release. Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM
- 4. Dimensions:**
- Span: 8,350 m
  - Length: 7,193 m
  - Height: 2,680 m
- 5. Engines:** Lycoming O-540-E4A5  
State of Design Engine TCDS No. FAA n°E-295  
EASA Engine TCDS No: N.A
- 5.1 Installed Engine Limits: For all operations, 2700 rpm (260 hp).
- 6. Propeller/s:** Hartzell HC-C2YK-1BF / F8477-8R or  
Hartzell HC-C2YK-4F / FC8477-8R  
State of Design Propeller TCDS No. FAA P-920 for both propellers  
EASA Propeller TCDS No: N.A
- 6.1 Propeller Limits: Diameter: Max. 1,930 m (76 in.), No reduction permitted.  
Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15°30'
- 7. Fluids (Fuel/Oil/Additives):**
- 7.1 Fuel Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N°1070.
- 7.2 Oil Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N°1014.
- 8. Fluid capacities (See Note 2):**
- 8.1 Fuel Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total Usable fuel 235 lt (62 US Gal)
- 8.2 Oil 11,4 lt (3 US Gal) at 0,800 m  
Usable 8,75 lt (9,25 qt.)
- 9. Airspeed limits:**
- Never Exceed Speed VNE: 236 Kts (272 mph ) IAS  
For detailed information see approved
- 10. Maximum Operating Altitude and Temperature:** N.A.
- 11. Operating Limitations:**
- 11.1 General: In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.
- 11.2 Additional limitations for take-off and landing: N.A.
- 12. Maximum Certified Weights:** Maximum gross weight 1200 Kg (2645 lbs) Utility & Acrobatic
- 13. Centre of Gravity Range:** See Flight Manual for CG envelope.
- 14. Datum:** Reference Datum: 1,50m forward of firewall.
- 15. Levelling Means:** Two screws on the left side of fuselage.
- 16. Minimum Flight Crew:** 1 pilot seated in the RH front seat.
- 17. Maximum Passenger Seating Capacity:** : 2 (two passengers )(1 at +2,550m 1 at +3,300m) – Utility Category,  
1 (one passenger) (1 at +2,550m) – Acrobatic Category, (See Note a pag. 24).
- 18. Passenger Emergency Exit:** Emergency canopy release
- 19. Maximum Baggage/Cargo Loads:**

| Location             | Max Load/Loading |
|----------------------|------------------|
| Baggage compartments | 41Kg at +3,800m. |

20. **Auxiliary Power Unit (APU):** N.A.

21. **Life-limited parts:** F.260F RAI-Approved Airplane Maintenance Manual - Chap 4. and Chap. 5 97/2262/MAE.

#### **IV. Operating and Service Instructions**

1. **Flight Manual, Document No:** F.260F RAI-Approved Airplane Flight Manual 281.522/SCMA dated May 29, 1992 and subsequent approved revisions.  
F.260F EASA - AFM Approval n°10033166 dated 21-12-2010
2. **Maintenance Manual, Document No:** F.260F RAI-Approved Airplane Maintenance Manual - Chap 4. and Chap. 5 97/2262/MAE.  
F260D/SF260D-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
4. **Required Equipment:** Refer to equipment list in Flight Manual.

#### **V. Notes**

1. Applicable Serial Number: Aermacchi S/N 3001 to 3999.
2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:

|                     |                                |
|---------------------|--------------------------------|
| Unusable fuel:      | Wing tanks 4 lbs at +105,5 in. |
|                     | Tip tanks 8,6 lbs at +95,3 in. |
| Trapped engine oil: | 5,07 lbs at +31,5 in.          |

## SF260TP

### I. General

1. **Data Sheet No:** EASA.A. 586
2. **Type / Variant or Model**
  - (a) **Type:** F260
  - (b) **Variant or Model:** SF260TP
3. **Airworthiness Category:** Acrobatic
4. **Type Certificate Holder:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
5. **Manufacturer:** Alenia Aermacchi S.p.A. *Via Paolo Foresio, 1 - 21040 Venegono Superiore (VA) - Italy*
6. **National Certification Date:** 29.October.1993 (ENAC C.O. A132)
7. **ENAC Application Date:** 30.July.1993 (ENAC C.O. A132)
8. **ENAC Recommendation Date:** N.A.
9. **EASA Type Certification Date:** N.A.

