

# *European Aviation Safety Agency*

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**EASA**

## **TYPE-CERTIFICATE DATA SHEET**

**Cessna Model 206**

### **Type Certificate Holder:**

#### **Cessna Aircraft Company**

P.O. Box 7704  
Wichita, Kansas 67277  
USA

### **Manufacturer:**

#### **Cessna Aircraft Company**

P.O. Box 7704  
Wichita, Kansas 67277  
USA

For variants: 206H  
T206H

Issue 5: 14 April 2008

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**SECTION 1: GENERAL, Model 206H Type Design**

**A. General**

Data Sheet No.:	EASA IM A 053	Issue:	04	Date:	17 March 2008
1 a) Type:		Model 206H			
b) Variant:		N/A			
2 Airworthiness Category:		FAR-23 Normal Category			
3 Type Certificate Holder:		Cessna Aircraft Company P O. Box 7704 Wichita, Kansas 67277 U S A			
4 Manufacturer:		Cessna Aircraft Company P O. Box 7704 Wichita, Kansas 67277 U S A			
5 JAA Certification Application Date:		N/A			
6 JAA recommendation Date:		N/A			
7 EASA Type Certification Date:		28 September 2003			

**B. Certification Basis**

1. Reference Date for determining the applicable requirements:	FAA Application date 24 February 1998
2. (Reserved)	
3. (Reserved)	
4. Certification Basis:	As defined in FAA TCDS A4CE
5. Airworthiness Requirements:	FAR 23 as defined in FAA TCDS A4CE, and JAR-23, Change 1, plus Special Conditions as defined in Garmin G-1000 EASA CRI A-01, Issue 5, dated 17 March 2008 for the Nav III avionics option
6. Requirements elected to comply:	None
7. EASA Special Conditions:	As defined in CRI A-01 for the Nav III avionics option only.
8. EASA Exemptions:	None
9. EASA Equivalent Safety Findings:	None
10. EASA Environmental Standards:	CS 36 (ICAO Annex 16, Volume I, as applicable )

**C. Technical Characteristics and Operational Limitations**

- 1 Type Design Definition: Master Drawing List, Document No 206-97-001, latest FAA Approved Revision
- 2 Description: Single-engine, all-metal, six-place, high-wing airplane, fixed tricycle landing gear
- 3 Equipment: See Original delivery documents
- 4 Dimensions:
- |           |  |
|-----------|--|
| Span      | 10.9728 m (36.00 ft.)                            |
| Length    | 8.52424 m (27.97 ft.)                            |
| Height    | 2.54635 m (8.35 ft.)                             |
| Wing Area | 16.3045 m <sup>2</sup> (175.50 ft <sup>2</sup> ) |
- 5 Engines: Lycoming IO-540-AC1A5
- The EASA Engine Type Certification standard includes that of FAA TC 1E4, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003, Other standards conforming to TC/TCDS standards Certificated by individual EU member States prior to 28 September 2003 are also acceptable.
- 5.1 Engine Limits: For all operations: 2700 RPM (300 hp)
- For power-plants limits refer to Airplane Flight Manual and Pilot's Operating Handbook, Part No 206HPHUS00 or 206HPHAUS00 (Garmin) or 206HPHBUS00 (GFC-700) or latest revision
- 6 Propellers: a McCauley Constant Speed propeller limits  
McCauley Model: B3D36C432/80VSA-1
- The EASA Propeller Type Certification standard includes that of FAA TC P58GL, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003, Other standards conforming to TC/TCDS standards Certificated by individual EU member States prior to 28 September 2003 are also acceptable
- Maximum Diameter: not over 2 0066 m (79 in.)  
Minimum Diameter: not under 1 9685 m (77 5 in.)  
Number of Blades: 3  
(1) McCauley Governor DC290D1/T37  
(2) Cessna Spinner: 2150151
- 7 Fluids:
- 7 1 Fuel: 100/100LL minimum grade aviation gasoline
- 7 2 Oil: Engine MIL-L-6082 or SAE J1966 Aviation Grade Straight Mineral Oil or MIL-L-22851 or SAE J1899 Aviation Grade Ashless Dispersant Oil Oil conforming to Textron

Lycoming Service Instruction No. 1014, latest revision, must be used after first 50 hours or once oil consumption has stabilized.

- 7.3 Coolant: Not Applicable
8. Fluid capacities:
- 8.1 Fuel: Units 20608001 through 20608173
- |         |                                |
|---------|--------------------------------|
| Total:  | 348 258 liters (92 US Gallons) |
| Usable: | 333 116 liters (88 US Gallons) |
- Units 20608174 and on
- |         |                                |
|---------|--------------------------------|
| Total:  | 348 258 liters (92 US Gallons) |
| Usable: | 329 331 liters (87 US Gallons) |
- 8.2 Oil: Maximum: 10 4099 liters (11 0 qts )  
at 0 32512 m (12 8 in ) forward of datum
- Minimum: 5 67812 liters (6 0 qts ) usable
9. Air Speeds:      Maneuvering                      125 KIAS (123 KCAS)  
                         Maximum Structural Cruising      149 KIAS (147 KCAS)  
                         Never Exceed                              182 KIAS (180 KCAS)  
                         Flaps Extended                            100 KIAS (100 KCAS)
10. Maximum Operating Altitude: With a portable oxygen system, the aircraft is limited to 4785.36 m (15,700 ft MSL) Oxygen must be provided as required by the operating rules.
11. Operational Capability: VFR Day and Night  
IFR Day and Night
12. Maximum Masses: Maximum Ramp: 1639.28 kg (3614 lbs )  
Maximum Takeoff: 1632.98 kg (3600 lbs )  
Maximum Landing: 1632.98 kg (3600 lbs )
13. Centre of Gravity Range:
- Forward Limits:** Limer variation from 1 0795 m (42.5 in ) aft of datum at 1632 93 kg (3600 lbs ) to 0.8382 m (33 0 in .) aft of daum at 1133 98 kg (2500 lbs ); 0 8382 m (33 0 in .) aft of datum at 1133 98 kg (2500 lbs.) or less.
- Aft Limits:** 1 26238 m (49 7 in ) aft of datum at 1632 93 kg (3600 lbs ) or less
14. Datum: Front Face of Firewall (Fuselage Station 0 0)
15. (Reserved)
16. Levelling Means: Left side of Tailcone at 3.85699 m (151 85 in ) and 4.57835 m (180 25 in ) aft of datum
17. Minimum Flight Crew: 1 (Pilot)

18	Maximum Passenger Seating Capacity:	6 [2 at 0.8636 m (34.0 in.) to 1.2192 m (48.0 in.) aft of datum, 2 at 1.7526 (69.0 in.) to 2.0066 m (79.0 in.) aft of datum, 2 at 2.4892 m (98.0 in.) aft of datum]
19	Baggage / Cargo Compartment	81.6466 kg (180 lbs) at 2.7686 m (109 in.) to 3.6830 m (145 in.) aft of datum
20	Wheels and Tires	
	Nose Wheel Tire Size	5.00 x 5
	Main Wheel Tire Size	6.00 x 6
21	Control Surface Movements	
	Wing Flaps:	Down 40° +1°, -2°
	Elevator Tab:	Up 25° +1°, -0°      Down 5° +1°, -0°
	Ailerons:	Up 21° ± 2°              Down 14°30' ± 2°
	Elevator:	Up 21° ± 1°              Down 17° ± 1°
	(Relative to stabilizer)	
	Rudder:	Right: 24° ± 1°      Left: 24° ± 1° (Parallel to 0.00 W L ) Right: 27°13' ± 1°      Left: 27°13' ± 1° (Perpendicular to hinge line)

#### **D. Operating and Service Instructions**

Airplane Flight Manual (AFM):	Document No.206HPHUS00 or 206HPHAUS00 (Garmin) or 206HPHBUS00 (GFC-700), latest approved revision
Airplane Maintenance Manual (AMM) (Including Airworthiness Limitations)	Document No. 206HMM00 or latest revision

#### **E. Notes**

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis, Note 6) must be installed in the airplane for certification.