### II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 30.July.1993
2. **ENAC Certification Date:** 29.October.1993
3. **ENAC Type Certificate Data Sheet No** A 132 rev. 15
4. **ENAC Certification Basis:** Civil Air Regulation Part 23, dated May 15, 1966, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-251, 23-907, 23-1105 and including Amendment 23-7 for paragraph 23-729, 23-735, 23-937, 23-955, 23-1041, 23-1045, 23-1091, 23-1103, 23-1155, 23-1505, 23-1527; Amendment 23-14 for paragraph 23-173, 23-201, 23-572, 23-929, 23-1017, 23-1027; Amendment 23-15 for paragraph 23-951, 23-1013, 23-1015; Amendment 23-16 for paragraph 23-335; Amendment 23-17 for paragraph 23-787, 23-933, 23-977, 23-1111, 23-1165, 23-1303, 23-1309; Amendment 23-18 for paragraph 23-939, 23-943, 23-959, 23-1121, 23-1141, 23-1145, 23-1337; Amendment 23-20 for paragraph 23-1351(e); Amendment 23-21 for paragraph 23-45, 23-49, 23-65, 23-75(d), 23-77, 23-175, 23-1043, 23-1353, 23-1521; Amendment 23-23 for paragraph 23-863, 23-1545, 23-1557, 23-1583; Amendment 23-26 for paragraph 23-253, 23-361, 23-371, 23-991, 23-1305, 23-1529; Amendment 23-28 for paragraph 23-1549; Amendment 23-29 for paragraph 23-901, 23-903, 23-905, 23-995, 23-997, 23-1019, 23-1093, 23-1143, 23-1183, 23-1189; Amendment 23-31 for paragraph 23-629;
5. **Airworthiness Requirements:** Civil Air Regulation Part 23, dated May 15, 1966, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-251, 23-907, 23-1105 and including Amendment 23-7 for paragraph 23-729, 23-735, 23-937, 23-955, 23-1041, 23-1045, 23-1091, 23-1103, 23-1155, 23-1505, 23-1527; Amendment 23-14 for paragraph 23-173, 23-201, 23-572, 23-929, 23-1017, 23-1027; Amendment 23-15 for paragraph 23-951, 23-1013, 23-1015; Amendment 23-16 for paragraph 23-335; Amendment 23-17 for paragraph 23-787, 23-933, 23-977, 23-1111, 23-1165, 23-1303, 23-1309; Amendment 23-18 for paragraph 23-939, 23-943, 23-959, 23-1121, 23-1141, 23-1145, 23-1337; Amendment 23-20 for paragraph 23-1351(e); Amendment 23-21 for paragraph 23-45, 23-49, 23-65, 23-75(d), 23-77, 23-175, 23-1043, 23-1353, 23-1521; Amendment 23-23 for paragraph 23-863, 23-1545, 23-1557, 23-1583; Amendment 23-26 for paragraph 23-253, 23-361, 23-371, 23-991, 23-1305, 23-1529; Amendment 23-28 for paragraph 23-1549; Amendment 23-29 for paragraph 23-901, 23-903, 23-905, 23-995, 23-997, 23-1019, 23-1093, 23-1143, 23-1183, 23-1189; Amendment 23-31 for paragraph 23-629;
6. **Special Conditions:** N.A.
7. **Reversion and Exemptions:** N.A.
8. **Equivalent Safety Findings:** N.A.
9. **Environmental Standards including Noise:** FAR Part 36, App. F (Acoustic Certification).

**III. Technical Characteristics and Operational Limitations**

- 1. Type Design Definition:** Refer to Report N°598-260-059 Rev. / and subsequent approved revisions.
- 2. Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260D except for reinforced wing with aerodynamic modifications for stall speed reduction, and Allison Turboprop Model 250-B17D installation).
- 3. Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.  
Besides are required the following equipment:  
Stall warning , Safe Flight Instruments Cop. 164S  
Refer also to the Equipment list in FM
- 4. Dimensions:**
- Span: 8,350 m
  - Length: 7,400 m
  - Height: 2,410 m
- 5. Engines:** Allison Gas Turbine 250-B17D  
State of Design Engine TCDS No. FAA E10CE  
EASA Engine TCDS No: N.A
- 5.1 Installed Engine Limits:**
- |                                   |               |           |
|-----------------------------------|---------------|-----------|
| TO Power and Mac Cont.            | 320 SHP       |           |
| Propeller speed (Np):             |               |           |
| -TO and Max Cont Power            | 100%          | 2030 rpm. |
| -transient (15 sec.)              | 110%          | 2233 rpm. |
| Turbine Outside Temperature (TOT) |               |           |
| -TO and Max Cont Power            | 810°C         |           |
| -transient (6 sec)                | 843°C         |           |
| -starting (10 sec)                | 810°C – 927°C |           |
| Torque:                           |               |           |
| -TO and Max Cont Power            | 78 PSI        |           |
- 6. Propeller/s:** Hartzell HC-B3TF-7A / T10173-25R  
State of Design Propeller TCDS No. FAA P15EA  
EASA Propeller TCDS No: N.A
- 6.1 Propeller Limits:** Diameter: max 1,930 m (76 in.), No reduction permitted  
Pitch at R = 0,762 m (30 in.),  
Flight Idle: +10° ± 0,5°
- 7. Fluids (Fuel/Oil/Additives):**
- 7.1 Fuel** JP-4 o JP-5, according to MIL-T-5624; Jet A, Jet A1 o Jet B according to ASTM D-1655. Anti-icing additive, according to MIL-I-27686E, must be used in fuel not actually premixed for operation at ambient temperature below 5°C. The maximum concentration is authorized between 0,06% and 0,15%.
- 7.2 Oil** Oil in accordance with:  
MIL-L-23699 above - 40°C (ambient temperature).  
MIL-L-7808G below -40 °C (ambient temperature).
- 8. Fluid capacities (See Note 2):**
- 8.1 Fuel** Total fuel 243 lt (64 US Gal), stored in:  
2 wing tanks of 49,5 lt (13 US Gal) each at 2,420 m  
2 tip tanks of 72 lt (19 US Gal) each at 2,680 m  
Total Usable fuel 220,6 lt (58,2 US Gal)
- 8.2 Oil** 8,7 lt (2,3 US Gal) at 0,800 m  
Usable 6,8 lt (1,8 US Gal)
- 9. Airspeed limits:**
- Maximum Structural Cruising Speed VNO: 229 Kts (264 mph ) up to 12.000 ft IAS  
196 Kts (226 mph ) up to 20.000 ft IAS
- 10. Maximum Operating Altitude and Temperature:** 20000 ft.  
43,3 °C at sea level.
- 11. Operating Limitations:**
- 11.1 General:** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.
- 11.2 Additional limitations for take-off and landing:** N.A.