NOTE 1:      Weight and Balance:

Serial Nos.20608001 through 20608173 :

The certificated basic empty weight and corresponding center of gravity location must include unusable fuel of 10 8862 kg (24 lbs) at 1.2192 m (48 in.) aft of datum, and full oil of 9 344 kg (20.6 lbs) at 0.32513 (12.8 in.) forward of datum

Serial Nos.20608174 and on :

The certificated basic empty weight and corresponding center of gravity location must include unusable fuel of 13 6078 kg (30 lbs) at 1.2192 m (48 in.) aft of datum, and full oil of 9 344 kg (20.6 lbs) at 0.32513 (12.8 in.) forward of datum

NOTE 2:      FAA Approved Airplane Flight Manual (AFM), or Pilot's Operating Handbook and FAA Approved Airplane Flight Manual (POH/AFM): Part number 206HPHUS00 or later FAA approved revisions are applicable to the Model 206H. The Airplane must be operated according to the appropriate AFM or POH/AFM. Required placards are included in the AFM or POH/AFM

FAA Approved Airplane Flight Manual (AFM): Part Number 206HPHAUS-00 (or later FAA approved revisions) are applicable to the Model 206H equipped with Garmin G1000 Integrated Cockpit System. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM.

FAA Approved Airplane Flight Manual (AFM): Part Number 206HPHBUS-00 (or later FAA approved revisions) are applicable to the Model 206H equipped with Garmin G1000 Integrated Cockpit System and Garmin GFC-700 AFCS. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM

NOTE 3: The CHI probe must be installed on Head #3

NOTE 4: Model 206H airplanes, serial numbers 20608060 through 20608091 may differ structurally and are, therefore, not eligible for any weight increases above the approved maximum takeoff weight limit of 3,600 pounds unless compliance with Cessna Document AI206-57-01 (latest revision) has been accomplished and documented with an appropriate logbook entry. Any exceptions must first be coordinated with the Wichita Aircraft Certification Office

NOTE 5: Special Ferry Flight Authorization. Flight Standards District Offices are authorized to issue Special overweight ferry flight authorizations. This airplane is structurally satisfactory for ferry flight if maintained within the following limits: (1) Takeoff weight must not exceed 130% of the maximum weight for Normal Category, and (2) The Never Exceed Airspeed ( $V_{NE}$ ) and Maximum Structural Cruising Speed ( $V_C$ ) must be reduced by 30%; and (3) Forward and aft center of gravity limits may not be exceeded; and (4) Structural load factors of 2.5 g to -1.0 g may not be exceeded. Requirements for any additional oil should be established in accordance with Advisory Circular AC23.1011-1. Increased stall speeds and reduced climb performance should be expected for the increased weights. Flight characteristics and performance at the increased weights have not been evaluated. Flight Permit for operations of overweight aircraft may be found in Advisory Circular AC21-4B

NOTE 6: FAA Certification Basis (Model 206H)

Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-6, except as follows:

FAR 23 423; 23 611; 23.619; 23 623; 23 689; 23 775; 23 871; 23.1323; and 23.1563 as amended by Amendment 23-7. FAR 23 807 and 23 1524 as amended by Amendment 23-10. FAR 23 507; 23 771; 23 853(a),(b) and (c); and 23 1365 as amended by Amendment 23-14. FAR 23 951 as amended by Amendment 23-15. FAR 23 607; 23 675; 23 685; 23 733; 23.787; 23.1309 and 23 1322 as amended by Amendment 23-17. FAR 23 1301 as amended by Amendment 23-20. FAR 23 1353; and 23 1559 as amended by Amendment 23-21. FAR 23 603; 23 605; 23 613; 23 1329 and 23.1545 as amended by Amendment 23-23. FAR 23.441 and 23.1549 as amended by Amendment 23-28. FAR 23 1093 as amended by Amendment 23-29. FAR 23 779 and 23 781 as amended by Amendment 23-33. FAR 23 1; 23 51 and 23 561 as amended by Amendment 23-34. FAR 23 301; 23 331; 23 351; 23 427; 23 677; 23 701; 23 735; and 23 831 as amended by Amendment 23-42. FAR 23 961; 23 1107(b); 23 1143(g); 23 1147(b); 23 1303; 23.1357; 23 1361 and 23 1385 as amended by Amendment 23-43. FAR 23 562(a), 23 562(b)2, 23.562(c)1, 23 562(c)2, 23.562(c)3, and 23 562(c)4 as amended by Amendment 23-44. FAR 23 33; 23 53; 23 305; 23 321; 23 485; 23 621; 23 655 and 23 731 as amended by Amendment 23-45.

FAR 36 dated December 1, 1969, as amended by Amendments 36-1 through 36-21.

Additions for the Garmin G1000 Integrated Cockpit System (ICS) Only:

14 CFR 23 303; 23 307; 23 601; 23 1163(a)(1)(2); 23.1367 and 23.1381 as amended by Amendment 23-N/C  
14 CFR 23.1589 as amended by Amendment 23.13. 14 CFR 23 771(a) as amended by Amendment 23 14. 14 CFR 23 607 and (Electrical System) 23 1309(a)(1)(2), (c) as amended by Amendment 23-17. 14 CFR 23 1301; 23.1327 and 23.1547(e) as amended by Amendment 23-20. 14 CFR 23.1501 and 23 1541(a)(1), (a)(2), (b)(1), (b)(2) AS AMENDED BY Amendment 23-21. 14 CFR 23 603 and 23 605 as amended by Amendment 23-23. 14 CFR 23 1529 as amended by Amendment 23-26. 14 CFR 23 561(e); 23 1523; 23 1581(a)(2); 23 1538(a)(1), (a)(2), (b)(h) and 23.1585(a)(b)(d) as amended by Amendment 23-34. 14 CFR 23 301 as amended by Amendment 23-42. 14 CFR 23 1322; 23.1331 and 23.1357(a)(b)(c)(d) as amended by Amendment 23-43. 14 CFR 23 305; 23 773(a)(1), (a)(2); 23.1525 and 23 1549 as amended by Amendment 23-45. 14 CFR 23 1303(a)(b)(c)(f); 23.1309(a)(1)(i), (a)(1)(ii), (a)(2), (b)(1), (b)(2)(i), (b)(2)(ii), (b)(3), (b)(4)(i), (b)(4)(ii), (b)(4)(iii), (b)(4)(iv), (c)(1), (c)(2)(iii), (c)(3), (d), (e), (f)(1); 23.1311; 23.1321(a)(c)(d)(e); 23 1323(a), (b)(1), (b)(2), (c); 23 1329(g)(h); 23 1351(a)(1), (a)(2)(i), (b)(1)(iii), (b)(2)(3), (c)(4), (d)(1); 23 1353(a)(b)(c)(d)(e); 23.1359(c); 23 1361; 23 1365(a)(b)(d)(e)(f) and 23.1431 (a)(b)(d)(e) as amended by Amendment 23-49. 14 CFR 23 1325(a), (b)(1), (b)(2)(i), (b)(3), (c)(d)(e); 23 1543(b)(c); 23 1545(a), (b)(1), (b)(2), (b)(3), (b)(4); 23 1553; 23 1555(a)(b); 23 1563(a) and 23 1567(a) as amended by Amendment 23 50.

14 CFR 23.777(a)(b); 23.955(a)(2); 23.1337(a)(1), (a)(2), (b)(1), (c) as amended by Amendment 23-51 14 CFR 23.1305(a)(1), (a)(2), (a)(3), (b)(2), (b)(3)(i), (b)(4)(i), (b)(5), (b)(6)(i) as amended by Amendment 23-52 14 CFR 23.901 (a)(b) as amended by Amendment 23-53.

Additions for the Garmin GFC-700 Automatic Flight Control System (AFCS) Only:

14 CFR 23.1335 as amended by Amendment 23-20, 14 CFR 23.1329 (a)(c)(d)(e)(f) as amended by Amendment 23-49

Equivalent Safety Items:

- |                          |                    |  |
|--------------------------|--------------------|--|
| (1) Throttle Control     | FAR § 23.1143(g)   | Number 97-4, FAA letter November 25, 1997                            |
| (2) Mixture Control      | FAR § 23.1147(b)   | Number 97-4, FAA letter November 25, 1997                            |
| (3) Fuel Tank Sump       | FAR § 23.971       | Number ACE-02-03, FAA letter January 3, 2002 (Units 20608174 and on) |
| (4) Anticollision Lights | FAR § 23.1401(d)   | Number ACE-02-02, FAA letter January 3, 2002 (Units 20608174 and on) |
| (5) Aviation White Color | 14CFR § 23.1397(c) | Refer to ACE-07-12, FAA letter 11/29/07                              |

Date of Application for Amended Type Certificate was October 25, 1996  
Type Certificate No. A4CE was amended November 26, 1997

Special Conditions as follows:

No 23-150-SC, (Special Conditions: Cessna Aircraft Company; Cessna Model 206H Airplane; Installation of Electronic Flight Instrument System and the Protection of the System From High Intensity Radiated Fields (HIRF))