- 12. **Maximum Certified Weights:** Maximum gross weight 1160 Kg (2560 lbs) Acrobatic
- 13. **Centre of Gravity Range:** See Flight Manual for CG envelope.
- 14. **Datum:** Reference Datum: 1,50m forward of firewall.
- 15. **Levelling Means:** Two screws on the left side of fuselage.
- 16. **Minimum Flight Crew:** 1 pilot seated in the RH front seat.
- 17. **Maximum Passenger Seating Capacity:** 1 (one passenger) (1 at +2,550m)
- 18. **Passenger Emergency Exit:** N.A.
- 19. **Maximum Baggage/Cargo Loads:**

| Location             | Max Load/Loading |
|----------------------|------------------|
| Baggage compartments | 20Kg at +3,300m. |

- 20. **Auxiliary Power Unit (APU):** N.A.
- 21. **Life-limited parts:** Refer to Maintenance Manual

**IV. Operating and Service Instructions**

- 1. **Flight Manual, Document No:** SF.260TP RAI-Approved Airplane Flight Manual 93/3051/MAE dated October 28, 1993 and subsequent approved revisions.
- 2. **Maintenance Manual, Document No:** SF.260TP RAI-Approved Airplane Maintenance Manual Chap 4 and Chap 5 97/2262/MAE.  
SF260TP-2 Airplane Maintenance Manual (Not RAI Approved).
- 3. **Service Letters and Service Bulletins:** As published by Alenia Aermacchi and approved by ENAC (for SB only).
- 4. **Required Equipment:** Refer to equipment list in Flight Manual.

**V. Notes**

- 1. Applicable Serial Number: S/N 661 and S/N 4001 and subsequent.
- 2. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.  
The certificated empty weight and corresponding center of gravity location must include:
  - Unusable fuel: Wing tanks 9,26 lbs at +61,46 in.  
Tip tanks 29,32 lbs at +68,07 in.
  - Trapped engine oil: 8,82 lbs at +34,29 in.

**NOTES :**

- a. For utility category, aircraft F260, F260B, F260C, F260D, rear seat may be occupied by two persons and/or baggage provided:
  - The total weight on rear seat (including baggage) is under 250 lb
  - The rear seat is equipped with two separate safety belts
  - Weight and G position are within limits
  - No baggage is on the baggage compartment
  - No radio equipment is in the radio compartment. (for F260C and F260 D)
- b. The following placards must be displayed as indicated:
  - On Instrument Panel:
  - This airplane must be operated as a Utility or Acrobatic category airplane in compliance with Approved Airplane Flight Manual.
  - All markings and placards on this airplane apply to its operation as a Utility category airplane
  - For Acrobatic category operations, refer to Airplane Flight Manual.
  - All Placards required in the basic Approved Airplane Flight Manual, installed in the appropriate location.
- c. Each individual airplane will be supplied with a placard that specifies the kind of operation such as VFR and IFR, Day or Night, to which the operation of the airplane is limited by the equipment installed.
- d. The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. In addition, the following items of equipment are required: (all items notes as standard equipment in the Manufacturer's approved equipment list).
  - Pre-stall warning indicator, Safe Flight Indicator Corp. 164S.
  - Applicable F260() and SF260TP flight manuals. (See relative paragraphs "Operating and Service Instructions" of this document).
- e. As of November 30, 1989, SIAI Marchetti Spa became a member of Agusta S.p.A. Group.
- f. As of January 1, 1997, AERMACCHI S.p.A. has acquired SIAI Marchetti S.r.L.
- g. As of July 13, 2006, AERMACCHI S.p.A. became Alenia Aermacchi S.p.A.

**ADMINISTRATIVE SECTION**

I. Acronyms

N/A

II. Type Certificate Holder Record

Alenia Aermacchi S.p.A.  
Via Paolo Foresio, 1  
21040 Venegono Superiore (VA)  
Italy

III. Change Record

| Issue | Date              | Changes                      |
|-------|-------------------|------------------------------|
| 01    | 21. December 2011 | Transfer to EASA Type Design |
|       |                   |                              |