**Production Basis (Model 206H)**

Production Certificate No PC-4 issued November 25, 1998 Applies to airplane serial numbers 20608001 and on Cessna is authorized to issue airworthiness certificates under the delegation provisions of Delegation Option Authorization No CE-1 in accordance with Part 21 of the Federal Aviation Regulations

## **SECTION 2: GENERAL, Model T206H Type Design**

### **A. General**

Data Sheet No.:	EASA IM.A 053	Issue:	04	Date:	17 March 2008
1. a) Type:		Model T206H			
b) Variant:		N/A			
2. Airworthiness Category:		FAR-23 Normal Category			
3. Type Certificate Holder:		Cessna Aircraft Company P O Box 7704 Wichita, Kansas 67277 U S A			
4. Manufacturer:		Cessna Aircraft Company P.O. Box 7704 Wichita, Kansas 67277 U S A			
5. JAA Certification Application Date:		N/A			
6. JAA recommendation Date:		N/A			
7. EASA Type Certification Date:		28 September 2003			

### **B. Certification Basis**

1. Reference Date for determining the applicable requirements:		FAA application date 24 February 1998
2. (Reserved)		
3. (Reserved)		
4. Certification Basis:		As defined in FAA TCDS A4CE
5. Airworthiness Requirements:		FAR 23 as defined in FAA TCDS A4CE, and JAR-23, Change 1, plus Special Conditions as defined in Garmin G-1000 EASA CRI A-01, Issue 5, dated 17 March 2008 for the Nav III avionics option.
6. Requirements elected to comply:		None
7. EASA Special Conditions:		As defined in CRI A-01 for the Nav III avionics option only
8. EASA Exemptions:		None
9. EASA Equivalent Safety Findings:		None
10. EASA Environmental Standards:		CS 36 (ICAO Annex 16, Volume I, as applicable)

### **C. Technical Characteristics and Operational Limitations**

1. Type Design Definition: Master Drawing List, Document No T206-98-001, latest FAA Approved Revision.
2. Description: Single-engine, all-metal, six-place, high-wing airplane, fixed tricycle landing gear
3. Equipment: See Original delivery documents
4. Dimensions:

Span	10.9728 m (36 00 ft.)
Length	8.52424 m (27 97 ft.)
Height	2.54635 m (8.35 ft.)
Wing Area	16.3045 m <sup>2</sup> (175.50 ft <sup>2</sup> )
5. Engines: Lycoming IO-540-AJ1A

The EASA Engine Type Certification standard includes that of FAA IC E14EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003, Other standards conforming to TC/TCDS standards Certificated by individual EU member States prior to 28 September 2003 are also acceptable.

#### 5.1 Engine Limits:

Maximum Takeoff: 2500 RPM (310 hp)  
Maximum Continuous Power: 2500 RPM (310 hp)

For power-plants limits refer to Airplane Flight Manual and Pilot's Operating Handbook, Part No T206HPHUS00 or T206HPHAUS00 (Garmin) or T206HPHBUS00 (GFC-700) or latest revision

#### 6 Propellers: a.

McCauley Constant Speed  
McCauley Model: B3D36C432/80VSA-1

The EASA Propeller Type Certification standard includes that of FAA IC P58GL, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003, Other standards conforming to TC/TCDS standards Certificated by individual EU member States prior to 28 September 2003 are also acceptable.

Maximum Diameter: not over 2.0066 m (79 in )  
Minimum Diameter: not under 1.9685 m (77.5 in )  
Number of Blades: 3  
(1) McCauley Governor DC290D1/T25  
(2) Cessna Spinner: 2150151

#### 7. Fluids:

##### 7.1 Fuel:

100/100LL minimum grade aviation gasoline

- 7.2 Oil: Engine MIL-L-6082 or SAE J1966 Aviation Grade Straight Mineral Oil or MIL-L-22851 or SAE J1899 Aviation Grade Ashless Dispersant Oil. Oil conforming to Textron Lycoming Service Instruction No. 1014, latest revision, must be used after first 50 hours or once oil consumption has stabilized.
- 7.3 Coolant: Not Applicable
8. Fluid capacities:
- 8.1 Fuel: Units T20608001 through T20608361  
Total: 348 258 liters (92 US Gallons)  
Usable: 333 116 liters (88 US Gallons)
- Units 20608174 and on  
Total: 348 258 liters (92 US Gallons)  
Usable: 329 331 liters (87 US Gallons)
- 8.2 Oil: Maximum: 10 4099 liters (11.0 qts.)  
at 0 32512 m (12 8 in.) forward of datum  
Minimum: 5 67812 liters (6 0 qts ) usable
9. Air Speeds: Maneuvering 125 KIAS (123 KCAS)  
Maximum Structural Cruising 149 KIAS (147 KCAS)  
Never Exceed 182 KIAS (180 KCAS)  
Flaps Extended 100 KIAS (100 KCAS)
- 10 Maximum Operating Altitude: With a portable oxygen system, the aircraft is limited to 8229 6 m (27,000 ft MSL). Oxygen must be provided as required by the operating rules
11. Operational Capability: VFR Day and Night  
IFR Day and Night
- 12 Maximum Masses: Maximum Ramp: 1640 64 kg (3617 lbs )  
Maximum Takeoff: 1632 93 kg (3600 lbs )  
Maximum Landing: 1632 93 kg (3600 lbs )
13. Centre of Gravity Range:  
**Forward Limits:** Limer variation from 1 0795 m (42 5 in.) aft of datum at 1632 93 kg (3600 lbs.) to 0 8382 m (33 0 in ) aft of datum at 1133 98 kg (2500 lbs.): 0 8382 m (33 0 in ) aft of datum at 1133.98 kg (2500 lbs ) or less  
**Aft Limits:** 1 26238 m (49 7 in.) aft of datum at 1632 93 kg (3600 lbs ) or less
14. Datum: Front Face of Firewall (Fuselage Station 0 0)
- 15 (Reserved)
- 16 Levelling Means: Left side of Tailcone at 3 85699 m (151 85 in ) and 4 57835 m (180 25 in ) aft of datum
17. Minimum Flight Crew: 1 (Pilot)

- 18 Maximum Passenger Seating Capacity: 6 [2 at 0 8636 m (34 0 in.) to 1.2192 m (48.0 in.) aft of datum; 2 at 1 7526 (69 0 in) to 2.0066 m (79.0 in ) aft of datum; 2 at 2 4892 m (98.0 in) aft of datum].
- 19 Baggage / Cargo Compartment 81.6466 kg (180 lbs ) at 2 7686 kg (109 in ) to 3 683 m (145 in ) aft of datum.
- 20 Wheels and Tires
- |                      |          |
|----------------------|----------|
| Nose Wheel Tire Size | 5 00 x 5 |
| Main Wheel Tire Size | 6.00 x 6 |
21. Control Surface Movements
- |                          |                               |                   |
|--------------------------|-------------------------------|-------------------|
| Wing Flaps:              |                               | Down 40° +1°, -2° |
| Elevator Tab:            | Up 25° +1°, -0°               | Down 5° +1°, -0°  |
| Ailerons:                | Up 21° ± 2°                   | Down 14°30' ± 2°  |
| Elevator:                | Up 21° ± 1°                   | Down 17° ± 1°     |
| (Relative to stabilizer) |                               |                   |
| Rudder:                  | Right: 24° ± 1°               | Left: 24° ± 1°    |
|                          | (Parallel to 0.00 W L )       |                   |
|                          | Right: 27°13' ± 1°            | Left: 27°13' ± 1° |
|                          | (Perpendicular to hinge line) |                   |

#### **D. Operating and Service Instructions**

Airplane Flight Manual (AFM): Document No T206HPHUS00 or T206HPHAUS00 (Garmin) or T206HPHBUS00 (GFC-700), latest approved revision

Airplane Maintenance Manual (AMM) (Including Airworthiness Limitations) Document No. T206HMM00 or latest revision

#### **E. Notes**

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis, Note 6) must be installed in the airplane for certification.

NOTE 1: Weight and Balance:

Serial Nos. T20608001 through 20608361 :

The certificated basic empty weight and corresponding center of gravity location must include unusable fuel of 10 8862 kg (24 lbs ) at 1 2192 m (48 in ) aft of datum, and full oil of 9.344 kg (20 6 lbs.) at 0 32513 (12 8 in ) forward of datum.

Serial Nos. T20608362 and on :

The certificated basic empty weight and corresponding center of gravity location must include unusable fuel of 13 6078 kg (30 lbs.) at 1 2192 m (48 in ) aft of datum, and full oil of 9.344 kg (20 6 lbs.) at 0 32513 (12 8 in ) forward of datum.

NOTE 2: FAA Approved Airplane Flight Manual (AFM), or Pilot's Operating Handbook and FAA Approved Airplane Flight Manual (POH/AFM): Part number T206HPHUS00 or later FAA approved revisions are applicable to the Model T206H. The Airplane must be operated according to the appropriate AFM or POH/AFM Required placards are included in the AFM or POH/AFM

FAA Approved Airplane Flight Manual (AFM): Part Number T206HPHAUS-00 (or later FAA approved revisions) are applicable to the Model T206H equipped with Garmin G1000 Integrated Cockpit System. The airplane must be operated according to the appropriate AFM Required placards are included in the AFM

FAA Approved Airplane Flight Manual (AFM): Part Number T206HPHBUS-00 (or later FAA approved revisions) are applicable to the Model T206H equipped with Garmin G1000 Integrated Cockpit System and Garmin GFC-700 AFCS. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM.

NOTE 3: The CHI probe must be installed on Head #5

NOTE 4: Model T206H airplanes, serial numbers T20608101 through T20608158 may differ structurally and are, therefore, not eligible for any weight increases above the approved maximum takeoff weight limit of 3,600 pounds unless compliance with Cessna Document A1206-57-01 (latest revision) has been accomplished and documented with an appropriate logbook entry. Any exceptions must first be coordinated with the Wichita Aircraft Certification Office.

NOTE 5: Special Ferry Flight Authorization Flight Standards District Offices are authorized to issue Special overweight ferry flight authorizations. This airplane is structurally satisfactory for ferry flight if maintained within the following limits: (1) Takeoff weight must not exceed 130% of the maximum weight for Normal Category, and (2) The Never Exceed Airspeed ( $V_{NE}$ ) and Maximum Structural Cruising Speed ( $V_C$ ) must be reduced by 30%; and (3) Forward and aft center of gravity limits may not be exceeded; and (4) Structural load factors of 2.5 g to -1.0 g may not be exceeded. Requirements for any additional oil should be established in accordance with Advisory Circular AC23 1011-1. Increased stall speeds and reduced climb performance should be expected for the increased weights. Flight characteristics and performance at the increased weights have not been evaluated. Flight Permit for operations of overweight aircraft may be found in Advisory Circular AC21-4B.

NOTE 6: FAA Certification Basis (Model T206H)

Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-6, except as follows:

FAR 23 423; 23 611; 23 619; 23 623; 23 689; 23 775; 23 871; 23 1323; and 23.1563 as amended by Amendment 23-7

FAR 23 807 and 23.1524 as amended by Amendment 23-10 FAR 23.507; 23 771; 23.853(a),(b) and (c); and 23 1365 as amended by Amendment 23-14. FAR 23 951 as amended by Amendment 23-15. FAR 23 607; 23 675; 23 685; 23 733; 23 787; 23 1309 and 23.1322 as amended by Amendment 23-17. FAR 23.1301 as amended by Amendment 23-20 FAR 23 1353; and 23 1559 as amended by Amendment 23-21 FAR 23.603; 23 605; 23 613; 23 1329 and 23.1545 as amended by Amendment 23-23 FAR 23.441 and 23.1549 as amended by Amendment 23-28 FAR 23 779 and 23.781 as amended by Amendment 23-33. FAR 23.1; 23.51 and 23.561 as amended by Amendment 23-34 FAR 23 301; 23 331; 23.351; 23 427; 23 677; 23 701; 23 735; and 23.831 as amended by Amendment 23-42 FAR 23 961; 23 1093; 23 1107(b); 23 1143(g); 23.1147(b); 23.1303; 23.1357; 23 1361 and 23 1385 as amended by Amendment 23-43 FAR 23 562(a), 23.562(b)2, 23 562(c)1, 23.562(c)2, 23.562(c)3, and 23 562(c)4 as amended by Amendment 23-44 FAR 23 33; 23 53; 23 305; 23 321; 23 485; 23 621; 23 655 and 23 731 as amended by Amendment 23-45

FAR 36 dated December 1, 1969, as amended by Amendments 36-1 through 36-21

Additions for the Gamin G1000 Integrated Cockpit System (ICS) Only: 14 CFR 23.303; 23.307; 23 601; 23 1163(a)(1)(2); 23 1367 and 23.1381 as amended by Amendment 23-N/C. 14 CFR 23.1589 as amended by Amendment 23 13 14 CFR 23 771(a) as amended by Amendment 23.14. 14 CFR 23.607 and (Electrical System) 23.1309(a)(1)(2), (c) as amended by Amendment 23-17 14 CFR 23.1301; 23 1327 and 23.1547(e) as amended by Amendment 23-20 14 CFR 23.1501 and 23 1541(a)(1), (a)(2), (b)(1), (b)(2) AS AMENDED BY Amendment 23-21. 14 CFR 23 603 and 23.605 as amended by Amendment 23-23. 14 CFR 23.1529 as amended by Amendment 23-26 14 CFR 23 561(e); 23.1523; 23 1581(a)(2); 23 1538(a)(1), (a)(2), (b)(h) and 23.1585(a)(b)(d) as amended by Amendment 23-34 14 CFR 23 301 as amended by Amendment 23-42. 14 CFR 23 1322; 23.1331 and 23 1357(a)(b)(c)(d) as amended by Amendment 23-43. 14 CFR 23 305; 23 773(a)(1), (a)(2); 23.1525 and 23.1549 as amended by Amendment 23-45 14 CFR 23.1303(a)(b)(c)(f); 23 1309(a)(1)(i), (a)(1)(ii), (a)(2), (b)(1), (b)(2)(i), (b)(2)(ii), (b)(3), (b)(4)(i), (b)(4)(ii), (b)(4)(iii), (b)(4)(iv), (c)(1), (c)(2)(iii), (c)(3), (d), (e), (f)(1); 23 1311; 23.1321(a)(c)(d)(e); 23 1323(a), (b)(1), (b)(2), (c); 23 1329(g)(h); 23.1351(a)(1), (a)(2)(i), (b)(1)(iii), (b)(2)(3), (c)(4), (d)(1); 23 1353(a)(b)(c)(d)(e); 23 1359(c); 23 1361; 23 1365(a)(b)(d)(e)(f) and 23.1431 (a)(b)(d)(e) as amended by Amendment 23-49. 14 CFR 23 1325(a), (b)(1), (b)(2)(i), (b)(3), (c)(d)(e); 23 1543(b)(c); 23 1545(a), (b)(1), (b)(2), (b)(3), (b)(4); 23 1553; 23 1555(a)(b); 23 1563(a) and 23 1567(a) as amended by Amendment 23 50. 14 CFR 23.777(a)(b); 23 955(a)(2); 23 1337(a)(1), (a)(2), (b)(1), (c) as amended by Amendment 23 51. 14 CFR 23 1305(a)(1),

(a)(2), (a)(3), (b)(2), (b)(3)(i), (b)(4)(i), (b)(5), (b)(6)(i) as amended by Amendment 23-52 14 CFR 23 901  
(a)(b) as amended by Amendment 23-53

Additions for the Garmin GFC-700 Automatic Flight Control System (AFCS) Only:

14 CFR 23 1335 as amended by Amendment 23-20 14 CFR 23 1329 (a)(c)(d)(e)(f) as amended by Amendment 23-49

Equivalent Safety Items:

- (1) Throttle Control FAR § 23 1143(g) Number 97-4, FAA letter October 1, 1998
- (2) Mixture Control FAR § 23 1147(b) Number 97-4, FAA letter October 1, 1998
- (3) Fuel Tank Sump FAR § 23 971 Number ACE-02-03, FAA letter January 3, 2002 (Units T20608362 and on)
- (4) Anticollision Lights FAR§23 140I(d) Number ACE-02-02, FAA letter January 3, 2002 (Units T20608362 and on)
- (5) Aviation White Color 14CFR§23.1397(c) Refer to ACE-07-12, FAA letter 11/29/07

Date of Application for Amended Type Certificate was October 30, 1996  
Type Certificate No A4CE was amended October 1, 1998

Special Conditions as follows:

No 23-150-SC, (Special Conditions: Cessna Aircraft Company; Cessna Model 206H Airplane; Installation of Electronic Flight Instrument System and the Protection of the System From High Intensity Radiated Fields (HIRF)

**Production Basis (Model T206H)**

Production Certificate No PC-4 issued November 25, 1998 Applies to airplane serial numbers T20608001 and on Cessna is authorized to issue airworthiness certificates under the delegation provisions of Delegation Option Authorization No CE-1 in accordance with Part 21 of the Federal Aviation Regulations

**SECTION 3: Reserved**